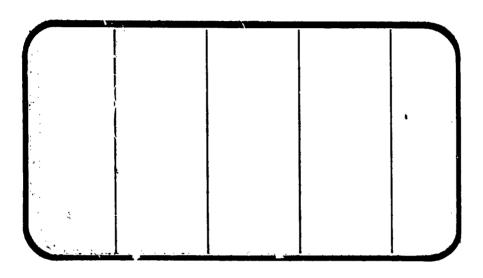


NATIONAL A .:. "NAUTICS AND SPACE ADMINISTRATION



(NASA-CF-147615) HEAT TRANSFER TEST OF AN 0.006-SCALE THIN-SKIN THERMOCOUPLE SPACE SHUTT'S HODEL (50-0, 41-T) IN THE NASA-AMES RESEARCH CHUTER 3.5-FOOT HYPERSONIC WIND TUNNEL AT MACH 5.3 (IH28), VOLUME 1

N76-32230 HC \$21.25

Unclas G3/18 05280

SPACE SHUTTLE

AEROTHERMODYNAMIC DATA REPORT



JOHNSON SPACE CENTER

HOUSTON, TEXAS

DATA MAÑagement services



DMS-DR-2180 NASA CR-147,615

VOLUME 1 OF 2

HEAT TRANSFER TEST OF AN 0.006-SCALE THIN-SKIN

THERMOCOUPLE SPACE SHUTTLE MODEL (50-0, 41-T) IN

THE NASA-AMES RESEARCH CENTER 3.5-FOOT HYPERSONIC

WIND TUNNEL AT MACH 5.3 (IH28).

bу

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Shuttle Aerosciences
Rockwell International Space Division
W. K. Lockman
NASA-Ames Research Center

Prepared Under Contract Number NAS9-13247

by

Data Management Services
Chrysler Corporation Space Division
New Orleans, La. 70189

for

Engineering Analysis Division

Johnson Space Center
National Aeronautics and Space Administration
Houston, Texas

WIND TUNNEL TEST SPECIFICS:

Test Number:

ARC 3.5-195

NASA Series Number:

IH28

Model Number:

50-0, 41-T May 17 through May 24, 1974

Test Dates: Occupancy Hours:

FACILITY COORDINATOR:

Joseph G. Marvin Mail Stop N-229-1

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Chrysler Corporation Space Division assumes no responsibility for the lata presented other than display characteristics.

HEAT TRANSFER TEST OF AN 0.006-SCALE THIN-SKIN

THERMOCOUPLE SPACE SHUTTLE MODEL (50-0, 41-T) IN

THE NASA-AMES RESEARCH CENTER 3.5-FOOT HYPERSONIC

WIND TUNNEL AT MACH 5.3 (1H28)

bу

J. W. Cummings/T. F. Foster
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Rockwell International Space Division
W. K. Lockman
NASA-Ames Research Center

ABSTRACT

This report presents data obtained from a heat transfer test conducted on an 0.006-scale Space Shuttle Orbiter and External Tank in the NASA-Ames Research Center 3.5-foot Hypersonic Wind Tunnel. The purpose of this test was to obtain data under simulated return-to-launch-site abort conditions. Configurations tested were integrated orbiter and external tank, orbiter alone, and external tank alone at angles of attack of 0, ±30, ±60, ±90, and ±120 degrees.

Runs were conducted at Mach numbers of 5.2 and 5.3 for Reynolds numbers of 1.0 x 10^6 and 4.0 x 10^6 per foot, respectively. Heat transfer data were obtained from 75 orbiter and 75 external tank iron-constantan thermocouples.

This report consists of 2 volumes. Volume 1 contains Figures 4+15; whereas, Volume 2 contains Figures 16+27 and the Tabulated Source Data.

TABLE OF CONTENTS

| A. B. G | | Pag |
|-----------|---------------------------------------|-------------|
| ABSTRACT | | ii : |
| INDEX OF | MODEL FIGURES | : |
| INDEX OF | DATA FIGURES | : |
| NOMENCLAT | TURE | • |
| CONFIGURA | ATIONS INVESTIGATED | 10 |
| MODEL INS | TRUMENTATION | 12 |
| TEST FACI | LITY DESCRIPTION | 13 |
| TEST PROC | EDURE | 14 |
| DATA REDU | CTION | 15 |
| REFERENCE | S | 18 |
| TABLES | | 10 |
| ı. | TEST CONDITIONS | 19 |
| II. | DATA SET/RUN NUMBER COLLATION SUMMARY | 20 |
| | MODEL DIMENSIONAL DATA | 22 |
| IV. | ORBITER THERMOCOUPLE LOCATIONS | 29 |
| v. | EXTERNAL TANK THERMOCOUPLE LOCATIONS | 31 |
| | RUN NUMBER/TUNNEL CONDITION SUMMARY | 32 |
| FIGURES | | 32 |
| MODEL | | 24 |
| DATA | (VOLUME 1 - FIGURES 4-15) | 34 |
| | (VOLUME ? - FIGURES 16-27) | 43 |
| PPENDIX | (FIGURES 10-2/) | 43 |
| | ATED SOURCE DATA (VOLUME 2) | |

INDEX OF MODEL FIGURES

| Figures | Title | Page |
|---------|---|------|
| 1. | Orbiter/External Tank General Layout | 34 |
| 2. | Model Instrumentation Skeuches | |
| | a. 50-0 Orbiter 147-B Configuration Thermo- couple Locations. | 35 |
| | b. 41-T 0.006-Scale External Tank T/C Locations. | 37 |
| | c. Specific Heat vs. Temperature for 17-4PH and 15-5PH stainless steel. | 38 |
| 3. | Model Installation Photographs | |
| | a. Orbiter/Tank at 0.0 degrees. | 39 |
| | b. Orbiter/Tank at -60.0 degrees. | 40 |
| | c. Orbiter/Tank at 90.0 degrees. | 41 |
| | d. Orbiter/Tank at 120.0 degrees. | 42 |

INDEX OF DATA FIGURES

| | INDEX OF DAIA FIGURES | | | |
|----------|---|--|--|---------|
| FIGURE | SCI COP TITLE I ITTE | SCHEDULE OF COEFFICIENTS PLOTTED | CONDITIONS VARYING | PAGES |
| VOLUME | | | | |
| 4 | TANK, ALONE | (¥) | HAW/HT,PHI,ALPHA, RN/L, X/L, MACH | 1-112 |
| ٦ | TANK, IN THE PRESENCE OF ORBITER | (A) | HAW/HT, PHI, ALPHA, RN/L,X/L,BETA,MACH | 113-336 |
| 9 | TANK, RATIO OF INTERFERENCE TO UNDISTURBED | (B) | PHI, ALPHA, X/L | 337-373 |
| 7 | ORBITER UNDERSIDE FUSELAGE, ORBITER ALONE | (3) | HAW/HT, BP, ALPHA | 374–395 |
| ∞ | ORBITER UNDERSIDE FUSELAGE, ORBITER IN THE PRESENCE OF THE TANK | (G) | HAW/HT, BP, ALPHA RN/L, BETA, MACH | 396-427 |
| σ'n | ORBITER UNDERSIDE FUSELAGE, RATIO OF INTERFERENCE TO UNDISTURBED | (a) | вР, АГРНА | 428–440 |
| 10 | ORBITER BODY SIDEWALL, ORBITER ALONE | (E) | HAW/HT,Z,ALPHA,X/L 441-528 | 441–528 |
| 11 | ORBITER BODY SIDEWALL, ORBITER IN PRESENCE OF THE TANK | (E) | HAW/HT, Z, ALPHA RN/L,X/L,BETA, MAJH | 529-656 |
| 12 | ORBITER BODY SIDEWALL, RATIO OF INTERFERENCE TO UNDISTURBED | (F) | Z, ALPHA, X/L | 657–690 |
| ដ | OMS PODS, ORBITER ALONE | (E) | HAW/HT, X/L, ALPHA, Z | 691-720 |
| 14 | OMS PODS, ORBITER IN PRESENCE OF THE TANK | (E) | HAW/HT,X/L,ALPHA, 721-768 RN/L,BETA,Z, MACH | 721–768 |
| 15 | OMS PODS, RATIO OF INTERFERENCE TO UNDISTURBED | (a) | Z, ALPHA | 769–789 |
| | | | | |

INDEX OF DATA FIGURES (Continued)

| | INDU TO WIGHT | Someon C OF | | |
|--------------|--|-----------------------------|--|------------|
| | | SCHEDULE OF COEFFICIENTS | CONDITIONS | |
| FIGURE | TITLE | PLOTTED | VARYING | PAGES |
| VOLUME 16 | CHINE, ORBITER ALONE | (၁) | HAW/HT, ALPHA | 790-800 |
| 17 | CHINE, ORBITER IN PRESENCE OF THE TANK | (3) | HAW/HT,ALPHA,RN/L,MACH | 801-811 |
| . 81 | CHINE, BATIO OF INTERFERENCE TO UNDISTURBED | (a) | ALPHA | 812-822 |
| 1 61 | LEFT WING LOWER SURFACE, C'BITER ALONE | (0) | HAW/HT, 2Y/B, ALPHA | 823-855 |
| 50 | LEFT WING LOWER SURFACE, ORBITER IN PRESENCE OF TANK | (9) | HAW/HT, 2Y/B, ALPHA, RN/L, BETA, MACH | 856-903 |
| 21 | LEFT WING LOWER SURFACE, RATIO OF INTERFERENCE TO UNDISTURBED | (H) | 2Y/B, ALPHA | 904-918 |
| 22 | RIGHT WING UPPER SURFACE, ORBITER ALONE | (I) | HAW/HI,2Y/B,ALPHA,X/C | 919-995 |
| 23 | RIGHT WING UPPER SURFACE, ORBITER IN PRESENCE OF TANK | (I) | HAW/HT, 2Y/B, ALPHA, RN/L, 996-1107 BETA, X/C, MACH | , 996–1107 |
| 24 | RIGHT WING UPPER SURFACE, RATIO OF INTERFERENCE TO UNDISTURBED | Ð | 2Y/B, ALPHA, X/C | 1108-1139 |
| 25 | VERTICAL TAIL, ORBITER ALONE | (9) | HAW/HT, Z, ALPHA | 1140-1161 |
| 26 | VERTICAL TAIL, ORBITER IN PRESENCE OF TANK | (9) | HAW/HT, Z, ALPHA RN/L, BETA, MACH | 1162-1193 |
| 27 | VERTICAL TAIL, RATIO OF INTERFERENCE TO UNDISTURBED | (н) | Z, ALPHA | 1194-1206 |
| | | | | |

INDEX OF DATA FIGURES (Concluded)

SCHEDULE OF COEFFICIENTS PLOTTED:

- (A) H/HREF versus X/L H/HREF versus PHI
- (B) HI/HU versus X/L HI/HU versus PHI
- (C) H/HREF versus X/L
- (D) HI/HU versus X/L
- (E) H/HREF versus X/L H/HREF versus Z
- (F) HI/HU versus X/L HI/HU versus Z
- (G) H/HREF versus X/C
- (H) HI/HU versus X/C
- (I) H/HREF versus X/C H/HREF versus 2Y/B
- (J) HI/HU versus X/C HI/HU versus 2Y/B

NOMENCLATURE

| SYMBOL | Plot Symbol | DEFINITION |
|--|-----------------|---|
| Ъ | | thickness of model skin, in. |
| В | BREF | span length, in. |
| C | | |
| C | | specific heat of model skin material, BTU/lbm-OR |
| c | | chord length, in. |
| C _o , C ₁ , C ₂ | 2 | constants in curve fit for C over model wall temperature range |
| сp | | specific heat of air stream (perfect gas value), $\rm BTU/1b_{m}\mbox{-}^{O}R$ |
| CHAN | CHAN | Recording-system channel |
| Haw | HAW | adiabatic wall enthalpy, BTU/1bm |
| Ht | ht | free-stream total enthalpy, BTU/1bm |
| | но | average of free-stream total enthalpy values of all tunnel runs incorporated into an aero dataset, $\rm BTU/lb_{m}$ |
| Hw ₁ | HW | enthalpy based on model wall temperature for given T/C location at initial time, BTU/lbm |
| h | Ħ | heat-transfer coefficient at model wall for given T/C location |
| hs | hs, href | stagnation-point heat-transfer coefficient for reference sphere |
| h/h _s | h/hs, h/href | ratio of model heat-transfer coefficient to heat-transfer coefficient of reference sphere for Haw/Ht = X.XXX |
| IML | | inner mold line |
| r | LREF, LENGTH | model reference length, in. or ft. |
| M∞ | MACH | free-stream Mach number |
| H _w | | enthalpy based on model wall temperature, $\mathrm{BTU/1b_m}$ |

NOMENCLATURE (Continued)

| SYMBOL | PLOT SYMBOL | DEFINITION |
|-------------------|----------------|--|
| Pt | PT | free-stream total pressure, psia |
| | 20 | average of free-stream total pressure valuer of all tunnel runs incorporated into an aero dataset, psia |
| q ₁ | QDOT, Q | heat-transfer rate at model wall for given T/C location at initial time, BTU/ft2-sec |
| ġ _s | QS, QREF | stagnation-point heat-transfer rate for reference sphere at initial time, BTU/ft2-sec |
| Rs | RS | reference sphere radius at model scale equivalent to 0.305 m (1 ft) for full-scale vehicle |
| Ře∞/ft | RE/FT | free-stream Reynolds number per foot |
| | rn/L | average of free-stream Reynolds number values (per foot) of all tunnel runs incorporated into an aero dataset |
| Re_{∞} , L | REL | free-stream Reynolds number based on model reference length, L |
| | s/R | body wetted running length |
| St | ST | Stanton number based on free-stream flow conditions and the model heat-transfer coefficient for Haw/Ht = X.XXX |
| T | | temperature, OR |
| Tt | TeI | free-stream total temperature, OR |
| | TO | average of free-stream total temperature values of all tunnel runs incorporated into an aero dataset, CR |
| Tw 1 | TW | model wall temperature for given T/C location at initial time, OR |

NOMENCIATURE (Continued)

| SYMBOL | PLOT SYMBOL | DEFINITION |
|--------|----------------|---|
| T/C | T/C | thermocouple |
| t | | time, sec |
| ti | TIME | initial time (before model insertion into flow) extrapolated from $f(T_w)$ vs. time, sec |
| u,V | | velocity, ft/sec |
| W | | density of model skin material lbm/ft3 |
| x | | axial distance measured from nose, in. |
| | x/c | chordwise location, fraction of local chord |
| | X/L | longitudinal location, fraction of body length |
| Y | | spanwise distance from centerline, in. |
| 2y/B | 2 Y /B | spanwise location, fraction of semi-span |
| Z | Z | water plane distance, in. |
| | z/ev | spanwise location on vertical tail, fraction of exposed span |
| α | ALPHA | angle of attack, degrees |
| β | BETA | angle of sideslip, degrees |
| μ | | viscosity of air, lb-sec/ft2 |
| ρ | | density of air, lbm/ft3 |
| θ | THETA | external tank angular surface coordinate, measured clockwise looking forward. O degrees at bottom centerline, degrees |
| ø | PHI | orbiter angular surface coordinate, measured clockwise looking forward. O degrees at bottom centerline, degrees |

NOMENCLATURE (Concluded)

| SYMBOL | PLOT SYMBOL | DEFINITION |
|------------|----------------|---|
| W.P. | | water plane, height measured along Z axis, in. |
| В.1 | ВР | butt plane, distance from orbiter centerline in the outboard direction, in. |
| | ні/ни | ratio of interference to undisturbed heat transfer coefficients |
| | ZMRP | moment reference point on Z axis |
| | YMRP | moment reference point on Y axis |
| | XMRP | moment reference point on X axis |
| | SREF | reference length or wing mean aerodynamic chord; ft. |
| SUBSCRIPTS | | |
| aw | | adiabatic vall |
| Ę | | initial value before model insertion |
| 0 | | Orbiter |
| PG | | perfect gas (calorically and thermally perfect gas) |
| ន | | reference sphere |
| t | | free-stream total condition |
| T | | tank |
| v | | vertical tail |
| W | | wall |
| œ | | free-stream |

CONFIGURATIONS INVESTIGATED

The model (Orbiter and External Tank) tested was a 0.006-scale representation of the Rockwell International Space Shuttle Vehicle. The Orbiter and External Tank are defined by Rockwell lines SS-H-O11:14 and SS-H-O1415.

The Orbiter and Tank were initially built by Grumman Aircraft,
Bethpage, New York, but the Orbiter was modified with additional thermocouples added to the upper surface of the left wing, vertical tail, and
OMS pod. Modifications of both Orbiter and External Tank stings were
accomplished to carry increased loading within the high angle of attack
range.

The Orbiter was a full span (cast stainless steel) model with thin-skin inserts. Thin-skin stainless steel (17-4PH) inserts were located on the underside region. left-hand wing (top and bottom), windshield area, left fuselage side, OMS pod, and vertical tail. These inserts were instrumented with 89 iron-constantan thermocouples of which only 75 were used during this test. The model was built with all control surfaces in the 0° deflection condition.

The External Tank was constructed of thin-skin (15-5PH) stainless steel. The Tank was instrumented with 111 iron-constantan thermocouples, of which only 75 were used.

The Orbiter and External Tank were designed so either could be tested alone or in the second stage configuration.

CONFIGURATIONS INVESTIGATED (Concluded)

The following configuration components were tested:

| Notation | Description |
|-----------------------|----------------------------|
| B ₂₂ | Fuselage (1-147B Lines) |
| c ₇ | Canopy |
| F ₅ | Body Flap |
| $M_{l_{4}}$ | OMS Pods |
| v ₇ | Vertical Tail |
| Wall | Wing |
| T 8 | External Tank (-139 Lines) |

MODEL INSTRUMENTATION

The Orbiter and External Tank were instrumented with 200 ironconstantan thermocouples, but only 150 were used for this test. All
thermocouples were spotwelded to thin-skin (nominal skin thickness of
0.030 in.) stainless steel inserts and the leads were clamped in bundles
within the model. The exact T/C locations for the Orbiter and External
Tank are presented in Tables IV and V, respectively, and illustrated in
Figures 2a and 2b, respectively. The T/C leads were 50 feet long and
fitted with Cannon Plug connectors.

TEST FACILITY DESCRIPTION

The NASA-Ames 3.5-foot Hypersonic Wind Tunnel is a closed-circuit, blowdown-type tunnel capable of operating at nominal Mach numbers of 5, 7, and 10 at pressures to 1800 psia and temperatures of 34000R for run times to four minutes. The major components of the facility include a gas storage system where the test gas is stored at 3000 psi, a storage heater filled with aluminum-oxide pebbles capable of heating the test gas to 34000R, axisymmetric contoured nozzles with exit diameters of 42 inches for generating the desired Mach number, and a 900,000 ft³ vacuum storage system which operates to pressures of 0.3 psia. The test section itself is an open-jet type enclosed within a chamber approximately 12-feet in diameter and 40-feet in length, arranged transversally to the flow direction.

A model support system is provided that can pitch models through an angle-of-attack range of -20 to +20 degrees, in a vertical plane, about a fixed point of rotation on the tunnel centerline. This rotation point is adjustable from 1 to 5 feet from the nozzle exit plane. The model normally is out of the test stream (strut centerline 37 inches from tunnel centerline) until the tunnel test conditions are established after which it is inserted. Insertion time is adjustable to as little as $\frac{1}{2}$ second and models may be inserted at any strut angle.

A high-speed, analog-to-digital data acquisition system is used to record test data on magnetic tape. The present system is equipped to measure and record the outputs from 80 transducers in addition to 20 channels of tunnel parameters.

TEST PROCEDURE

Heat Transfer Data were obtained by measuring the temperature rise over a period of time from a total of 150 iron-constantan thermocouples. The model was injected into the flow in approximately 1 second and held on tunnel centerline for approximately 1 second. Temperature measurements and tunnel conditions were recorded on magnetic tape at 0.07-second intervals by the data acquisition system from the start of model injection to the start of model retraction.

A maximum of 75 thermocouples could be recorded for any given run. The thermocouple leads were routed from the model through the tunnel model-injection mechanism, and connected to a junction box which was wired directly to a thermocouple reference-temperature (150°F) box. The junction box connectors were wrapped with asbestos for heat protection from the tunnel test-chamber ambient conditions (no free-stream flow on box). Thermocouple changes were accomplished by changing 5 Cannon Plugs containing 15 thermocouples each. Prior to testing, a thermocouple heat-response check, through the data-acquisition system, was performed on all thermocouples to assure proper hook-up, polarity and response.

Prior to each run with model attitude changes, the model was leveled in pitch and roll by means of leveling blocks which attach to the sting assembly of the Orbiter/External Tank. When leveling the models, an inclinometer was placed on the leveling plate. Proper roll relationships between the models were set using scribed lines on the model stings.

DATA REDUCTION

All test data were reduced at the NASA/Ames Research Center using the data-reduction techniques outlined below. The thermocouple data were reduced using the one-dimensional, thin-wall equation:

$$\dot{q} = WCb \frac{dT_W}{dt} = h \left(H_{dW} - H_W\right) \equiv hH_t \left(\frac{H_{dW}}{H_t} - \frac{H_W}{H_t}\right)$$
 (1)

which neglects heat-conduction losses.

Assuming that W and h are constant and

$$C = C_0 + C_1 T_W + C_2 T_W^2$$
 for T_W ranges (2)

the integration of equation (1) for $t = t_1$ to t and $T_W = T_{W_1}$ to T_W yields the linear equation:

$$f\left(T_{W}\right) = -\ln\left(\frac{T_{dW}^{2} - T_{W}}{T_{dW}^{2} - T_{W1}}\right) - \left[\frac{C_{1}}{C_{dW}^{2}} + \frac{C_{2}}{C_{dW}^{2}} \left(T_{dW}^{2} + \frac{T_{W} + T_{W1}}{2}\right)\right]\left(T_{W} - T_{W1}\right)$$

$$=\frac{hc_p}{WC_{a_n}^*b} \quad (t-t_1) \tag{3}$$

where it is defined that:

$$T_{AN}^{1} = \frac{R_{BN}}{op} = \frac{R_{BN}}{H_{2}} = \frac{R_{BN}}{6p} \ge (T_{BN}) \text{ pg}$$
(4)

$$C_{av}^{t} \equiv C_{o} + C_{1} T_{av}^{t} + C_{2} T_{av}^{t}$$
 (5)

≠ specific heat at adiabatic wall temperature

DATA REDUCTION (Continued)

The form of Eq (3) is $f(T_W) = mt + a$ where m is the slope and a is the intercept for a straight line if heat-conduction errors are negligible. Thus, deviations from a straight line can indicate heat-conduction effects.

The slope, m, of $f(\mathfrak{T}_W)$ vs t from Eq (3) is computed by a least-squares, straight-line fit over a finite time interval (approx. 1 sec) beginning when the model reaches uniform tunnel flow. The value of the heat-transfer coefficient, h, is then determined from:

$$h = \frac{WJ_{\hat{\mathbf{g}}W}^{\bullet}b}{\mathbf{g}_{\mathbf{p}}} \quad \mathbf{m} \tag{6}$$

Using this value of h, the heat-transfer rate is evaluated at the initial time, t_1 , when the model is isothermal at the initial wall enthalpy, H_{W_1}

$$\hat{q} = \hat{q}_{\underline{i}} = h \left(H_{\underline{n}\underline{i}} - H_{\underline{v}\underline{i}} \right) \equiv h H_{\underline{t}} \left(\frac{H_{\underline{n}\underline{i}}}{H_{\underline{t}}} - \frac{H_{\underline{v}\underline{i}}}{H_{\underline{t}}} \right) \tag{7}$$

where $H_{\rm BW}/H_{\rm t}$ is the same value used to evaluate h. The resultant value of d is independent of the value of $H_{\rm BW}/H_{\rm t}$ used for both the h and d evaluations.

The reference sphere heating is also evaluated at the initial wall enthalpy by the method of Fay and Riddell

$$\hat{q}_{g} = h_{g} \left(H_{t} - H_{e_{1}} \right) \equiv h_{g} \quad H_{t} \quad \left(1.0 - \frac{H_{e_{1}}}{H_{t}} \right) \tag{8}$$

The model-to-sphere ratio of heat-transfer coefficients is then determined from Eqs. (7) and (8) as

$$\frac{h}{h_0} = \frac{\dot{q}_1}{\dot{q}_0} \left[\frac{1.0 - \frac{H_{01}/H_0}{H_0}}{H_{00}/H_0 - \frac{H_{01}/H_0}{H_0}} \right] \tag{9}$$

DATA REDUCTION (Concluded)

where \dot{q}_i is constant for all values of H_{aw}/H_t . To determine h/h_s for various values of H_{aw}/H_t , the particular value of H_{aw}/H_t is substituted into Eq. (9).

The Stanton number is defined as

$$St = \frac{h}{\rho u} = \frac{\dot{q}_1}{\rho u(H_{aW} - H_{W1})}$$
 (10)

where for free-stream conditions, $\rho u = \rho_{\infty} V_{\infty}$.

The calculations of the model heating, reference sphere heating, and Reynolds number included the corrections of NACA report 1135 (Ref. 3) for calorically imperfect, thermally perfect air. Keyes' equation for viscosity (Ref. 4) was also used for the sphere heating and Reynolds number computations:

$$\mu = \frac{0.0232 \times 10^{-6} \text{T}^{0.5}}{1 + \frac{220}{\text{T}} \times 10^{-9/\text{T}}}$$
(11)

where the units for T and μ are $^{\circ}R$ and $^{1}b\text{-sec}/ft^2$, respectively.

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| T: IH28 (AI | RC 3.5-195) | | DATE: 6-8-74 |
|-------------------|----------------------------|----------------------------------|------------------------------------|
| | TEST CON | IDITIONS | |
| | | | • |
| MACH NUMBER | REYNOLDS NUMBER (per foot) | TOTAL PRESSURE (pounds/sq.inch) | TOTAL TEMPERATURE (degrees Rankine |
| 5.2 | 1.0 x 10 ⁶ | 100 | 1500°R |
| 5.3 | 4.0 x 10 ⁶ | 410 | 1500°R |
| | | | |
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| BALANCE UTILIZED: | | | |
| | CAPACITY: | ACCURACY: | CCEFFICIENT TOLERANCE: |
| NF | · | • | |
| SF | | | |
| AF | | | |
| PN | | | • |
| ŔN | | | |
| YN | | | |
| COMMENTS: 75 | iron-constantan T/C | 's on ORB | |
| 7 5 | 1ron-constantan T/C | YB OR AT | |

TABLE II.

| ſ | | | | | | | | | TE | ST F | NUI | NUM | BER | \$ | | | | | | | | | | 202 | |
|---------|---------------------------------------|-------------------|---------------|----------|-------------|----------|----------|------|----------|------|-----|----------|----------|----------|-----|----|----------|----------|---------|----------|----------|----------|----|------------------|-------------------------|
| | | | | | | | | | | | | | | | | | | | - | | 1 | 1 | | IDVAR (2) N | |
| ŀ | 1874 | | _ | _ | | | _ | _ | | | _ | | _ | _ | _ | _ | _ | | _ | | 4 | 4 | 4 | ij | |
| | DATE: JEWE, 1 | dh | | | | | | | | | | | | | - | - | - | | _ | _ | | | | IDVAR (1) | |
| | E : 7 | HOOKUP | | \dashv | ا - ا | 1 | | | | _ | | | | | | | | | - | | 4 | 4 | + | | |
| | DAT | ı | | - | | | | | | _ | _ | | | | | - | | _ | _ | | | | | PER FT. | |
| | | cour | | | | | | | | | | | | | | | | | \perp | _ | 1 | 4 | 4 | | |
| | ARŸ | THERMOCOUPLE | | | | | | | | | | | | | | | | | | | | | | 9-01 | |
| | SUMM | | | | | | | | | | | | | | | | | | | | 4 | - | - | 9.01X~* | |
| | DATA SET/RUN NUMBER COLLATION SUMMARY | MOJEL | C, | | 7 | 5. | 81 | 61 | 22 | 23 | 14 | 15 | 9 | " | 8 | 1 | 1 | | | | | | | | |
| | OLLA | ٧ | 7. | ~ | 3 | 4 | 11 | 10 1 | | 74 1 | 3 | / 9/ | 77 | , 0, | 6 | 33 | 31 | 32 | 37. | 00 | 1 | | 4 | SCHEDULES | |
| TT SHOW | ER C | 9 | | | | | ` | 7 | 121 | 7 | / | | / | <u> </u> | 1 | | ") | ί | 2 | 3 | | | | NT SC | 7.300 |
| 1 | LUMB | JES NO. | R.U | 7 | | | | | | | | | | | | 1 | | | | | | • | 1 | FFICIE ST. S. | T. S. |
| | RUN | PARAMETERS/VALUES | | | | | | | | | | | | | | | | | | | | | | COE | HOOKUP (CONST. SET 200) |
| | SET/ | ETER | 100 | 7.7 | | | | | | | | + | 5,3 | 5.3 | 5.2 | | | | | - | - | 1 | | gryker) | AMX |
| | ATA | ARAM | Re Ma | 23 01 | | | | | | | | * | 4.0 | ó | 0 | | | | | - | | | | 1 4 | Hac |
| | ۵ | SCHD. | | 0 | | | | | | | | | | • | Ŋ | 0 | | | , | * | - | - | 1 | 1 60 | 7672 |
| | _ | S | 8 | 0 | 30 | 09 | 90 | 82 | 3 | 90 | 18 | 30 | 3 | 38 | 8 | 0 | 30 | 60 | 90 | متر | | | | 1007 | RYON |
| | (5) | | z | | | | | | , | • | | | | | | | | : | | | - | - | •• | COEFFICIENT SCI | T TANK THERMOZONOLE |
| | 1-5 | | CONFIGURATION | | | | | | | | | | | - - | | | | | | | | | | 6 | 14 |
| | 18C 3 | | NFIGU | 70 | - | - | <u> </u> | | _ | | | | _ | <u> </u> | - | 1 | _ | | | - | | 1 | - | Ġ | لزاة |
| | 186 | | ပိ | 8 | | | | | | | | | <u>.</u> | | | | | | | | | | | | |
| | TEST : IH 28 (ARC 3.5-195, | SET | FIER | 10 | 70 | 20 | 70 | 50 | 90 | 70 | 80 | 8 | 9 | 11 | 1 | 13 | 1 | 15 | 16 | 11 | - | - | - | DATA | |
| | EST: | DATA SET | DENTIFIER | REVOO1 | $\ ar{F}$ | <u> </u> | | L | _ | | _ | | - | _ | | _ | | | _ | - | | | | TYPE OF DATA | |
| | F | 1_ | ñ | ٧ | 1 | | <u> </u> | | <u> </u> | | | L,_ | <u> </u> | L | | | <u> </u> | <u> </u> | | <u> </u> | <u> </u> | <u> </u> | | 15 | |

TEST RUN NUMBERS DATE: JUNE, 1974 THERMOCOUPLE HOOKUP DATA SET/RUN NUMBER COLLATION SUMMARY MODEL (Concluded) 36 36 33 25 28 Õ न्त 27 32 8 K 39 PARAMETERS/VALUES NO. REPUNS TABLE II. % 0 140 5.3 a B Re* Ma 0 0 1.0 5.2 30 8 8 8 8 8 2 TEST : IH28(ARC 3.5-195) CONFIGURATION 1 O DATA SET 25 23 2 27 REVO18 20 21 77 74

| [/C location. | er surface | G - wing upper surface | H - vertical tail |
|---|--|------------------------|-------------------|
| he T | low | ddn | ica1 |
| bes | wing | wing | vert |
| scri | ı | ı | 1 |
| qe | 14 | ტ | H |
| identifier | OMS pcds | D - chine | E - canopy |
| set | 1 | 1 | ı |
| data | ပ | Ω | Þì |
| 4th character of the dataset identifier describes the T/C location. | underside fuselage C - OMS pcds F - wing lower surface | external tank | body sidewall |
| The 4 | 1 | t | 1 |
| Ħ | A | Н | m |
| | | | |

202

IDVAR (2)

IDVAR (1)

*~ XWO -6 PER FT.

COEFFICIENT SCHEDULES

OL ORBITER HERHOCOUPLE HOOKUP (COUST.SET 100)

L-TANK THERHOCOUPLE HONKUP (COUST.SET 200)

TYPE OF DATA

TABLE III MODEL DIMENSIONAL DATA

| MODEL COMPONENT: BODY - B22 | | |
|---|------------------------|-------------|
| GENERAL DESCRIPTION: Fuselago, Configu VL70-000147B. | ration 3A per Rockwell | Lines |
| NOTE: Identical to Bl9, except unders | ide. | |
| Model Scale = 0.006 | | |
| DRAWING NUMBER: V1.70-000147 | <u>B</u> | |
| DIMENSIONS: | FULL-SCALE | MODEL SCALE |
| Length - in | 1290.3 | 7.742 |
| Max. Width - in | 267.6 | 1.606 |
| Max. Depth - in | 244.5 | 1.467 |
| Fineness Ratio | 4.84601 | 4.84601 |
| Area' - Ft ² | · | |
| Max. Cross-Sectional | | 0.0139 |
| Planform | ***************** | |
| Wetted . | | |
| Base | • | • |

TABLE II: (Continued) MODEL DIMENSIONAL DATA

| MODEL COMPONENT: Canopy - | <u>C7</u> | | |
|-------------------------------------|---------------|----------------|-------------|
| GENERAL DESCRIPTION:Configu | ration 3 per | Rockwell Lines | VL70-000139 |
| Model Scale = 0.006 | | | |
| DRAWING NUMBER . VI | 70-000139 | | |
| DIMENSION: | | FULL SCALE | MODEL SCALE |
| Length ($X_0 = .433$ to $X_0 = 67$ | 0) - in. FS | 237 | 1.422 |
| Max Width | | | |
| Mox Depth $(Z_0 = to Z_0)$ | , = 501) - in | FS | |
| Fineness Ratio | | | |
| Area | | | |
| Max Cross-Sectional | | | |
| Planform ' | | | |
| Wetted | | | |
| Base | | | • |

TABLE III (Continued) MODEL DIMENSIONAL DATA

| model component: | F5 Body Flap | | |
|--|--------------------|-------------------|-------------|
| GENERAL DESCRIPTION: | 3 Configuration pe | er Rockwell Lines | VL70-000139 |
| - | | | |
| Scale Model = 0.006 | | | |
| DRAWING NUMBER | VL70-000139 | | • |
| DIMENSION: | | FULL SCALE | MODEL SCALE |
| Length - in | • | 84.70 | .508 |
| Max Width - in | | 267.6 | 1.606 |
| Mox Depth | | | |
| Fineness Ratio Area - Ft ² | | | 4 |
| Max Cross-Sec | ctional . | | |
| Planform | | 142.5 | .005 |
| Wetted | | 38.0958 | .0014 |
| Base ' | | 30.0770 | |

TABLE III (Continued) MCDEL DIMENSIONAL DATA

| MODEL COMPONENT: ONS Pod - ML | | |
|--|------------------------|---------------|
| GENERAL DESCRIPTION: Configurati | on 3 per Rockwell Line | s V170-000139 |
| NOTE: M4 identical to M3, except in | tersection to fuselage | • |
| Model Scale = 0.006 | | |
| DRAWING NUMBER VL70-C | 000139 | • |
| DIMENSION: | FULL SCALE | MODEL SCALE |
| Length - IN | 346.0 | 2.076 |
| Max Width - IN | 108.0 | .648 |
| Max Depth - IN | 113.0 | .678 |
| Fineness Ratio Area - FT ² | | |
| Max Cross-Sectional | | |
| Planform . | | |
| Wetted | | |
| Dava. | • | • |

TABLE III (Continued) MODEL DIMENSIONAL DATA

| MODEL COMPONENT: T8 - EXTERNAL TANK | · · · · · · · · · · · · · · · · · · · | |
|---|---|-------------|
| GENERAL DESCRIPTION: 2A Configuration p VI.79-000018 and VL72-000061"C" Body of | | |
| Scale Model = 0.006 | | |
| DRAWING NUMBER VL'13-0060 | 018 | |
| DIMENSION: | FULL SCALE | MODEL SCALE |
| Length - In. (Nose @ $X_T = 309$) | 186.50 | 1.119 |
| Max Width (Dia) - In. | 324.0 | 1.944 |
| Max Depth | *************************************** | |
| Fineness Ratio L/D | 6.1389 | 6.1389 |
| Area - Ft. ² | | |
| Max Cross-Sectional | 572.56 | 0.0206 |
| Planform | | |
| Wetted | | *** |
| Base | • | |
| WP of tank centerline, (Zm) In. | 400.0 | 2.400 |

TABLE III (Continued) MODEL DIMENSIONAL DATA

| MODEL COMPONENT: VERTICAL - V 7 | | • |
|--|---|---|
| GENERAL DESCRIPTION: Centerline Vertical Tail, | Doublewedge Airfo | oil |
| with Rounded Leading Edge | | |
| NOTE: Same as V5, but with manipulator housin | g removed. | |
| Model Scale = 0.006 | | |
| DRAWING NUMBER: VL70-000139 | | , |
| DIMENSIONS: | FULL-SCALE | MODEL SCALE |
| TOTAL DATA | • • • | |
| Area (Theo) Ft ² Planform Span (Theo) In Aspect Ratio Rate of Taper Taper Ratio Sweep Back Angles, degrees Leading Edge Trailing Edge 0.25 Element Line Chords: Root (Theo) WP Tip (Theo) WP MAC Fus. Sta. of .25 MAC | 425.92 315.72 1.675 0.507 0.404 45.000 26.249 41.130 268.50 108.47 199.81 | 0.0153 1.894 1.675 0.507 0.404 45.000 26.249 41.130 1.611 0.651 1.199 |
| Fus. Sta. of .25 MAC W. P. of .25 MAC B. L. of .25 MAC Airfoil Section Leading Wedge Angle Deg Trailing Wedge Angle Deg Leading Edge Radius Void Area Blanketed Area | 1463.50 635.522 -0.00 10.000 -14.920 -2.0 -13.17 -0.00 | 8.781 3.813 0.00 10.000 14.920 2.0 0.0005 0.00 |

TABLE III (Concluded) MODEL DIMENSIONAL DATA

| GENERAL DESCRIPTION: Configuration 3A per Rockwell NOTE: Identical to W107, except lowered 3.5" and a | · | |
|---|---|---|
| | | |
| Model Scale = 0.006 | | |
| TEST NO. | DWG. NO. VL | 70-000147B |
| DIMENSIONS: | FULL-SCALE | MODEL SCALE |
| Area (Theo.) Ft2 Planform Span (Theo In. Aspect Ratio Rate of Taper Taper Ratio Dihedral Angle, degrees (@ T.E. of Elevon) Incidence Angle, degrees Aerodynamic Twist, degrees Sweep Back Angles, degrees Leading Edge Trailing Edge 0.25 Element Line Chords: Root (Theo) B.P.0.0. Tip, (Theo) B.P. MAC Fus. Sta. of .25 MAC W.P. of .25 MAC | 2690.00 936.68 2.265 1.177 0.200 3.500 0.500 +3.000 -10.24 35.209 689.24 137.85 474.81 1136.89 295.70 | 0.0968 5.620 2.265 1.177 0.200 3.500 0.500 +3.000 45.000 -10.24 35.209 4.135 0.827 2.49 6.821 1.774 |
| B.L. of .25 MAC EXPOSED DATA Area (Theo) Ft Span, (Theo) In. BP108 Aspect Ratio Taper Ratio Chords | 182.13 1752.29 720.68 2.058 0.2451 | 1.093 0.063 4.324 2.058 0.2451 |
| Root BP108 Tip 1.00 b MAC Fus. Sta. of .25 MAC W.P. of .25 MAC B.L. of .25 MAC Airfoil Section (Rockwell Mod HASA) | 562.40 137.85 393.03 1185.31 296.70 251.76 | 3.374 0.827 2.358 7.112 1.780 1.511 |
| XXXX-64 Root b = @ Yo 199 to NACA 0010 Tip b = | 0.10 | 0.10 |
| Data for (1) of (2) Sides Leading Edge Cuff 2 Planform Area Ft Leading Edge Intersects Fus M. L. @ Sta Leading Edge Intersects Wing @ Sta | 118.333 500.C 1083.5 | 0.0043 3.000 6.501 |

TABLE IV ORBITER THERMOCOUPLE LOCATIONS

| _ | 10 | Skin | | LC | CATI | ON | | | T/C | Skin | | | | LOC | | | | |
|--------------|-------------|--|--------------|--------------|----------|---------------|--|--|------|-------|--------|--------|----------|------|-------------|----------|------|----------|
| | /C | Thick | x/1 | x/c. | у | T | b/2 | 2 | No. | Thick | | x/ | 1 | x/c | y | | b/2 | 2 |
| | 1 | .035 | | | | | | | 35 | .0 | 35 | .6 | 0 | | | | | |
| | 2 | | .050 | | | | | | 36 | .0 | 34 | .7 | 0 | | | Ц | | |
| | 3 | | .075 | | | | | | 37 | .0 | 32 | .70 | | | | | | |
| | 4 | .034 | | 1 | | | | | 38 | ٠. | .035 | | 0 | | | | | |
| - | 5 | | .125 | | | | | | 39 | | 1 | ٤. | 25 | | 4 | | | |
| - | 6 | | .150 | | | | | | 40 | | | ٤. | 25 | - | OMS | P | DDS | |
| | 7 | .034 | | 1 | NDE | เลร | DE | | 41 | | | 3. | 325 | | | | | |
| \vdash | 8 | T | .20 | 1 | USE | Aq | EZ | | 42 | | | ٠, | 90 | | | | | |
| <u> </u> | <u> </u> | .035 | .25 | | вР | = (| 0.0 | | 43 | | | .9 | 90 | | | | | |
| - | .0 | + | .30 | | 1 | | | | 44 | | 1 | 1.9 | 90 | | | V | | |
| - | .1 | .034 | | | | . | | | 45 | | 35 | 1. | 10 | | | 4 | | |
| | .2 | .035 | · | · | | | | | 46 | | | • | 15 | | CH: | INE | | |
| - | 13 . | | .65 | <u> </u> | 1 | T | | | 47 | | | Ŀ | 20 | | <u> </u> | 1 | | |
| | 14 | | .80 | | | | | | 48 | | | | 17 | | | NOP | | |
| — | 15 | .036 | .95 | 1 | | 1 | | | 49* | | t | | 425 | | MID | FU | SE | <u> </u> |
| - | 16 | .030 | .35 | <u> </u> | | 1 | | | 50 | • | .031 | | <u> </u> | .05 | ļ | | 40% | |
| - | 17 | .027 | .40 | UNDERS | | DE | 1. | 51 | | 030 | | | .10 | | | 40% | 3 | |
| - | 18 | 1-1- | .50 | | FUSE | LA | GE | | 52 | • | 030 | | | .20 | | | 40% | |
| | 19 * | 1-1- | .60 | | BP | = | 117.0 | 0 | 53 | Τ. | 029 | | | .30 | | | 409 | 6 |
| 1 | 20 | 1 | .70 | | 1 | 1 | | | 54 | • | 028 | W | NG | .40 | | | 409 | 6 |
| | 21 * | 028 | 3 :80 | | | T | | | 55 | | | LV | IR | .50 | | | 409 | <u> </u> |
| + | 22 | .03 | | | - | 十 | | | 56 | * | \top | | | .60 | <u> </u> | | 409 | B |
| }- | 23 * | | 6 1.00 | , | | 1 | | _ | 57 | | 1 | | | .70 |] | | 409 | 78 |
| - | 24 - | | 4 .30 | | | 1 | 1 | | 58 | * . | .029 | 9 | | .80 | 1_ | | 40 | 76 |
| - | 25 | .03 | | - | 1. | 十 | 1 | | 59 | T | ł | | Ł | .90 | | | 40 | 70 |
| ŀ | 26 | .03 | | | В | ODY | , | | 60 | 7 | .03 | 4 | 4 | .10 | | | 60 | 70 |
| ł | 27 | .03 | | | | | VALL | | 61 | Ţ | .03 | 2 | | .20 | | | 60 | % |
| ł | 28 | .03 | _ | - | | 1 | 1 | | , 62 | * | .03 | 1 | | .30 | 1_ | , m, m- | . 60 | % |
| H | 29 | .03 | | _ | | + | | - | 63 | * | .03 | o w | ING | .40 | | | 60 | 90 |
| - | 30 | 1.03 | .50 | | _ | + | 1 | | 64 | * | 1 | L | WR | .50 | | | 60 | % |
| - | 31 | .03 | | | | + | | | 65 | 1 | 1 | | 1 | .60 | | | 60 | 976 |
| } | 32 | .03 | | | + | 十 | + | 1 | 66 | * | + | | 1 | .70 | | | 60 |)% . |
| | 33 | 1:03 | .6 | | + | + | | _ | 67 | 7# | 1 | \top | 1 | .80 | T | | 60 |)% |
| | | _ | | | | - | _ | | 68 | 7 | - | _ | 1 | .90 | 1 | | 60 |)% |
| | 34 | .03 | .60 | | | 7 | | | | | | | | 1.70 | | | | |

| | ADDITION | THERMOCOUPLE LOCATIONS | (Concluded) | 1 |
|----------|----------|--|-------------|---|
| TARLE IV | OUDTIEN | T THE PARTY OF THE | | |

| | - | कार्य का | APRT | TER TE | TERMOÖO! | UPLE | LOCAT | ions | (Concluded) | | | | | | | | | |
|----------------|---------------|---|----------------|--------------|----------|------|----------|--------|-------------|----------|--------------|--------------|---------------|--|--|--|--|--|
| | | TV PARTY | LO | CATION | 1 | | T/C | 1 2271 | 8 1 | | CATIO | 1-10 | | | | | | |
| · • 1 | Skin Thick | - | x/c | y | b/2 | 2 | No. | Thick | < x/1 | x/c | V. | 16/2 | 1 Z | | | | | |
| | | ~- | .20 | | 80% | | | | | | | | ! | | | | | |
| 00 | .034 | | .40 | | 80% | | | | | | | | | | | | | |
| 70 | | WING | | | 80% | 1 | | | | | | | | | | | | |
| 71 * | | LWR | .60 | | 80% | 1 | | | | | | | | | | | | |
| 72 | .035 | | .80 | <u> </u> | 40% | + | _ | _ | | | | | | | | | | |
| 73 | .035 | | .20 | | 40% | + | | | | | | | | | | | | |
| 74 | | WING | .40 | ļ | 40% | | | | | | | | | | | | | |
| 75 | .035 | UP | .60 | | | +- | | _ | _ | | | | <u> </u> | | | | | |
| 76 | .037 | 1 | .80 | | 40% | + | | | | | | | | | | | | |
| 77 | .034 | | .20 | | (0% | | | | | <u> </u> | 1 | | | | | | | |
| 78 | .030 | WING | .40 | | 60% | | | | | - | 1 | | | | | | | |
| 79 | .03 | | .60 | | 60% | - | | | | _ | _ | | 7 | | | | | |
| 80 | .03 | 1 | .80 | | 60% | | _ - | | | | | | 1 | | | | | |
| 81 | .02 | 7 | .20 | | 80% | | _ _ | | | | | | 1 | | | | | |
| 82 | .02 | 8 WING | .40 | | 80% | | | | | | | | - | | | | | |
| 83 | | 8 UP | .60 | | 80% | | | | | | | | | | | | | |
| 84 | .02 | | .80 | | 80% | | | | | | | | - | | | | | |
| | .03 | | .25 | | | 3. | 57 | | | | | | | | | | | |
| 85 | | 10 VER | | | | 3. | 57 | | | . i | | | | | | | | |
| 80 | .0: | | .75 | | | В. | .57 | | | | | | | | | | | |
| 87 | | | | | | 4. | .42 | | | | | | | | | | | |
| 88 | | 33 VER | | | _ | 4 | .42 | | | | | | | | | | | |
| 89 | - 0 | 34 | - • | | | | | | | | | | - | | | | | |
| | | | | | | | | | | | | | | | | | | |
| <u> </u> | | | | | | | | | | | | | | | | | | |
| | | | | | ed at t | hogo | m/C 1 | ocatio | ns. | | | | | | | | | |
| · | Dat | ta vere | not o | btaine | ea at c | nese | | | | | | | | | | | | |
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| $\cdot \Gamma$ | | • | | | | | | | | | | | | | | | | |
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| l | | | | | | | | | | | | | | | | | | |
| [| | | | | I | | | | | | | | 1 | | | | | |

TABLE V EXTERNAL TANK THERMOCOUPLE LOCATIONS

| | | TABLE | A EVIENN | | | | | | | | | | | |
|---|---|--|---|-------|---|---|---|--|--|--|--|--|--|--|
| t/c no. | SKIN THICK. | LOCA X/l | PION DEG. | | EKIN THICK | LOCATION X/1 | o:i p deg. | T/C NO. | SKIN THICK. | I,OCAT | ΙΟΝ Φ DEG. | | | |
| 1 2 3 4 5 6 7 8 9 10 1 12 13 14 5 6 7 8 9 10 1 12 13 14 5 6 7 8 9 10 1 12 13 14 5 6 7 8 9 10 12 12 13 14 5 6 7 8 9 10 12 12 13 14 5 6 7 8 9 10 12 12 13 14 15 15 15 15 15 15 15 15 15 15 15 15 15 | 0.030 0.030 0.030 0.039 0.028 0.028 0.028 0.030 0.030 0.030 0.030 0.030 0.030 0.030 0.030 0.030 0.030 0.030 0.030 0.030 0.030 0.030 0.030 | 0.005 0.010 0.020 0.04 0.06 0.08 0.10 0.125 0.175 0.20 0.25 0.275 0.275 0.275 0.30 0.30 0.30 0.325 0.325 | 112.5 90 180 135 112.5 90 67.5 180 | 35 | 0.032 0.033 0.033 0.039 0.031 0.031 0.031 0.031 0.033 0.033 0.033 0.033 0.031 0.031 0.031 0.031 0.031 0.031 0.031 0.031 0.031 | 0.40 0.440 0.445 0.445 0.455 0.555 0.555 0.555 0.555 0.555 0.60 0.60 | 135 112.5 67.5 180 157.5 180 157.5 112.5 90 67.5 180 157.5 180 157.5 180 157.5 180 157.5 180 157.5 180 157.5 180 157.5 180 157.5 180 180 157.5 180 180 157.5 180 180 157.5 180 180 157.5 180 180 180 180 180 180 180 180 180 180 | 71 72 73 74 75 76 77 78 79 81 82 83 84 85 86 87 88 89 99 19 19 19 19 19 19 19 19 19 19 19 19 | 0.030 0.033 0.032 0.031 0.039 0.031 0.033 | 0.60 0.65 0.65 0.65 0.65 0.65 0.65 0.65 | 45 0 180 187.5 135.5 5 180 157.5 135.5 5 180 157.5 135.5 5 180 157.5 135.5 12.5 135. | | | |
| . * [| ata wer | e not o | btained a | these | T/C 10 | cations | • | 107 108 109 110 | 0.031 0.039 0.029 0.033 | 0.90 0.90 0.90 | 90 67.5 45 180 180 | | | |

TABLE VI RUN NUMBER/TUNNEL CONDITION SUMMARY

| | | It of | 1500 | | | | | | | | | | | | | | | _ | #* - | | <u>.</u> | | | | | | - | - | | | |
|---------|---------|----------|-------------|--------|-------|--------|---------|------|-------|-----------------|-------------|-------|----------|-------|------------|--------------------|-------------|-----------|-----------------|-------|----------|-------------|----------------------|---------|-------|------------|-------|------------|----------------------------|---------------------------------------|-----------|
| Nowinal | | Pt psia | 100 | | | | - 3 | 014 | 001 | 001 | 001. | 410 | 410 | . 014 | 00. 100 | | | | | - 11- | - | | - | | | • | | _ | | | |
| | Re /ft | × 10-6 | 1.0 | | | | | 0.4 | 1.0 | | - | 0.4 | 0.4 | 0.4 | 1.0 | _ | - | | | | | | | <u></u> | | <u></u> | | - | | | |
| | | ∑ | 5.22 | | | | | 5.3 | 5.22 | | | 5.3 | | - | 5.55 | - | | | | - | | | | | | | | - | | s)/ce | 334 |
| | Const | Set** | 81 | 200 | 200 | % % | 100 | 100 | 100 | 100 | 800 800 | 800 | 100 | 200 | 200 | 100 | 100 | 00 800 | 8 | 100 | 901 | 200 | 00 00 00 00 | 8 | 700 | 200 200 | 100 | 100 | 37, 28" | -hiter T/Cs: 200 - External Tank T/Cs | the Amney |
| | | 8 | 0 | | | | | | - | ر ً. | <u>.</u> | 0 | - | | - | | | | | | | | | | | | | - | o 6 | Exter | |
| | astmit. | deg. | 0 | 0 | -10.0 | 20.0 | 20.0 | 20.0 | -10.0 | * | * | -10.0 | -10.0 | 20.0 | -50.0 | -20.0 | 10.0 | 10.0 | -10.0 | -10.0 | 20.0 | 20.0 | 20.0 | 20.0 | -10.0 | -10.0 | -10.0 | 20.0 | , αstrut = | ćs: 200 - | |
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| | | Config | 0, + 13, | - - | . • | | | | | | | | | | <u>-</u> | ,,,,,,, | | | | | | | • | | | - | 5 | 1 6 | $*\alpha_{m} = 30^{\circ}$ | * 100 L | *^ _ ^^ |
| | | Run No. | - | ι α |) (C | 14 | 7 | 'n | • | · c c | σ | 10 | 1 | 12 | 13 | 74 | <u>.</u> 51 |) 1 | 17 | 18 | 19 | 8 | 23 | 8 | 23 | 77 | 25 | % | | | • |

*** Actual test values are given in the Appendix

TABLE VI RUN NUMBER/TUNNEL CONDITION SUMMARY (Concluded)

| | Tt OR | 1500 | | | | | | | | - | _ | - | | - |
|------------|---|---------|-------|-------------|--------------|------------|------------|-----|--------------|----------|-----------|------------|------------|-------|
| Nominal*** | Pt psia | 100 | | • | | | | | | | | | - | 410 |
| | Re _∞ /ft x 10 ⁻⁶ | 1.0 | • | - · · · · · | | | | | | | | | - | 0.4 |
| | \mathbf{x}_{8} | 5.22 | _ | | | | | | | | | | - | 5.3 |
| | Const Set | 100 | 100 | 100 | 100 | 200 200 | 500 | 200 | 100 | 100 | 100 | 500 | 800 800 | 800 |
| | B | 0 | _ | | <u> </u> | | | | | | | | | - |
| | astrut, deg. | 20.0 | -10.0 | -20.0 | 10.0 | 10.0 | -20.0 | 0 | 0 | -10.0 | 20.0 | -10.0 | 20.0 | -10.0 |
| | απ. deg. | 120 | 8 | 9 | -30 | -30 | 9 | 0 | 0 | 8 | 9 | 96- | -120 | 8 |
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| | Run No. | 27 | 88 | % | <u>&</u> | 닭 | , X | 33 | * | 35 | * | 37 | % | 8 |

** 100 - Orbiter T/Cs; 200 - External Tank T/Cs *** Actual test values are given in the Appendix

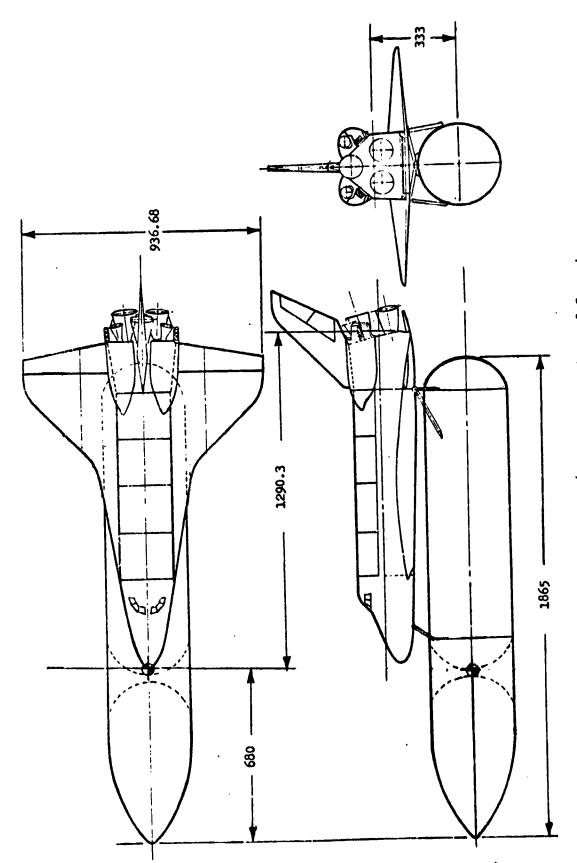
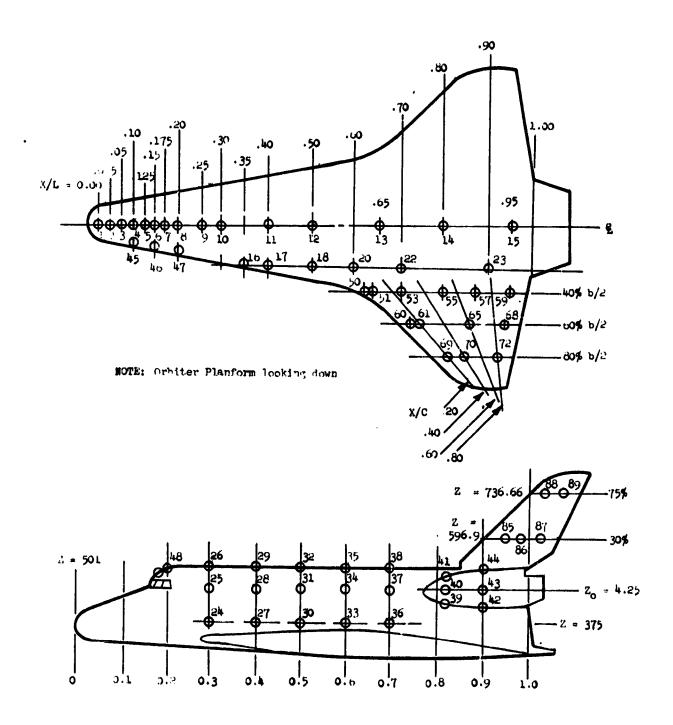
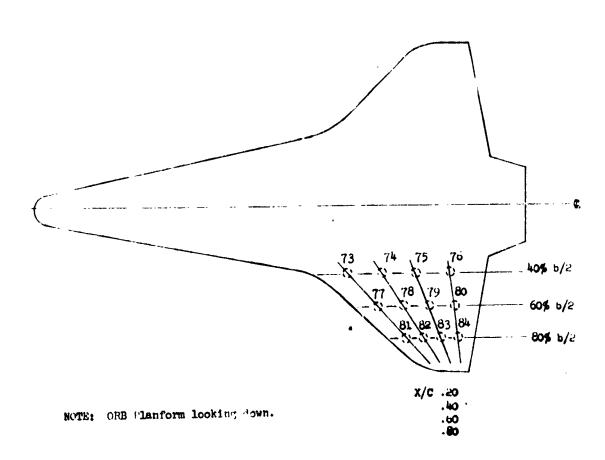


Figure 1. Orbiter/External Tank General Layout

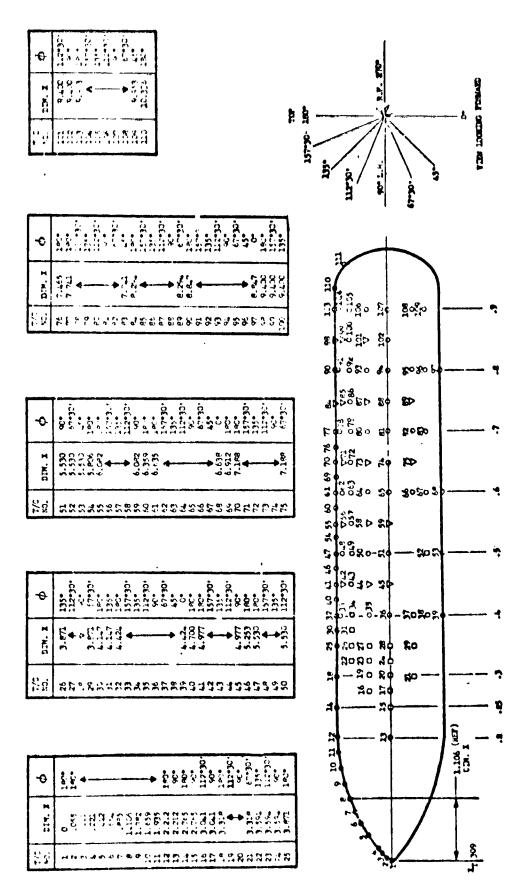


a. 50-0 Orbiter -- 147-B Configuration Thermocouple Locations Figure 2. Model Instrumentation Sketches

Upper Surface (Left Wing) Instrumentation

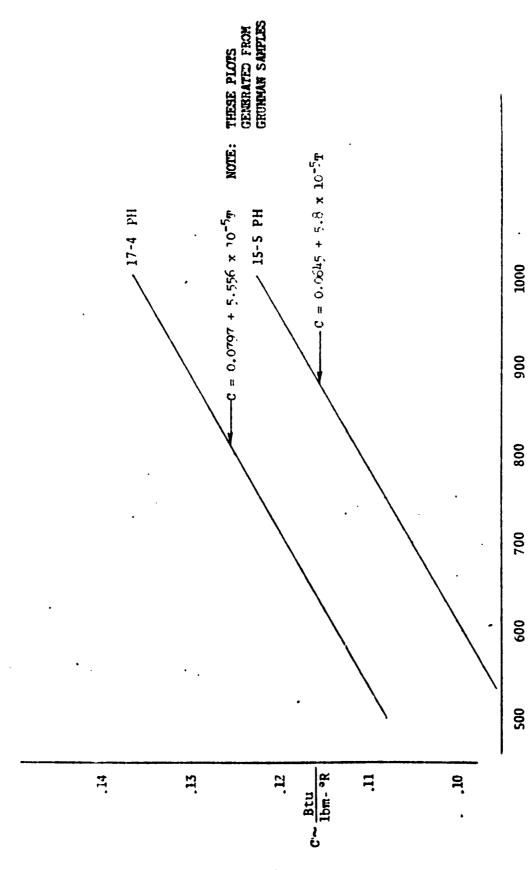


s. 50-0 orbiter -- 147-B Configuration Thermocourle Locations Figure 2. Model Instrumentation Sketches (Concluded).

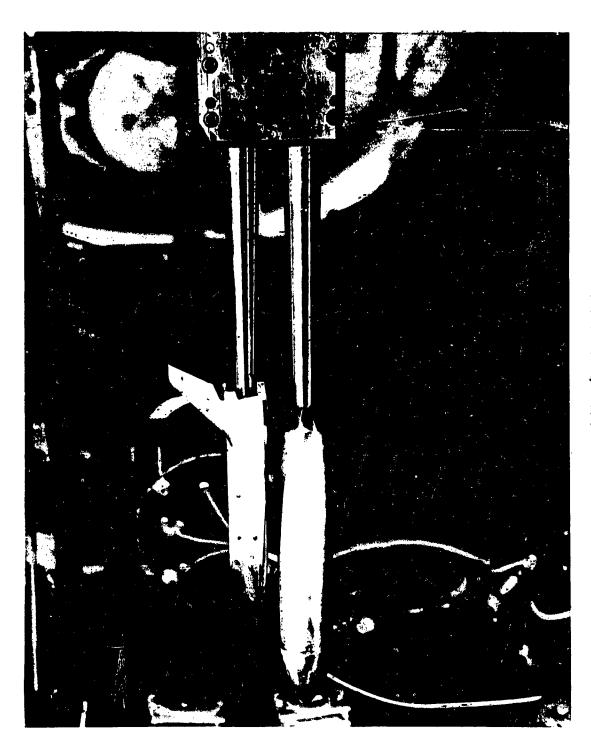


b. 41-T 0.006-Scale External Tank T/C Locations Fingre 0. Youel Instrumentation Cretifies

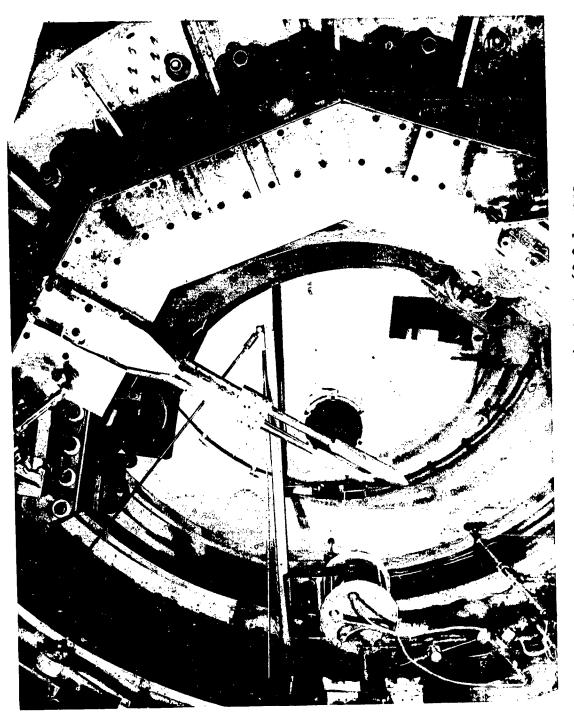
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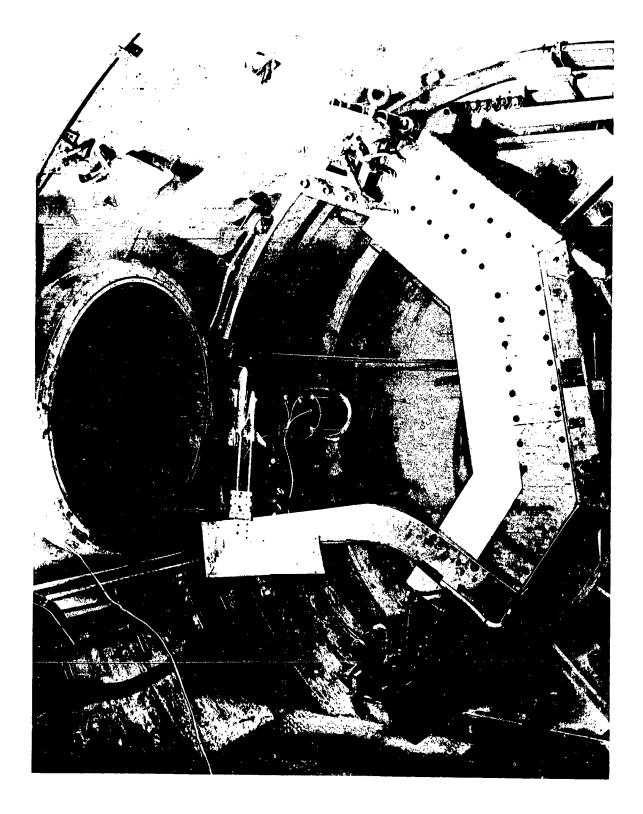
c. Specific Heat vs. Temperature for 17-4PH and 15-5PH stainless steel Figure 2. Model Instrumentation Sketches



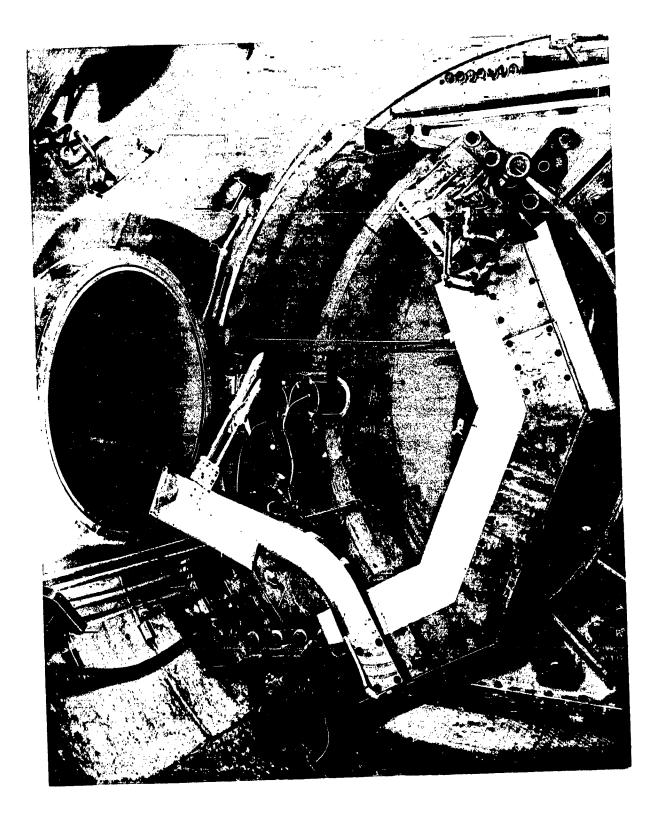
a. Orbiter/Tank at 0.0 degrees Figure 3. Model Installation Photographs



b. Orbiter/Tank at -60.0 degrees Figure 3. Model Installation Photographs



c. Orbiter/Tank at 90.0 degrees
Figure 3. Model Installation Photographs



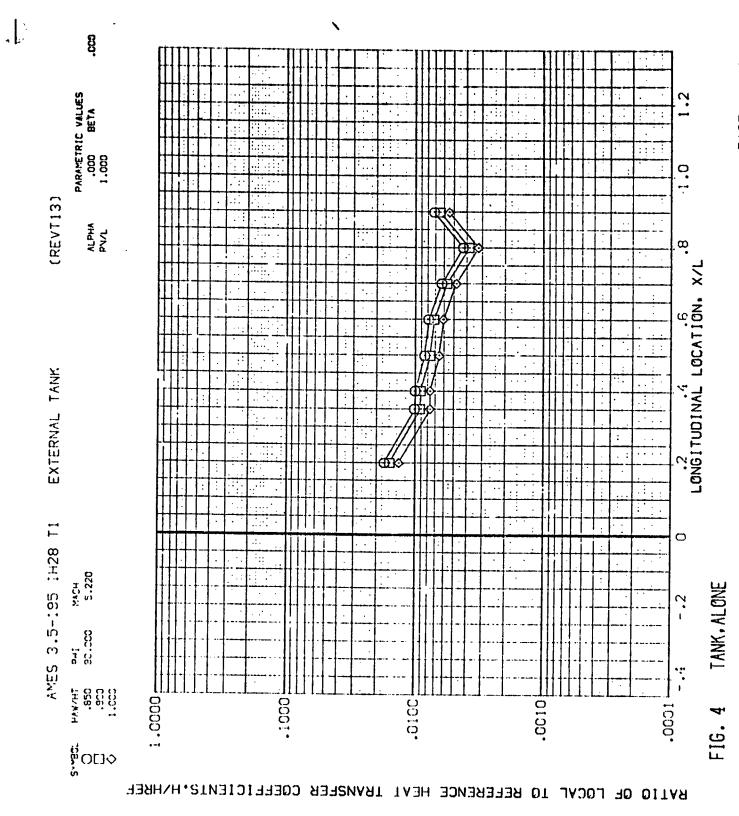
d. Orbiter/Tank at 120.0 degrees
Figure 3. Model Installation Photographs

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DATA FIGURES

Volume 1 - Figures 4-15

Volume 2 - Figures 16-27



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FIG. 4 TANK, ALONE

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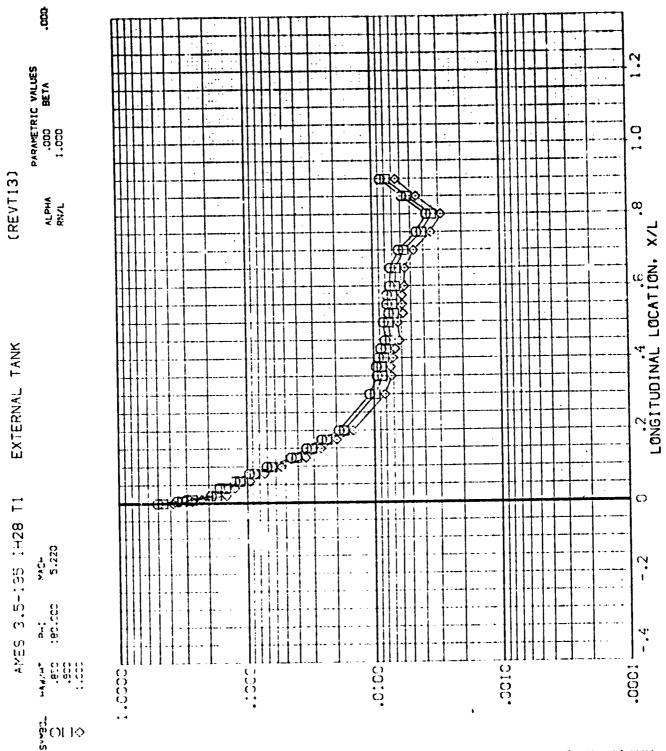
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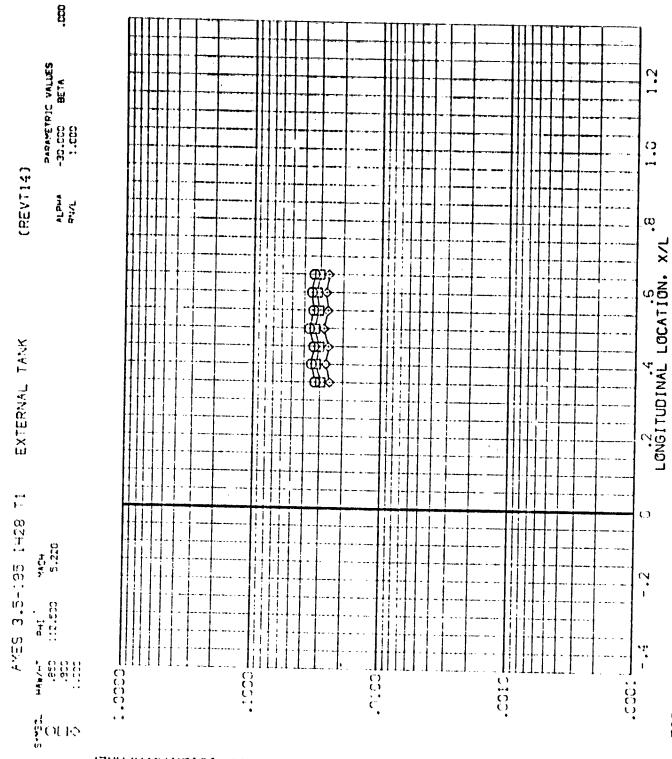


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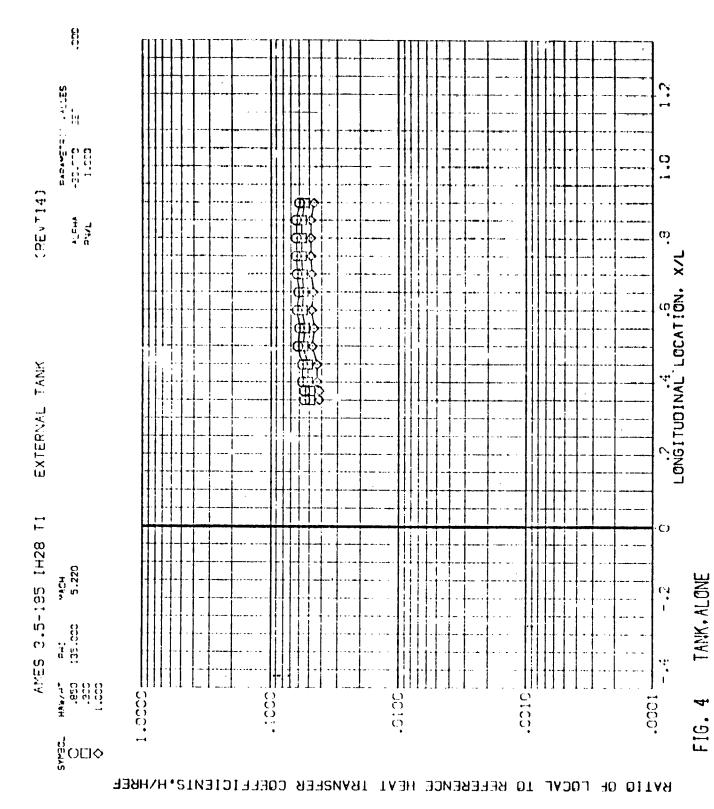
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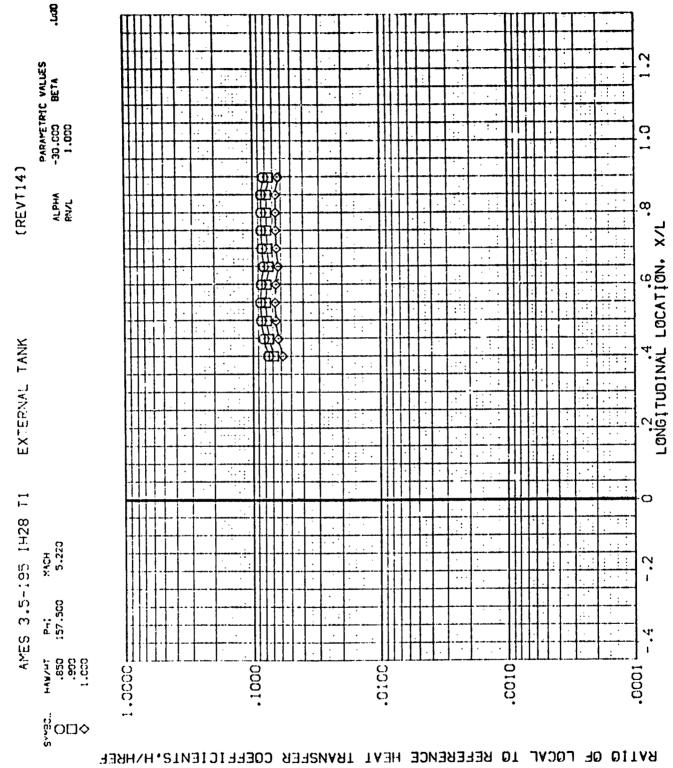


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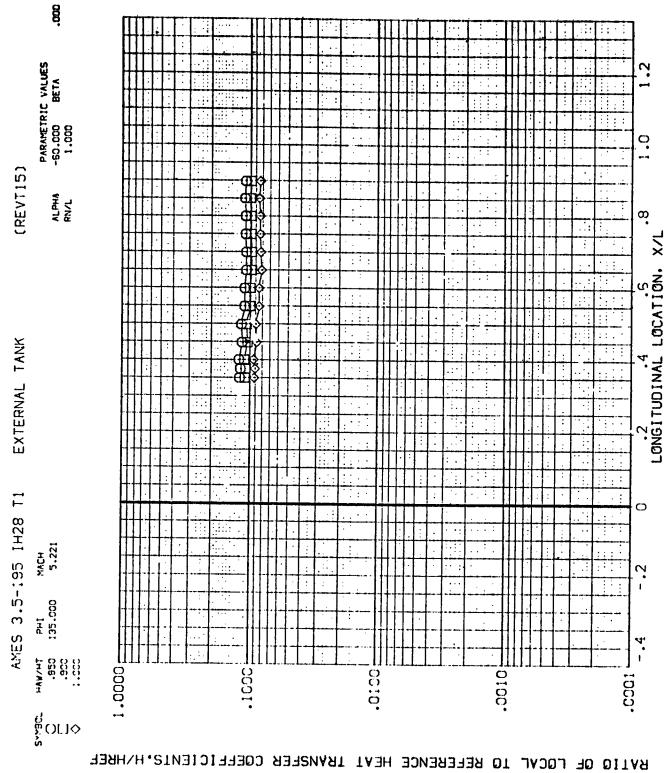
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FIG. 4 TANK, ALONE

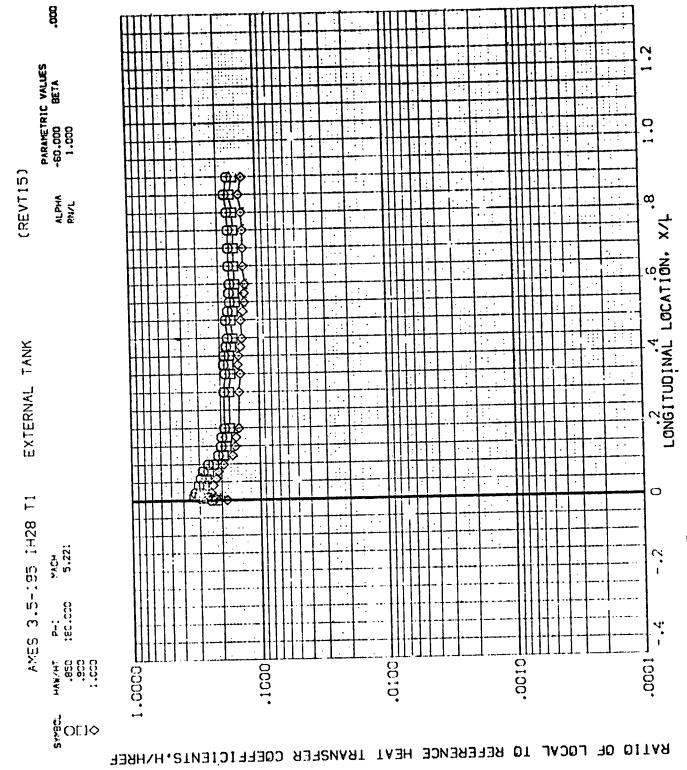
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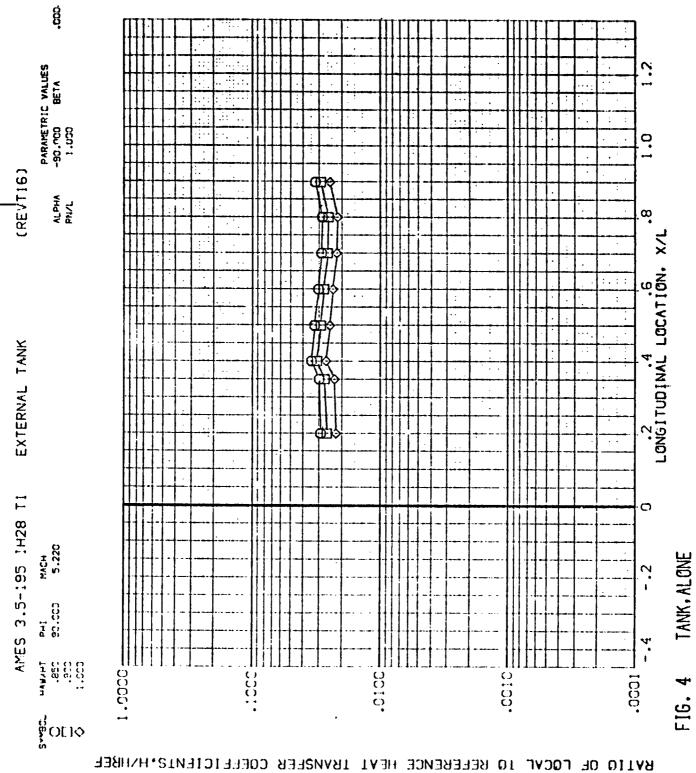


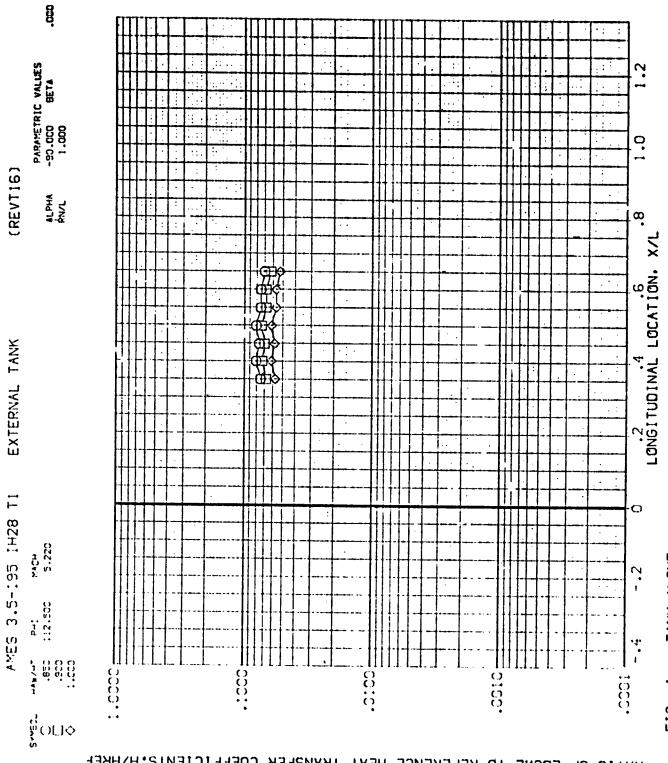
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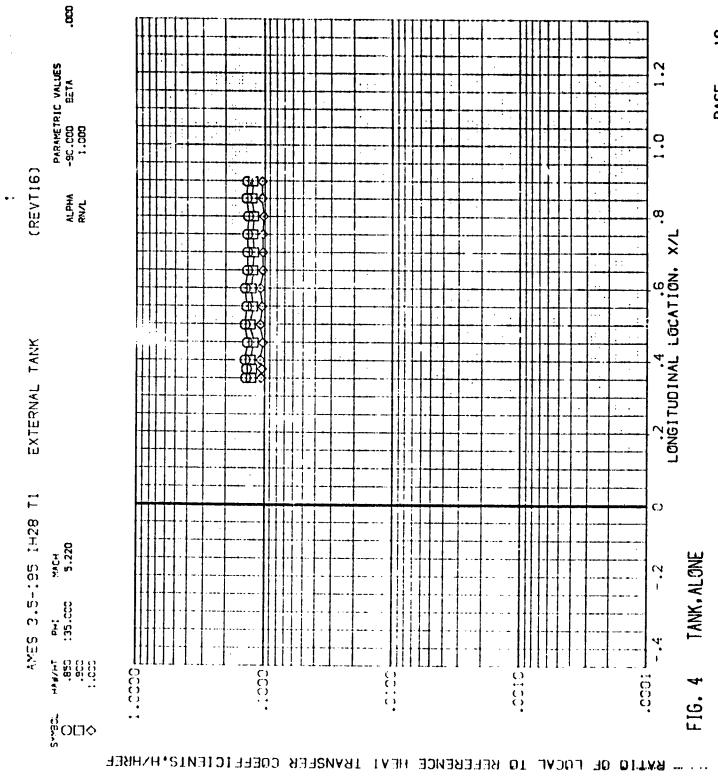


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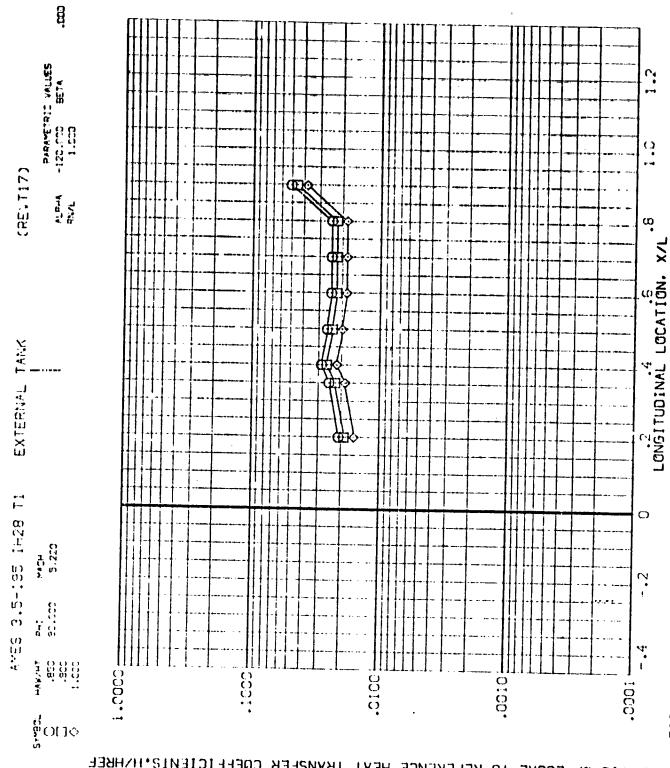
FIG. 4 TANK, ALON

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FIG. 4 TANK, ALONE



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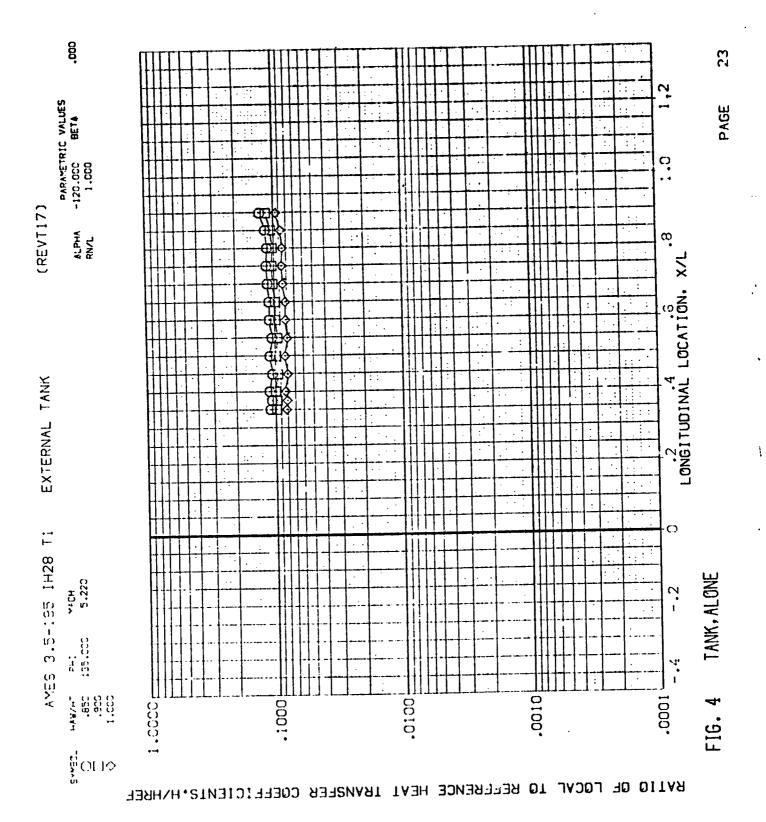
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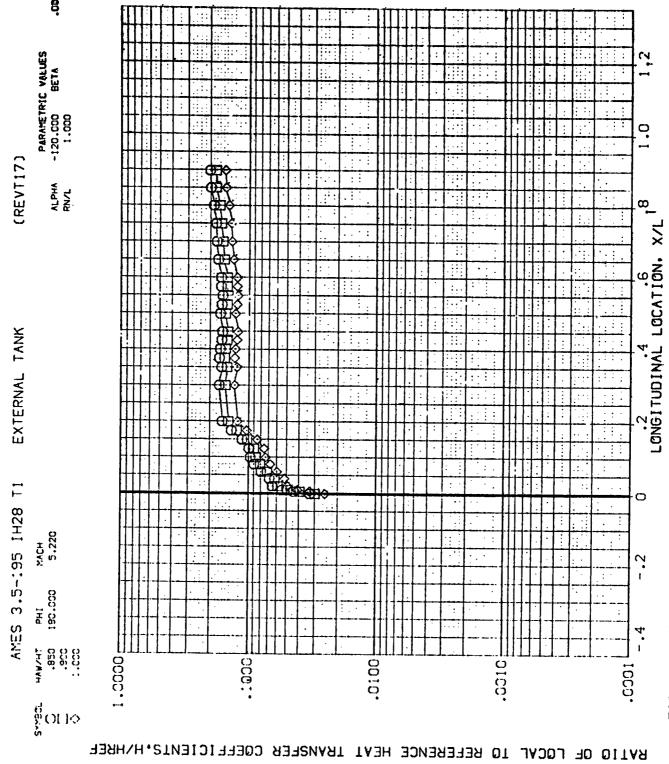


FIG. 4 TANK, ALONE

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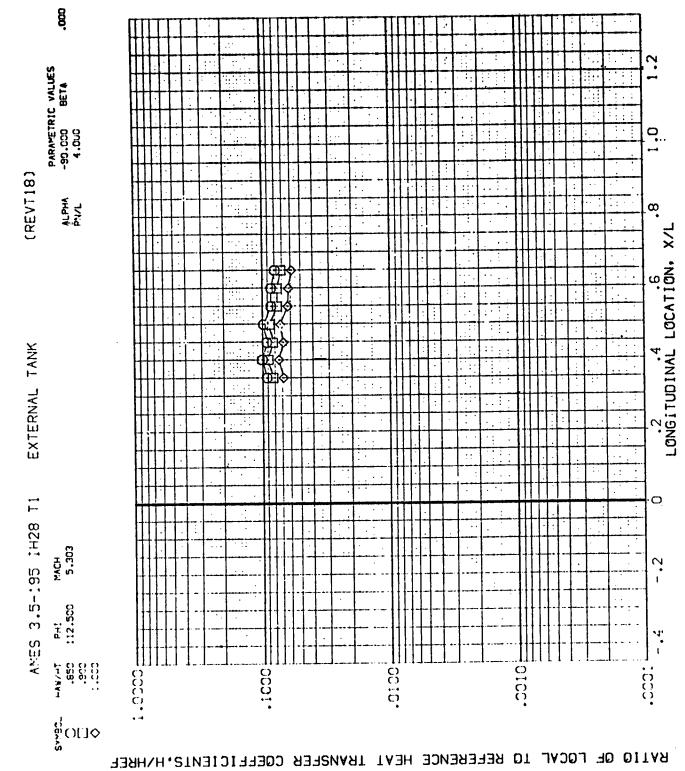
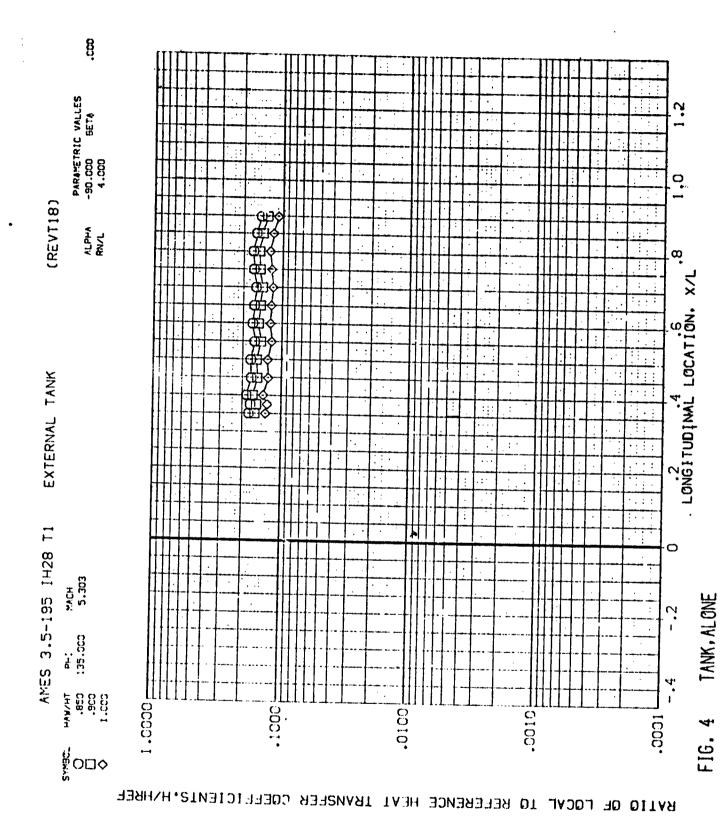
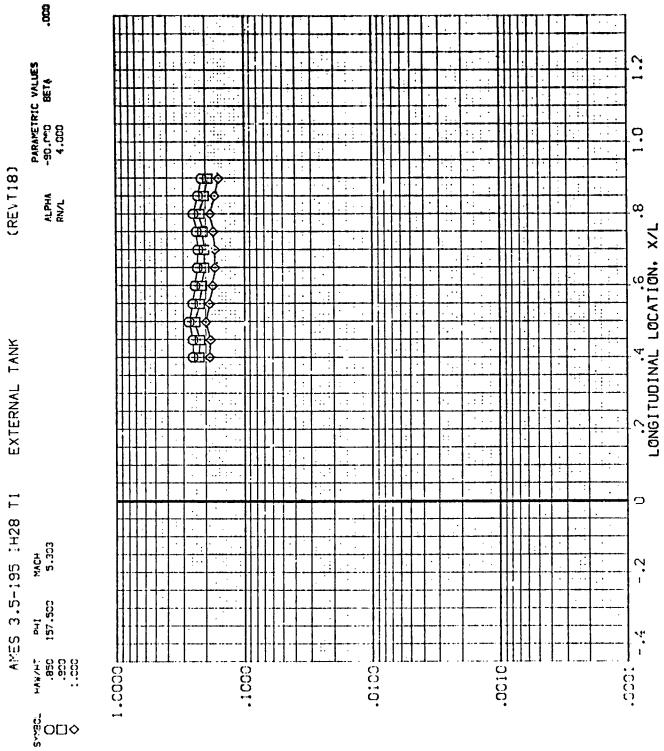


FIG. 4 TANK, ALONE

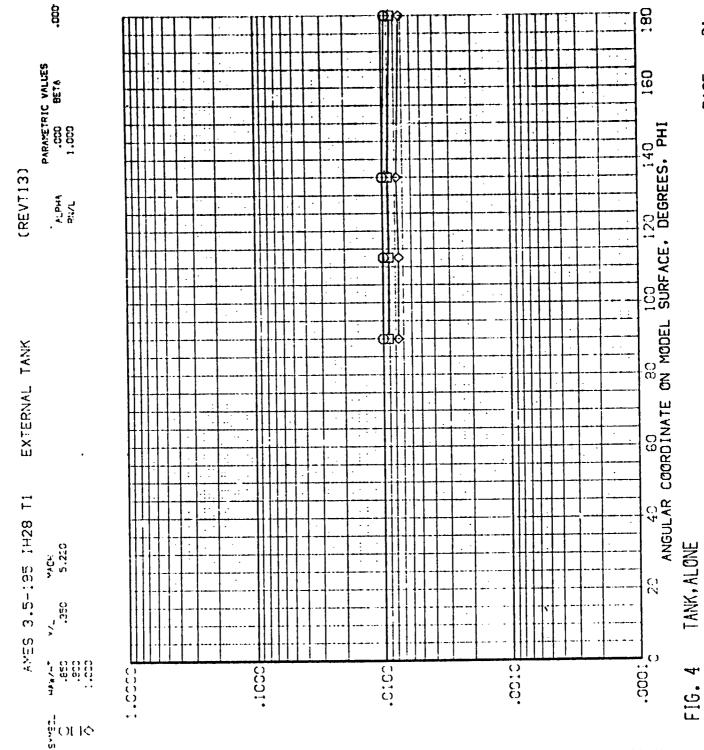




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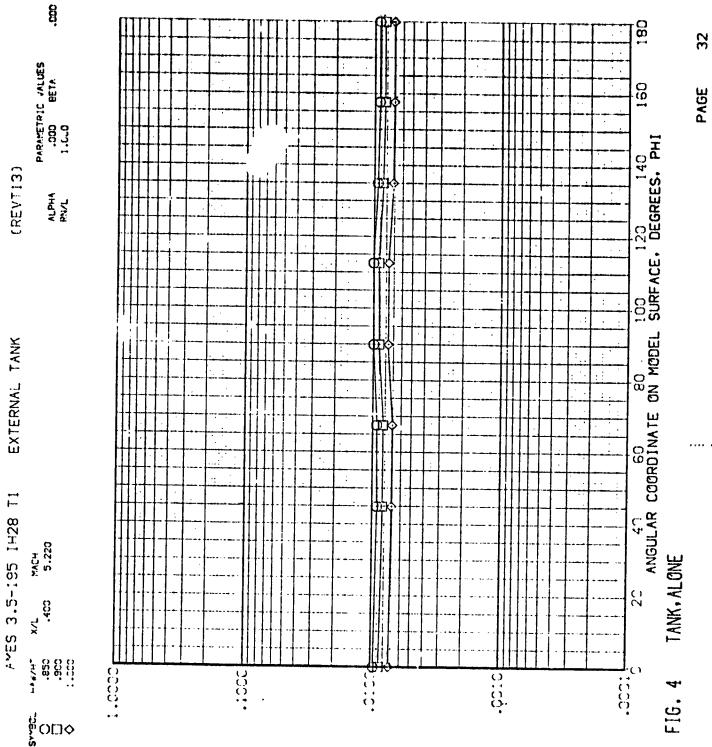
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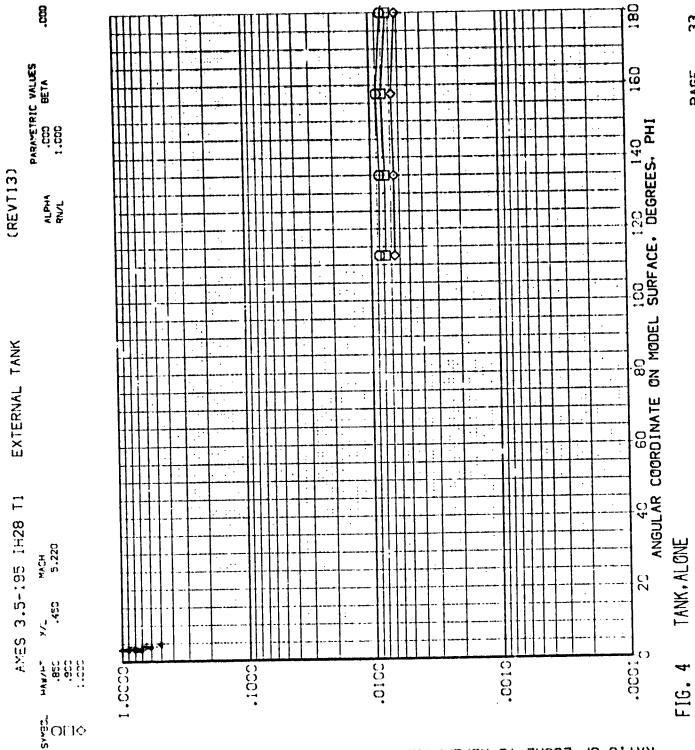


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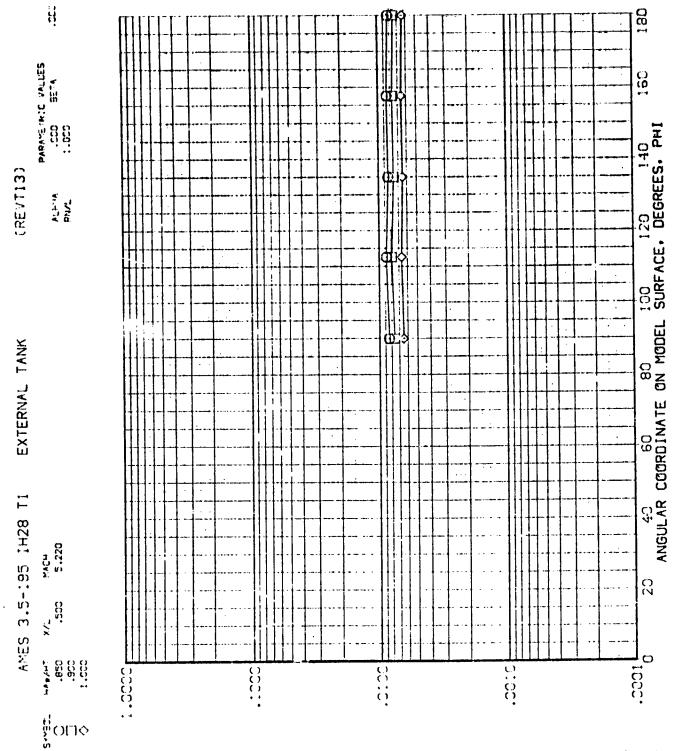


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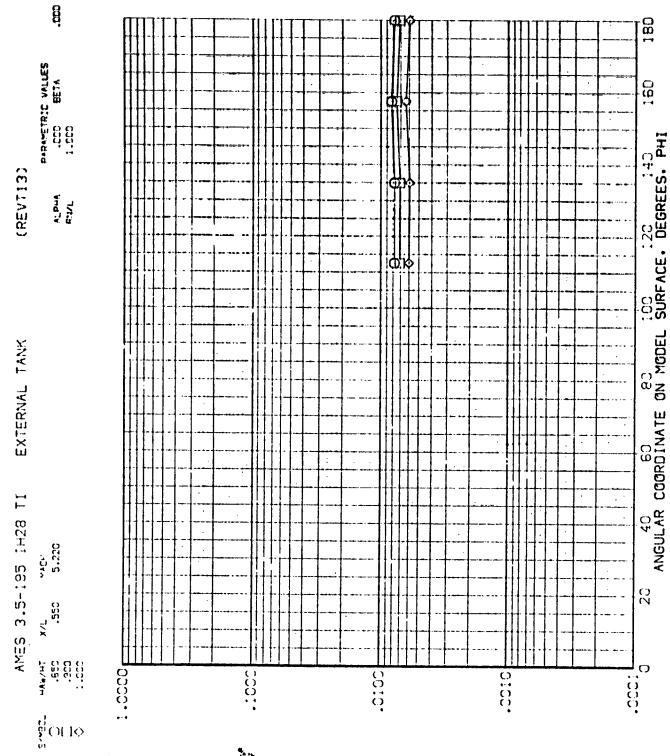


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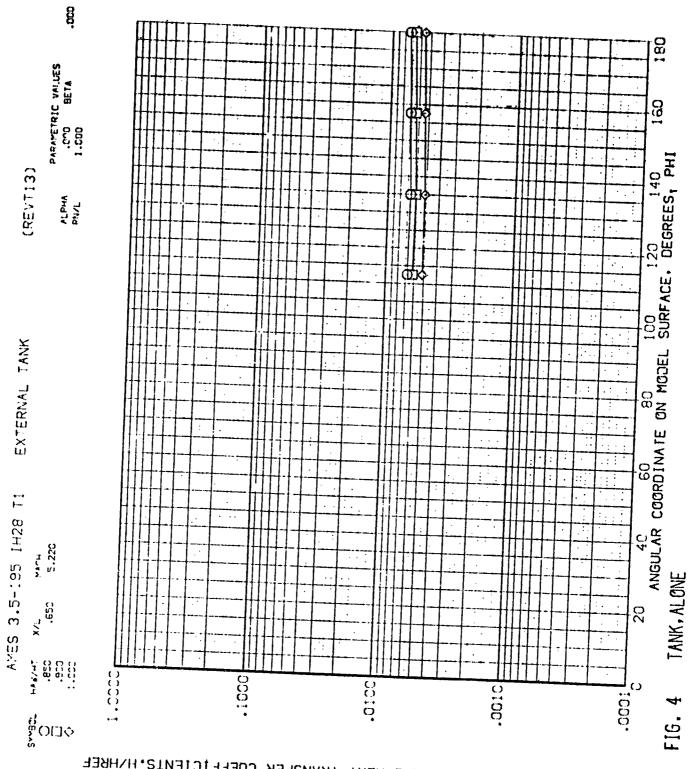
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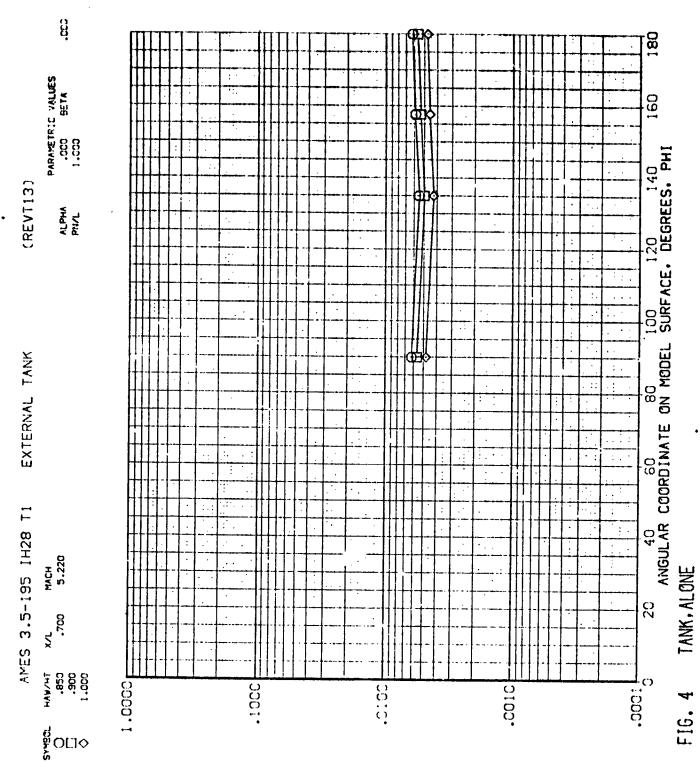
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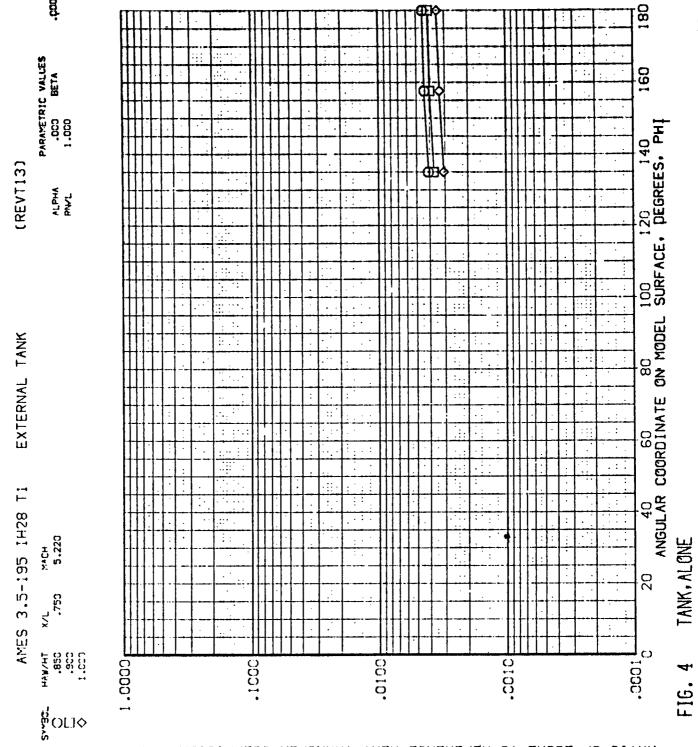
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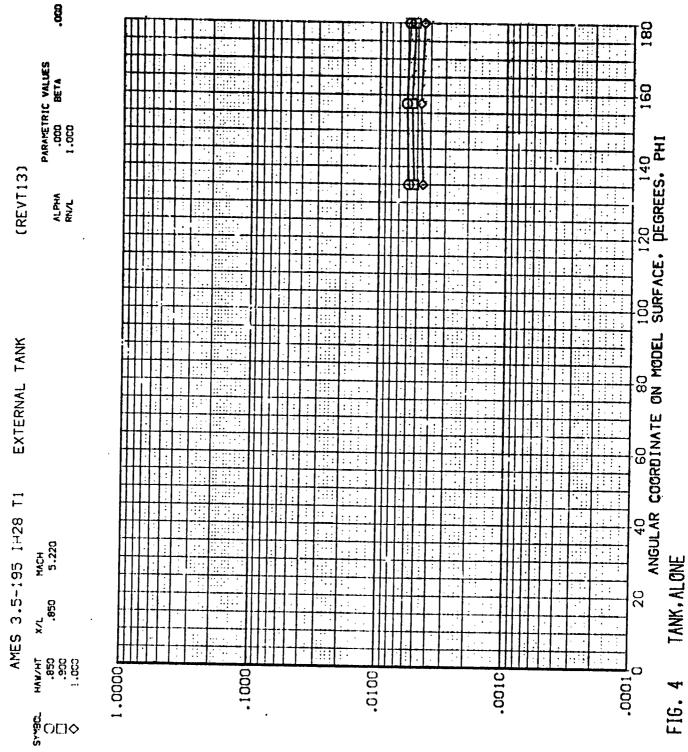
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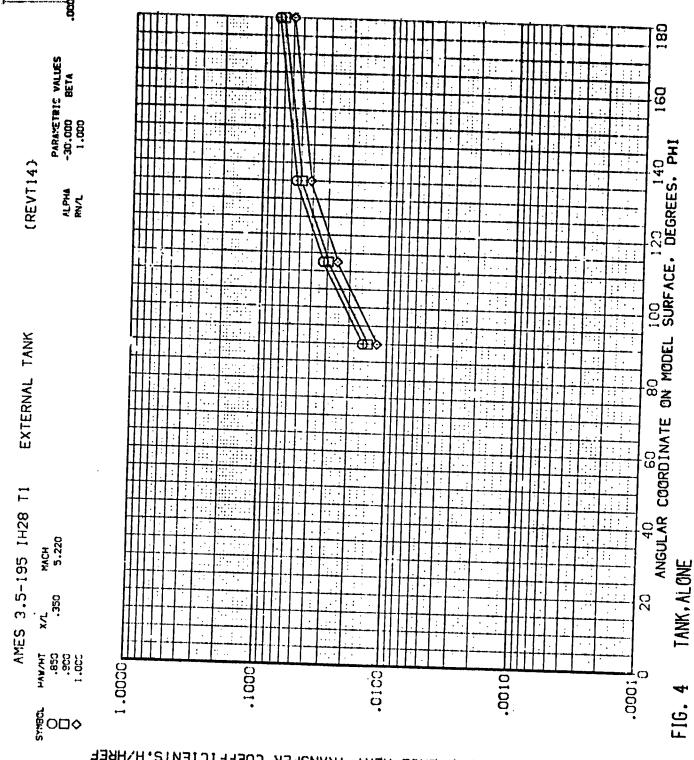
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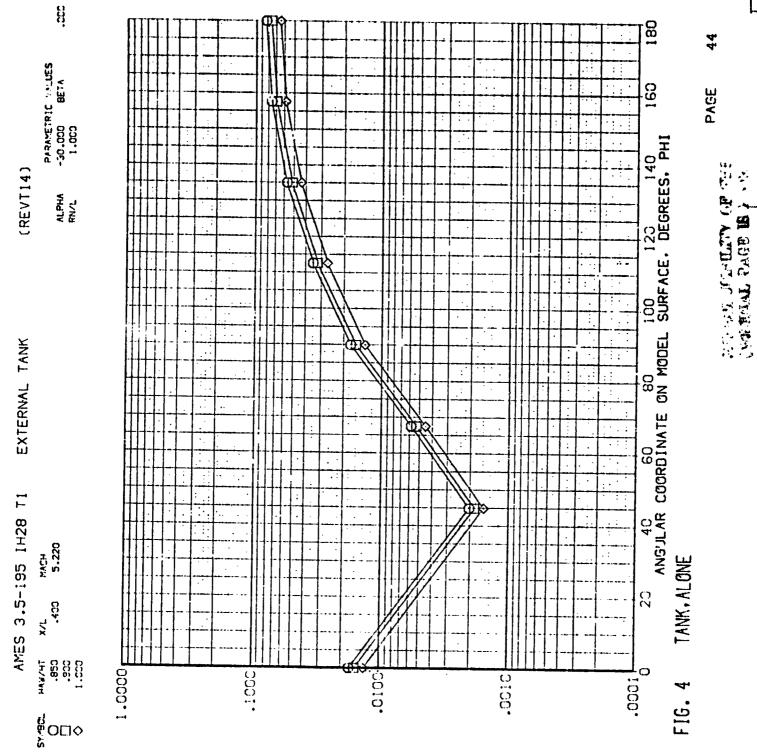
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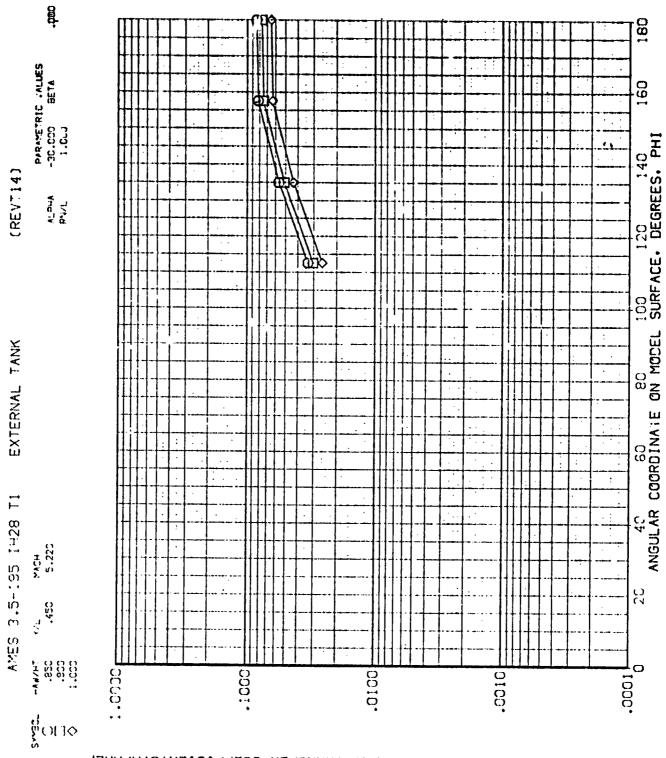
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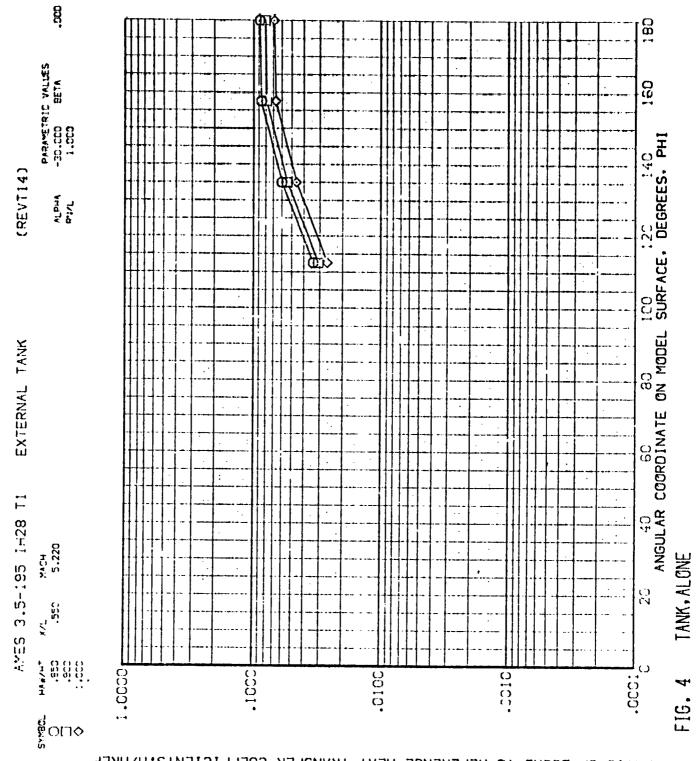
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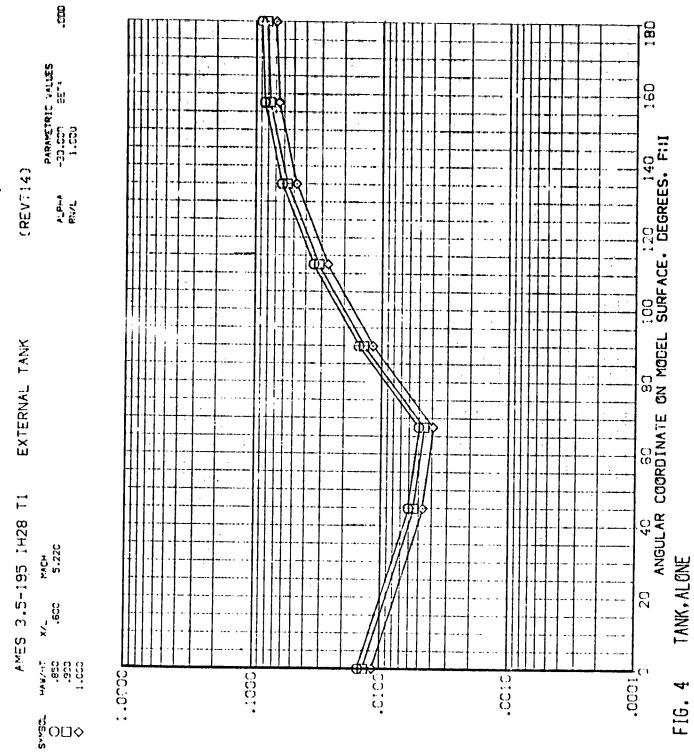
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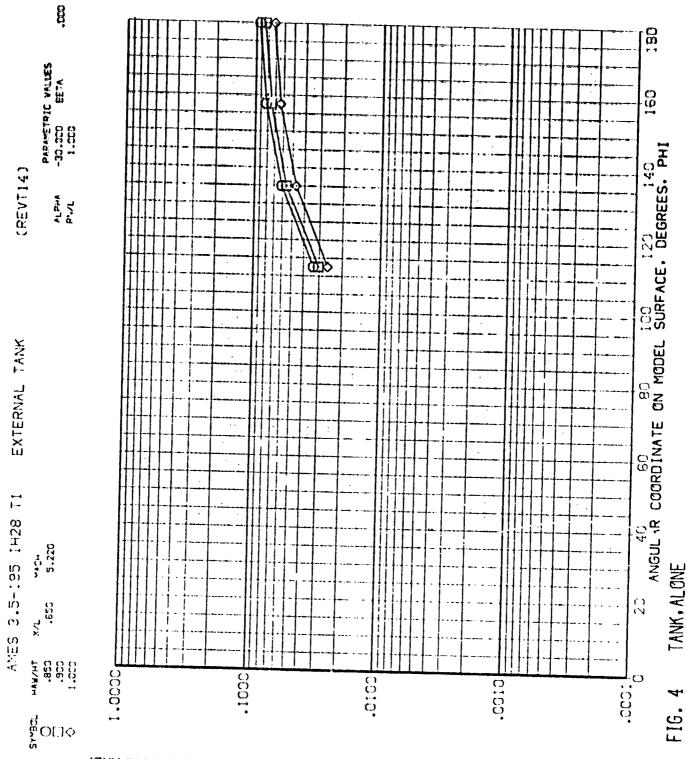
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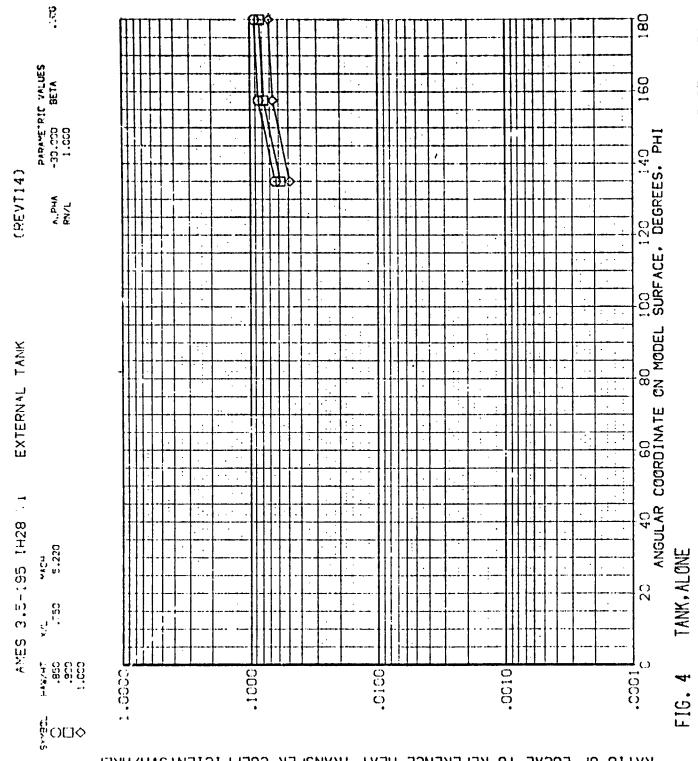


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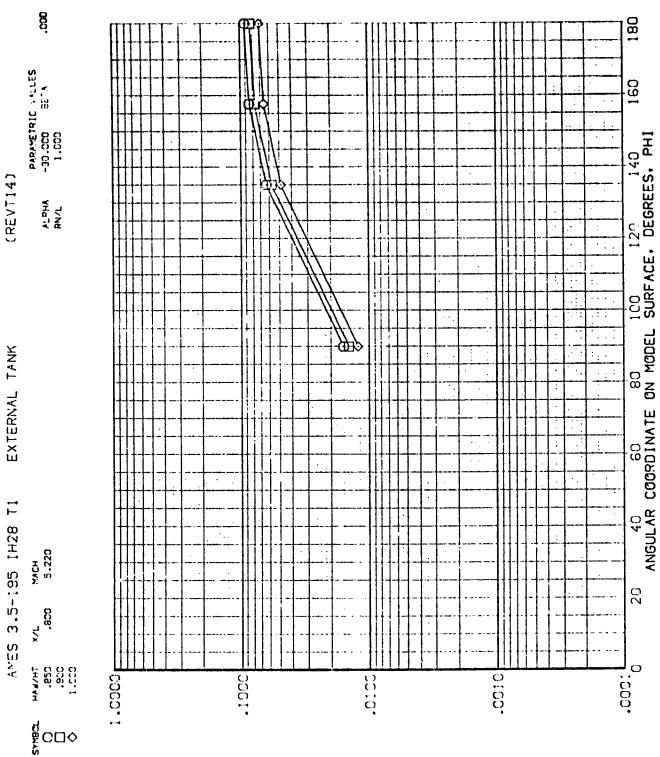


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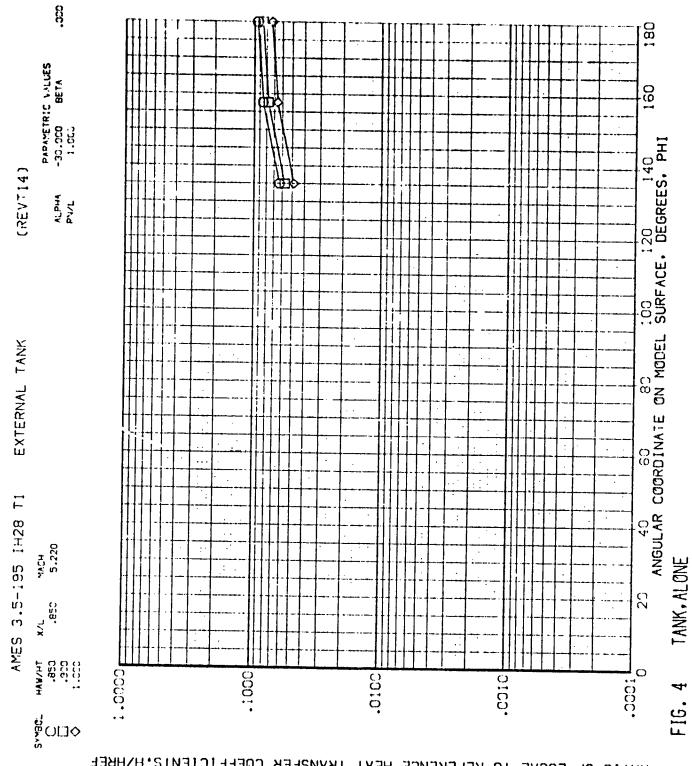
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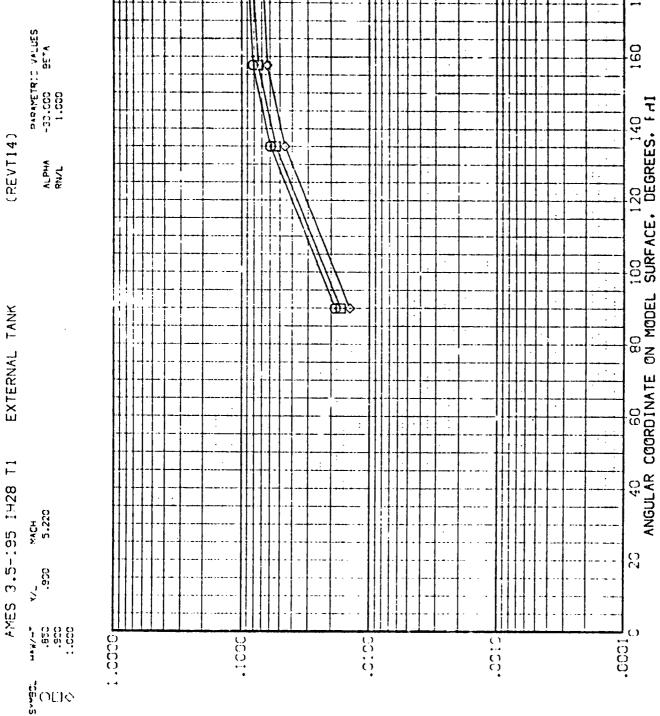


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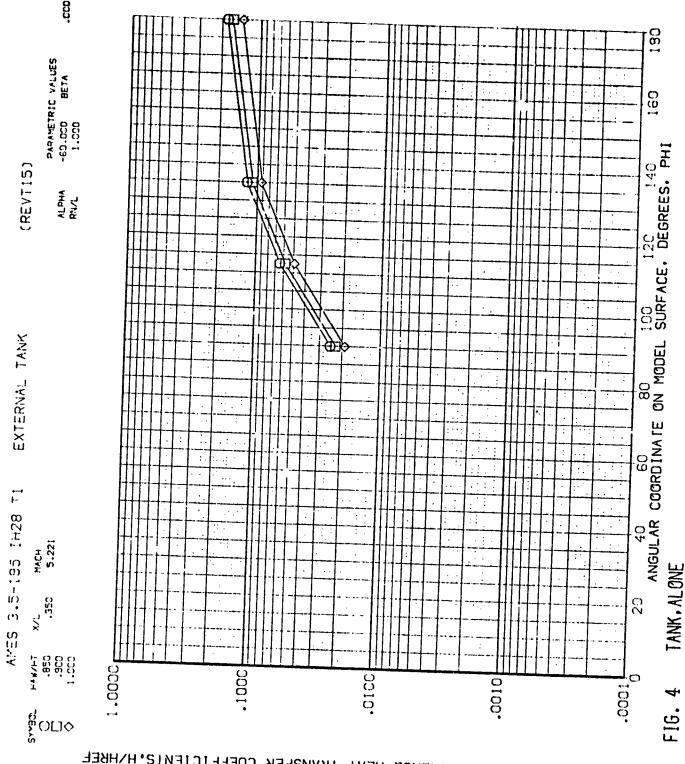
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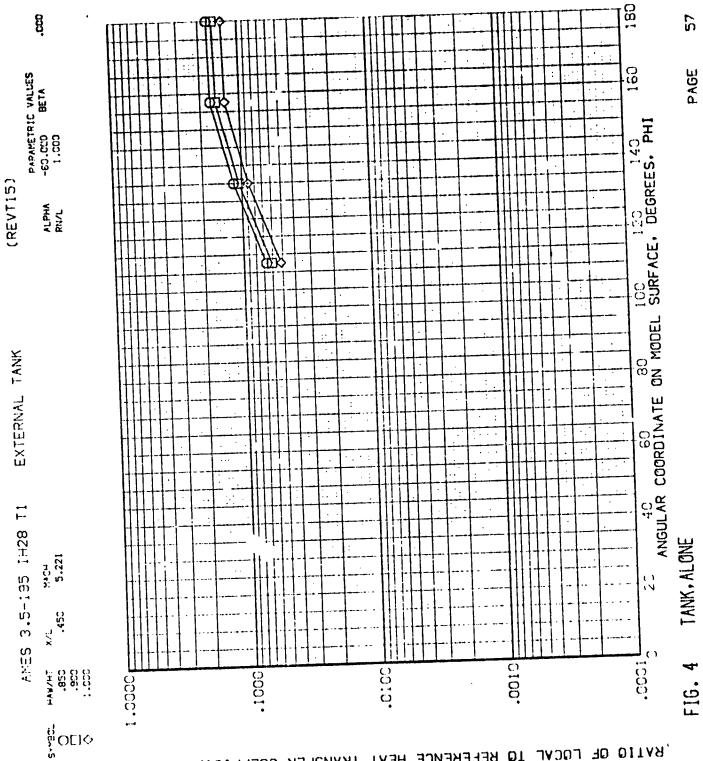
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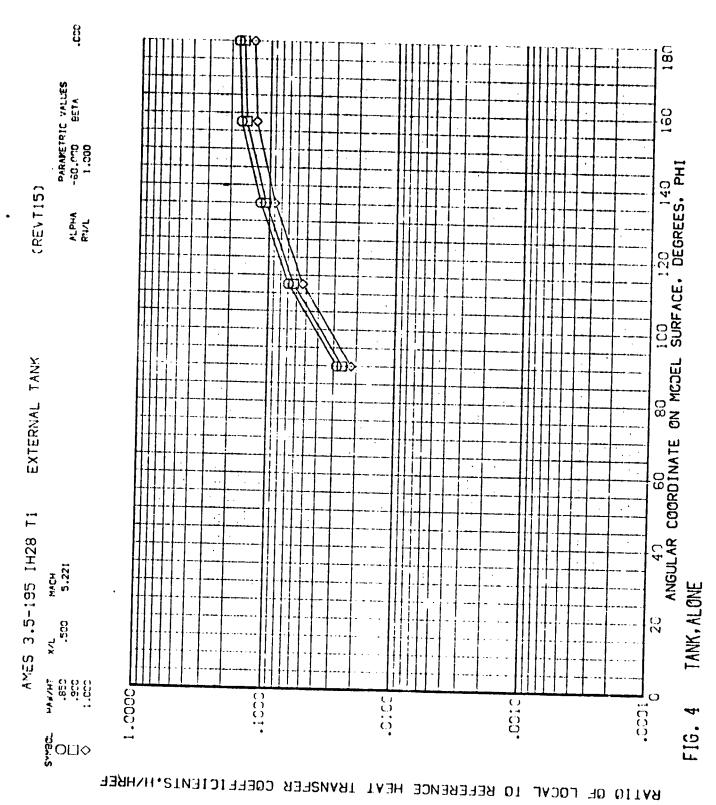
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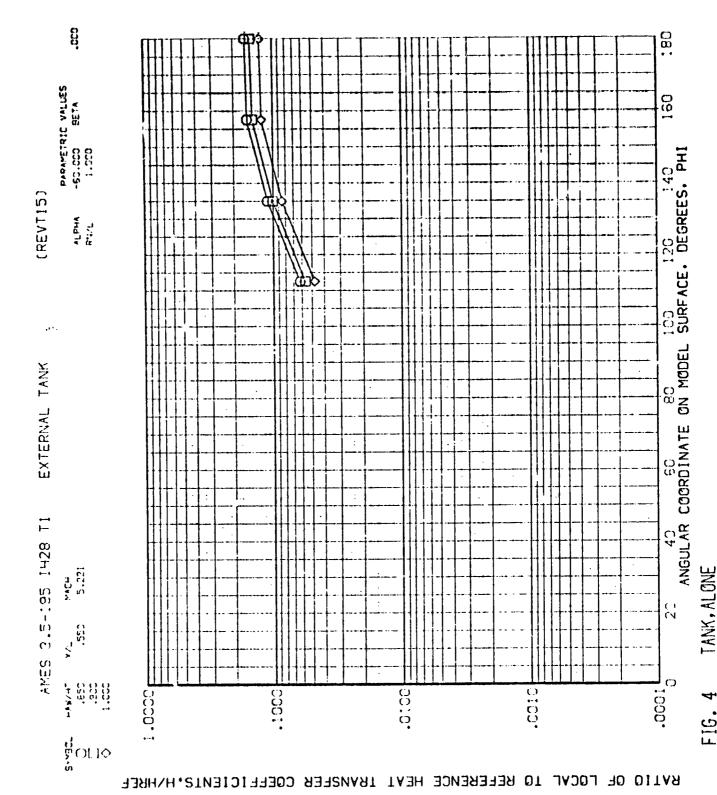
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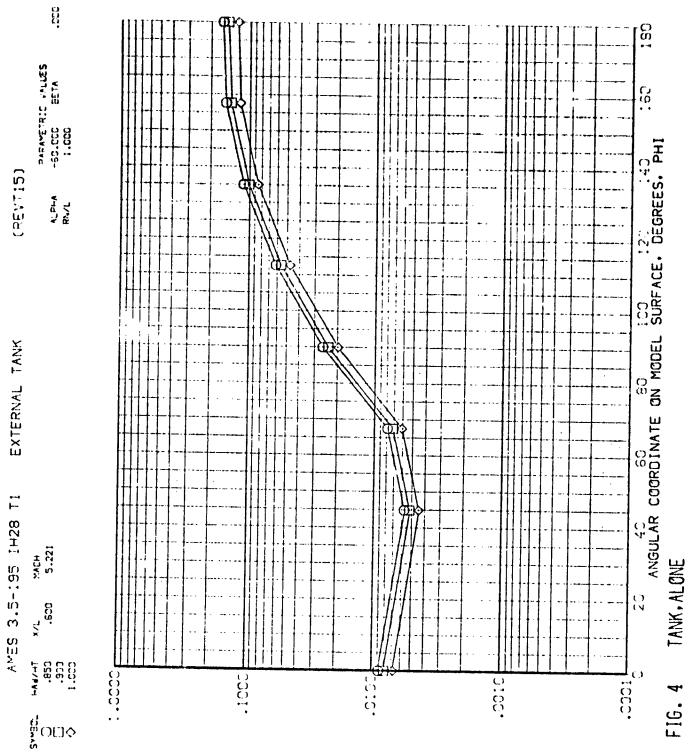
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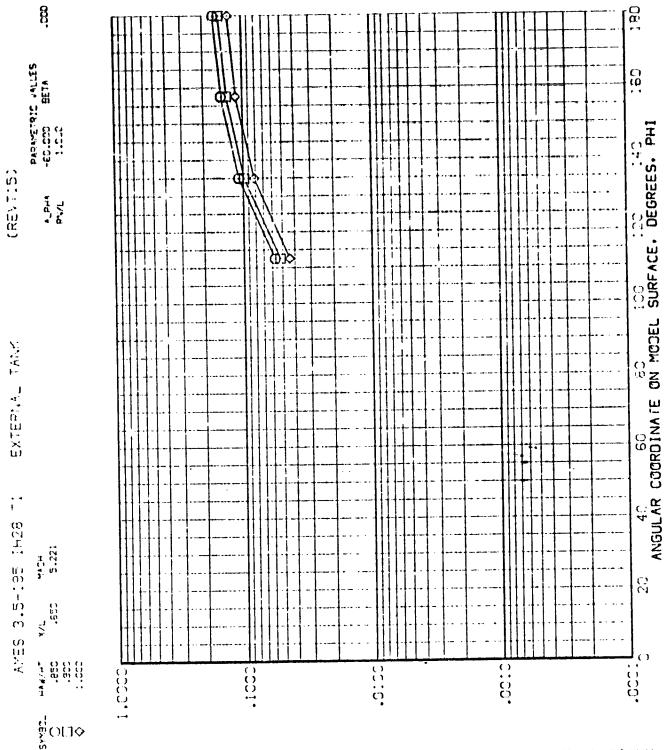
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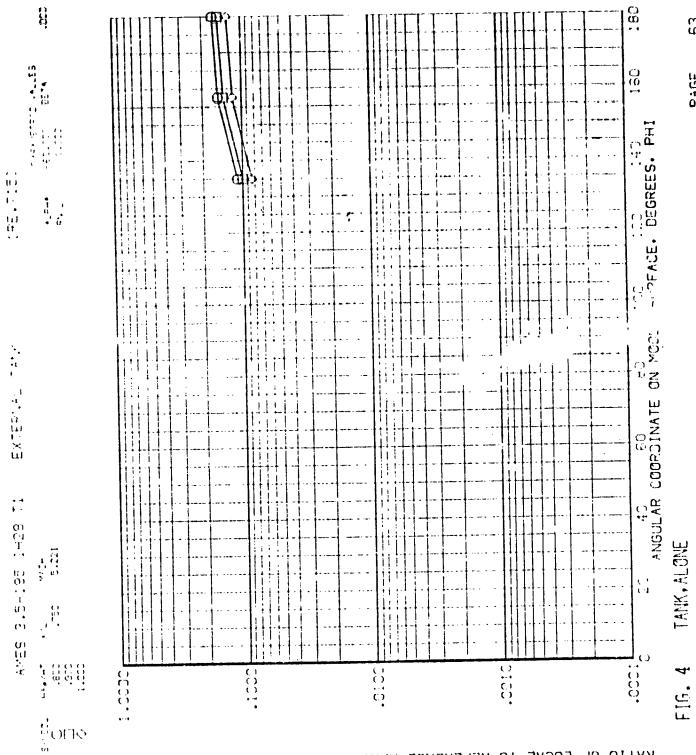


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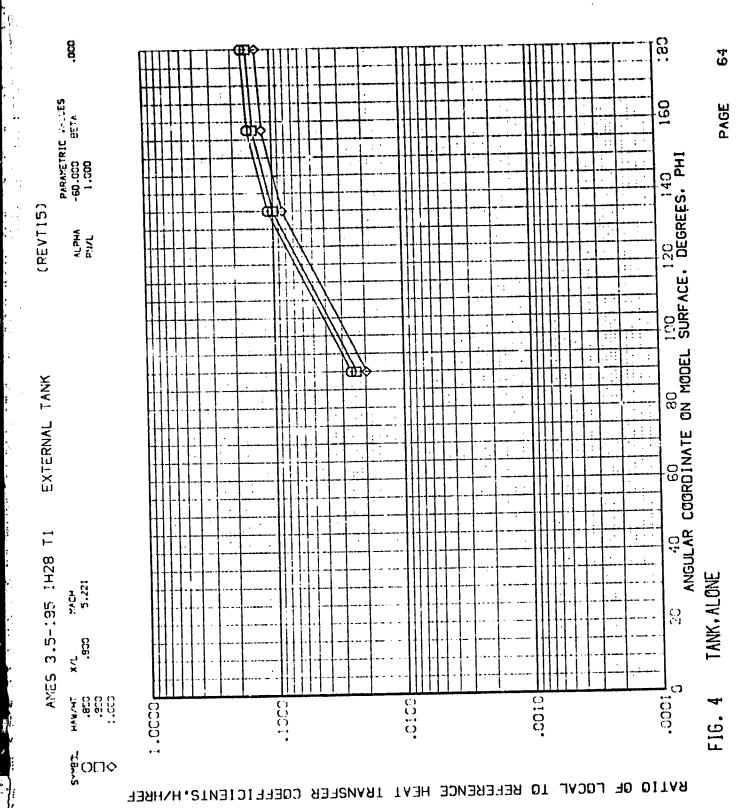


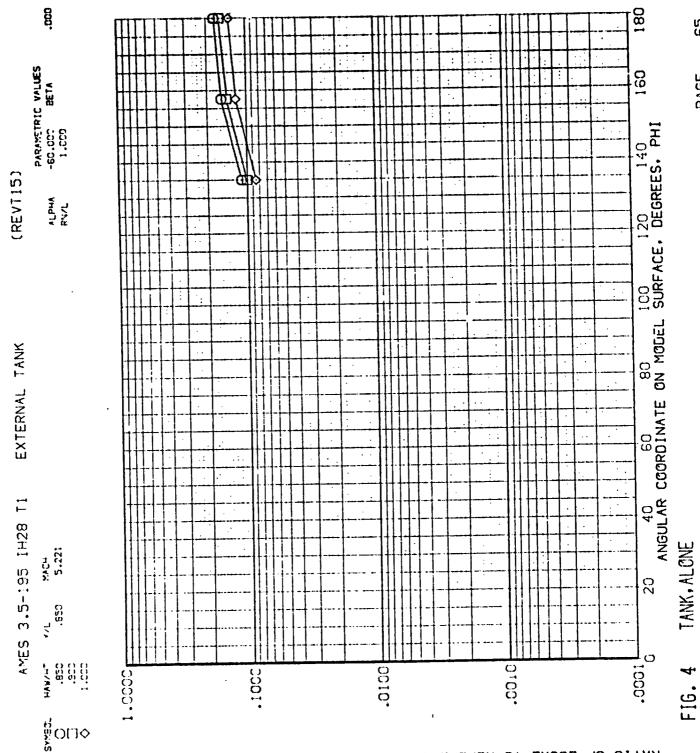
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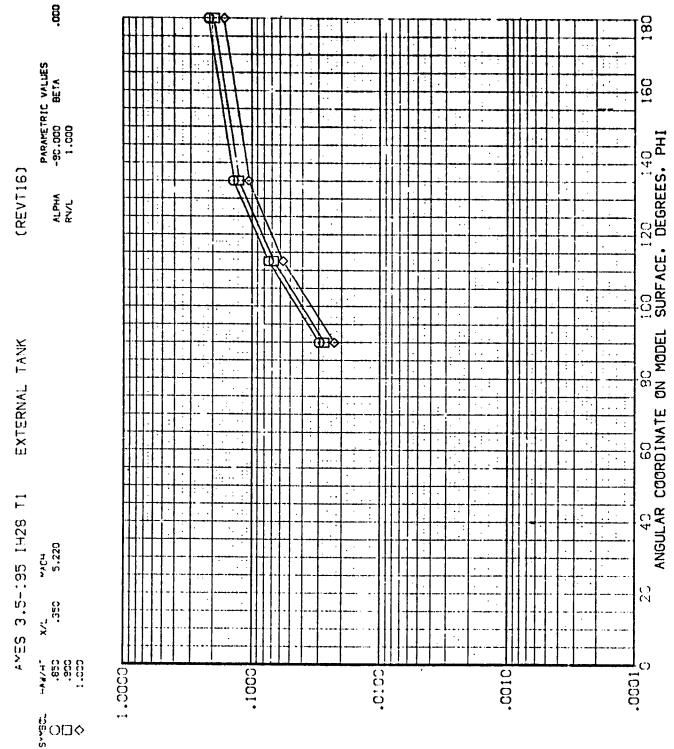


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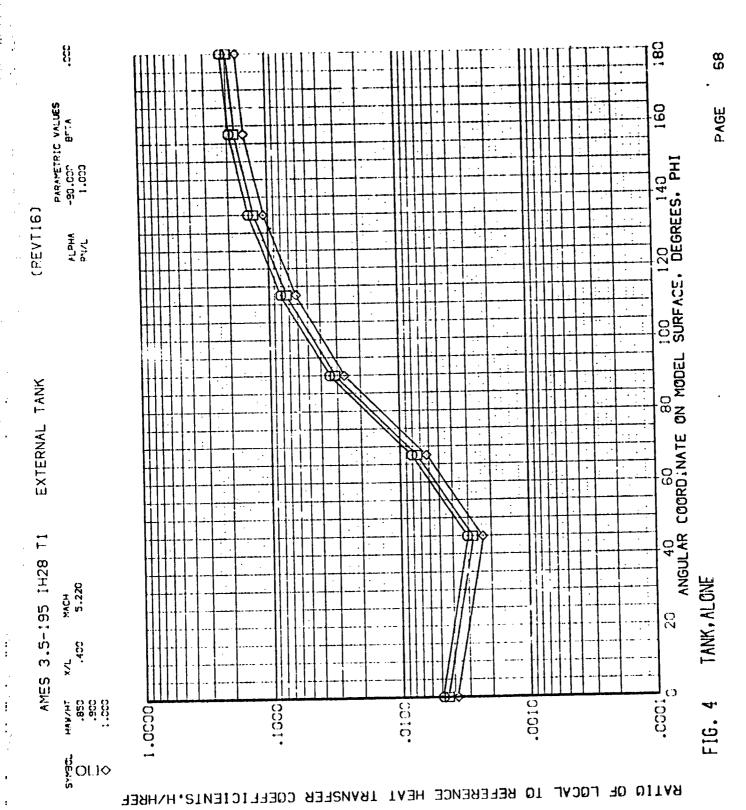


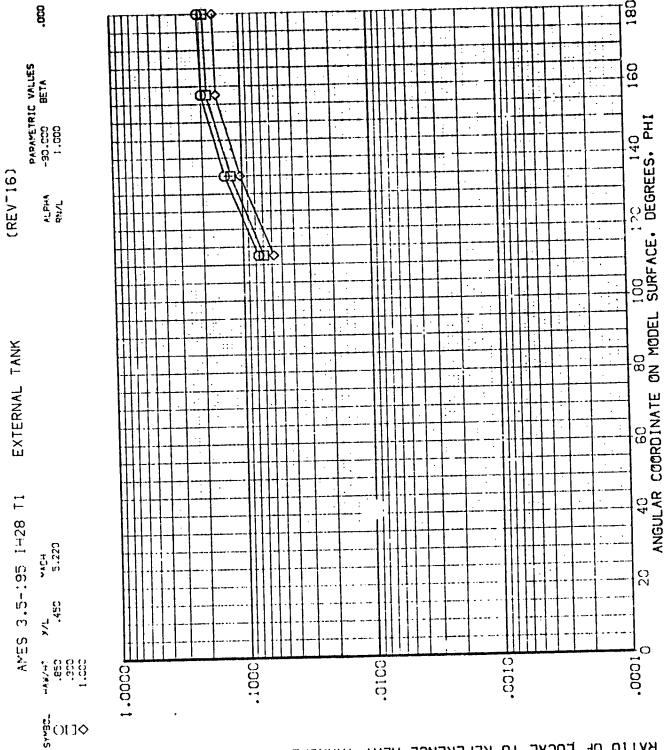


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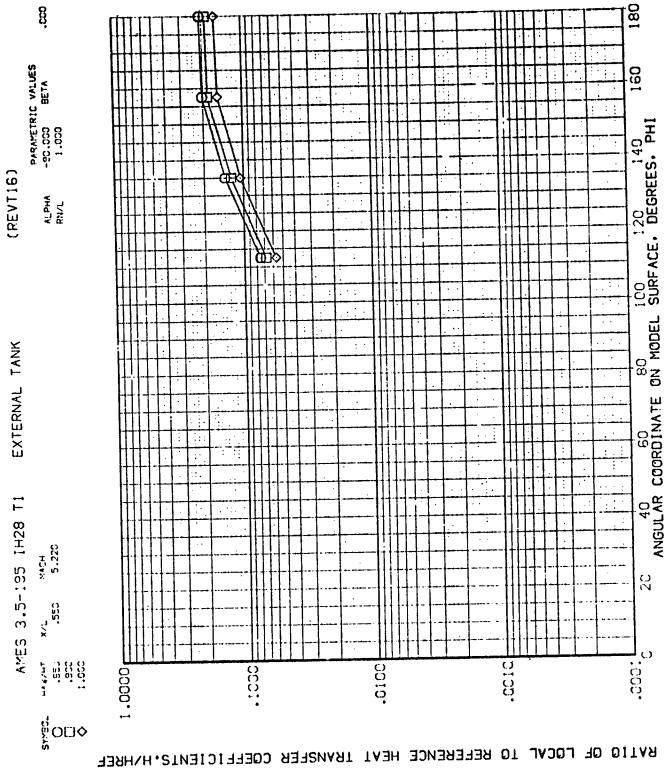
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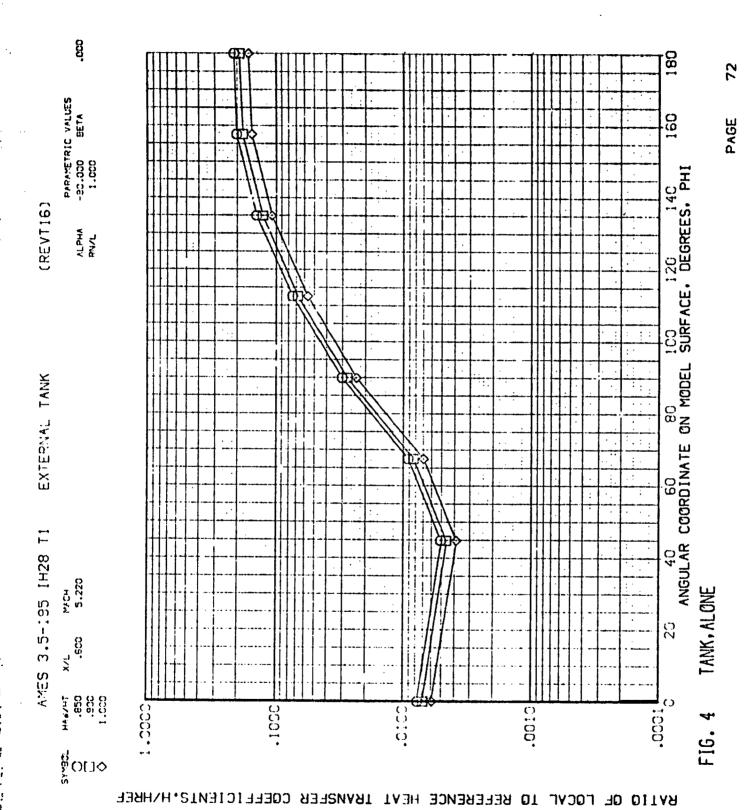
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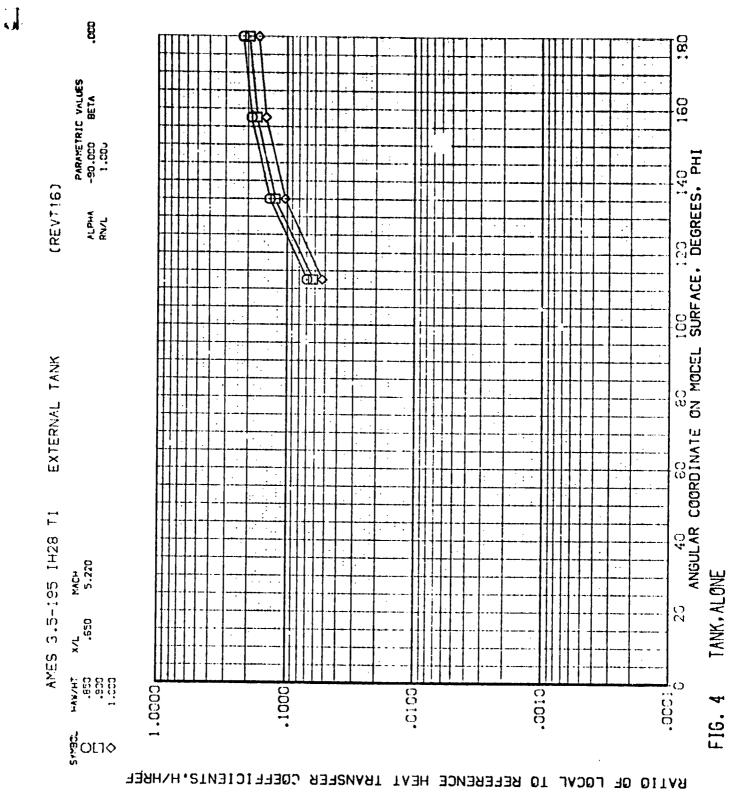
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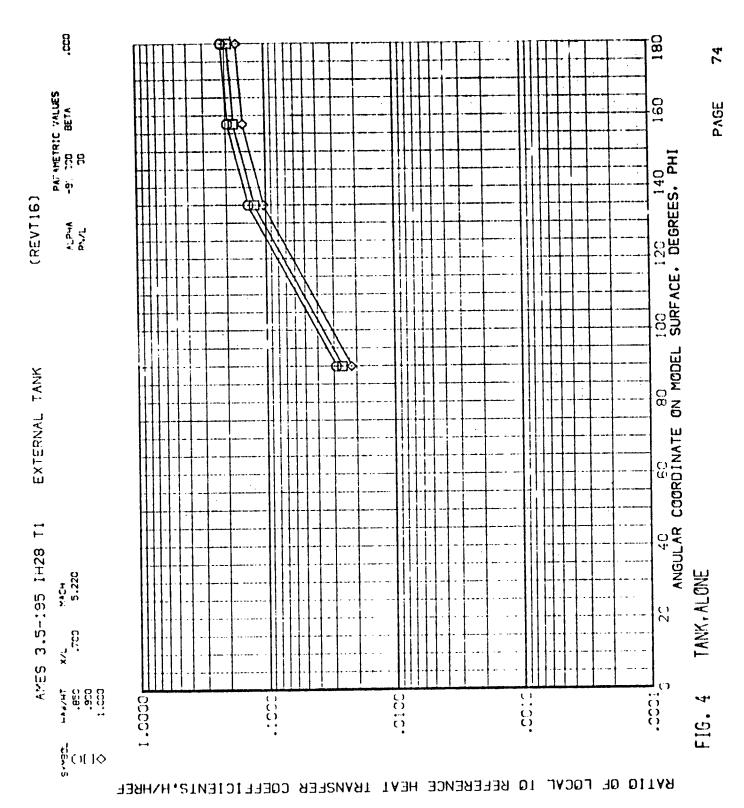
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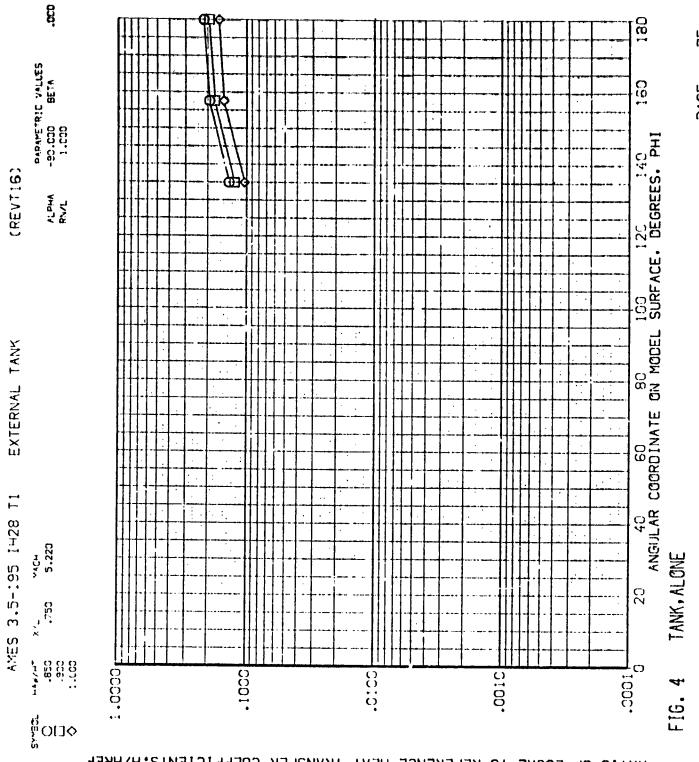
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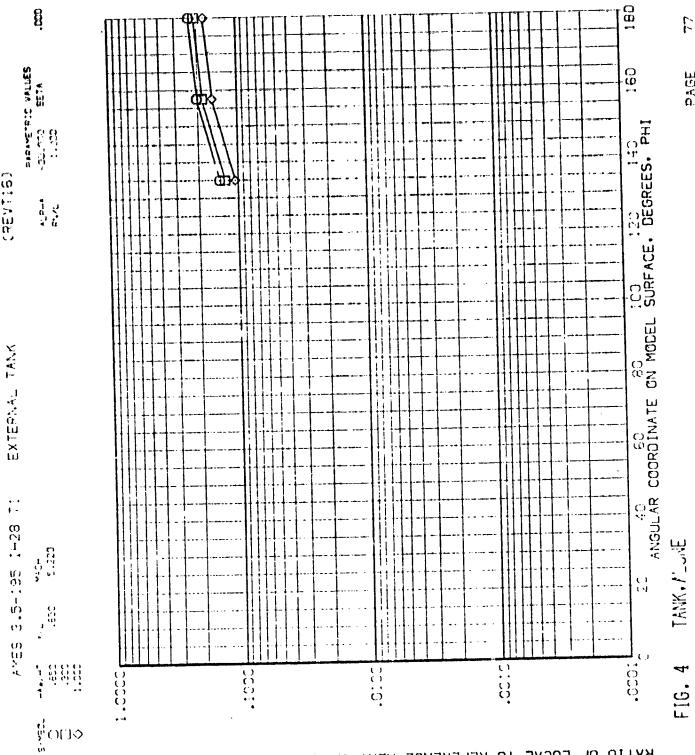




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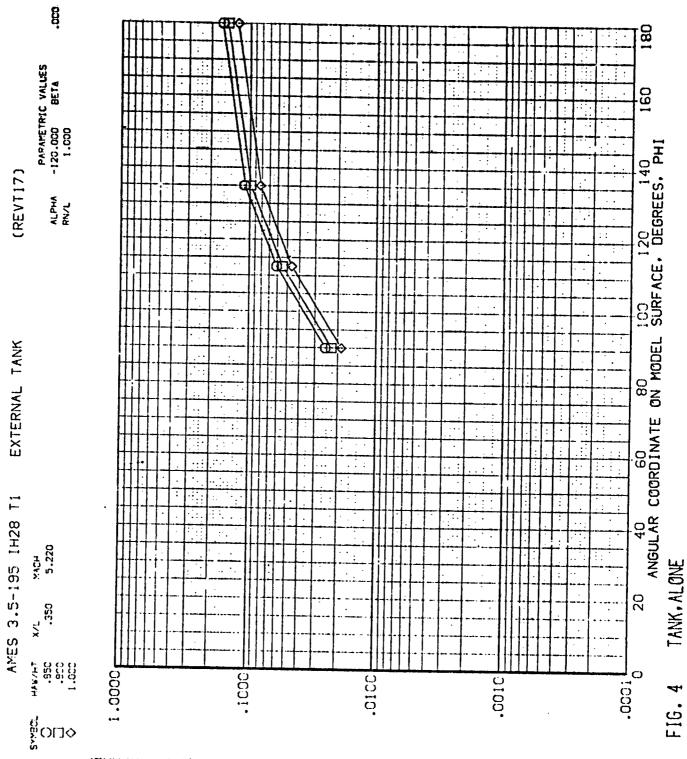
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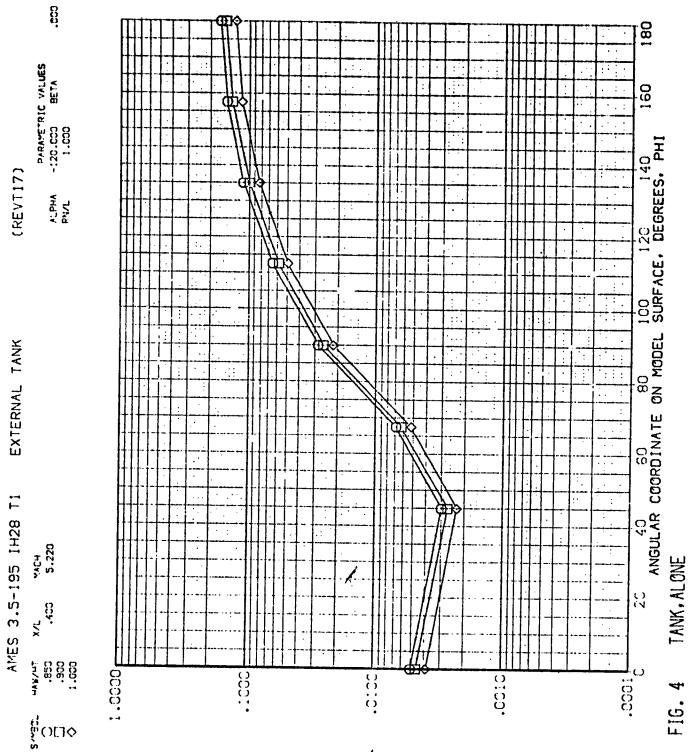
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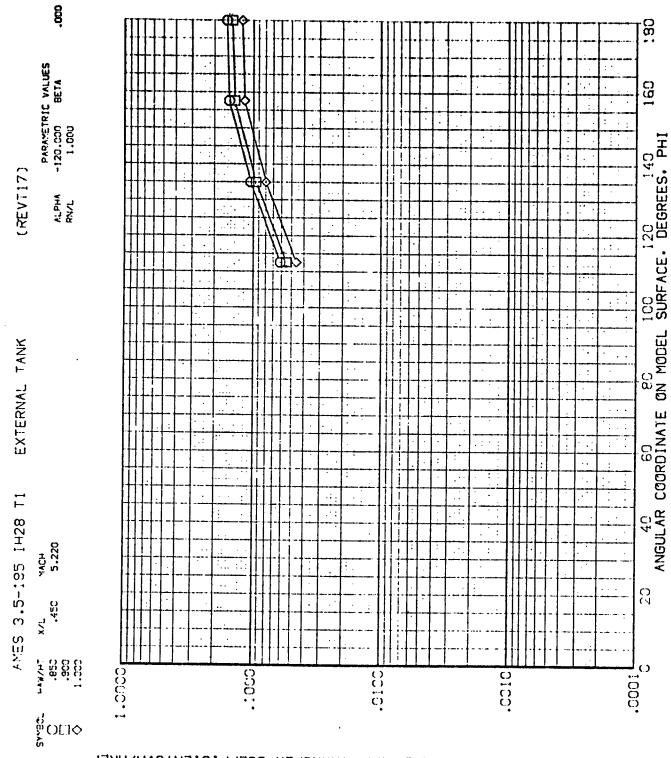
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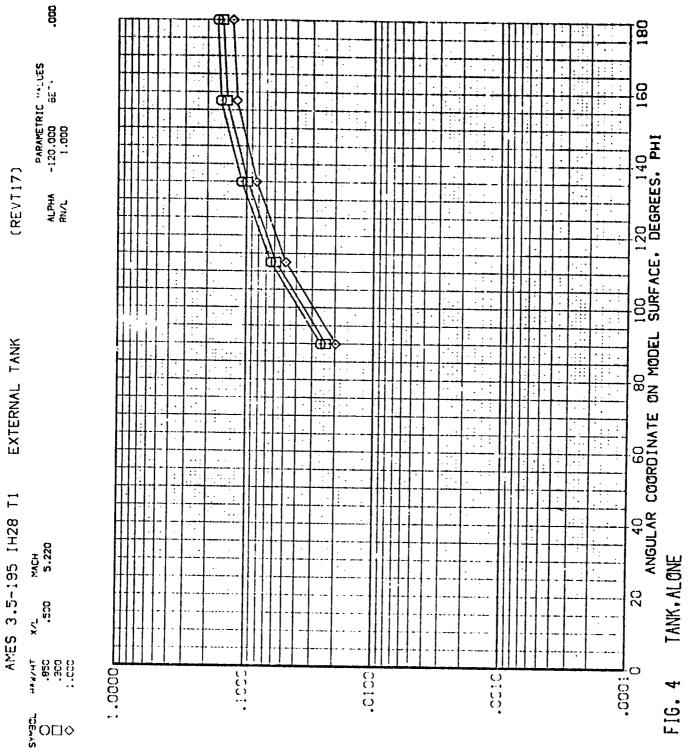
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FIG. 4



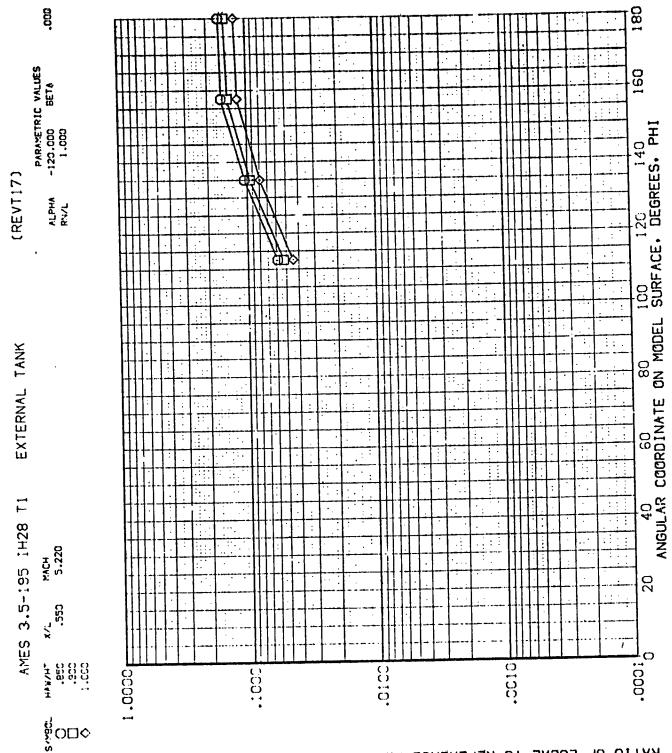
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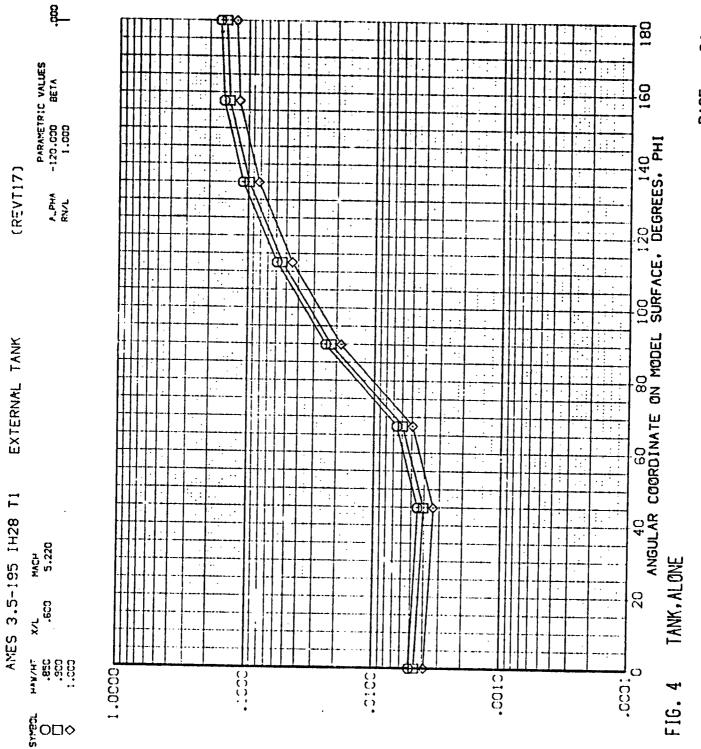
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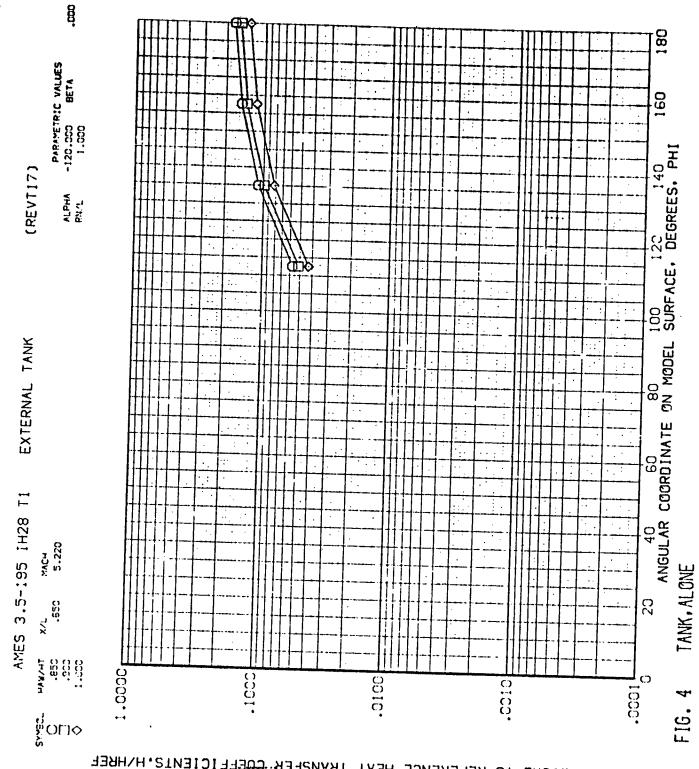
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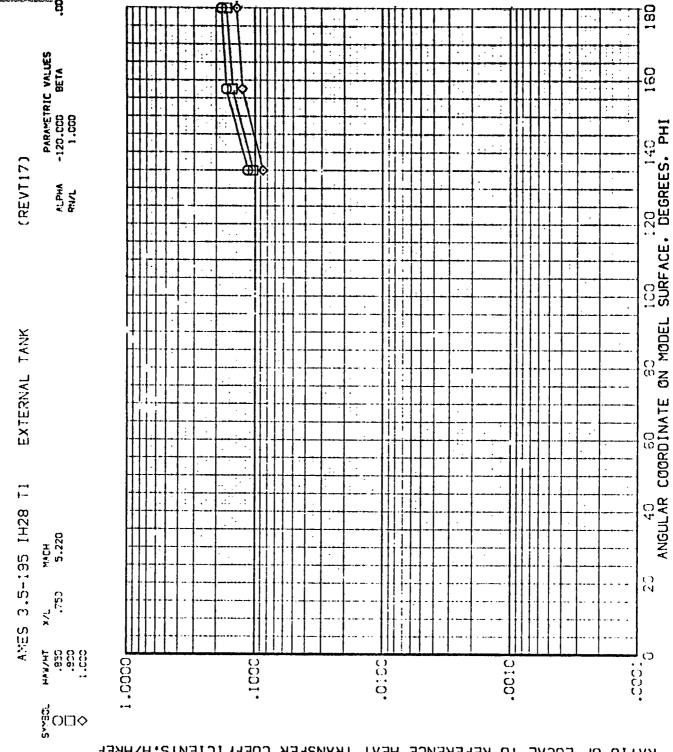


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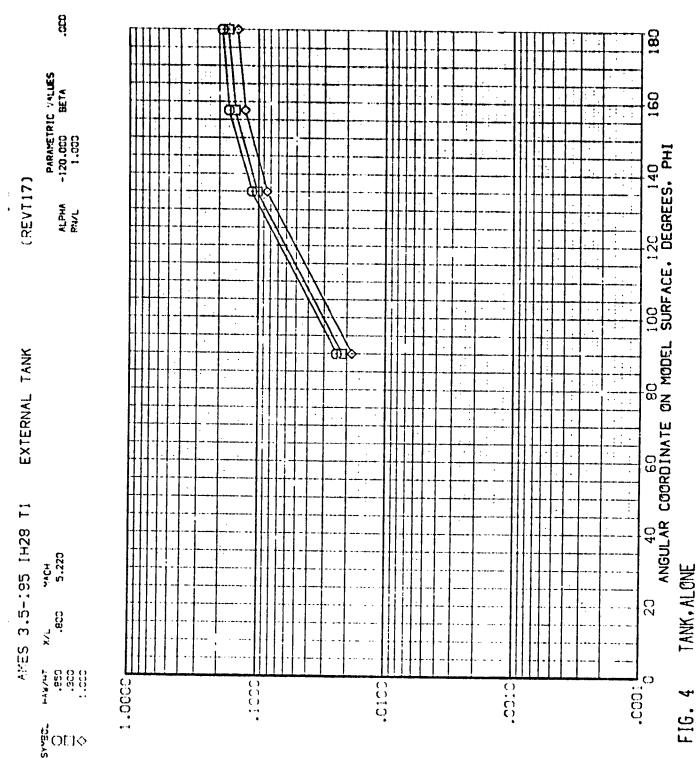
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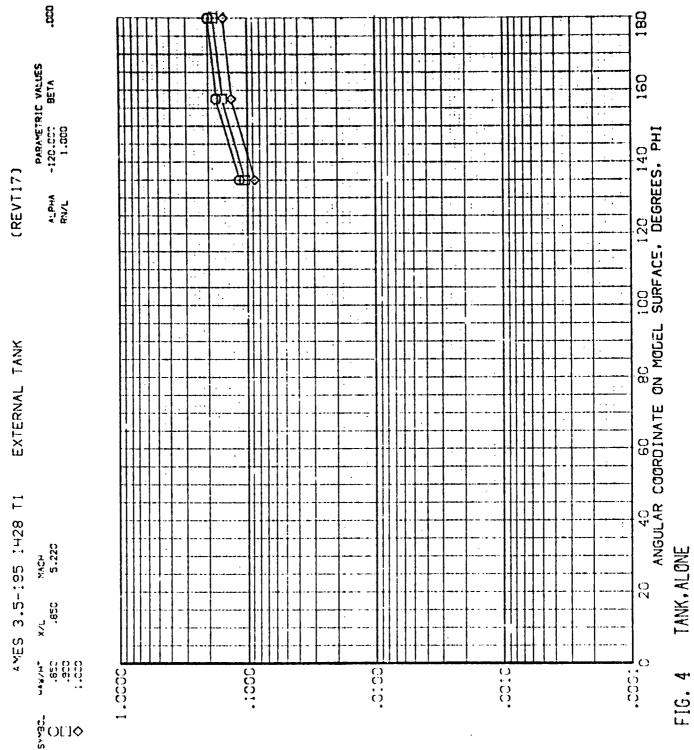
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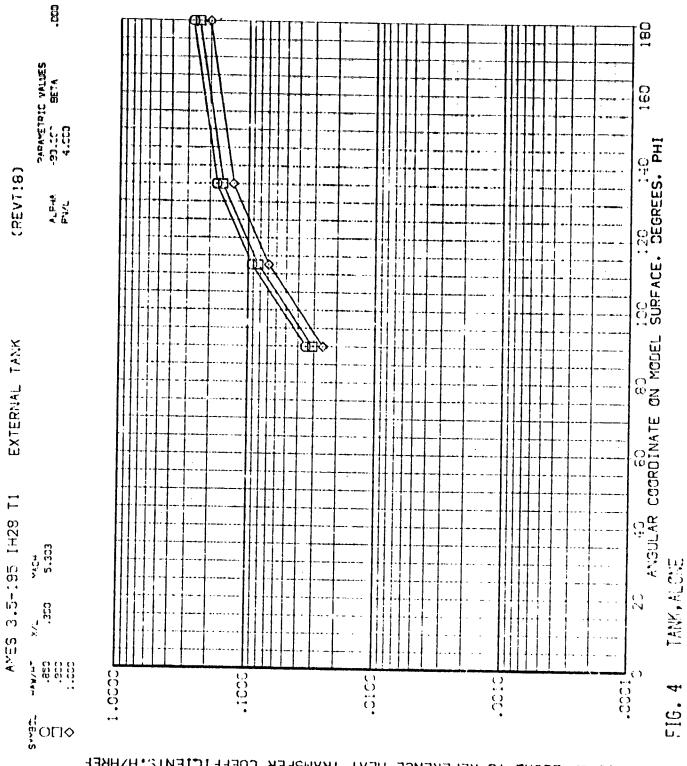
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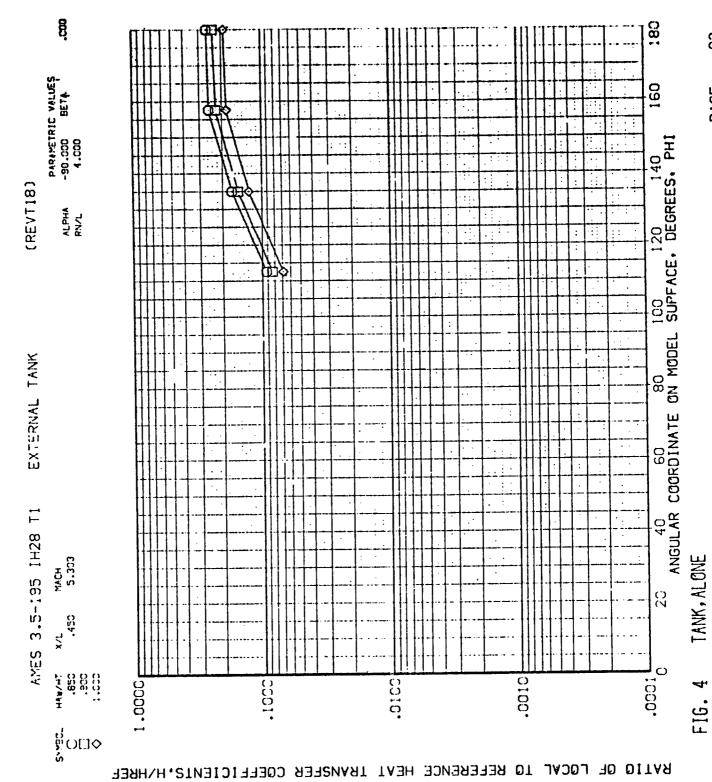
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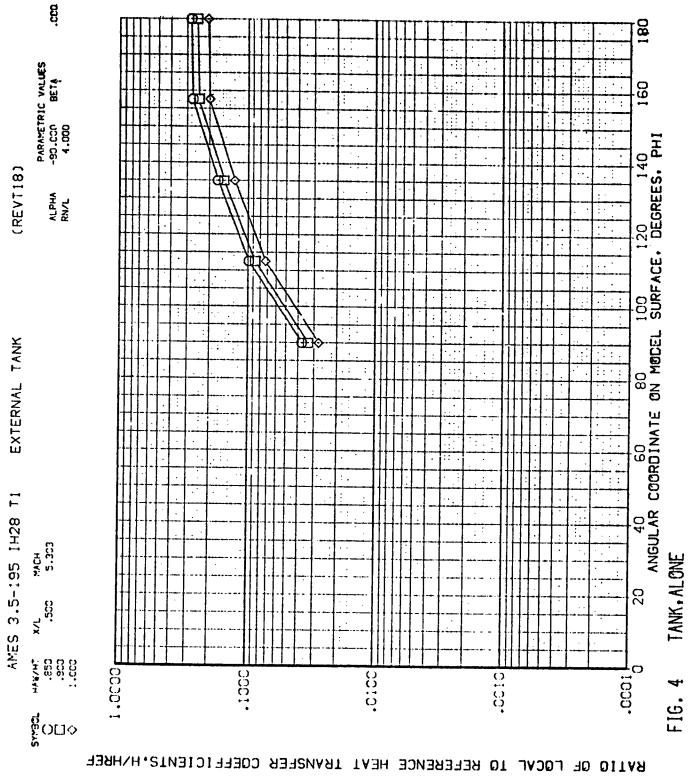
FIG. 4

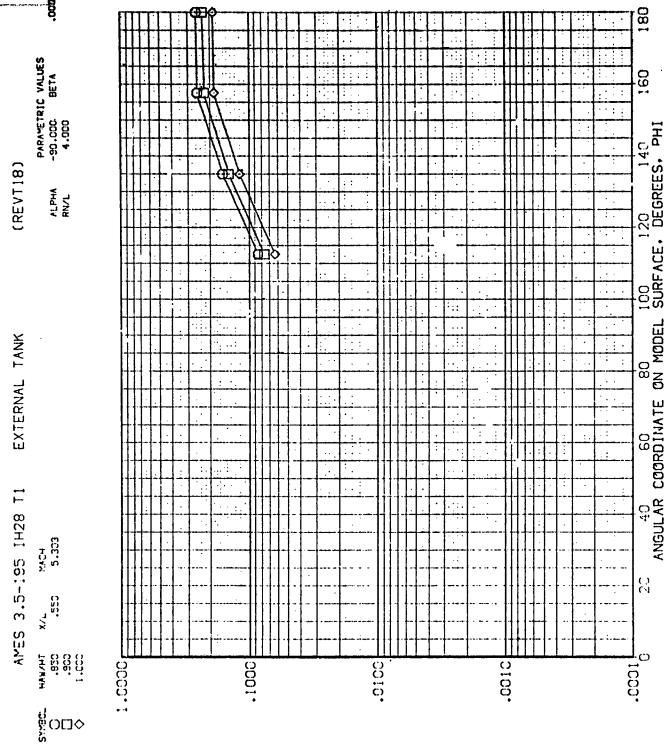
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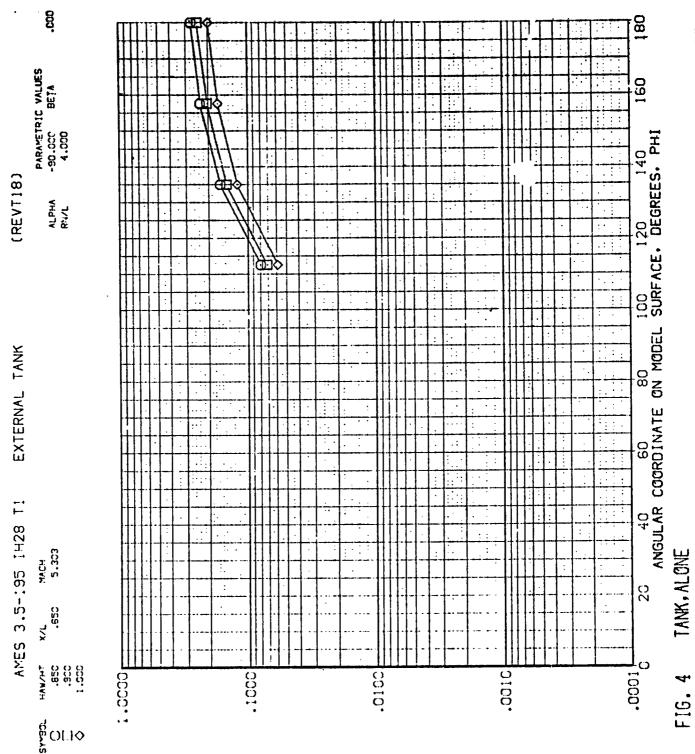
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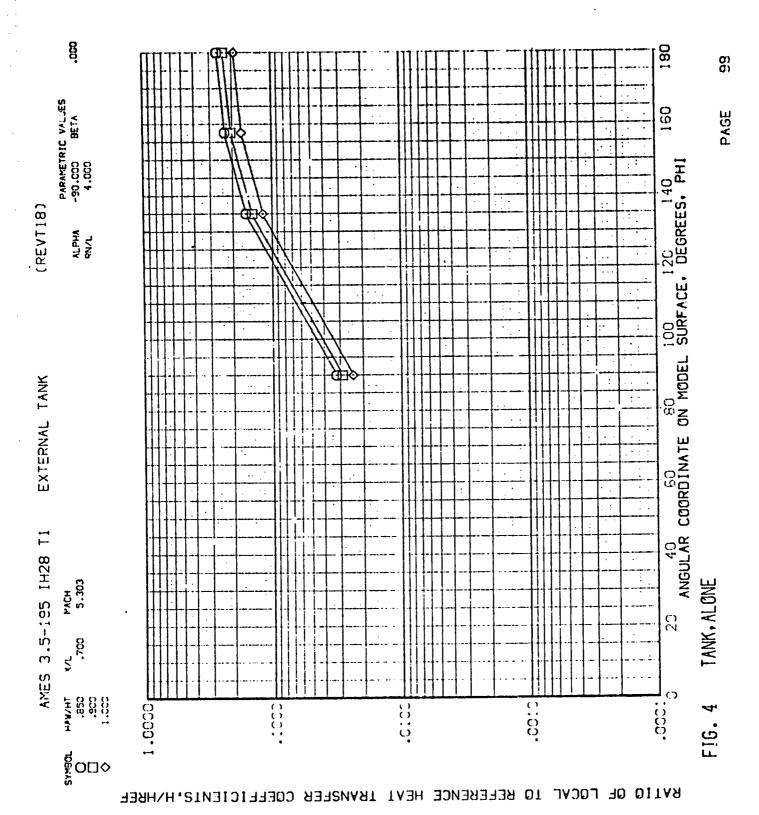
FIG. 4 TANK, ALONE

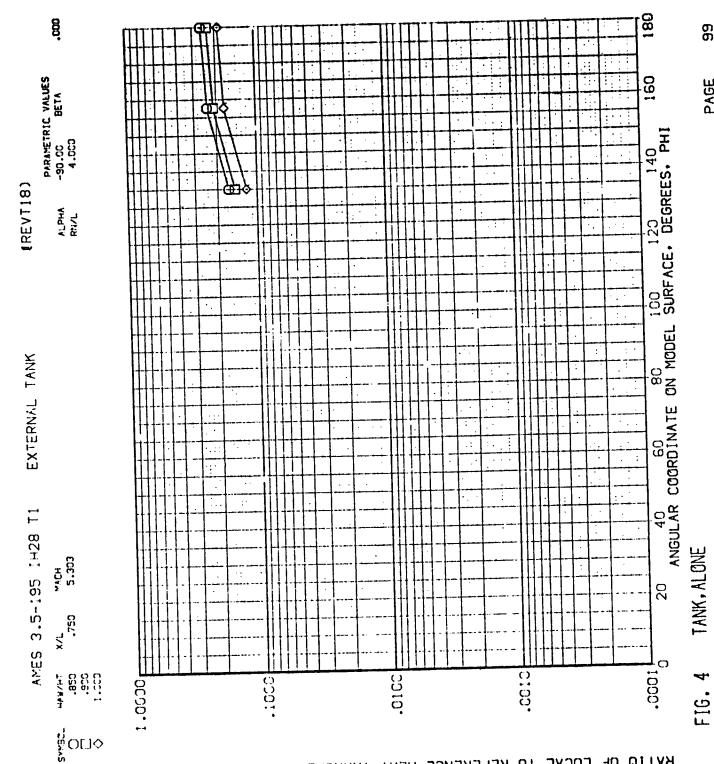
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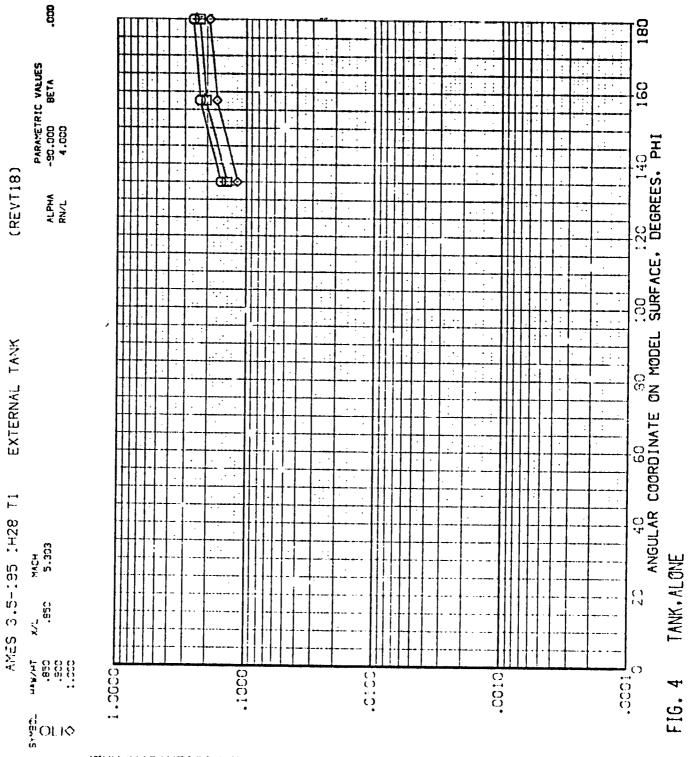




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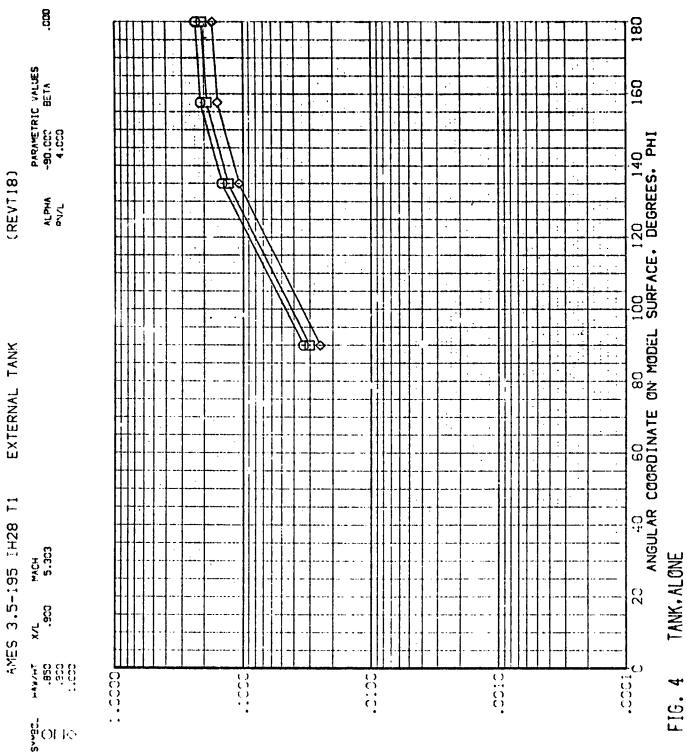


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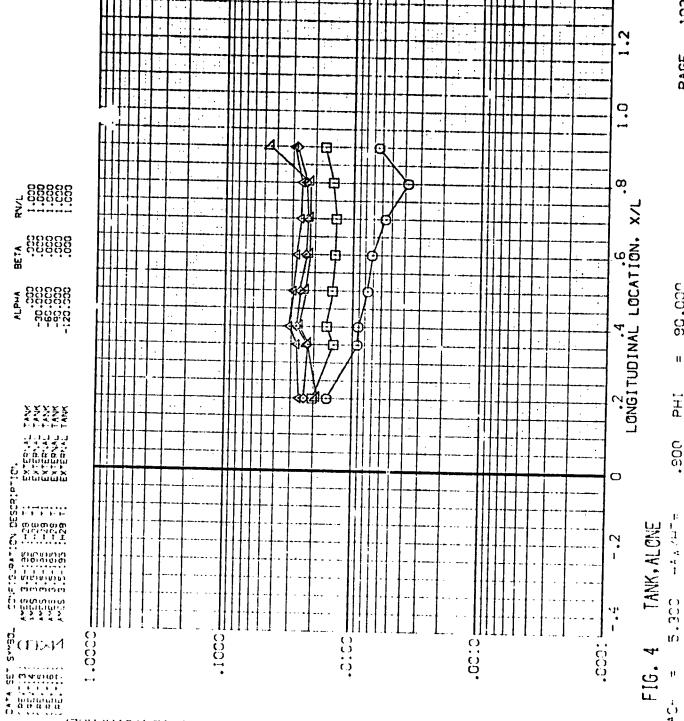
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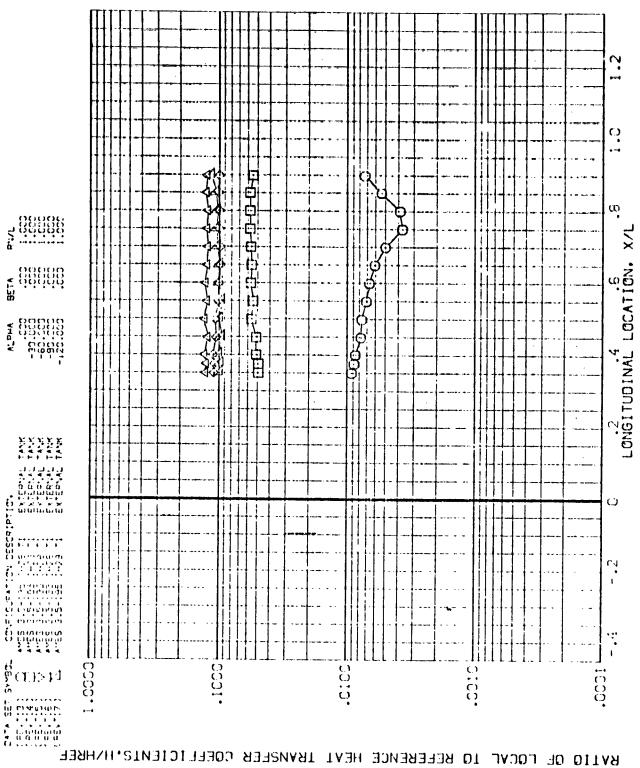


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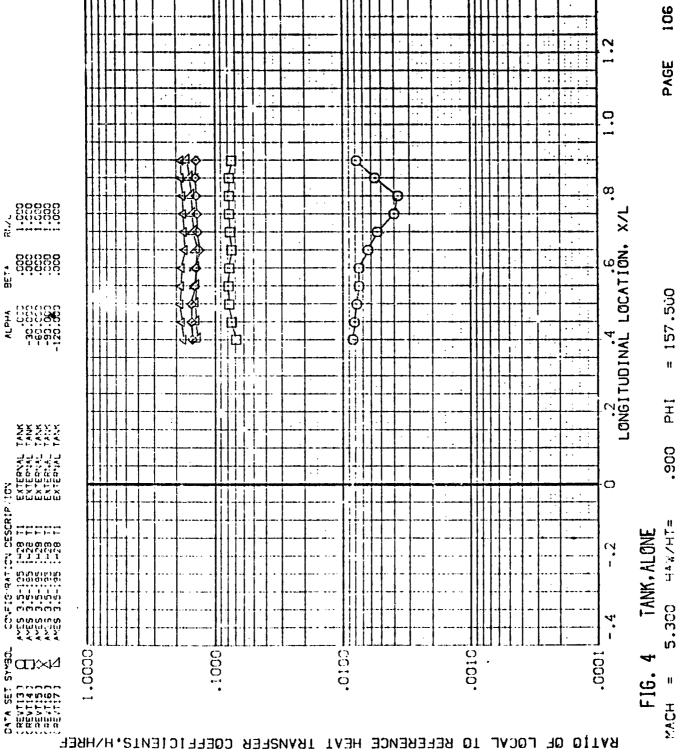
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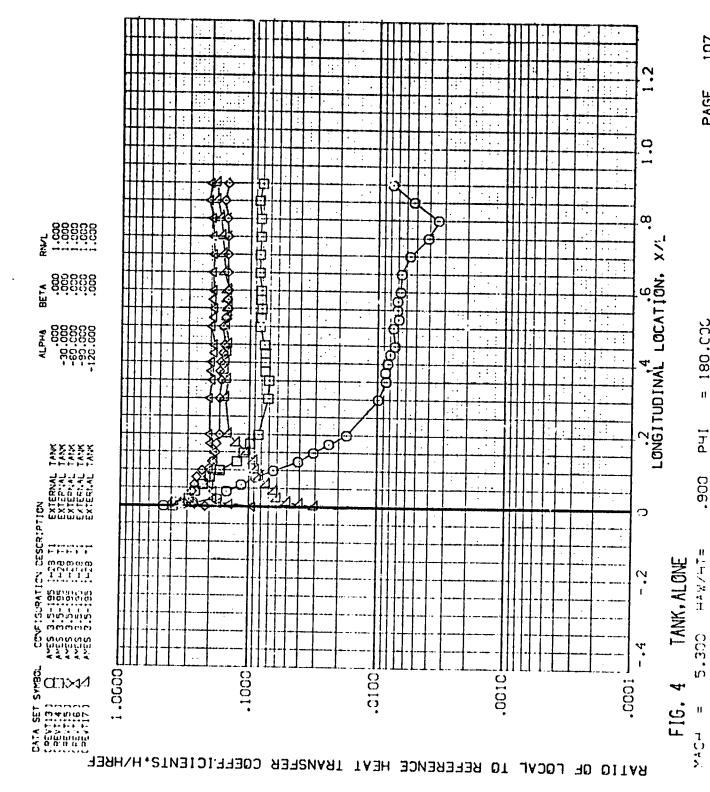
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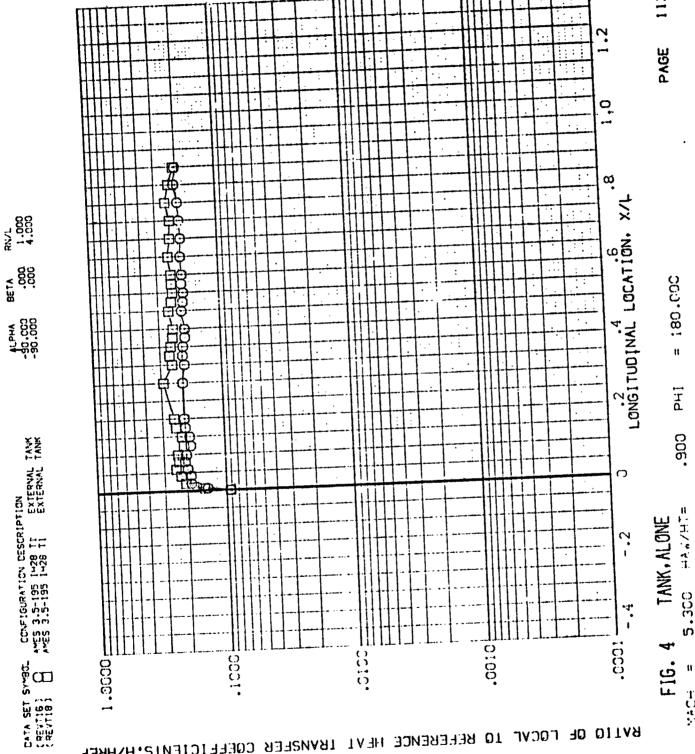
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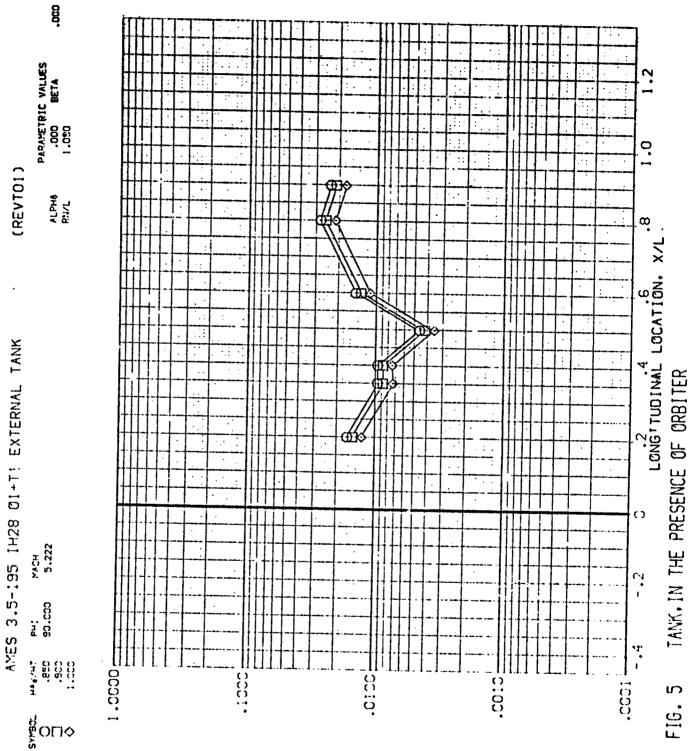
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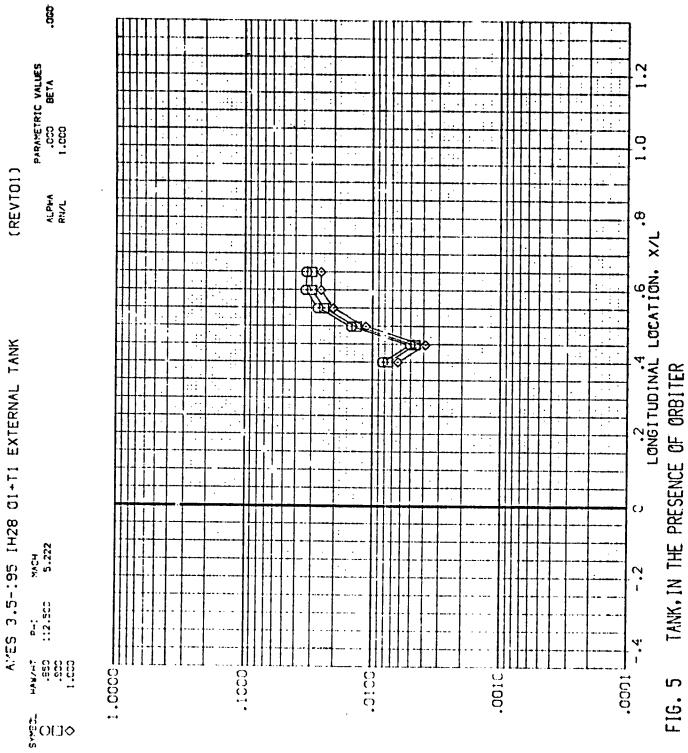


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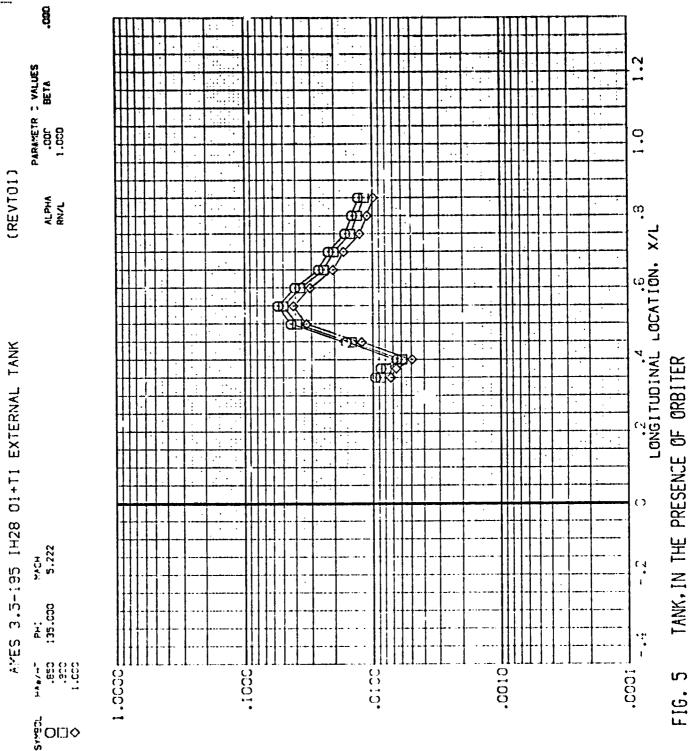
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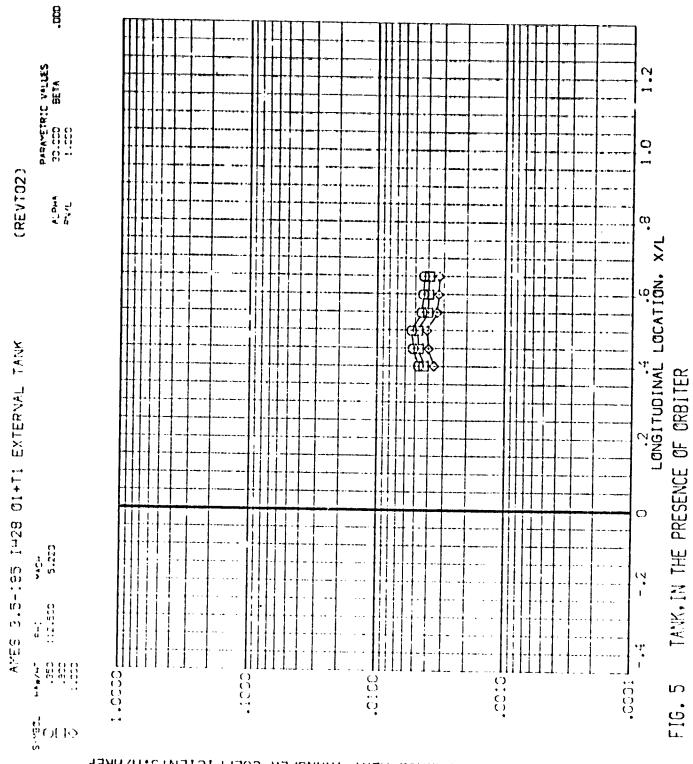
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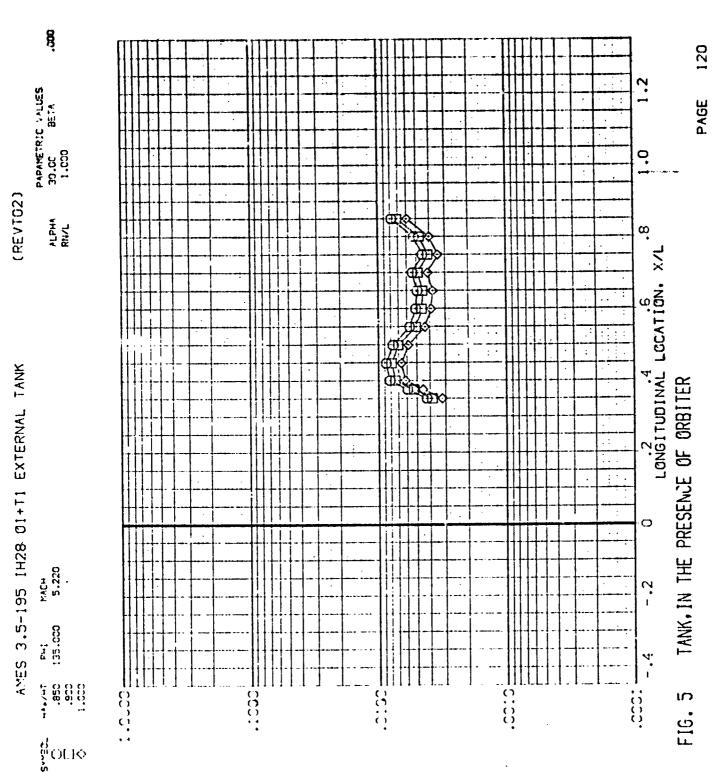
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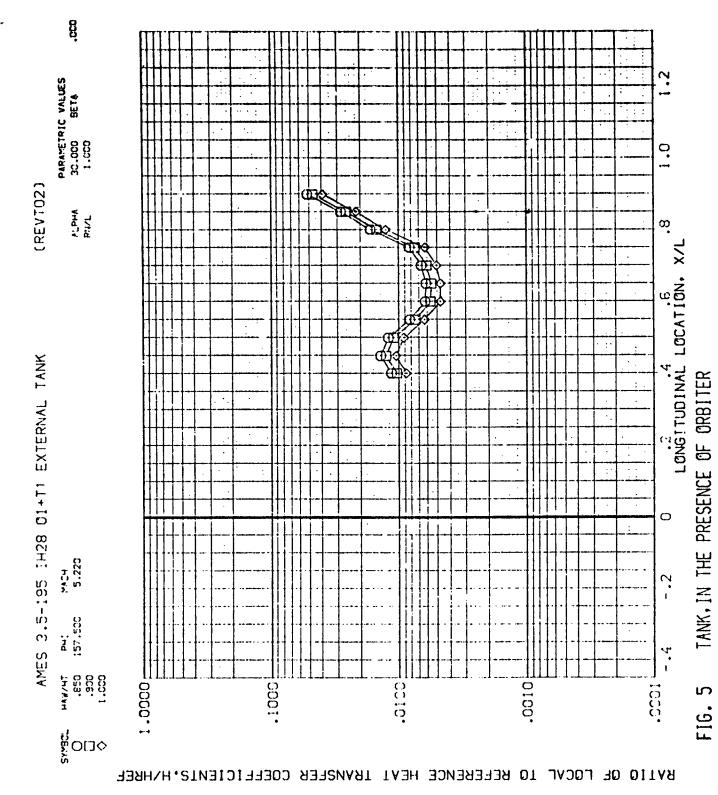
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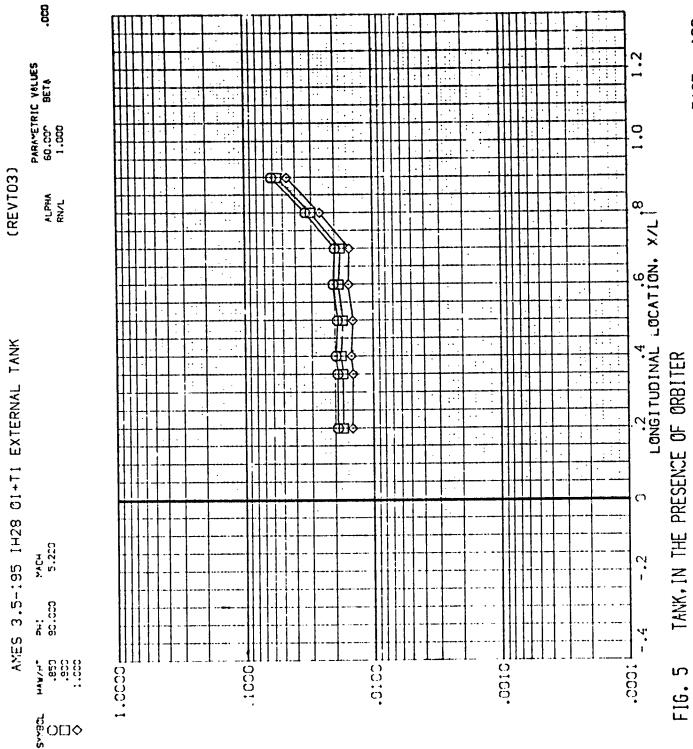
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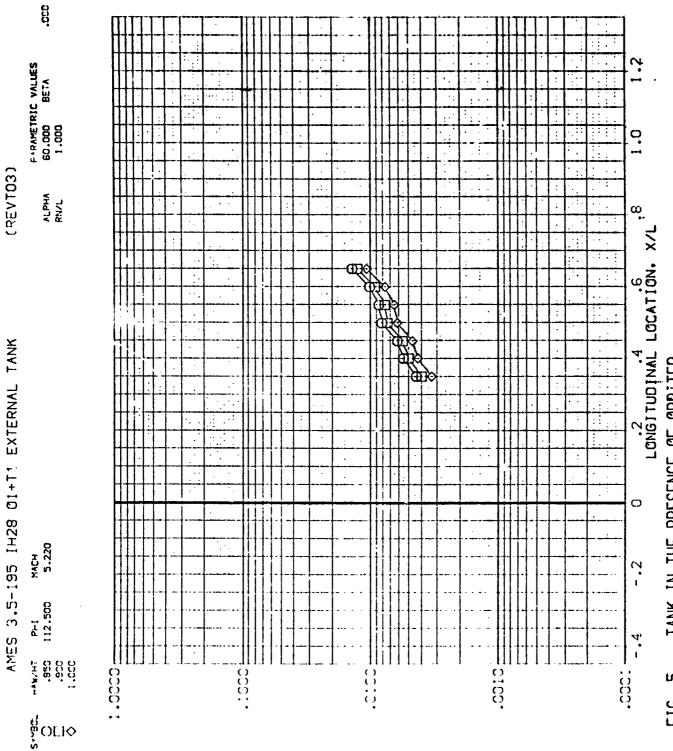
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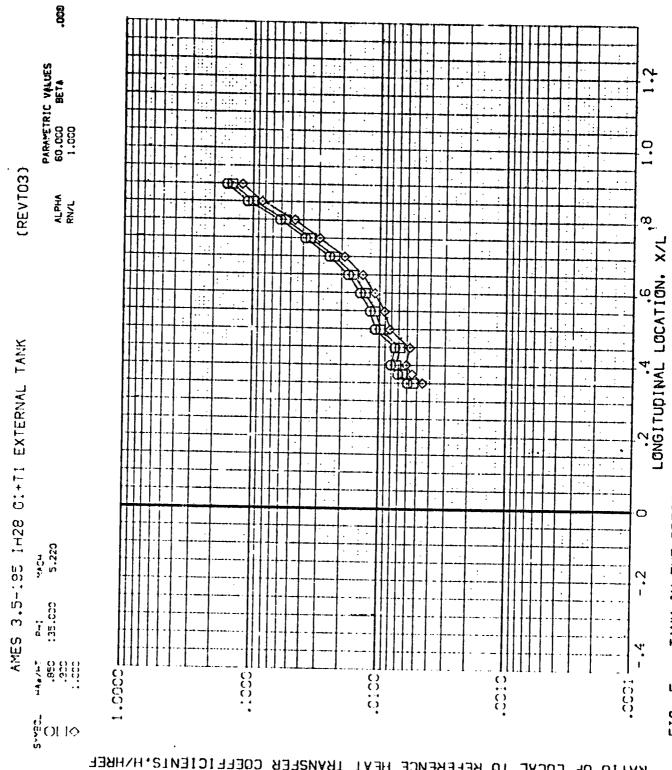


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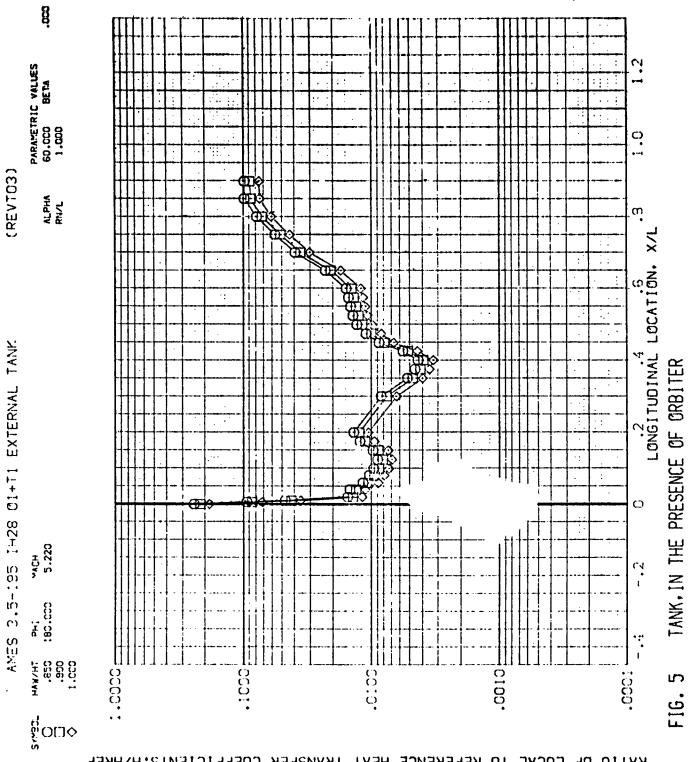
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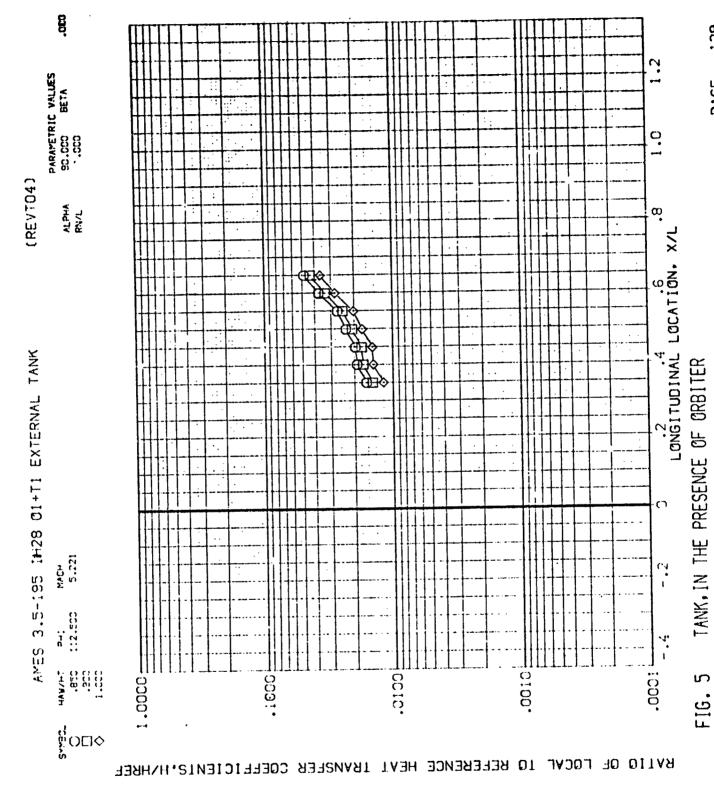


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FIG. 5 TANK, IN THE PRESENCE OF ORBITER

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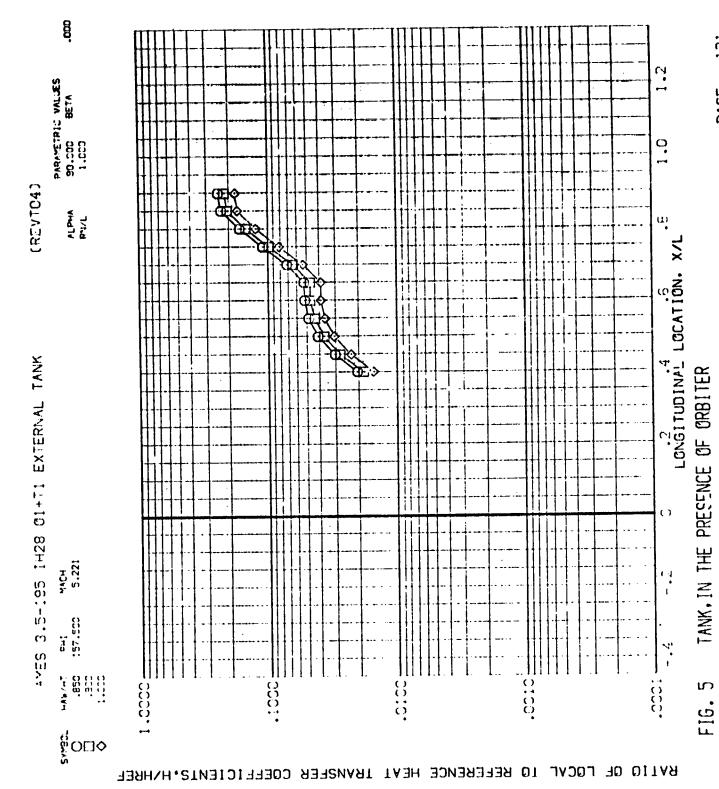
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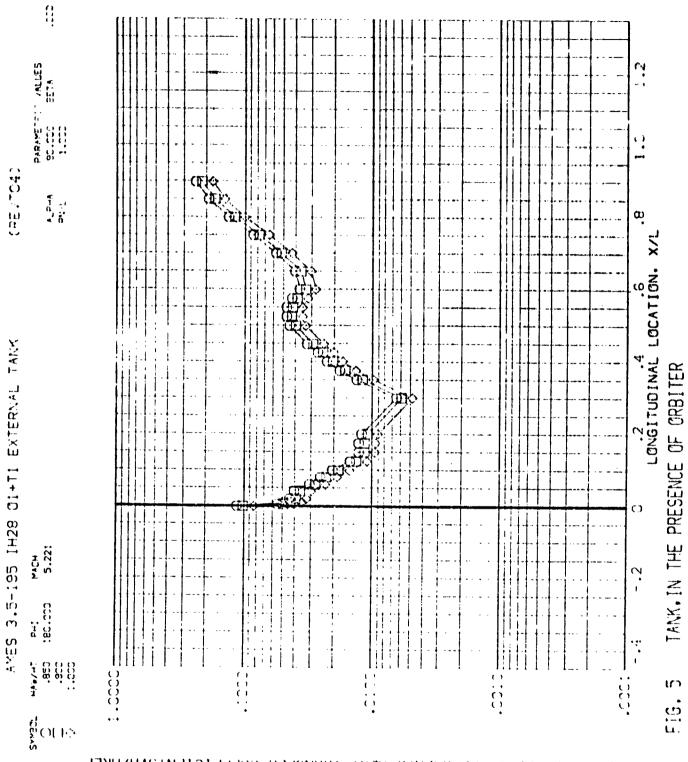
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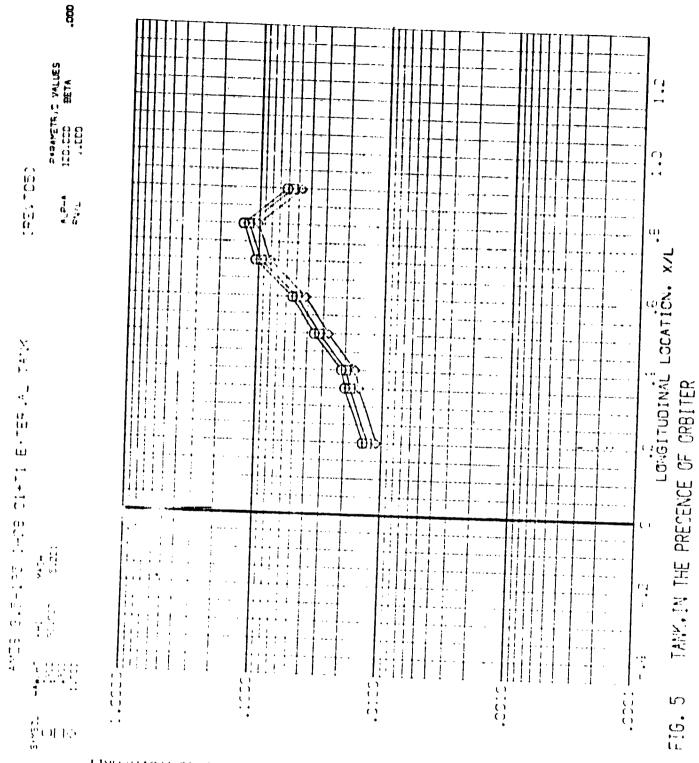
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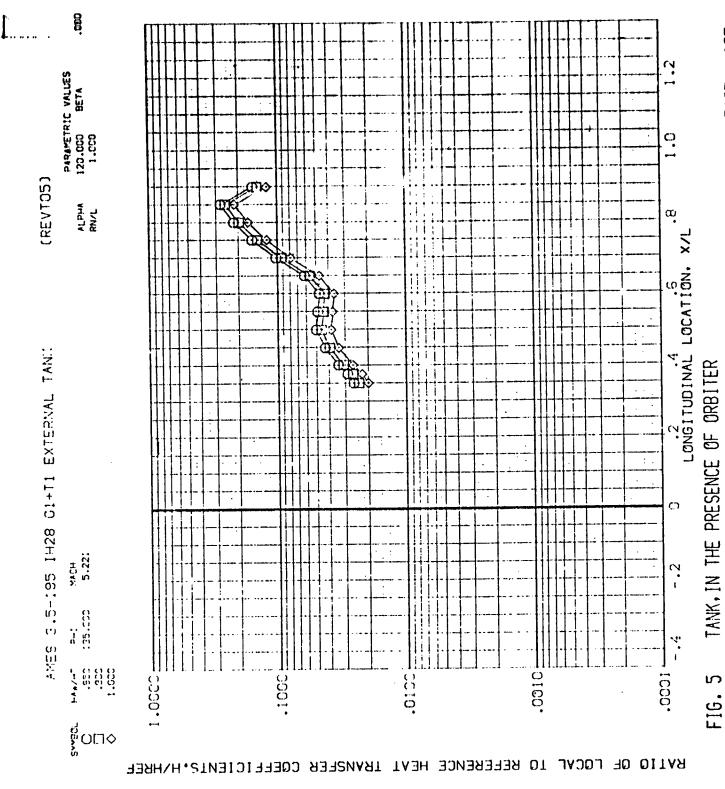
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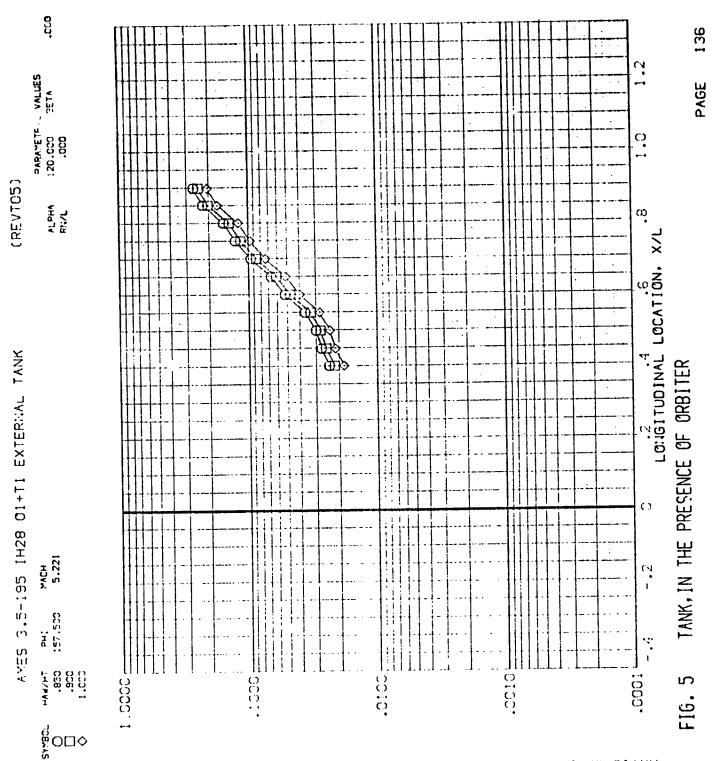
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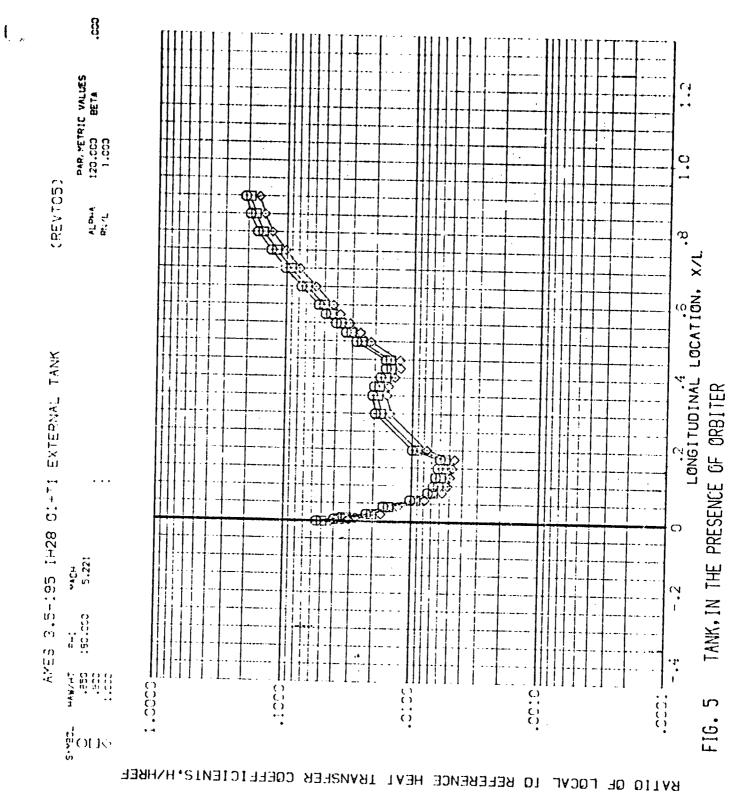
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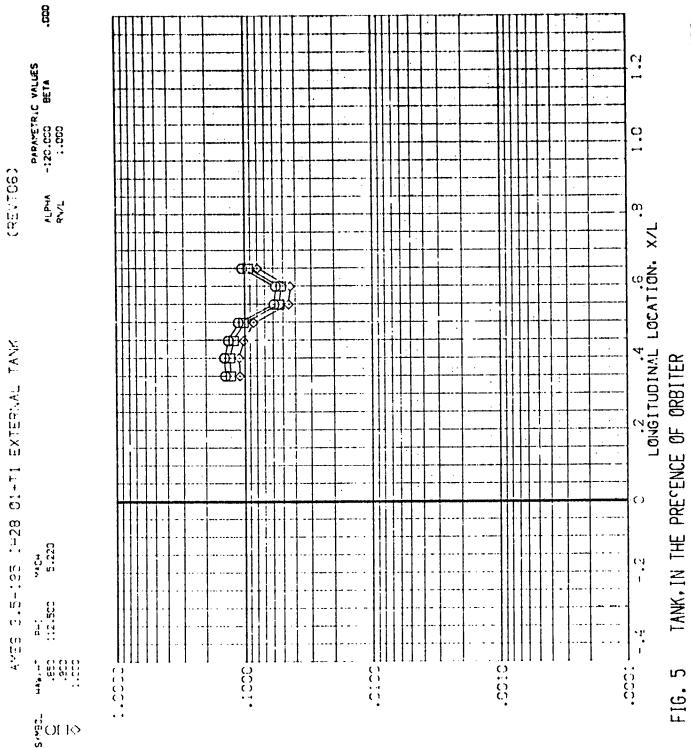


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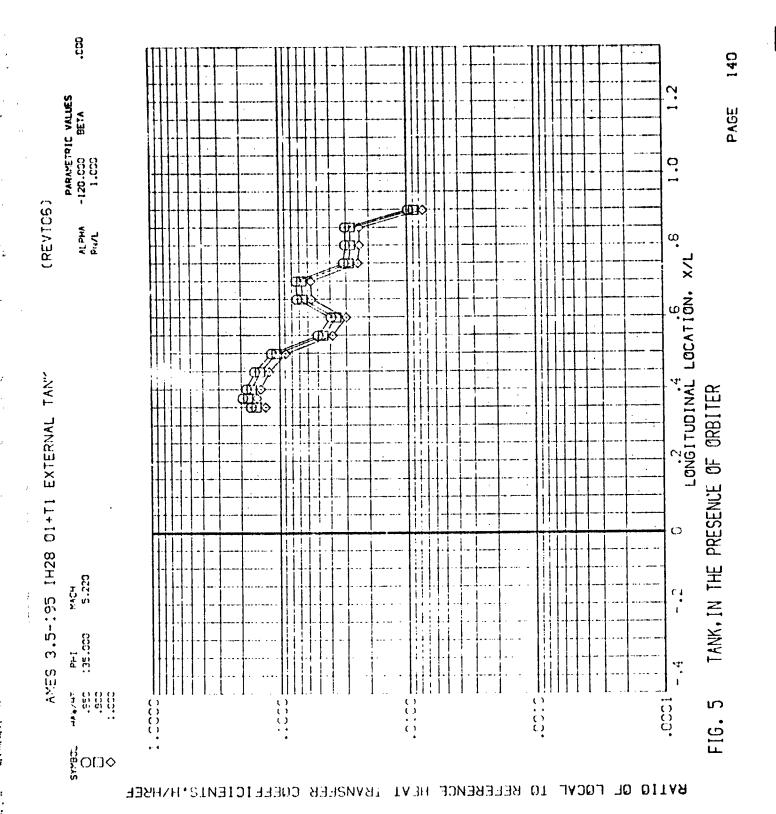
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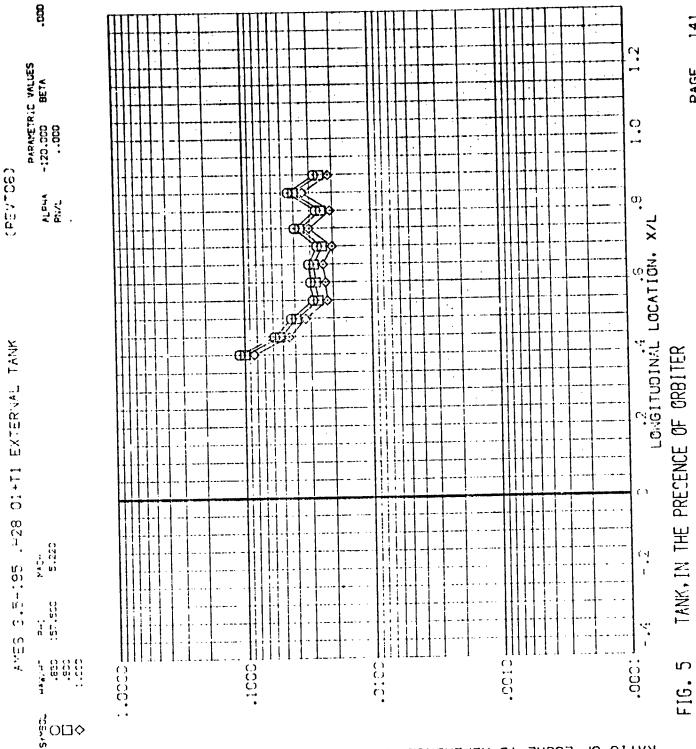


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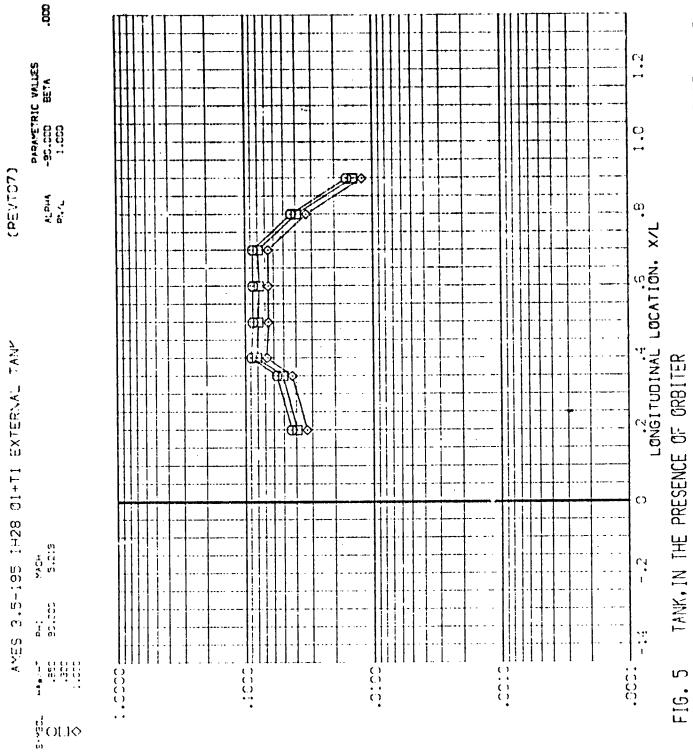
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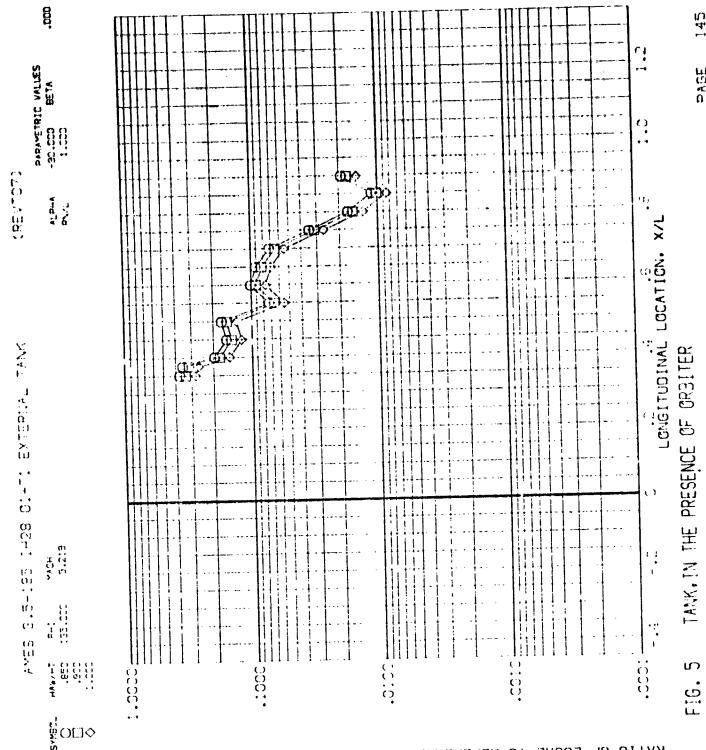
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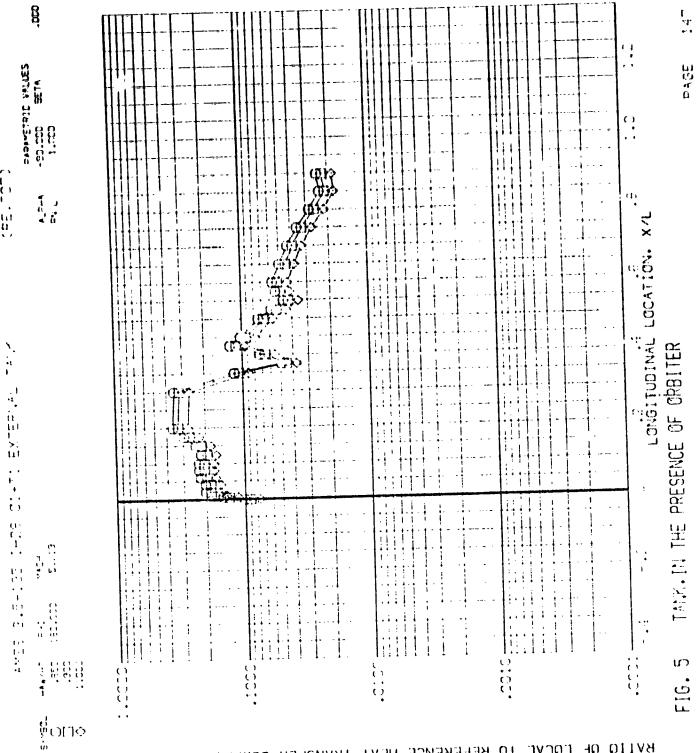
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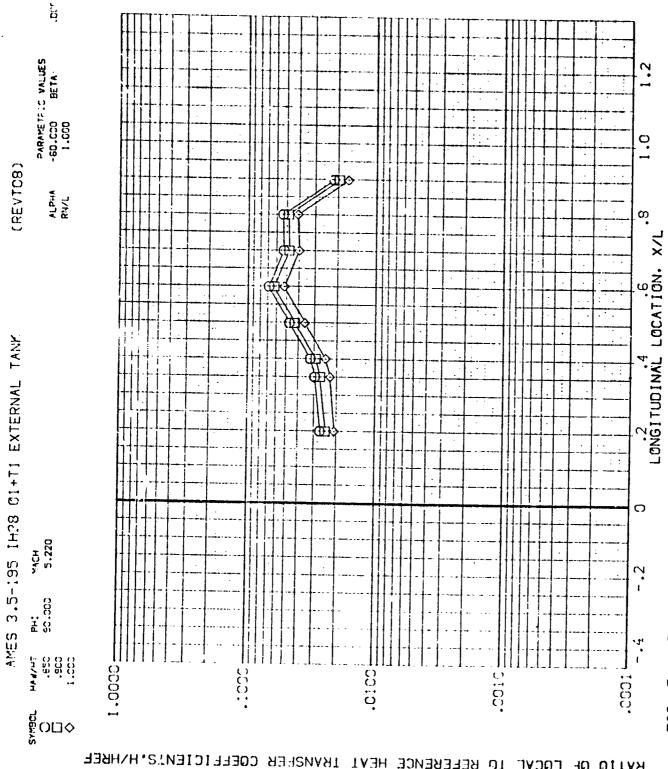
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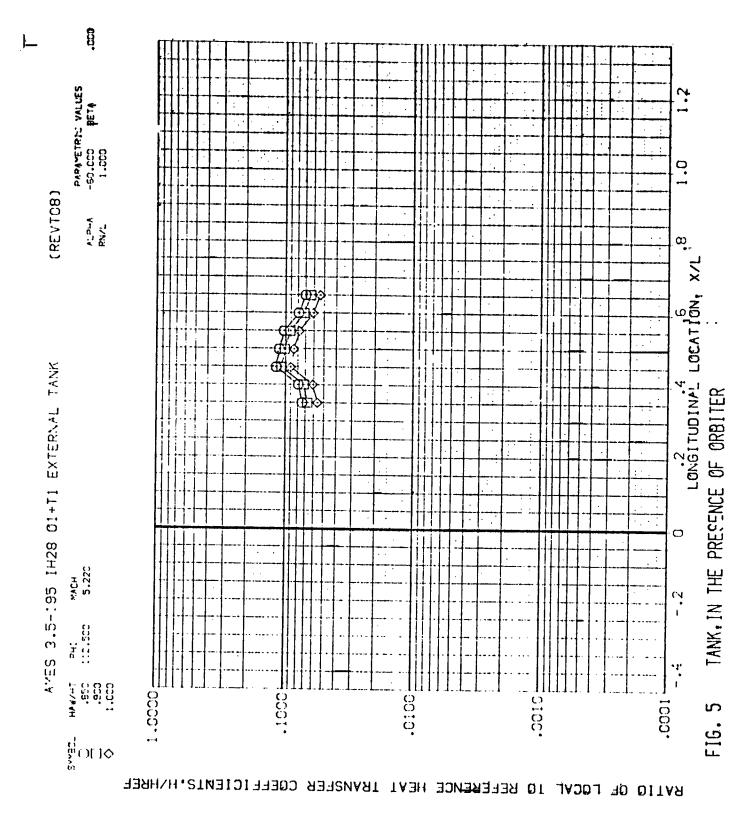
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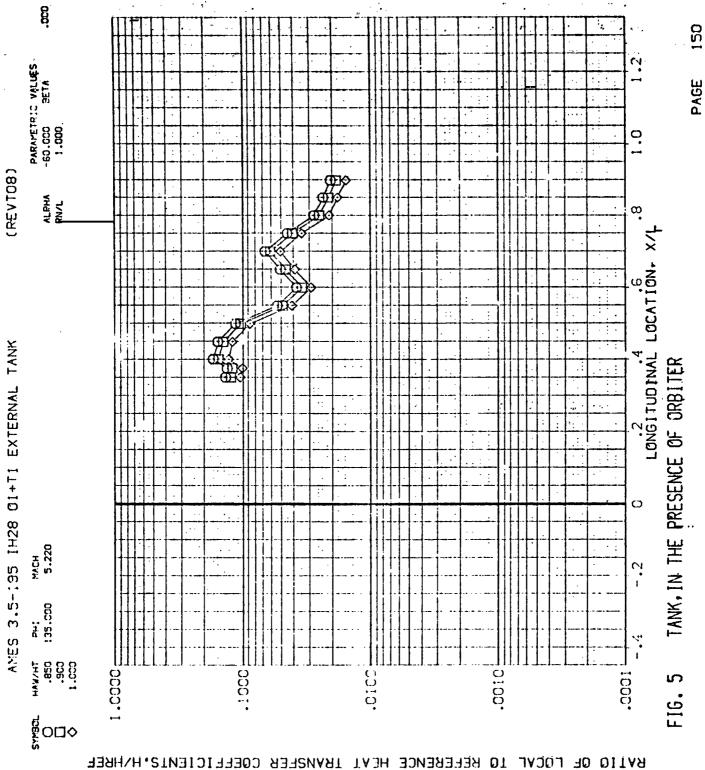


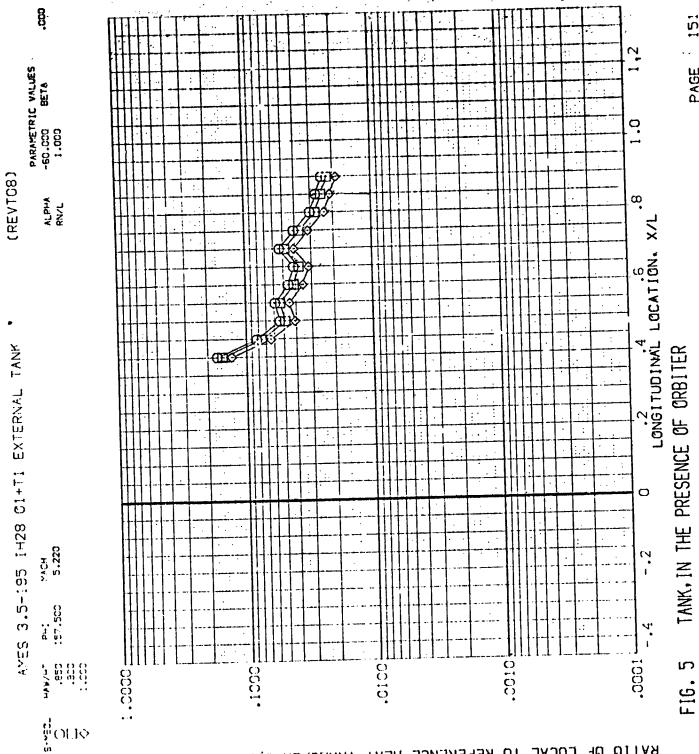
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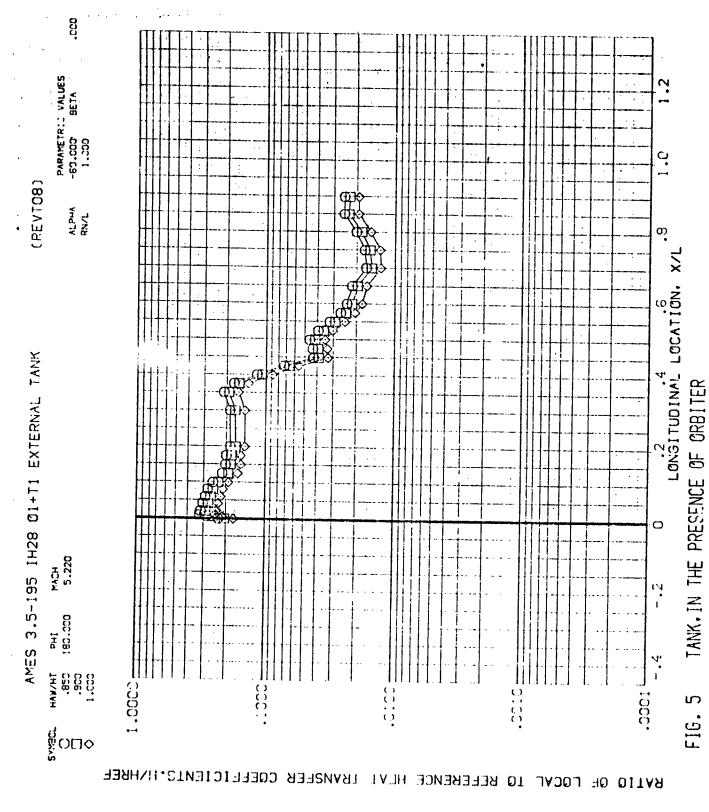
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RATIO OF LOCAL TO REFERENCE HEAT TRANSFER COEFFICIENTS, HAHREF



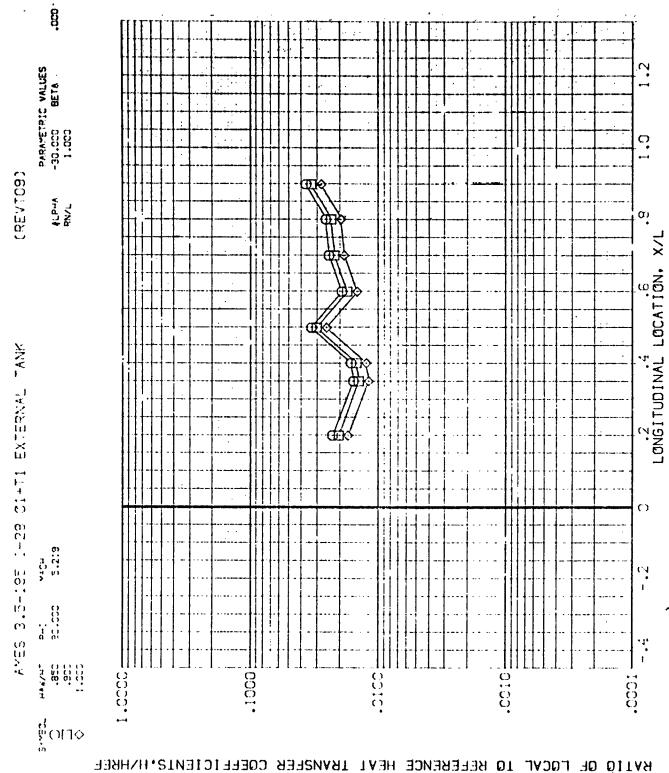
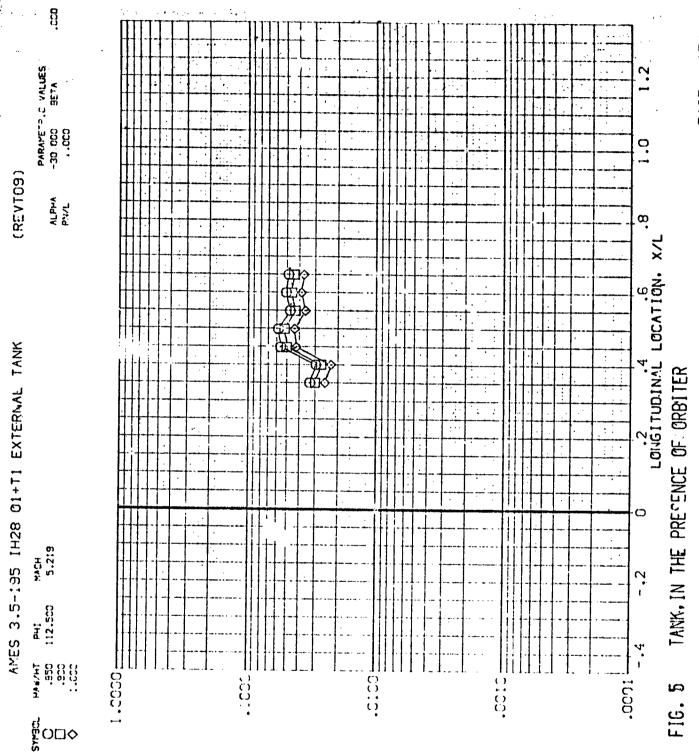
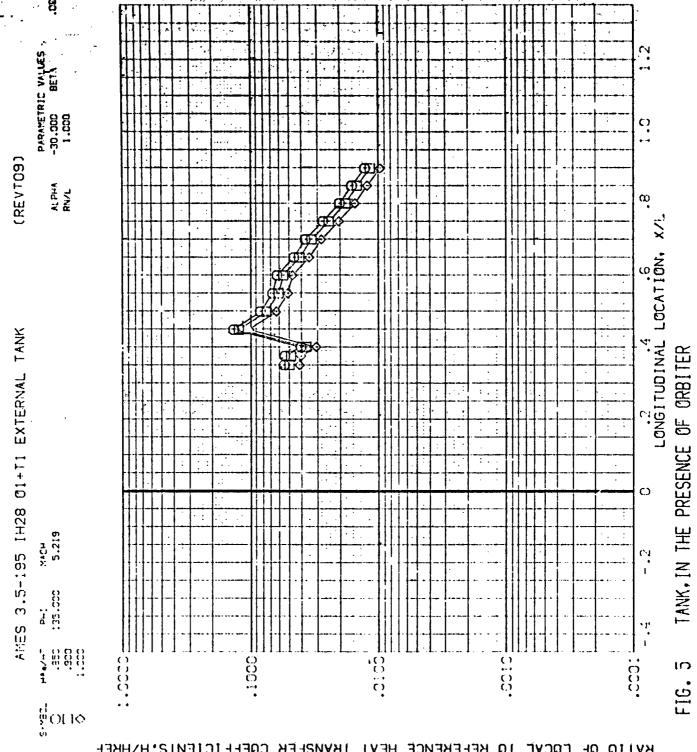


FIG. 5 TANK, IN THE PRESENCE OF ORBITER

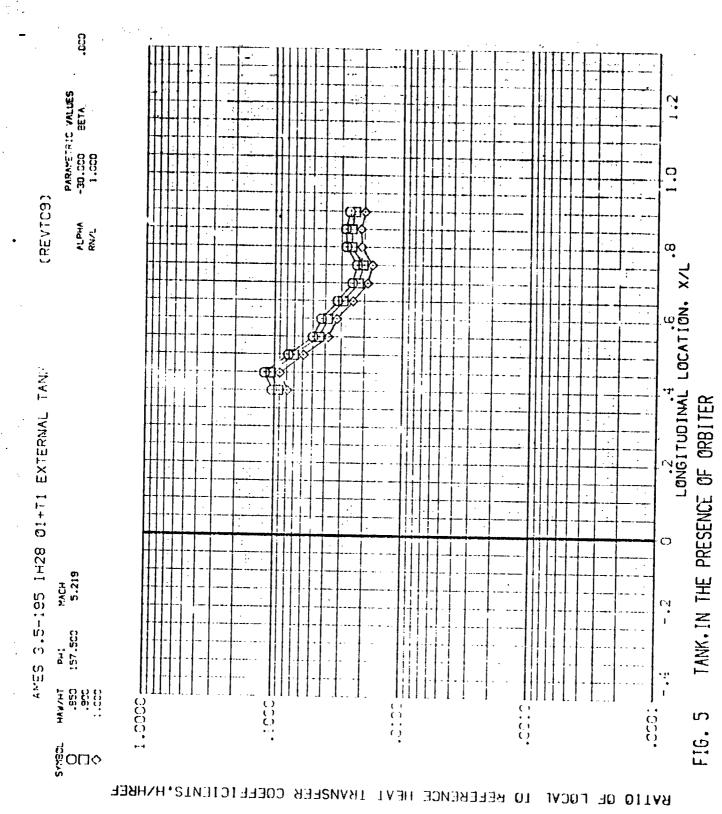
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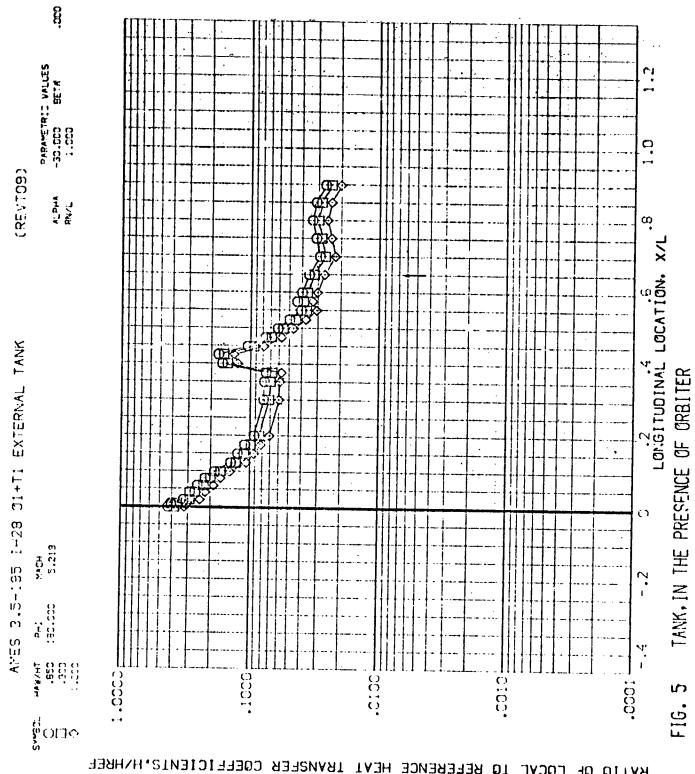


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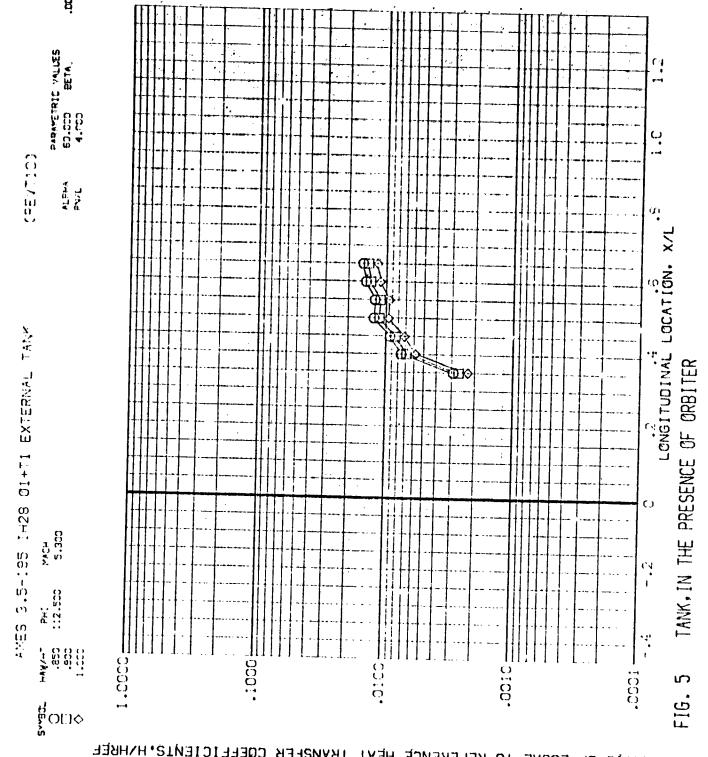
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TANK, IN THE PRESENCE OF ORBITER

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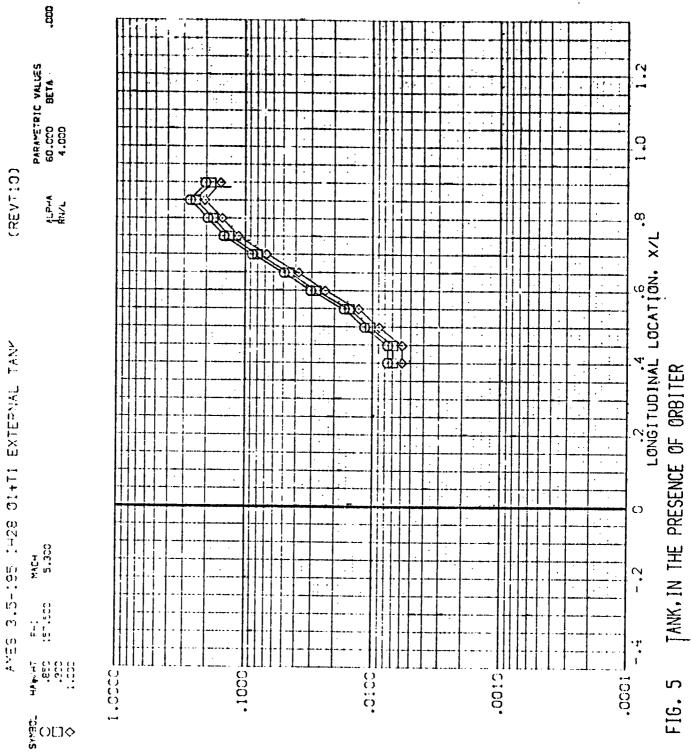
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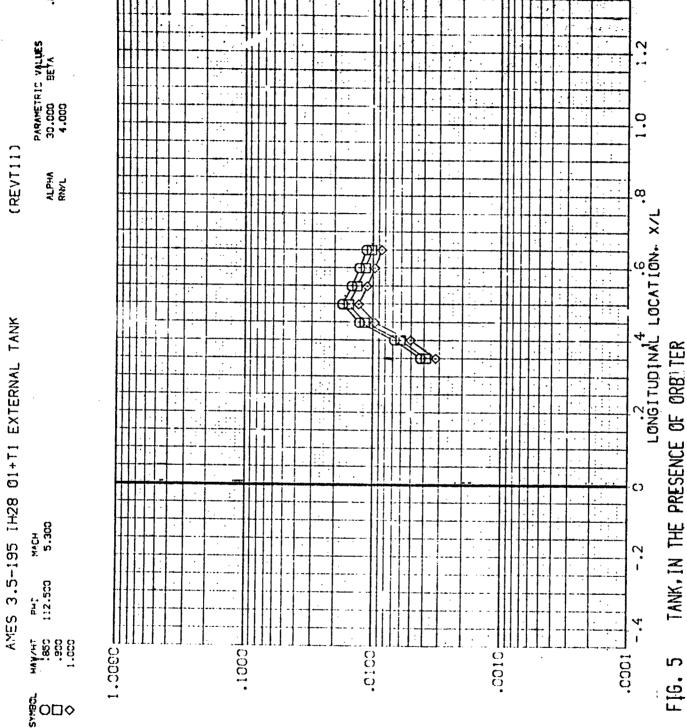
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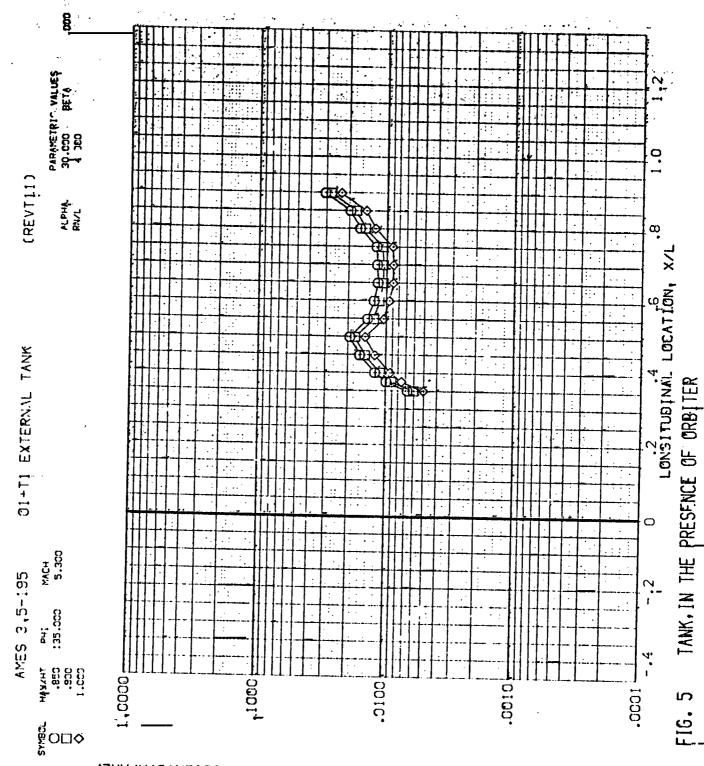
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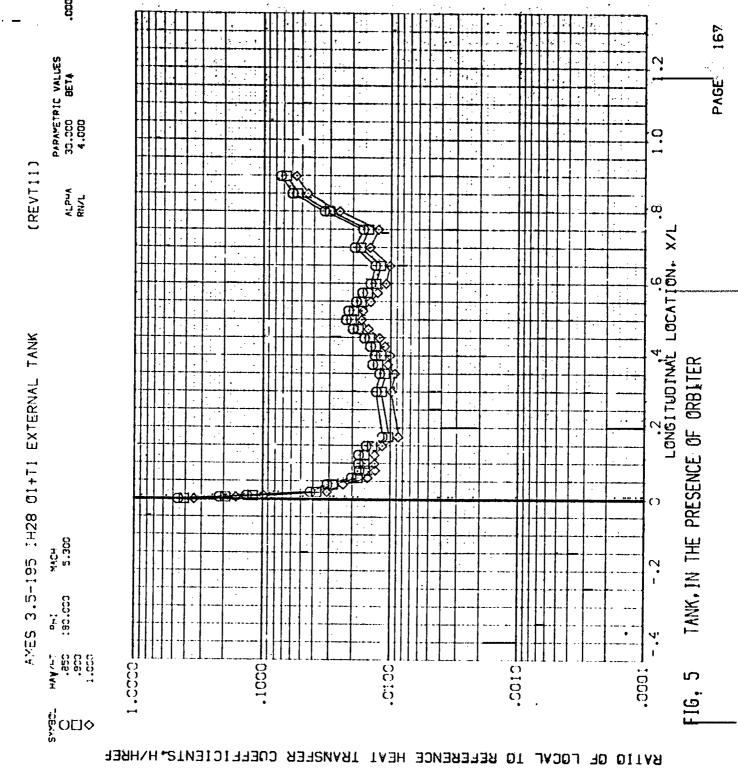
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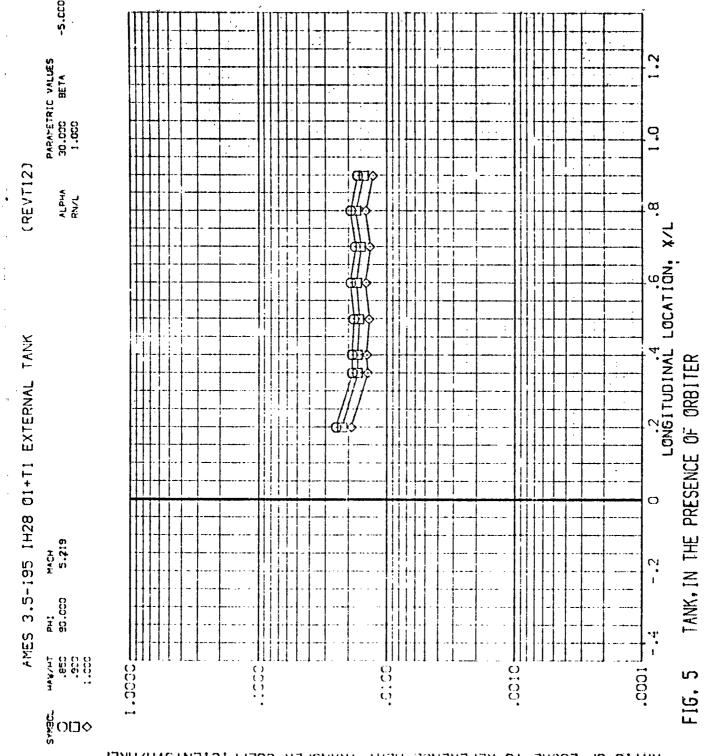


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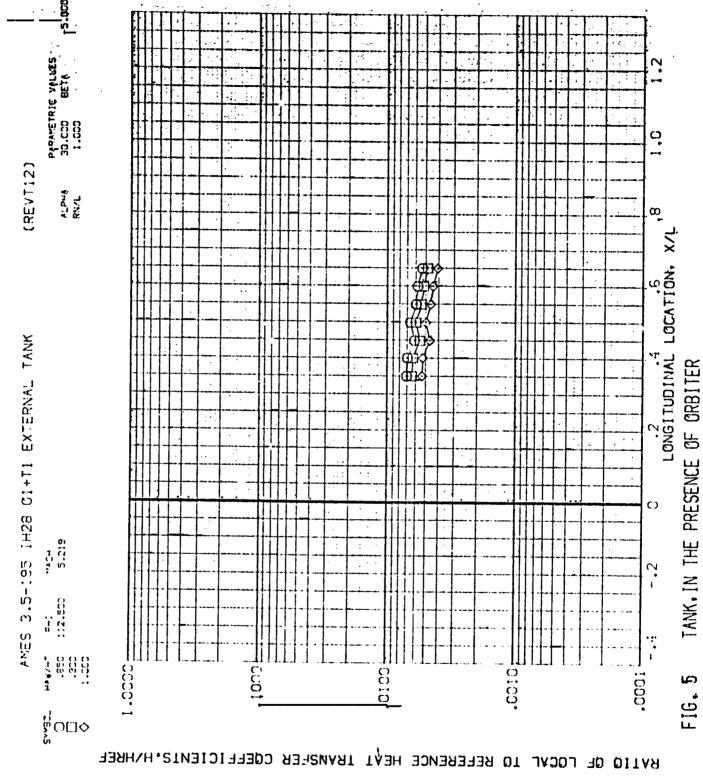
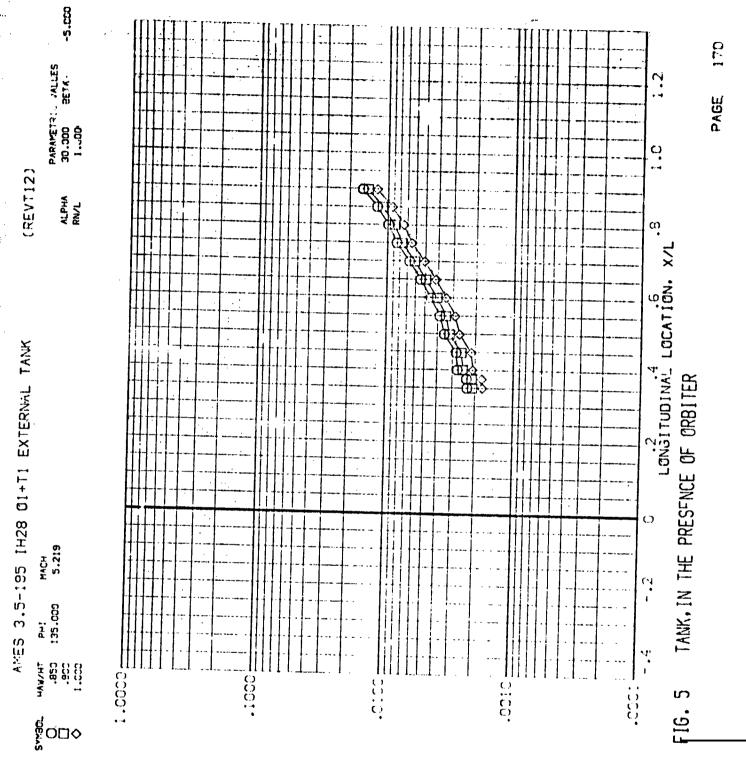


FIG. 5



RATIO OF LOCAL TO REFERENCE HEAT TRANSFER COEFFICIENTS, HYHREF

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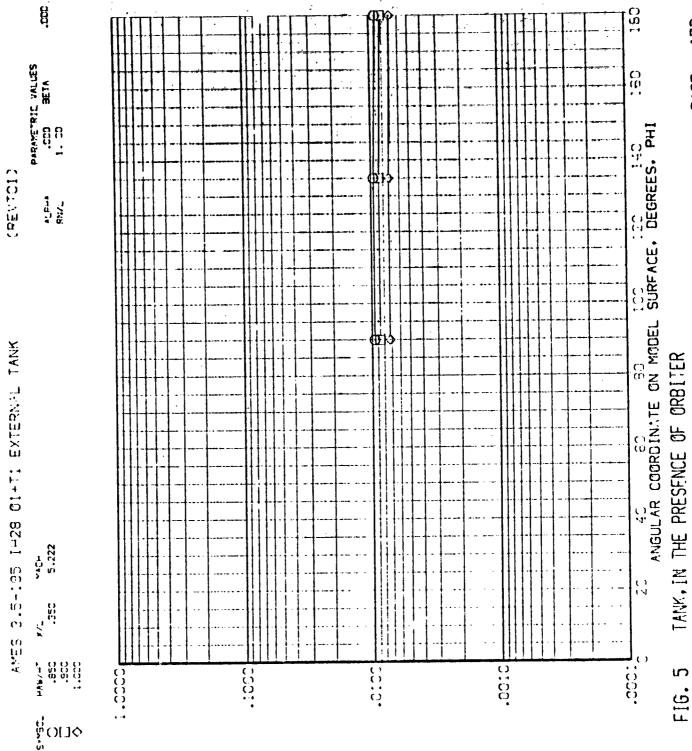
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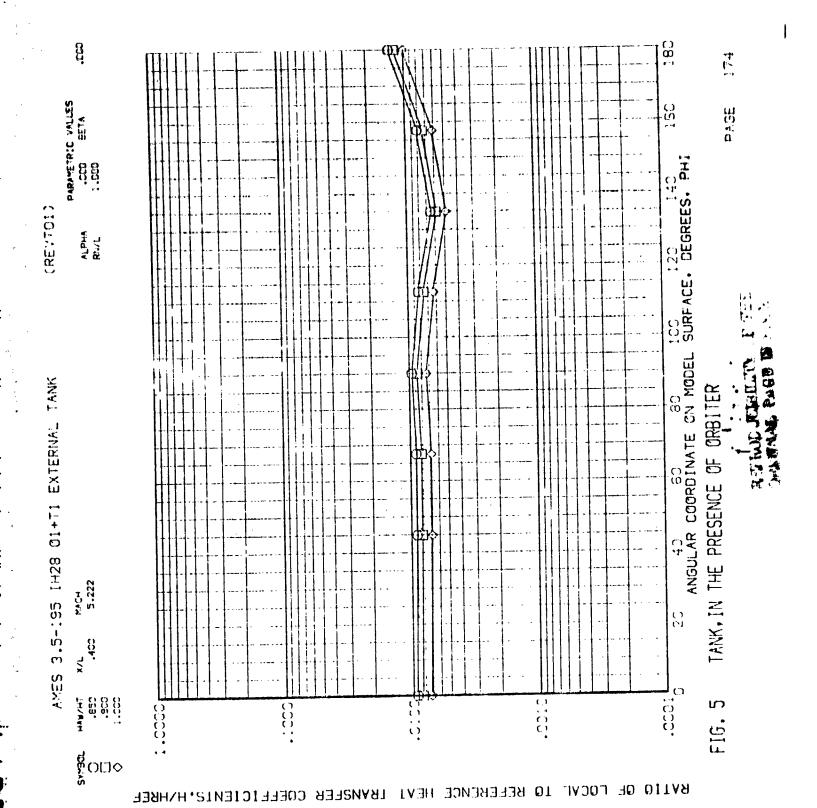
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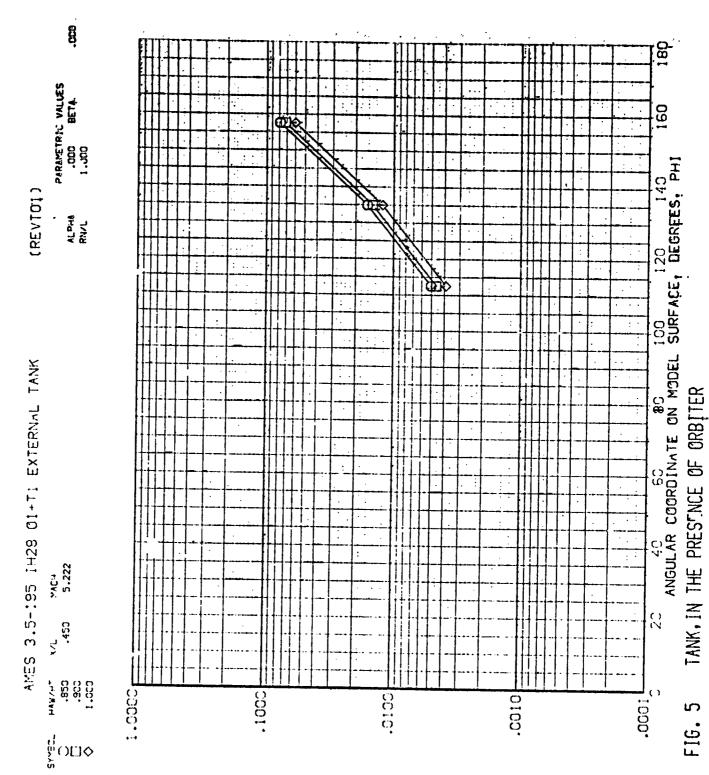
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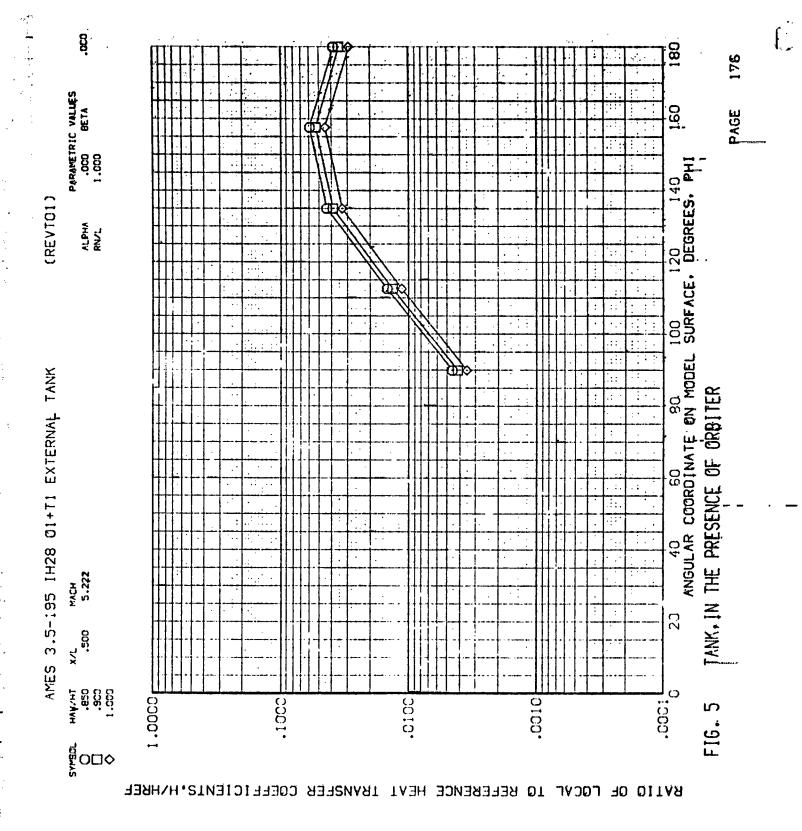


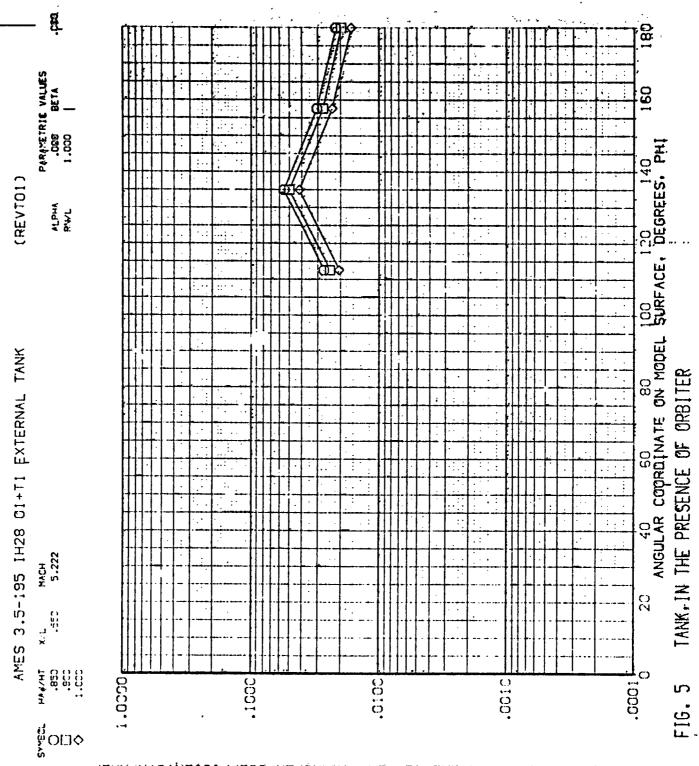
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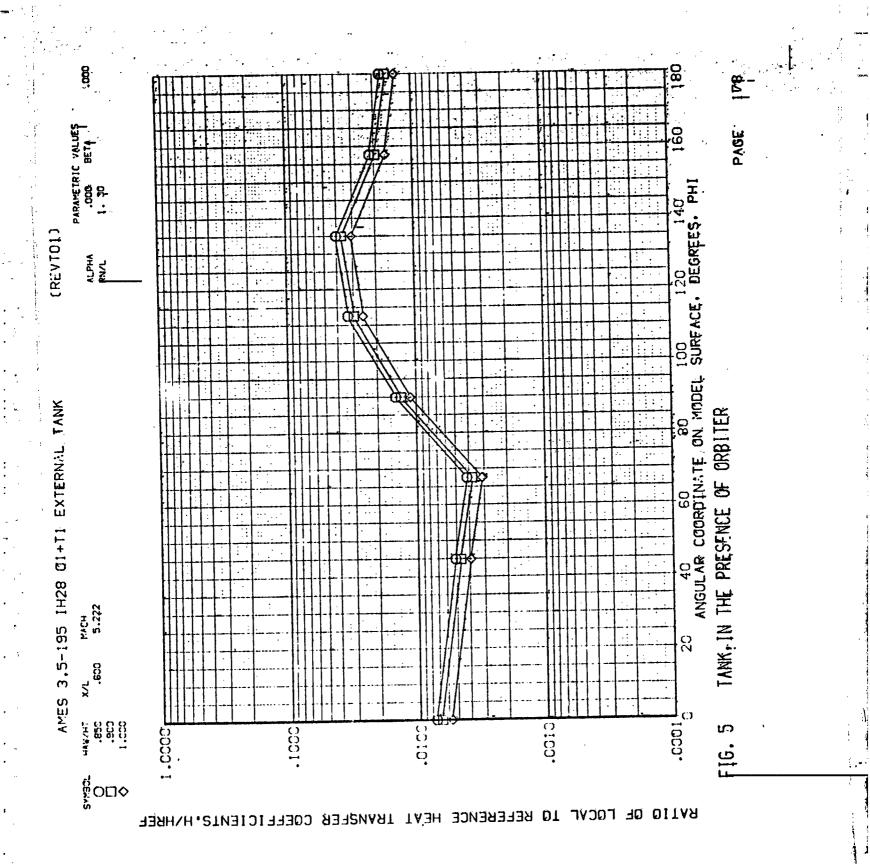
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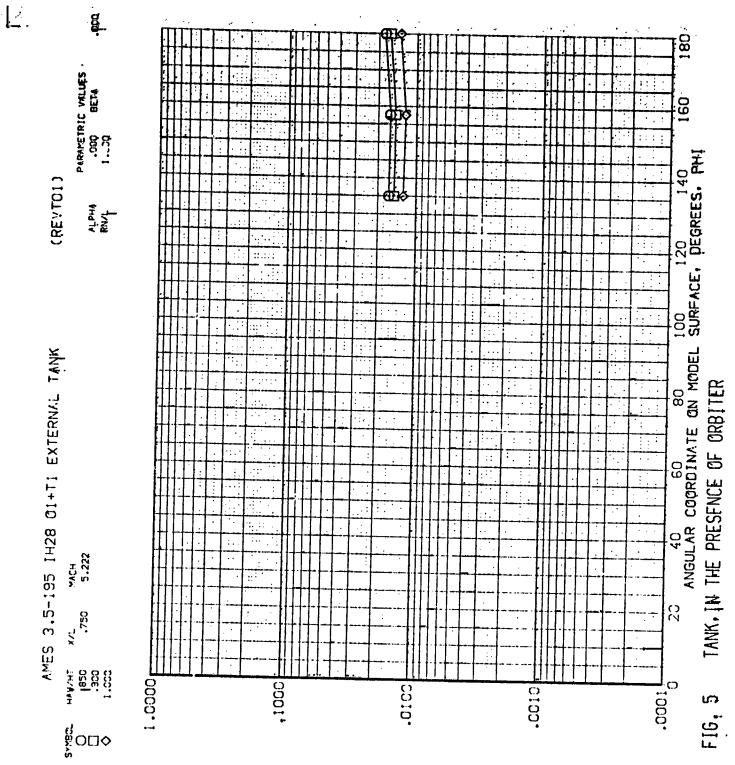


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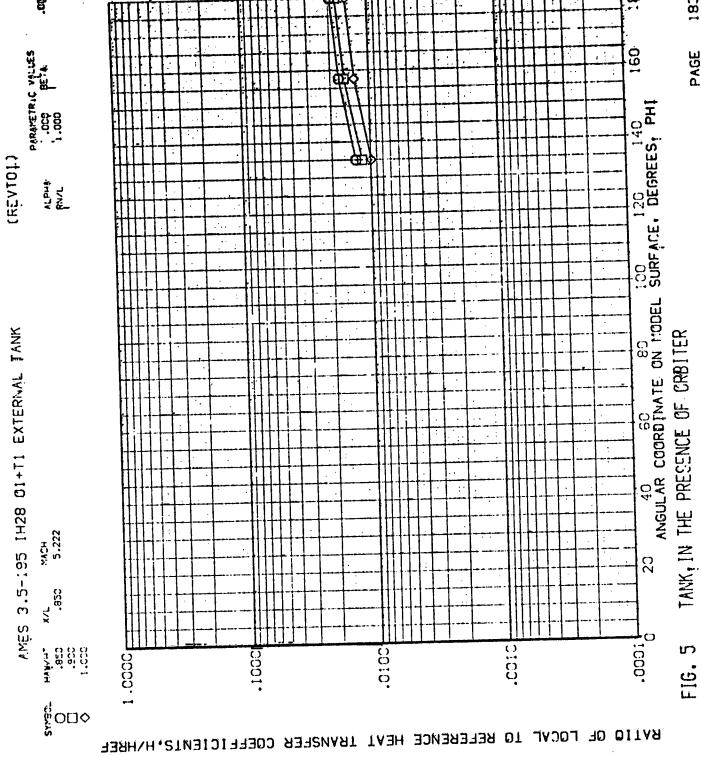
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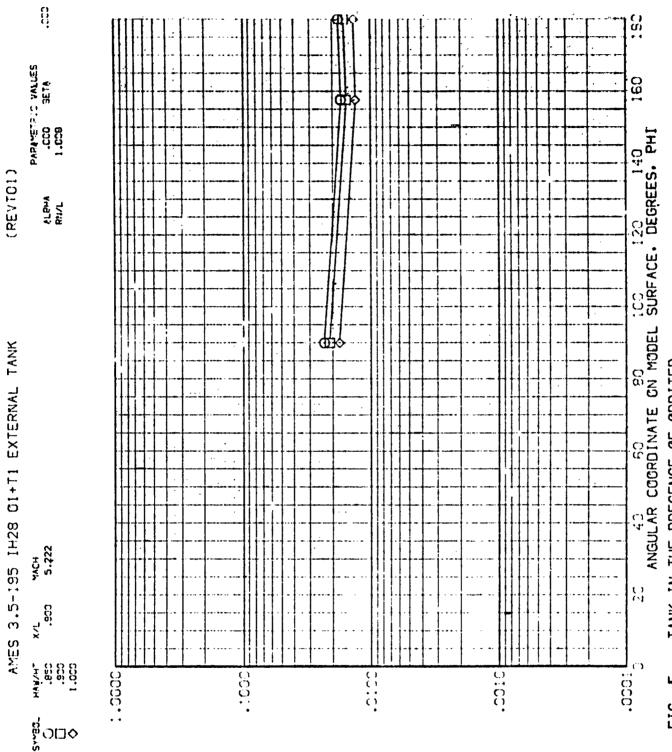


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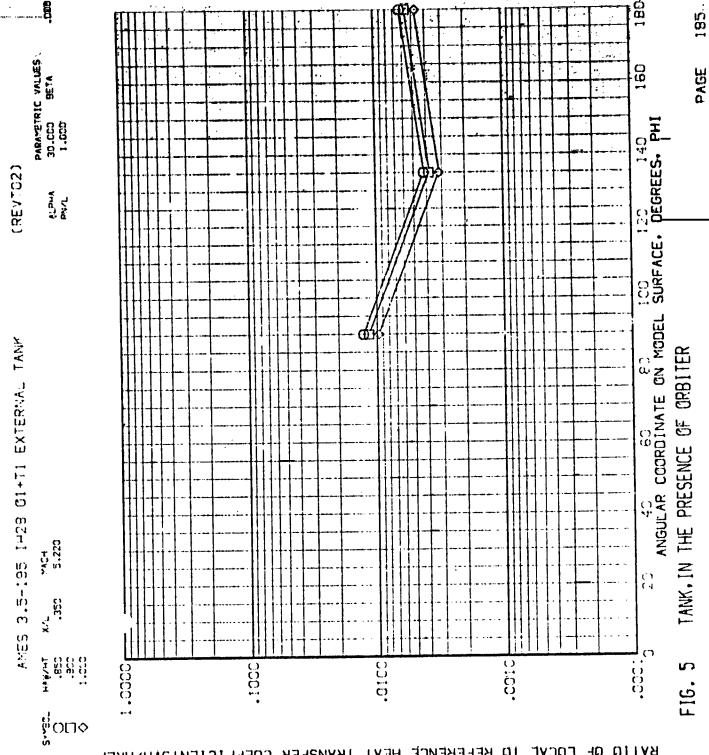
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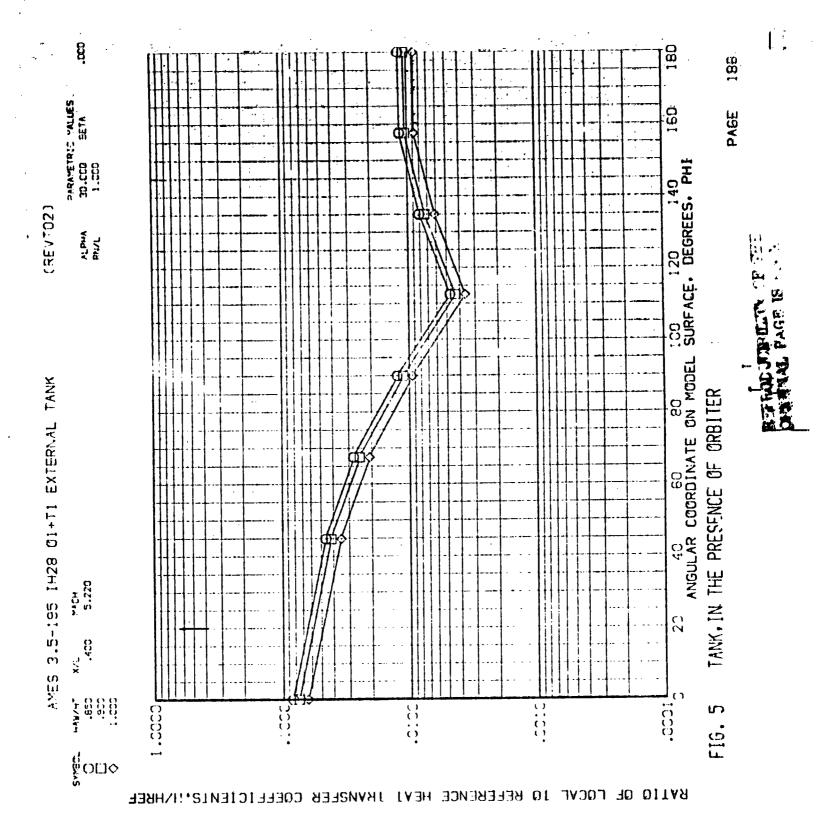


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FIG. 5 TANK, IN THE PRESENCE OF ORBITER

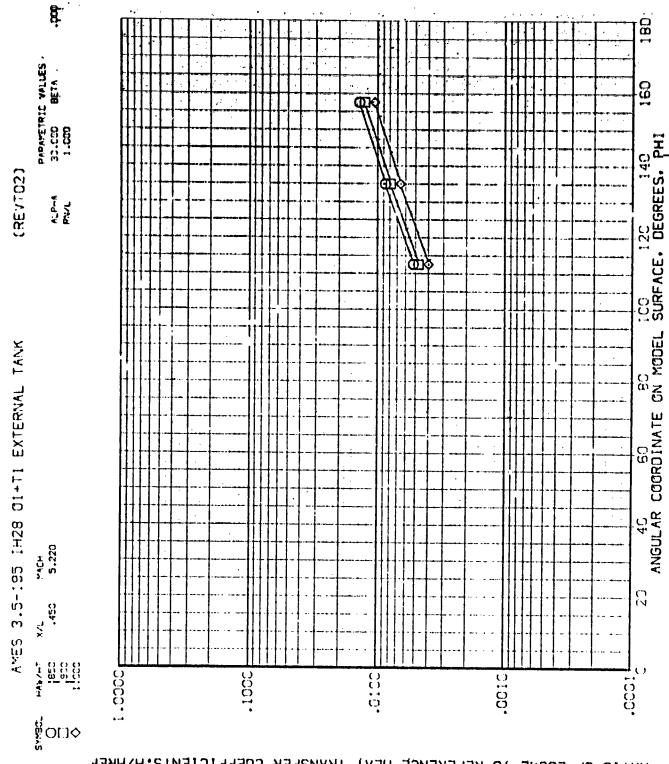


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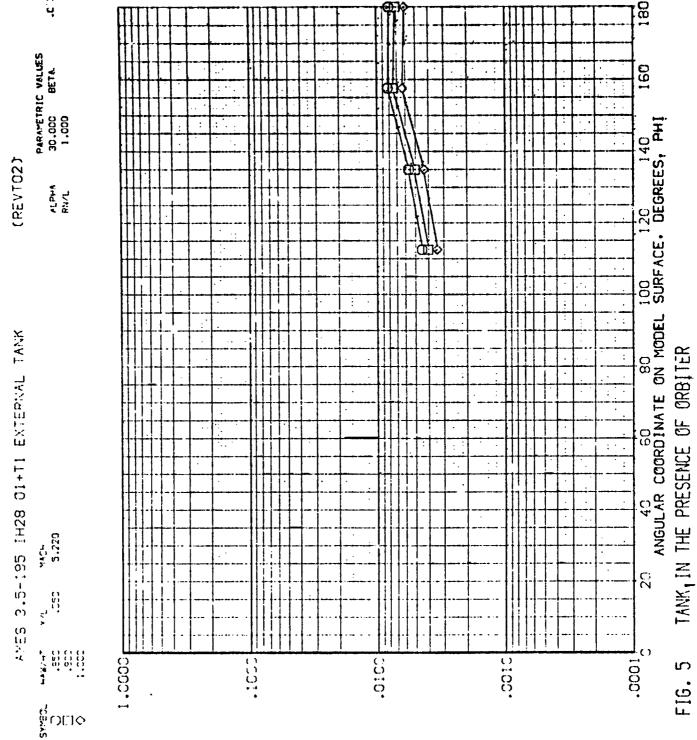
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FIG. 5 TANK, IN THE PRESENCE OF ORBITER

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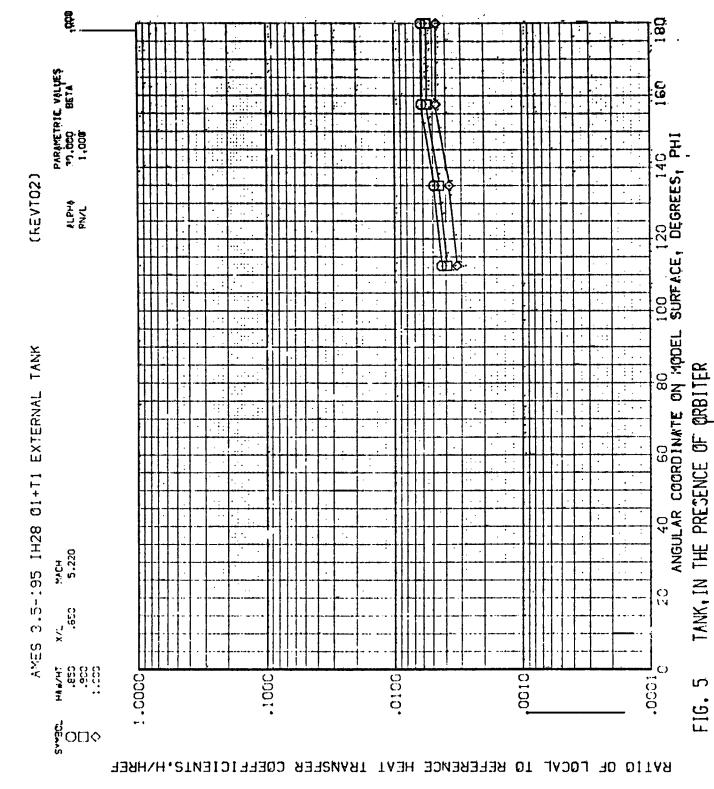
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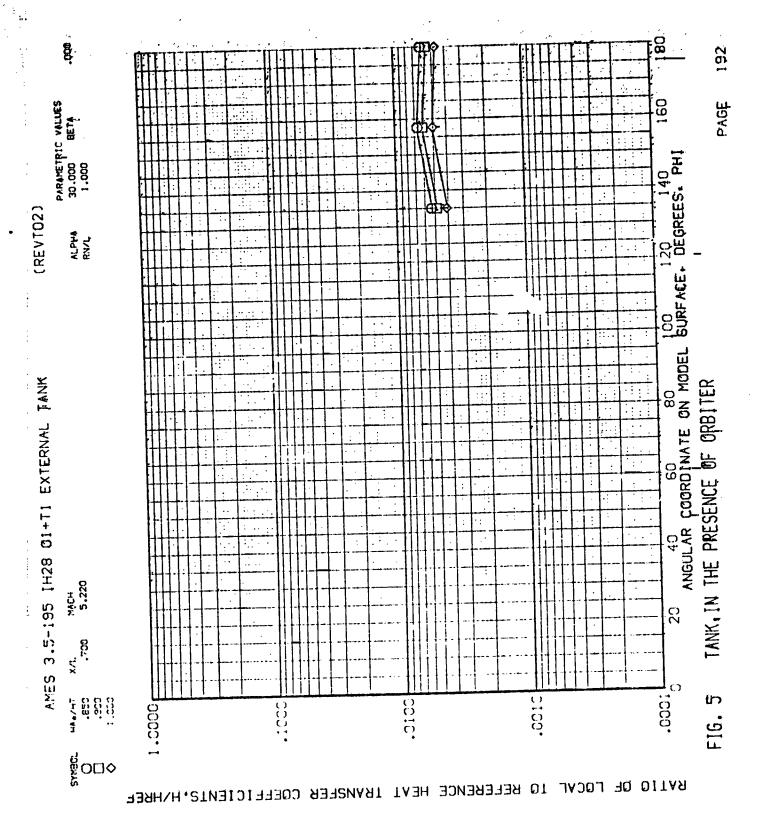
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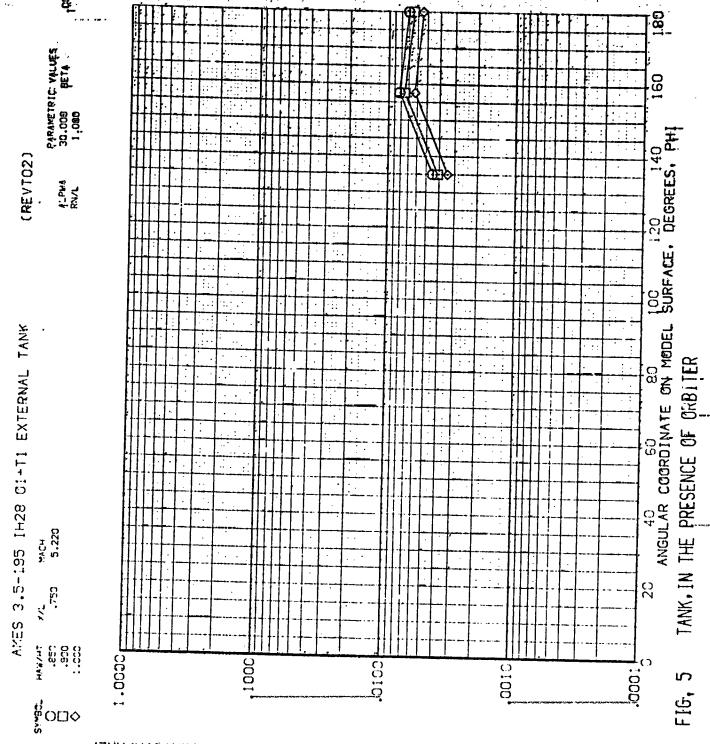
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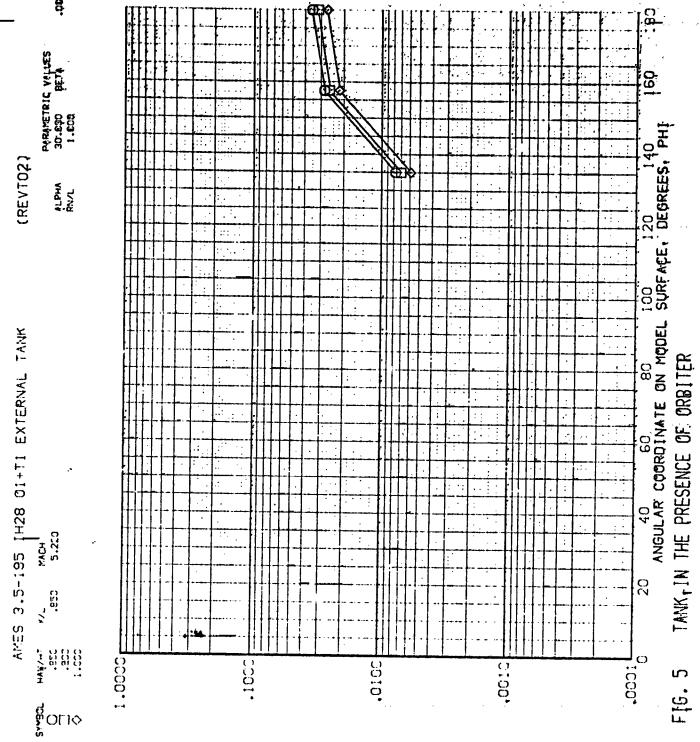
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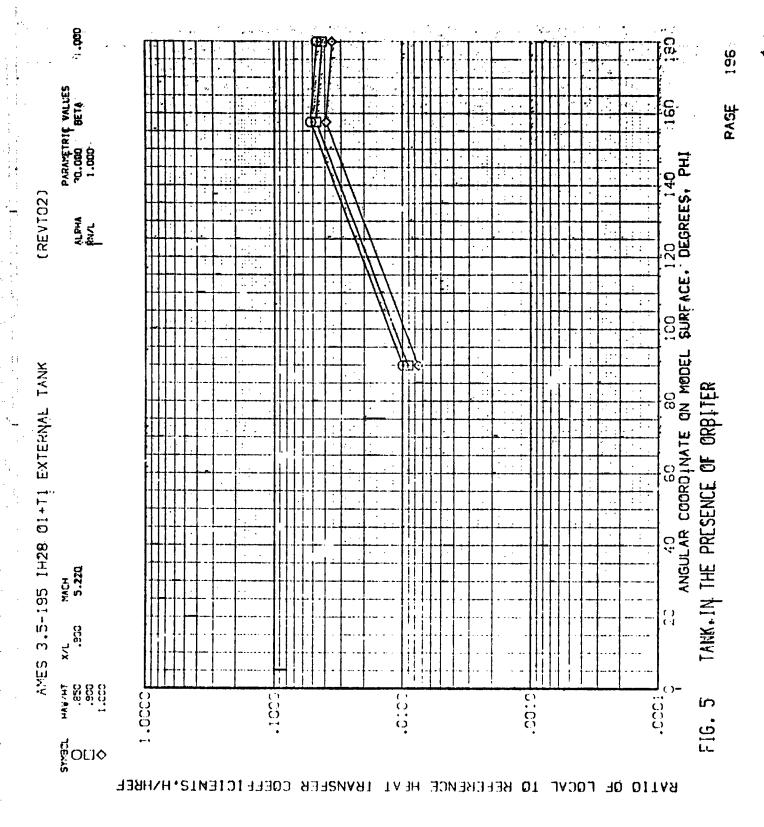


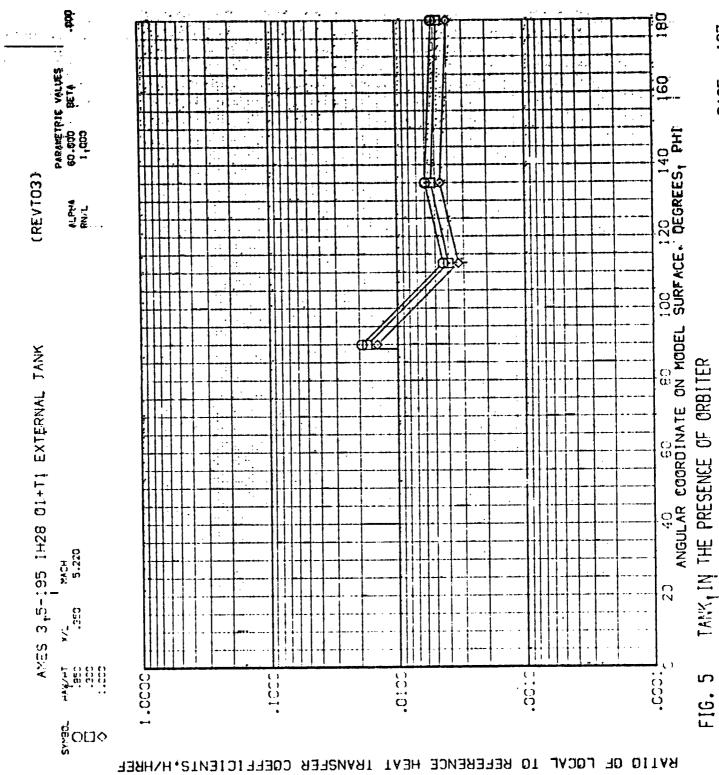


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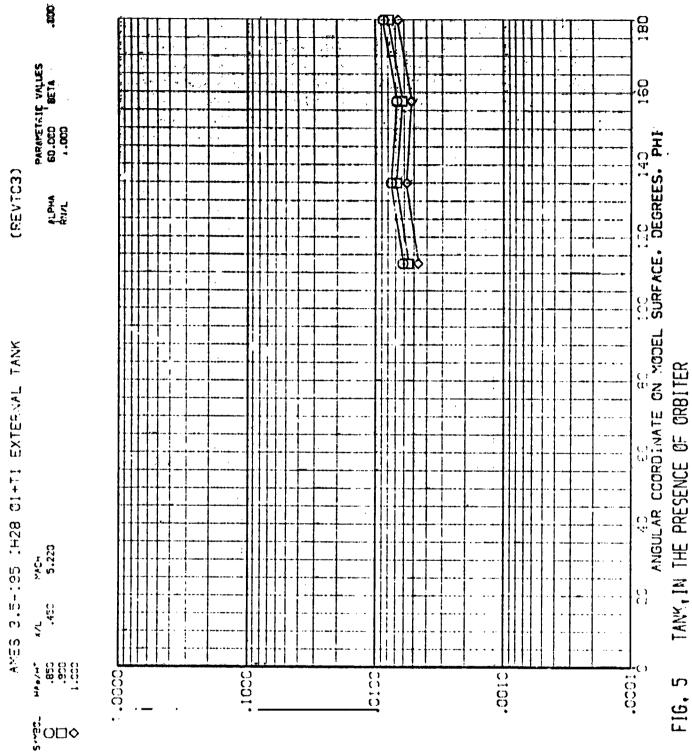
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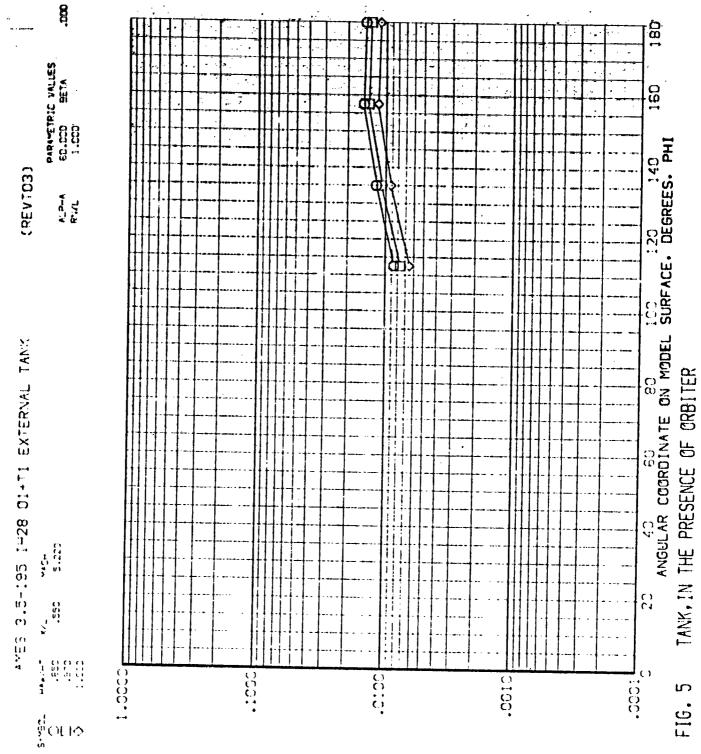


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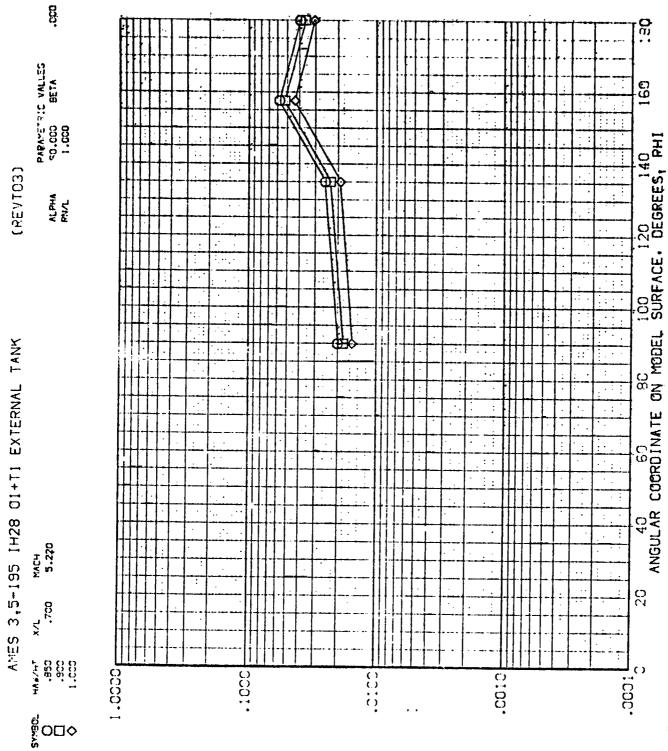


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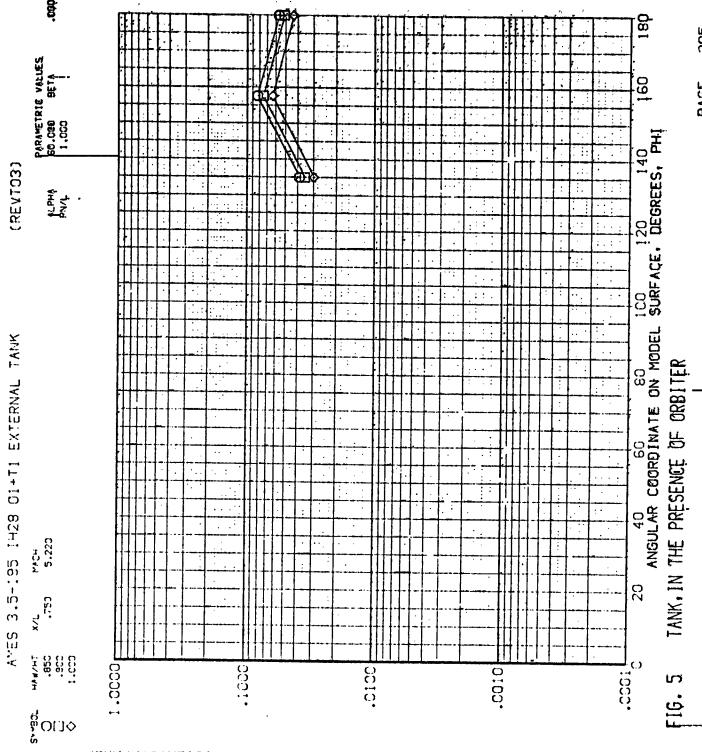
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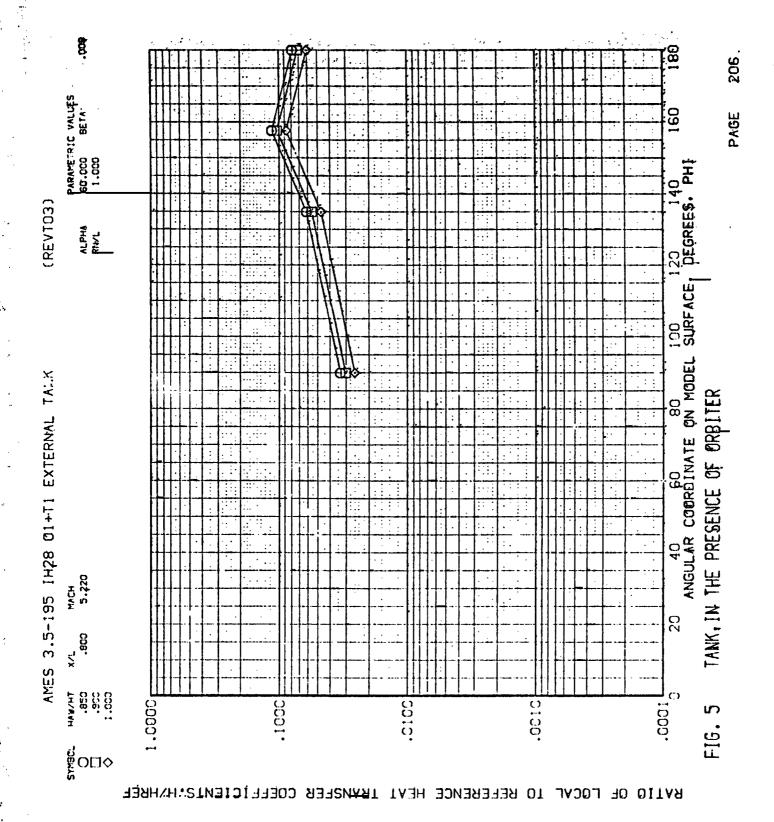


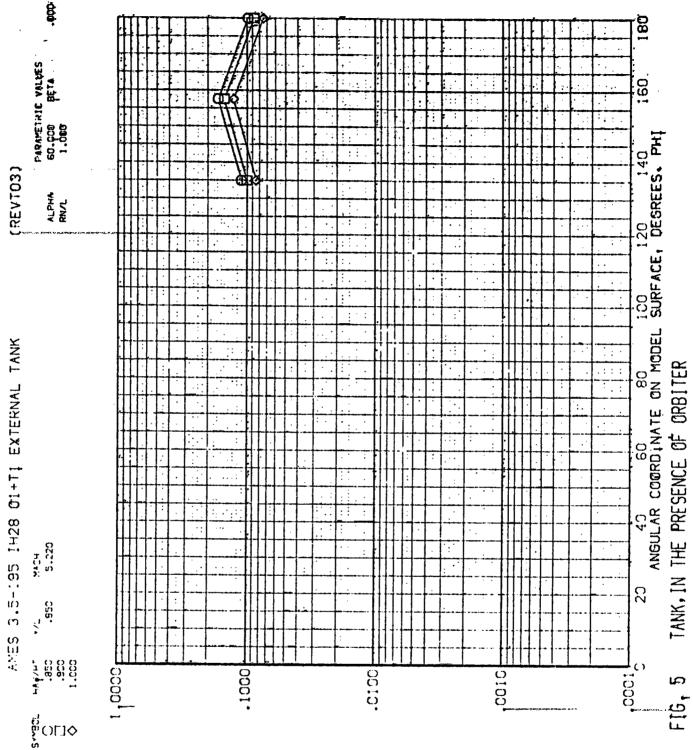
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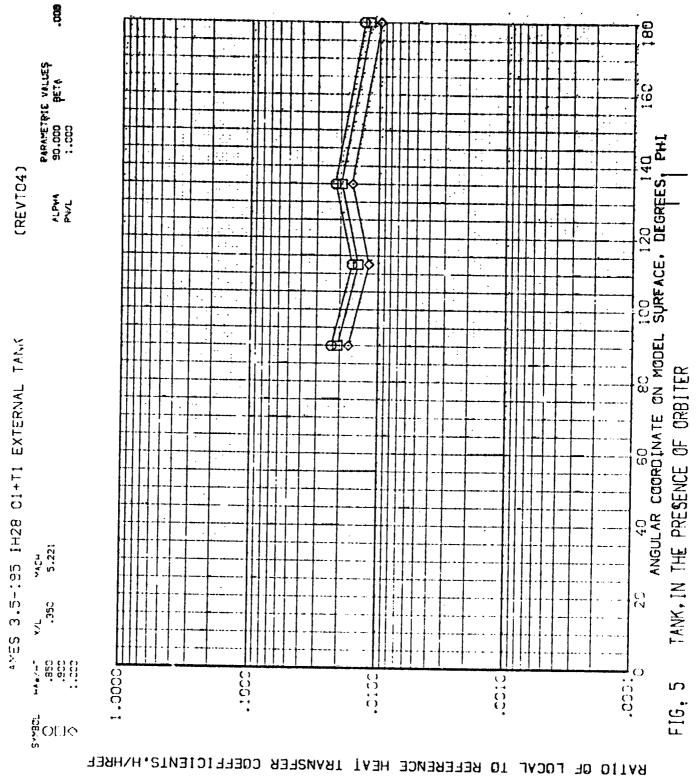
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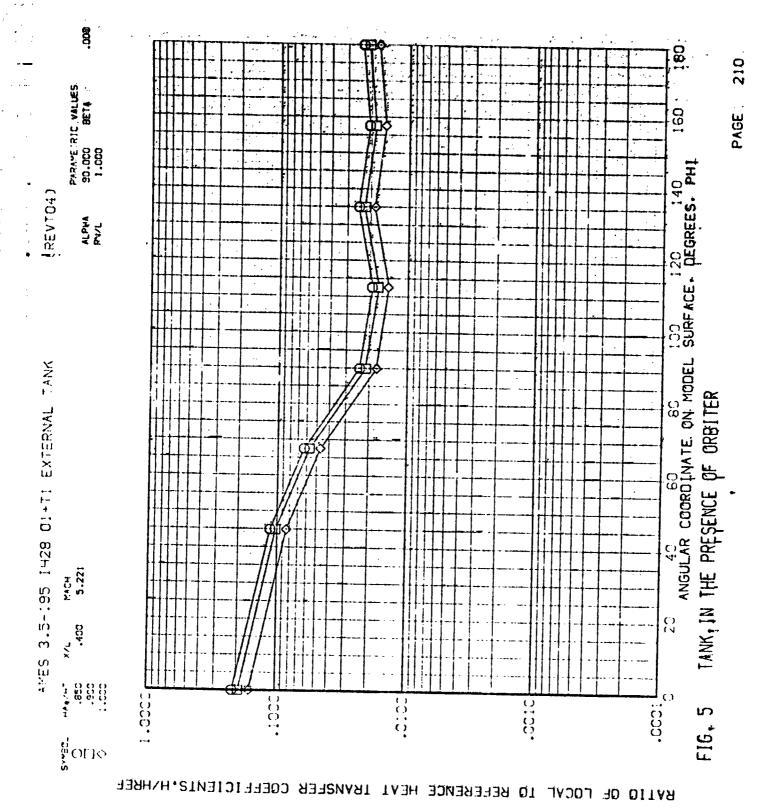


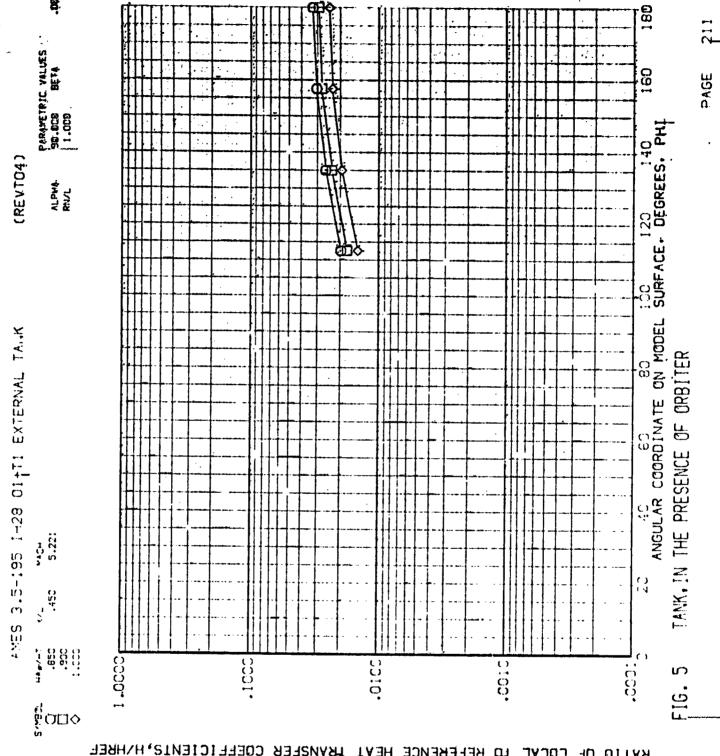
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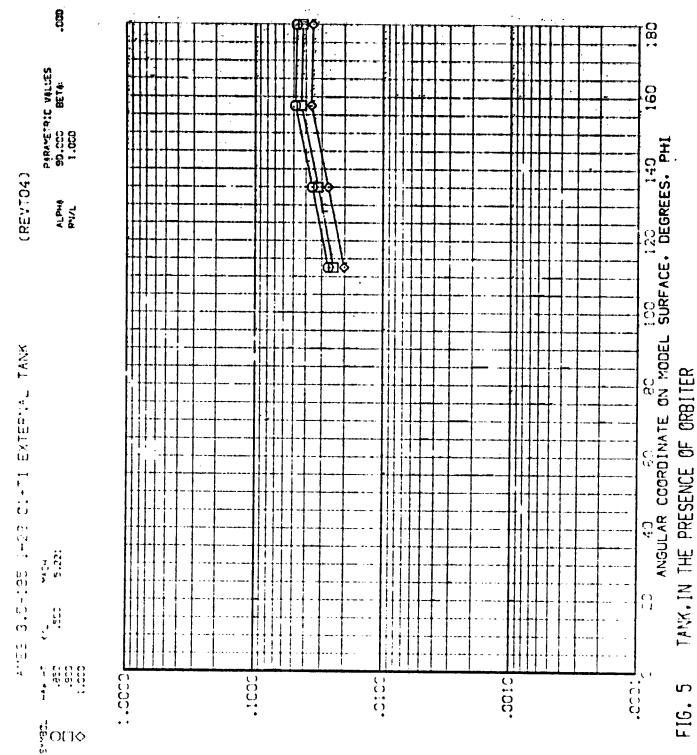
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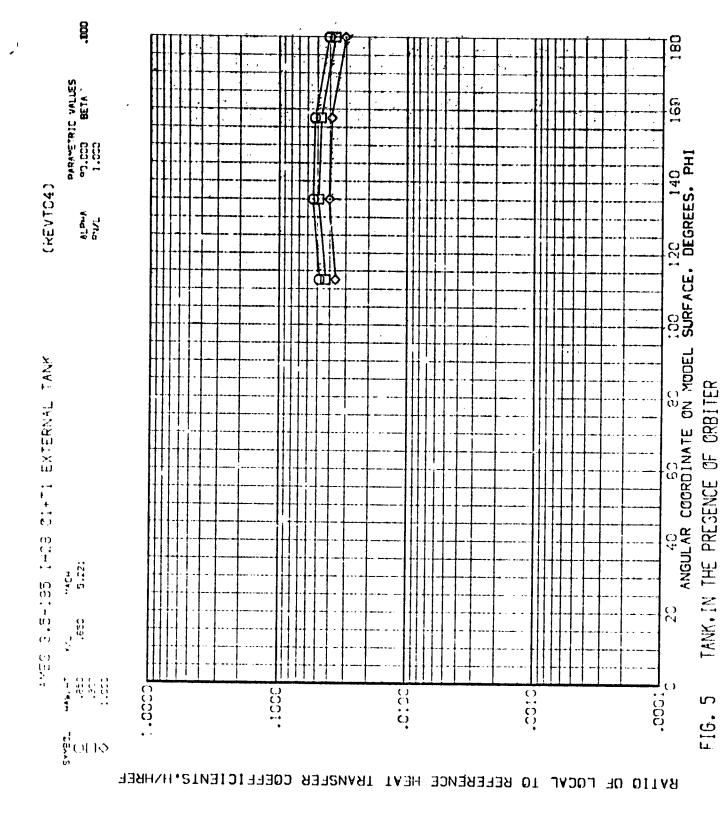
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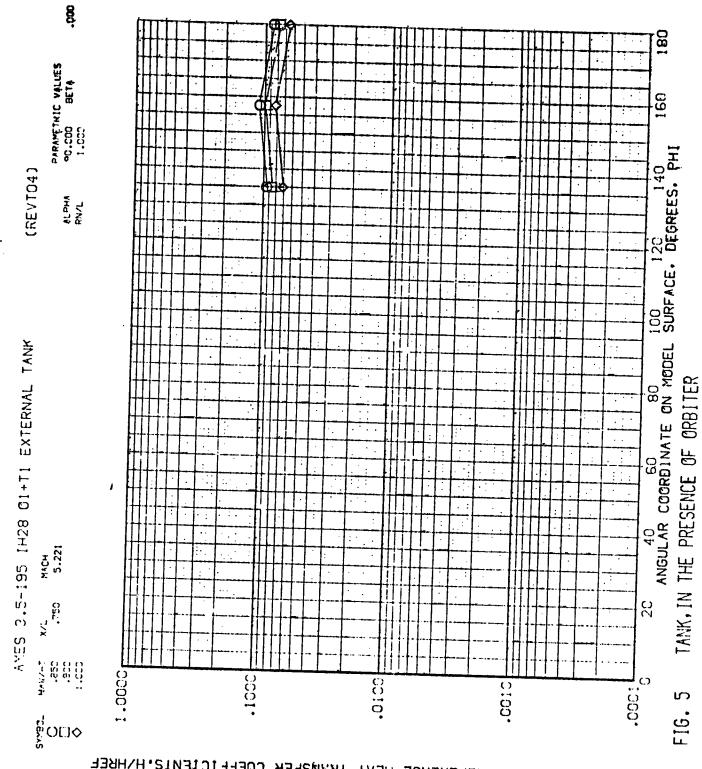
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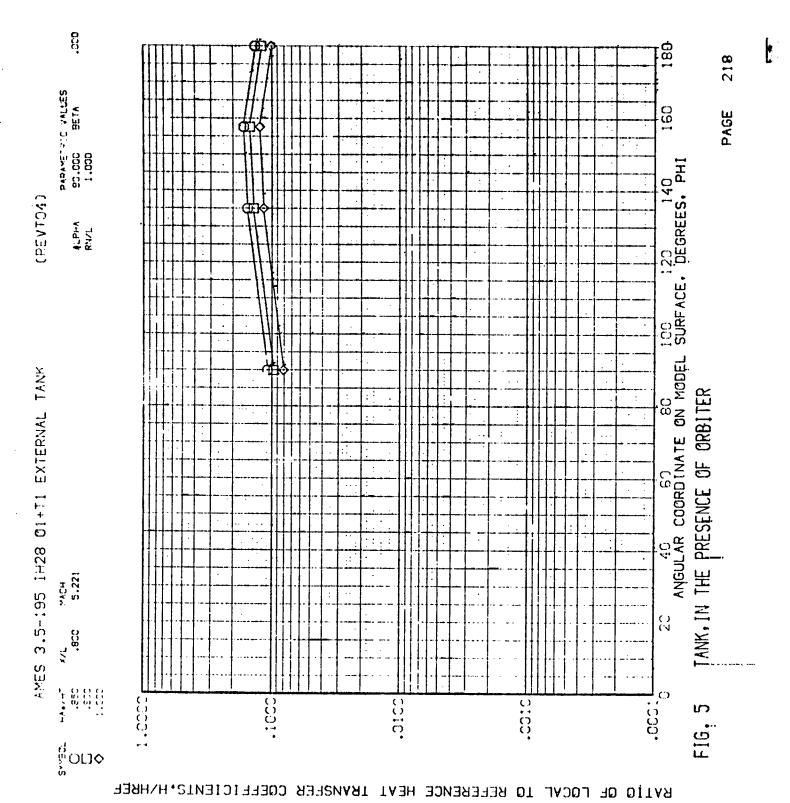
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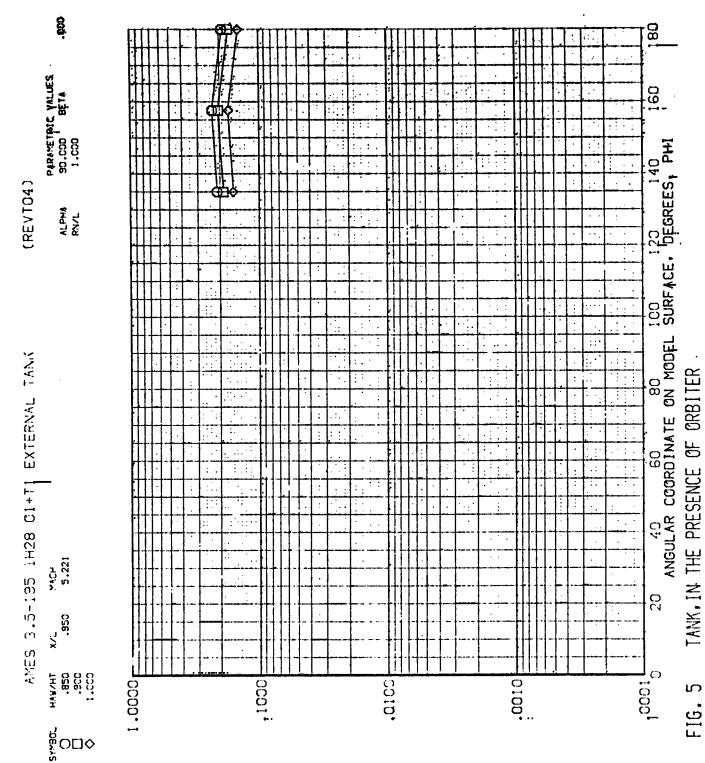




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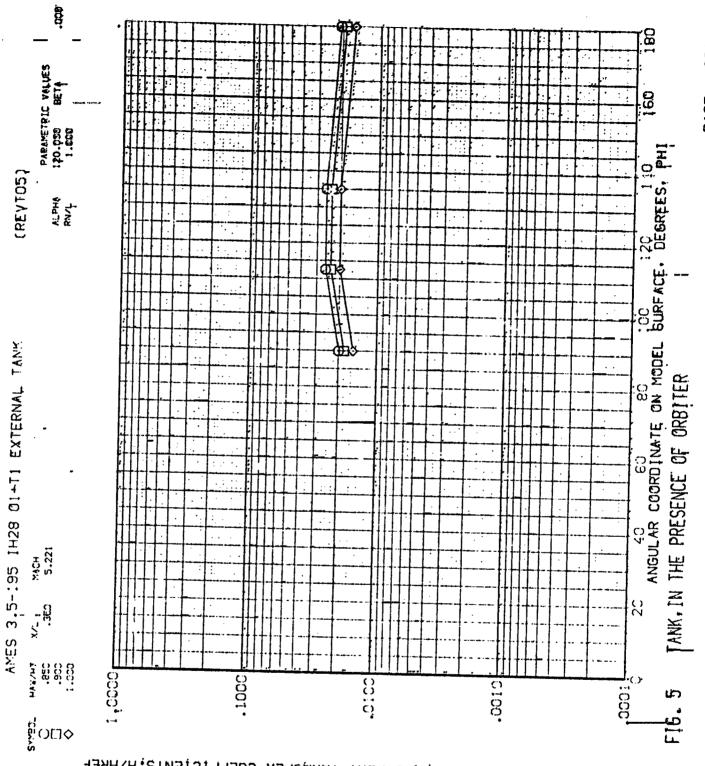




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RATIO OF LOCAL TO REFERENCE HEAT TRANSFER COEFFICIENTS, HZHREF



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8 222 160 ANGUĻĀR COORDĮNATE ON 19DEL SURFACE, DEGREES, PHI (REVT05) AMES 3.5-195 IH28 DI+TI EXTERNAL TANK TANK. IN THE PRECENCE OF ORBITER MACH 5.221 */. .433 0:0: .0010 FIG. 5 § O□♦ RATIO OF LOCAL TO REFERENCE HEAT TRANSFER COEFFICIENTS. HAHREF

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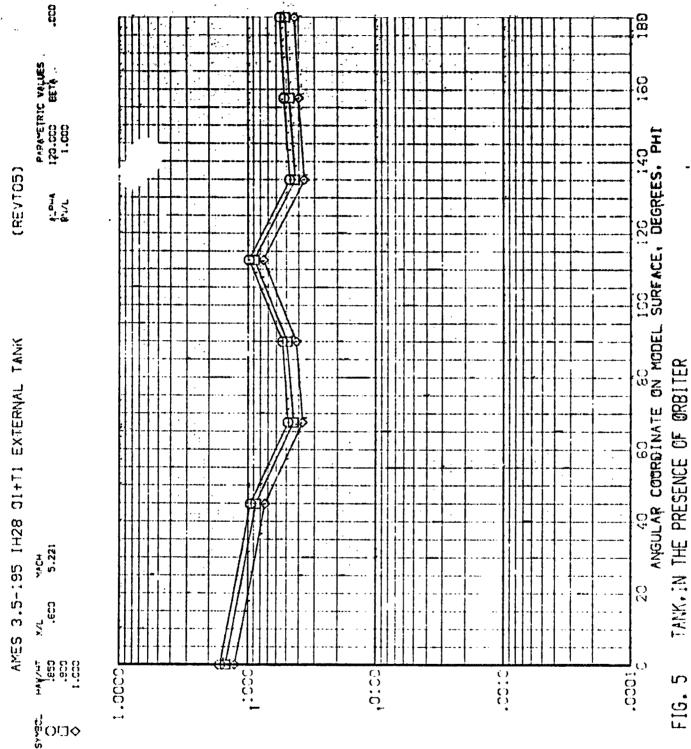
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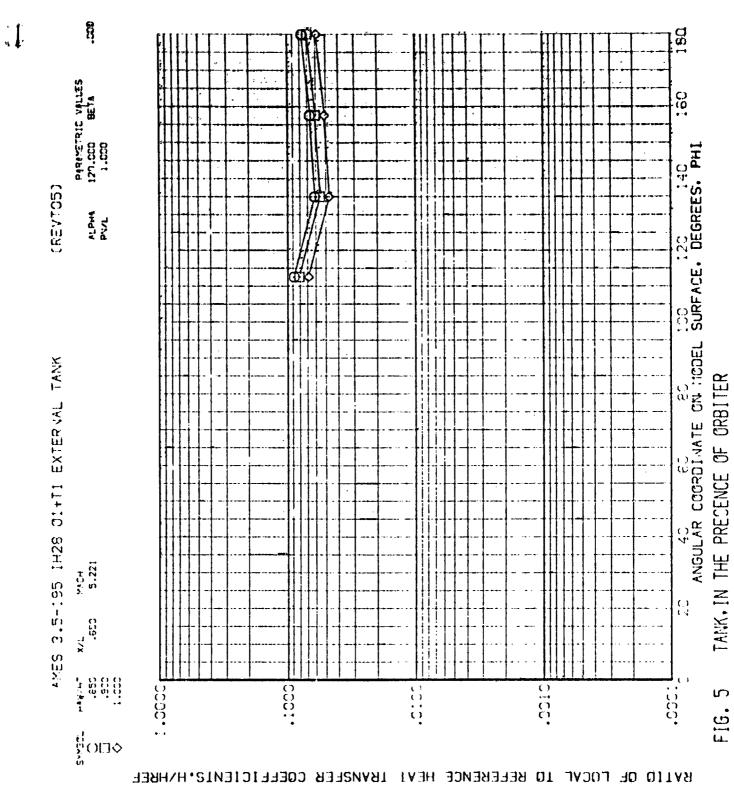


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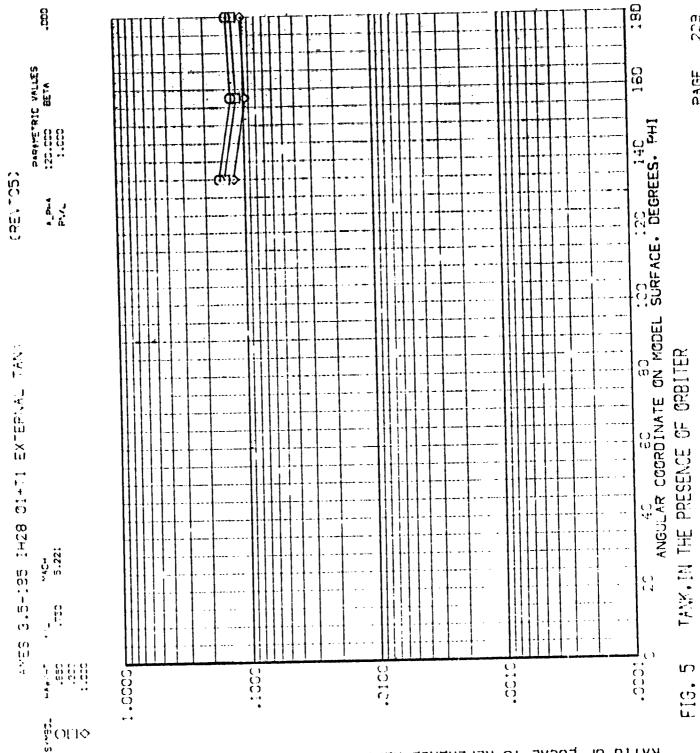
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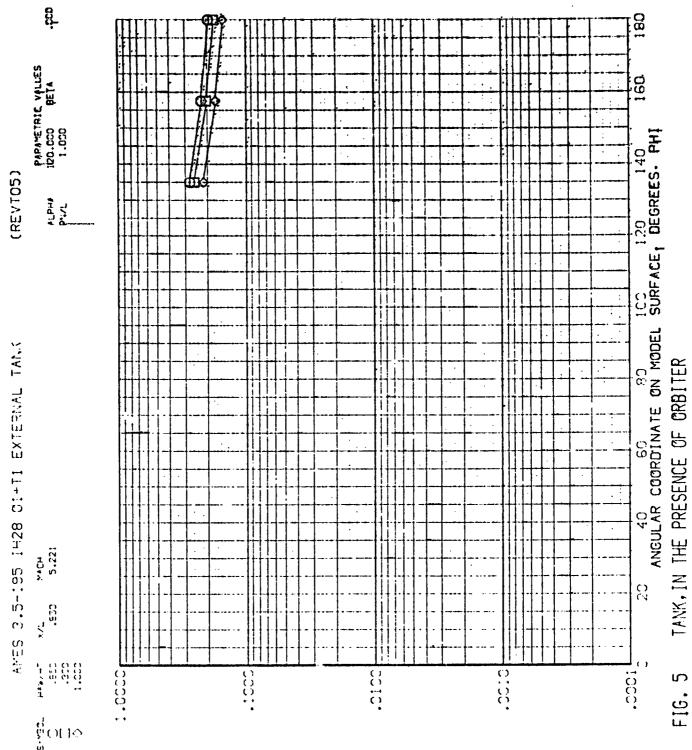
228 160 PAGE 40 60 80 140 ANGULAR COORDINATE ON MODEL SURFACE. DEGREES. PHI (REVT05) A.PHA *YES 3.5-195 IH28 C1+T1 EXTERNAL TANK TANK, IN THE PRESENCE OF GRBITER 44CH 5.221 ڈر 73 FIG. 5 \$COD BYITO DE FOCYF TO BEFEBENCE HEYT IBANGEER COEFFICIENTS, HZHREF

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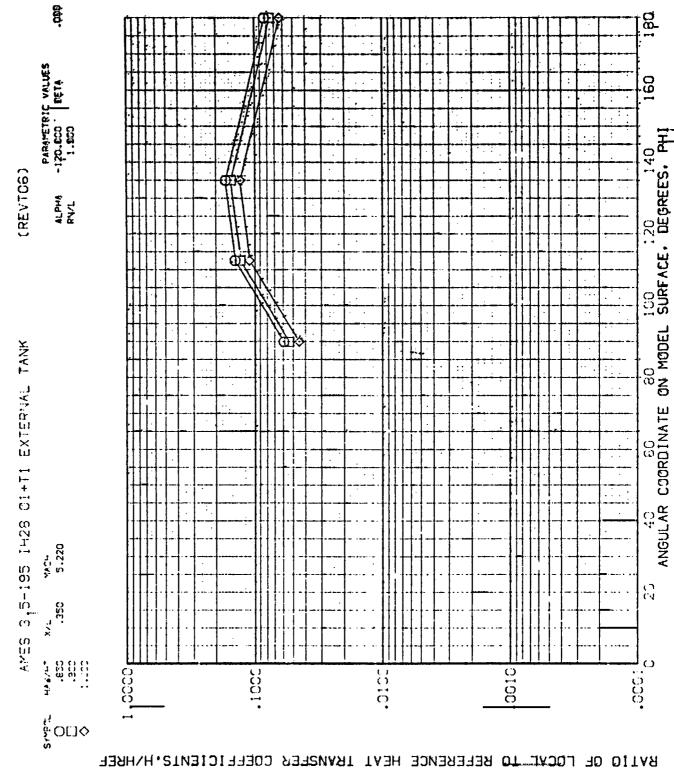
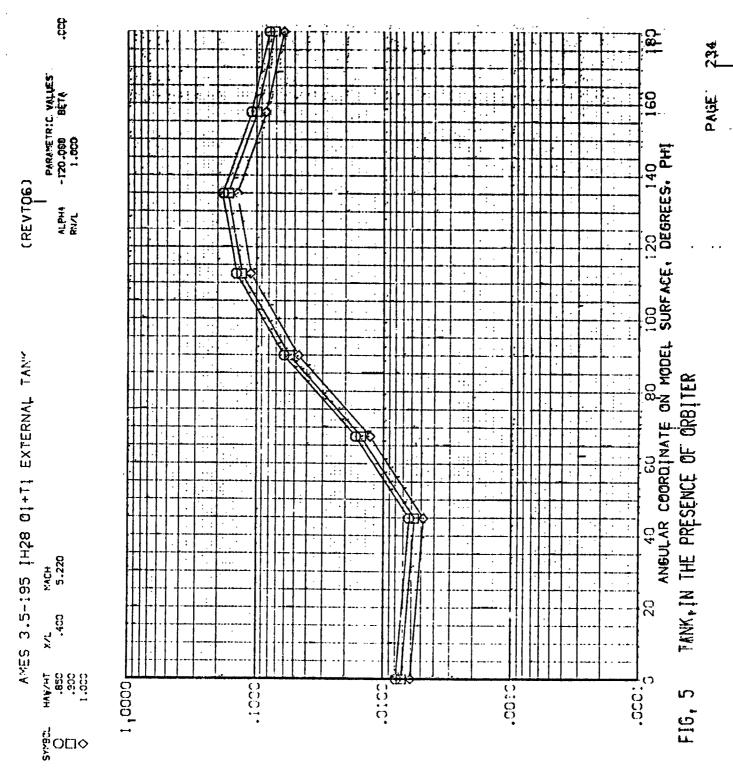
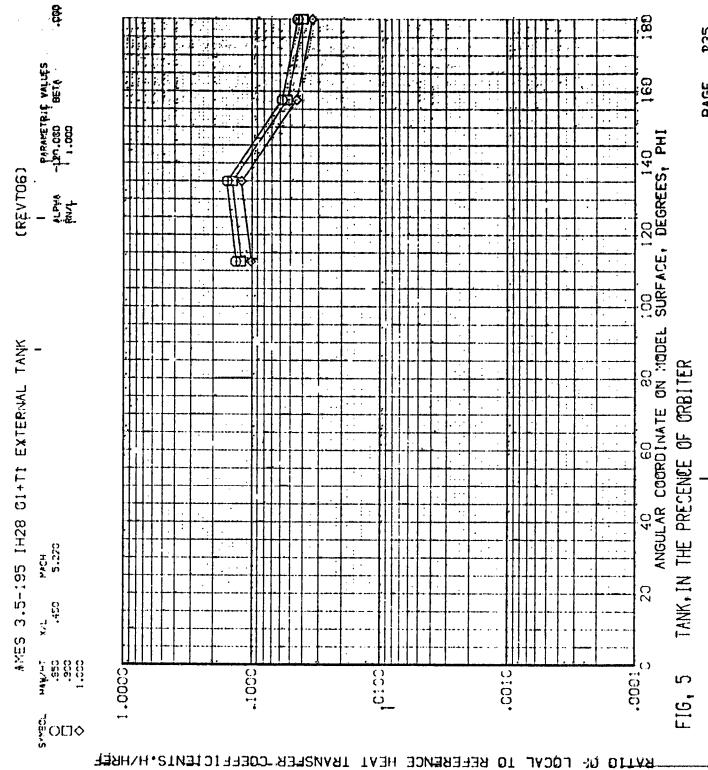
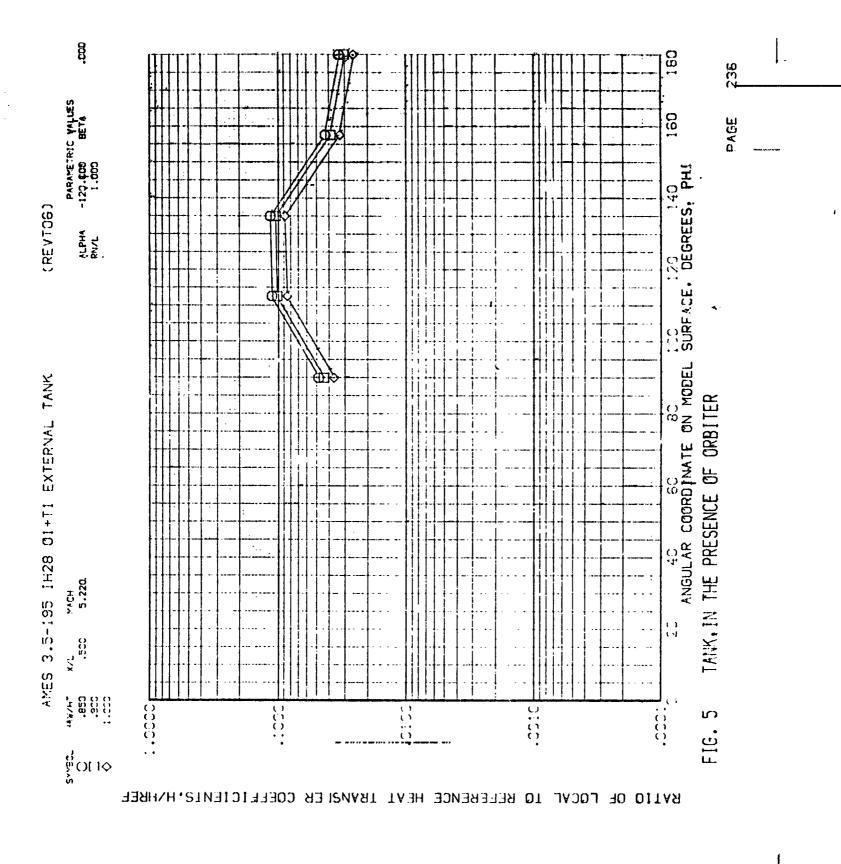


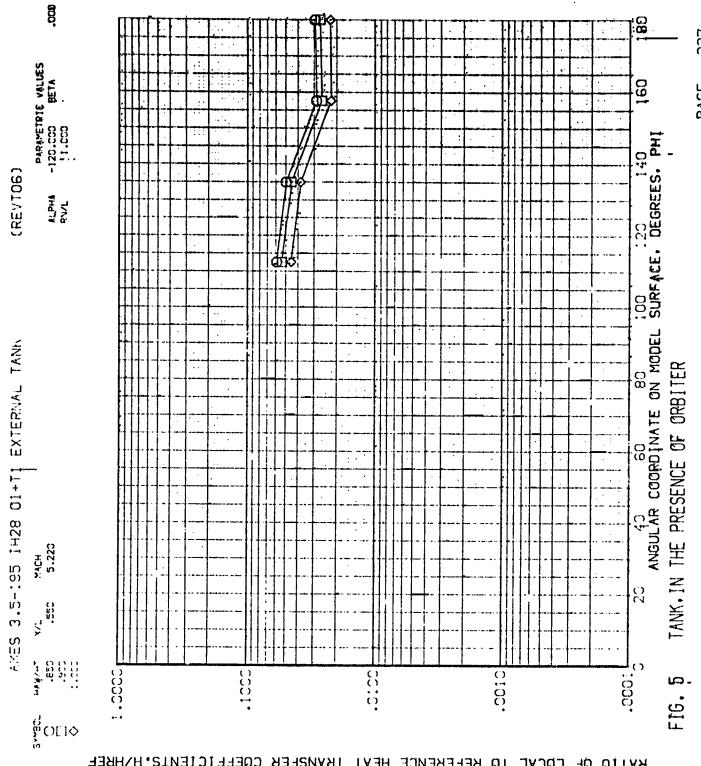
FIG. S TANK, IN THE PRESENCE OF ORBITER



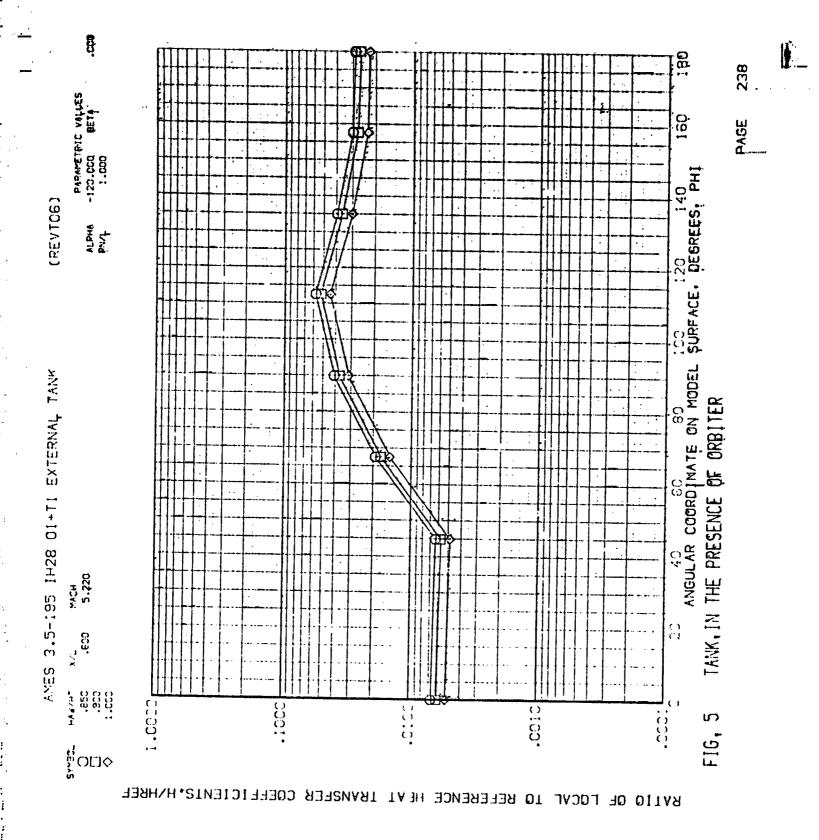
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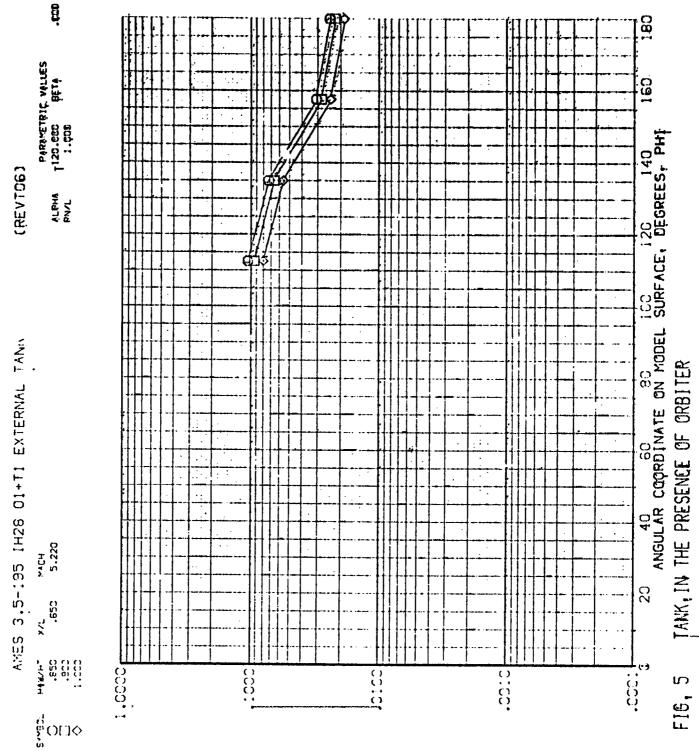




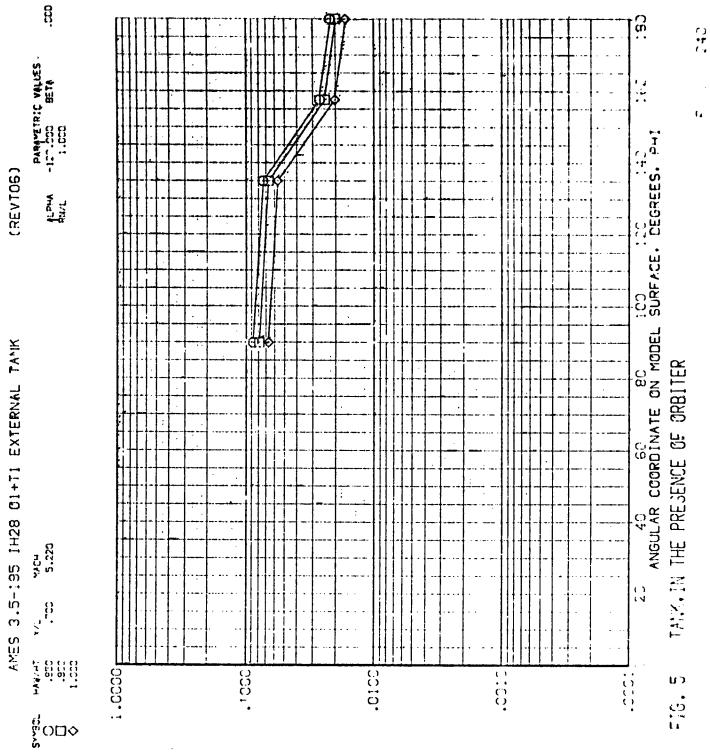


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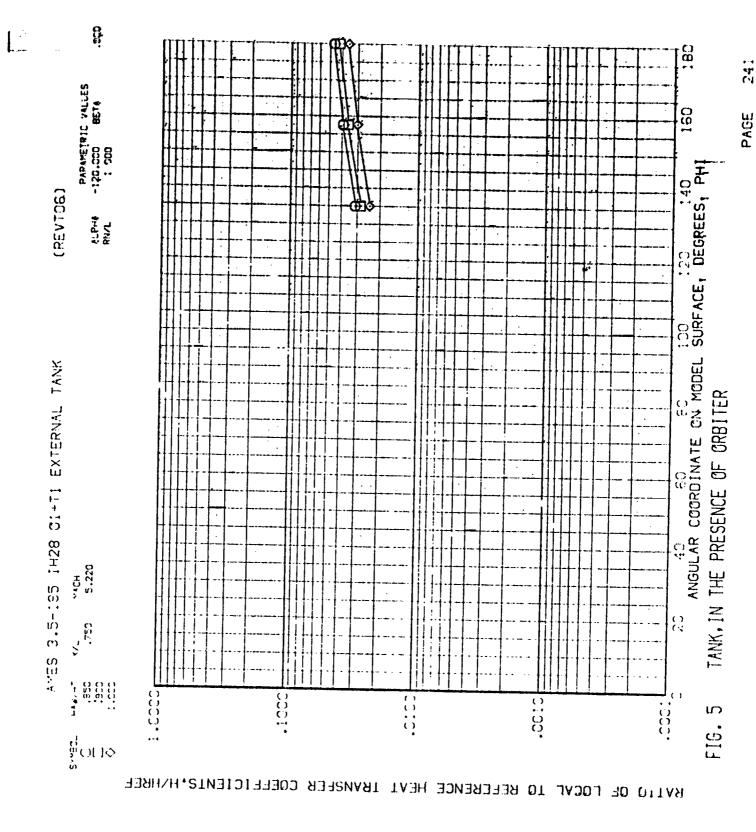




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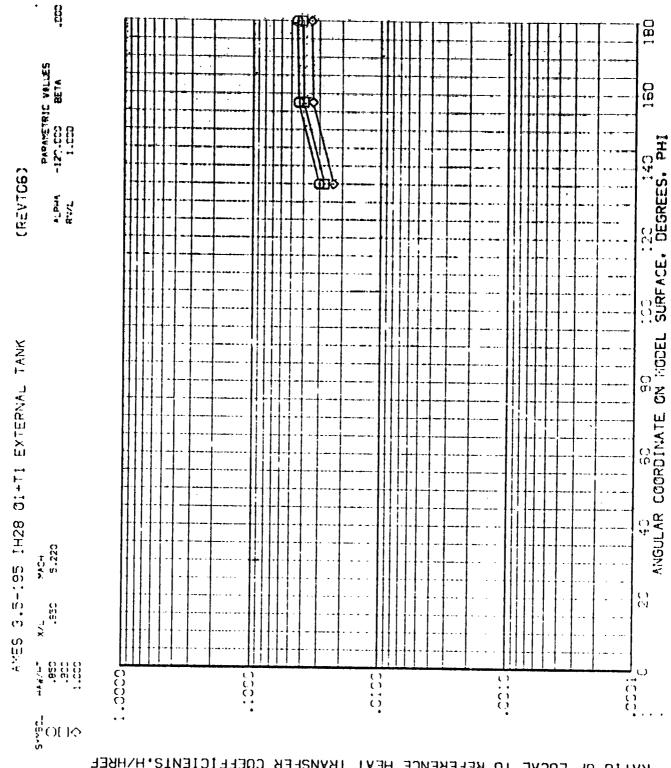
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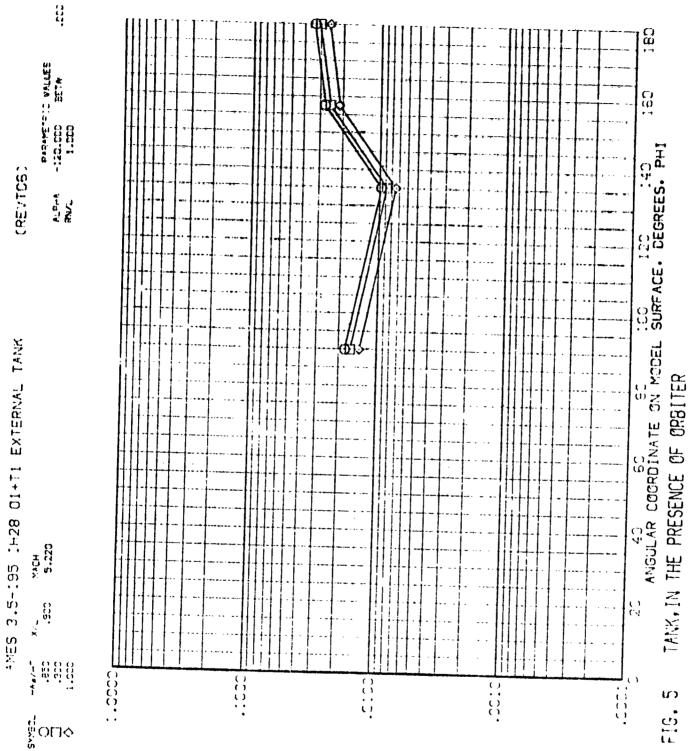
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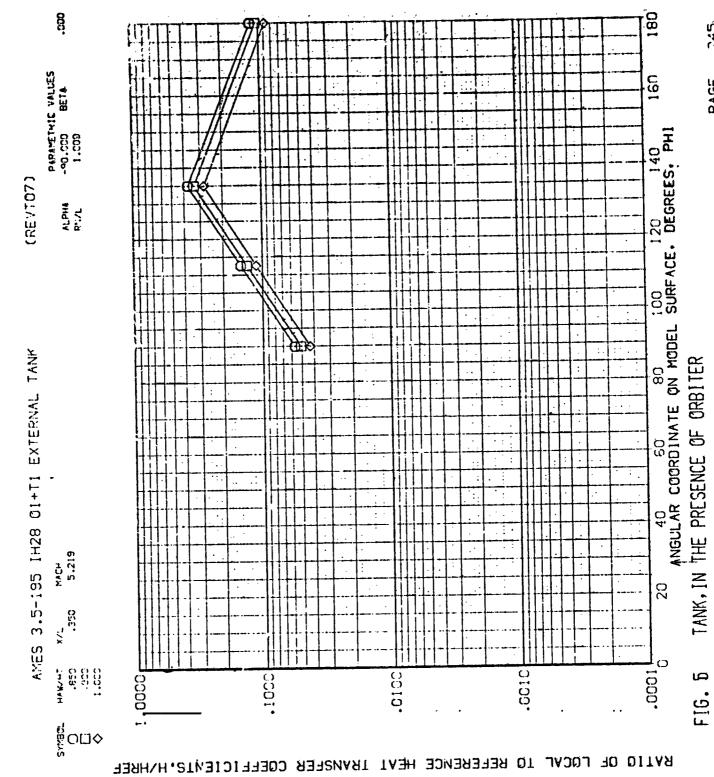
FIG. 5



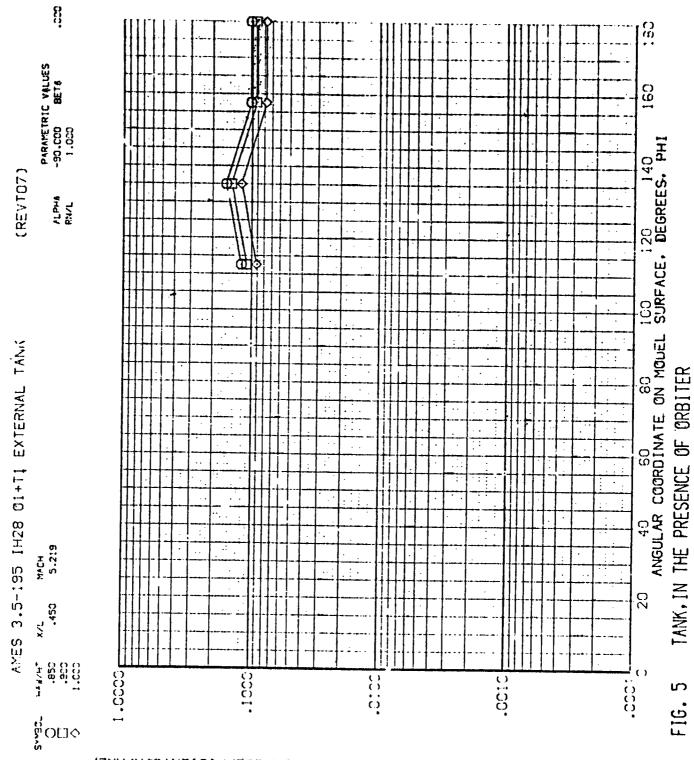
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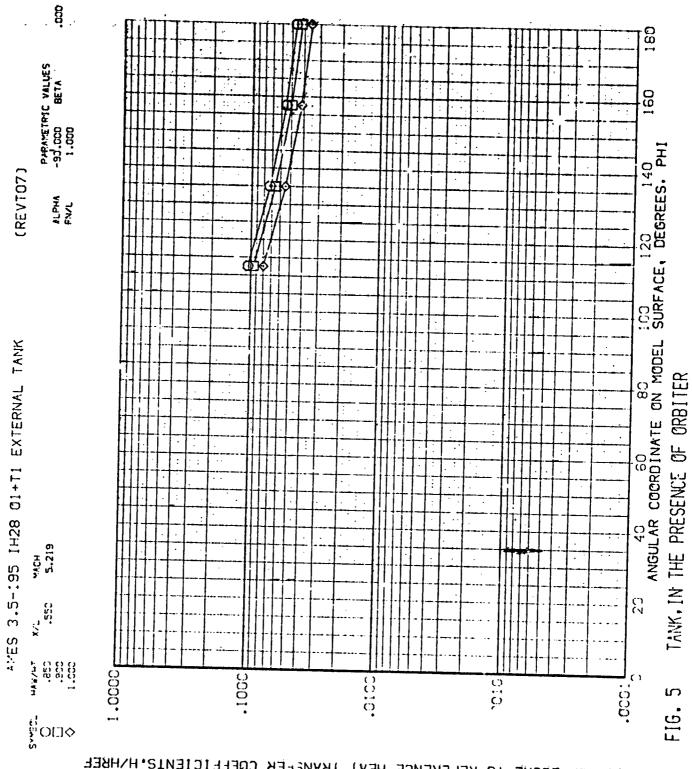


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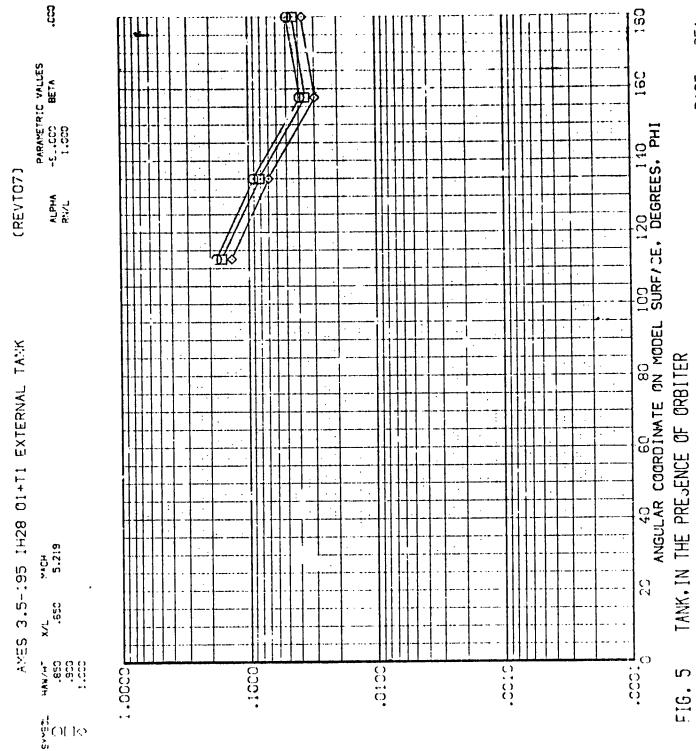
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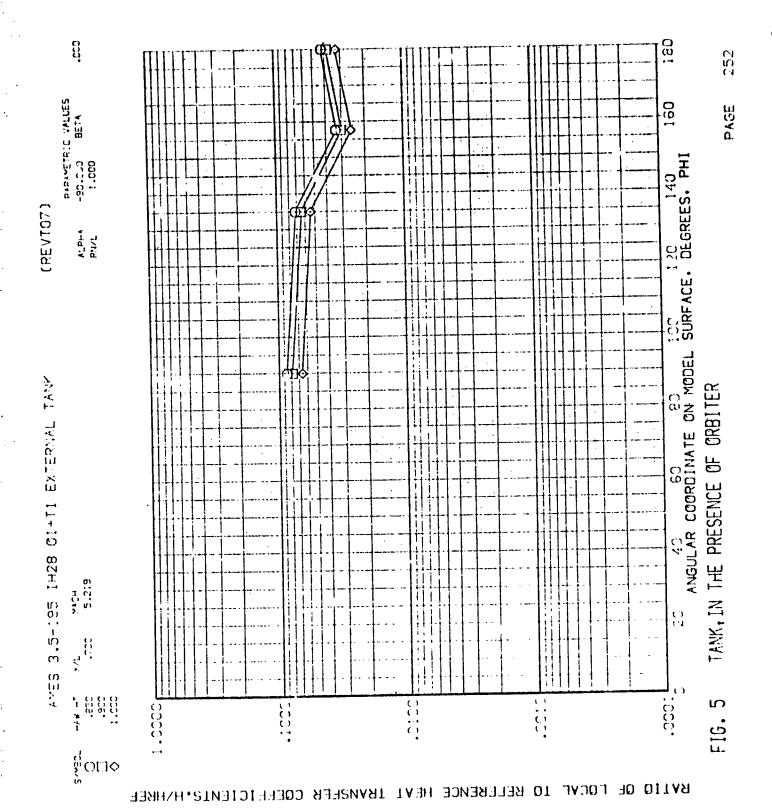
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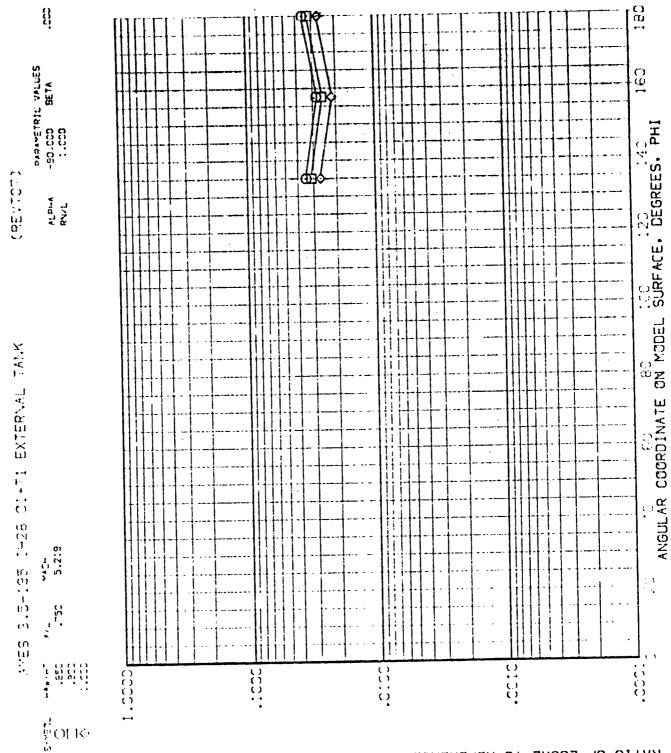


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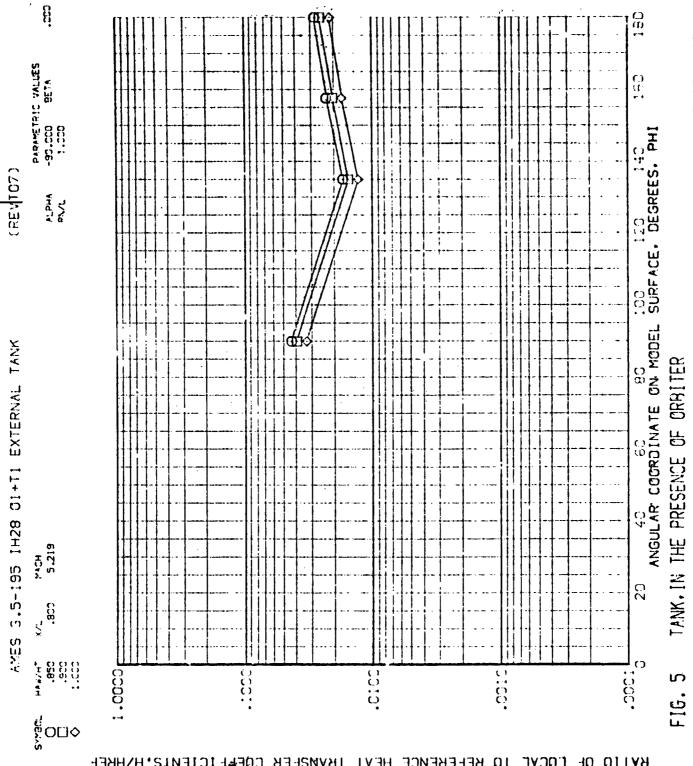


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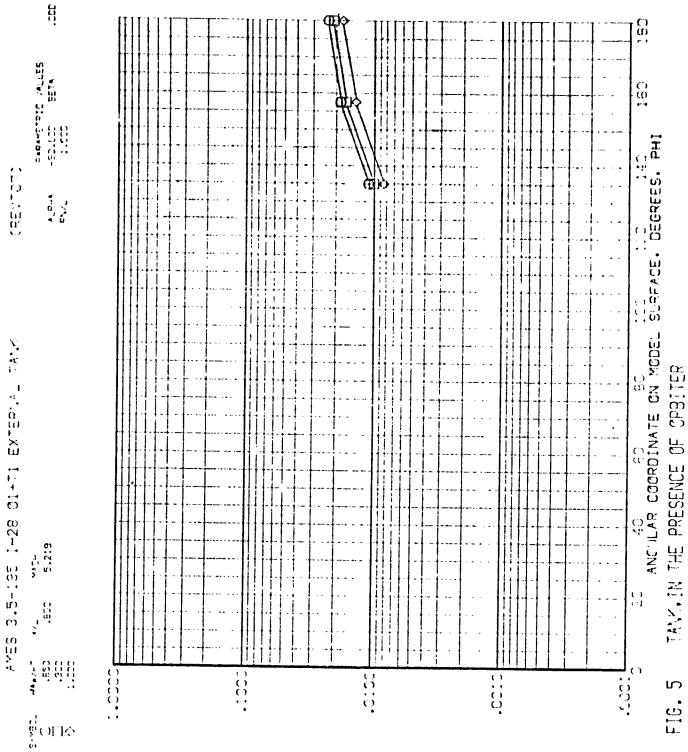
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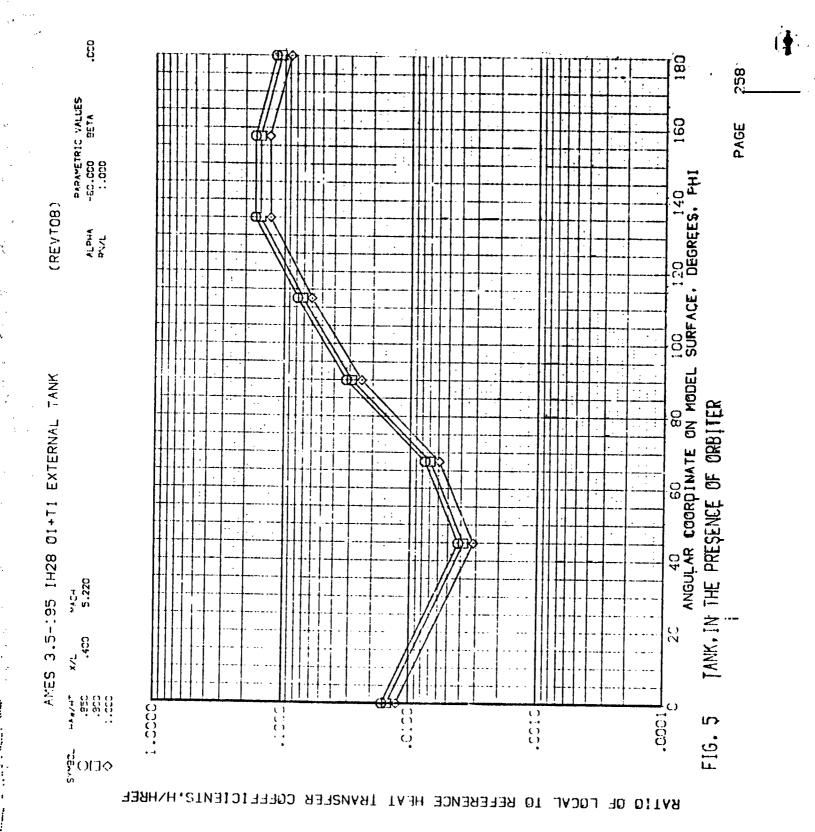
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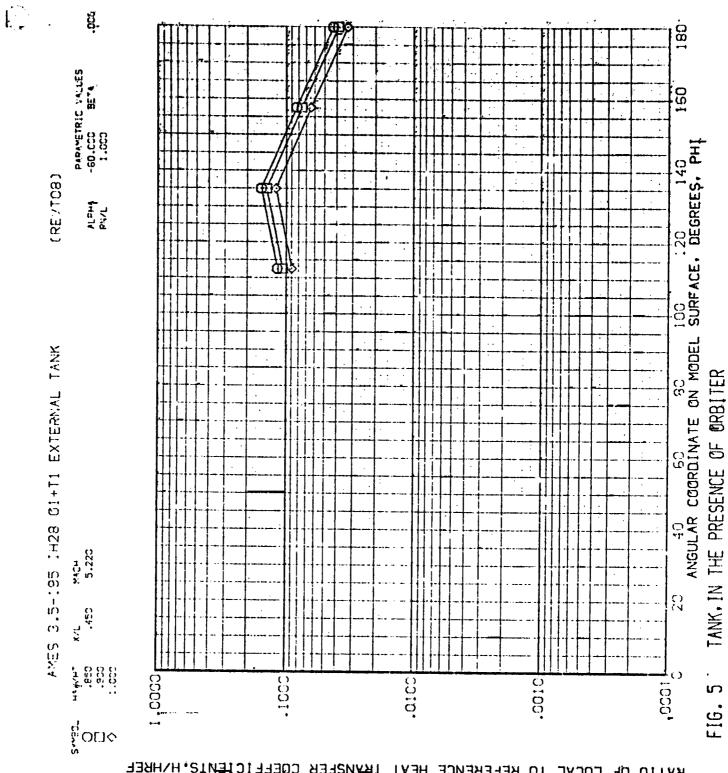
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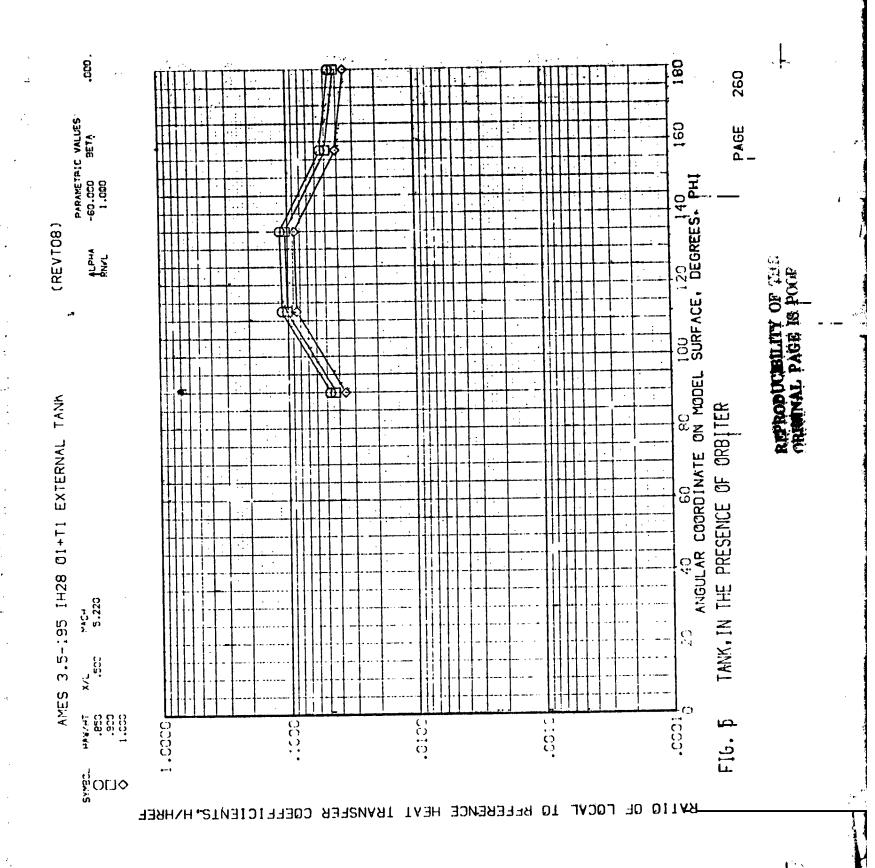
FIG. 5

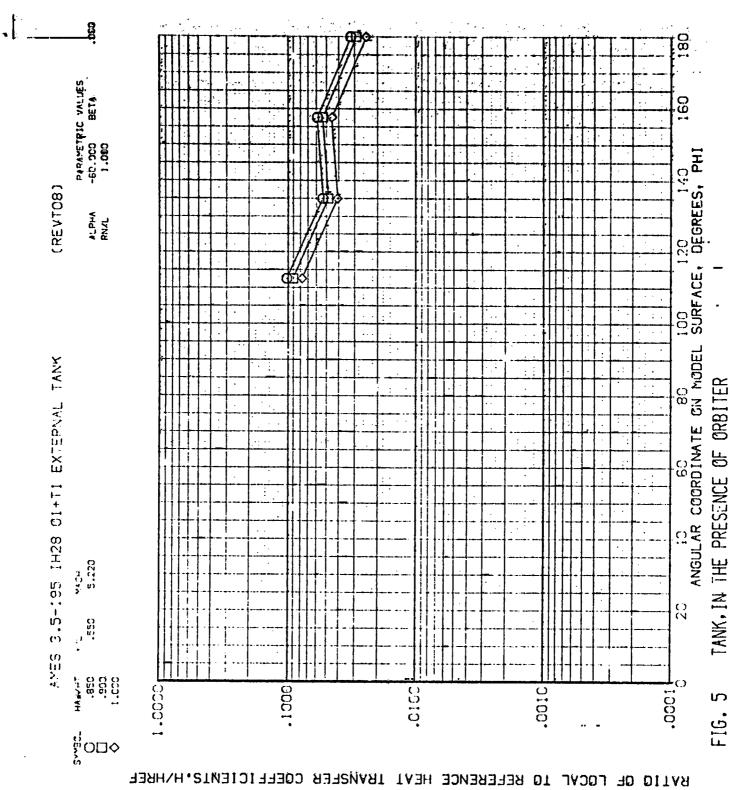
FIG. 5 TANK, IN THE PRESENCE OF CRBITER

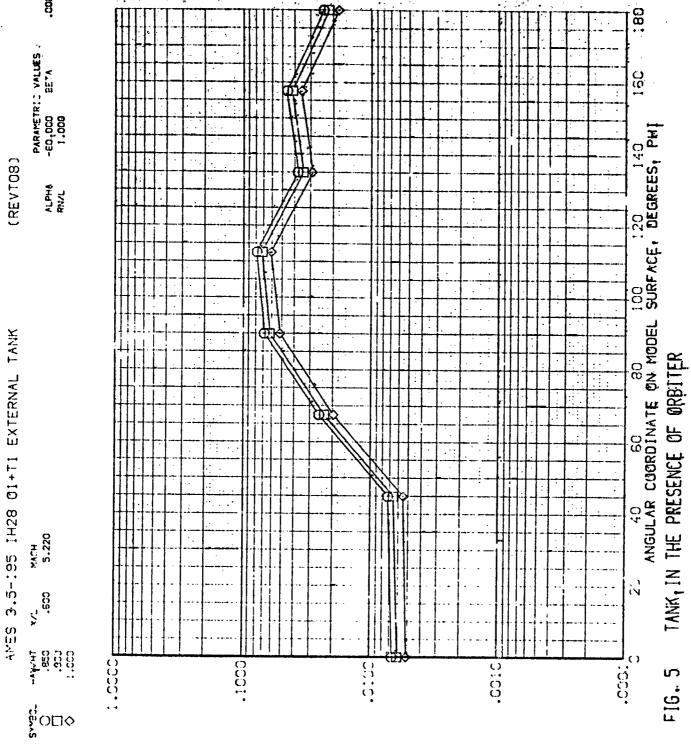




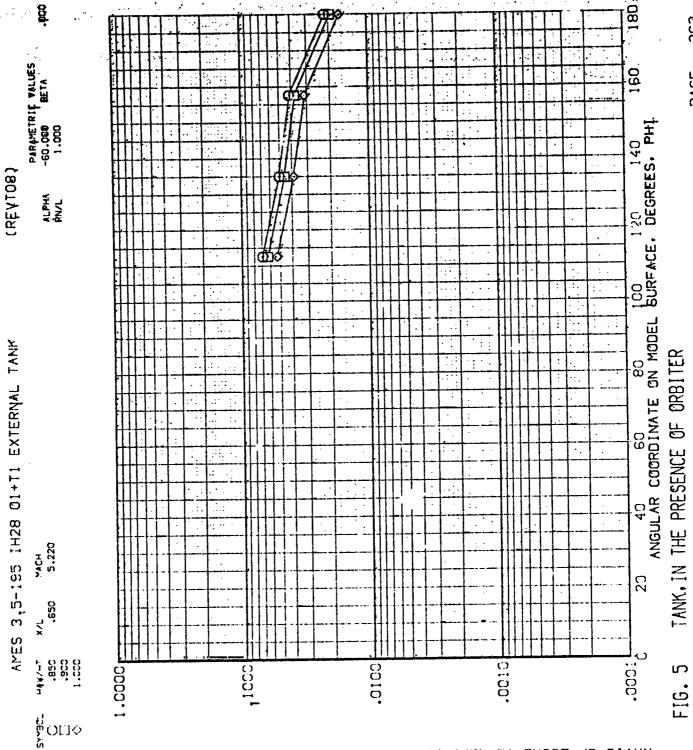
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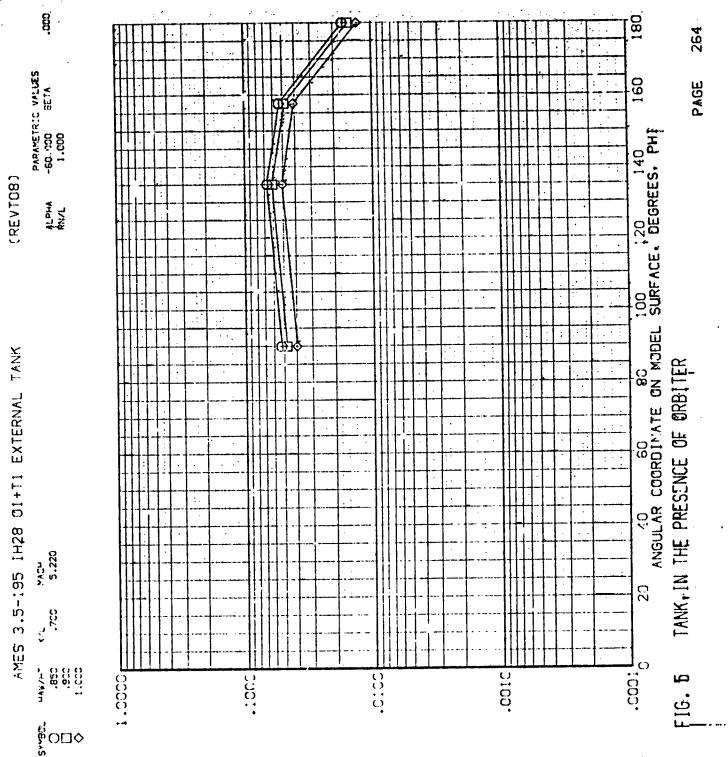




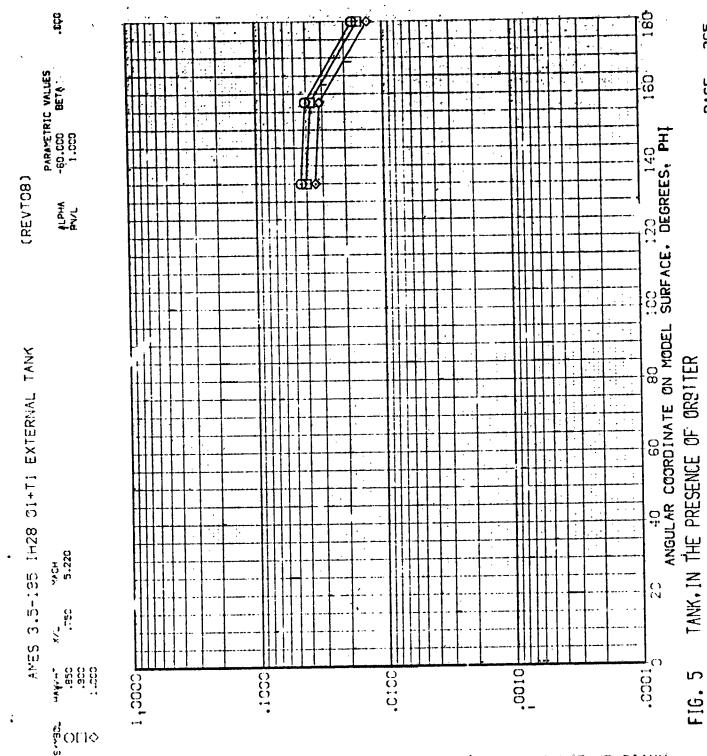
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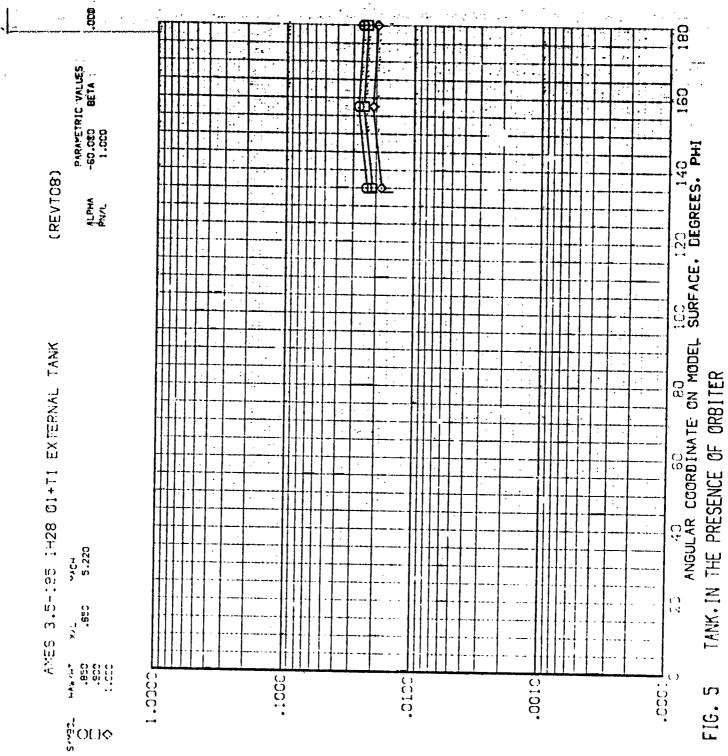


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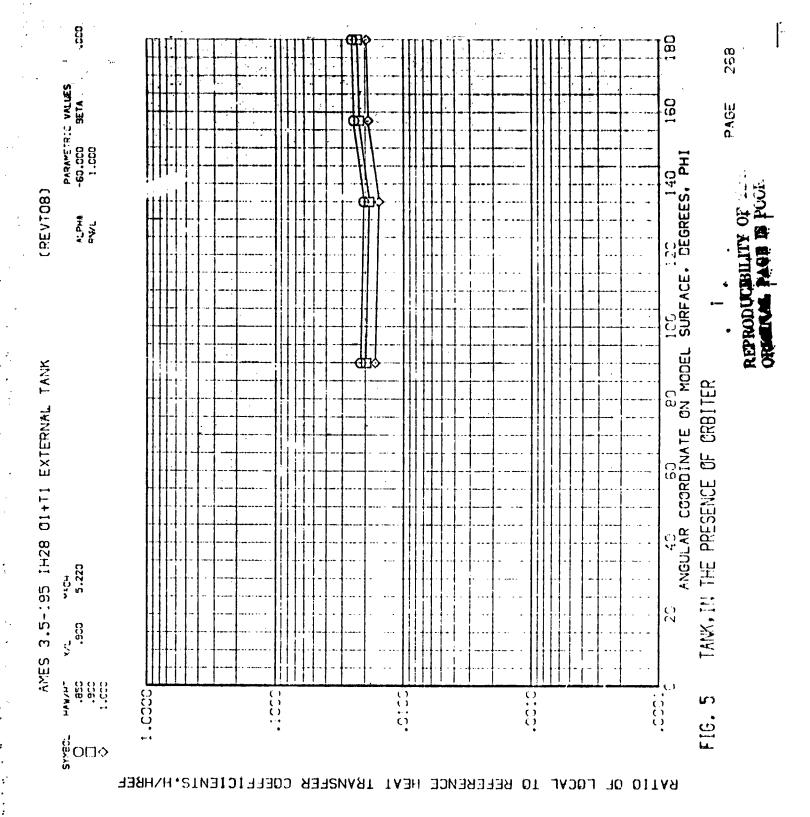


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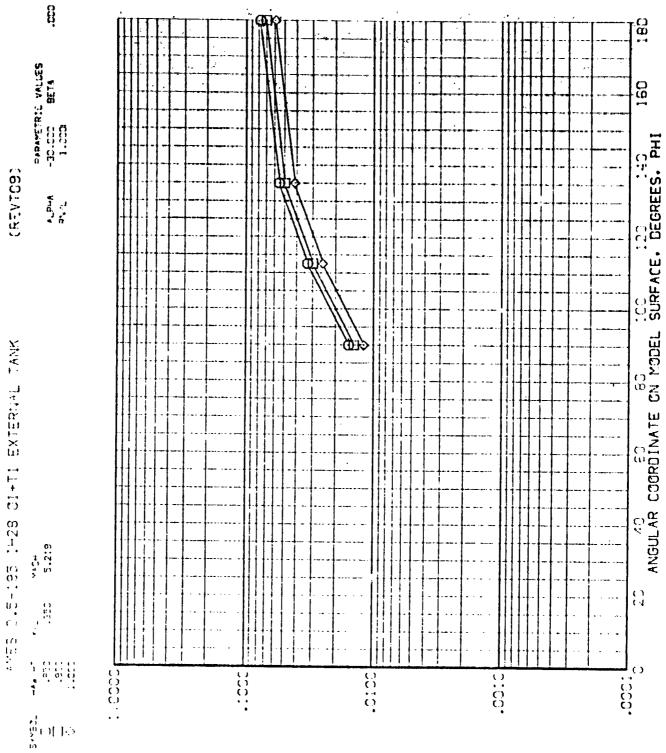


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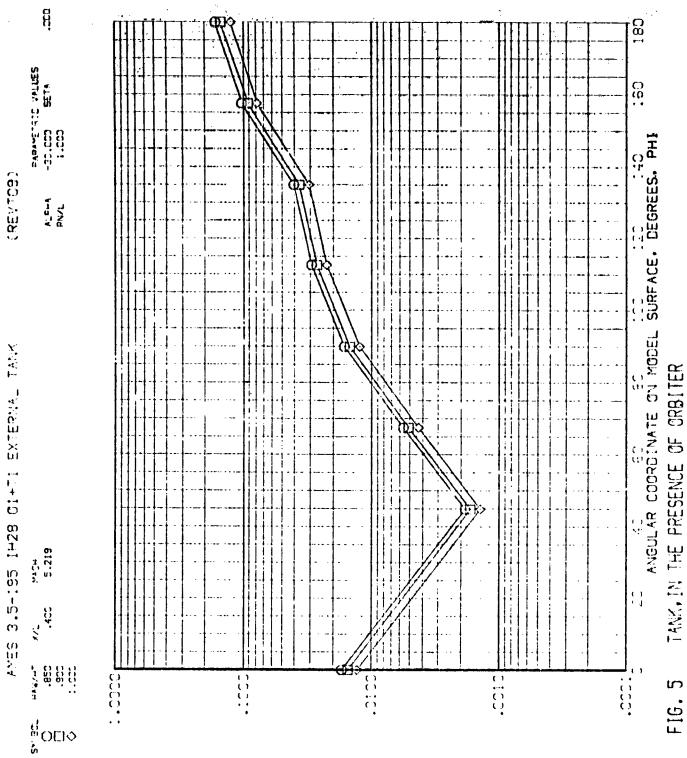
TANK, IN THE PRESENCE OF GRBITER

FIG. 5



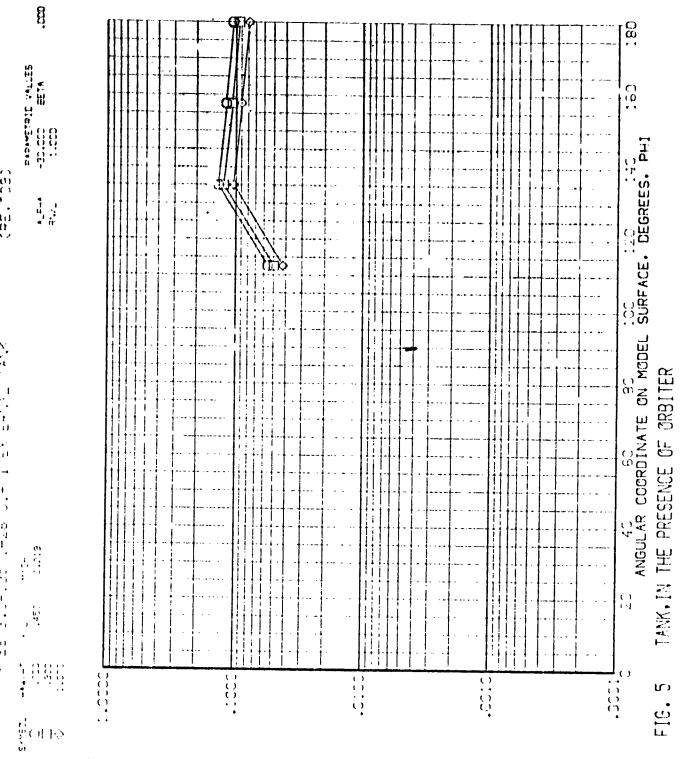
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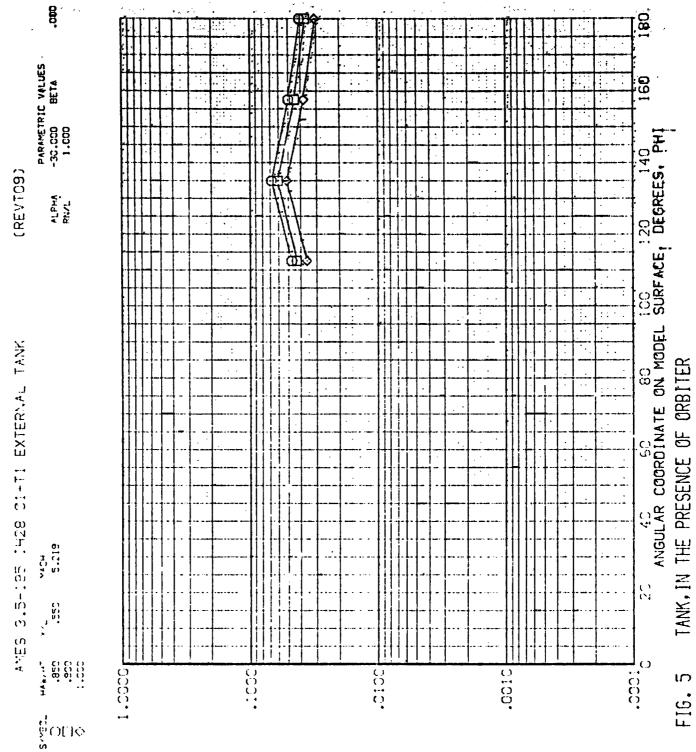


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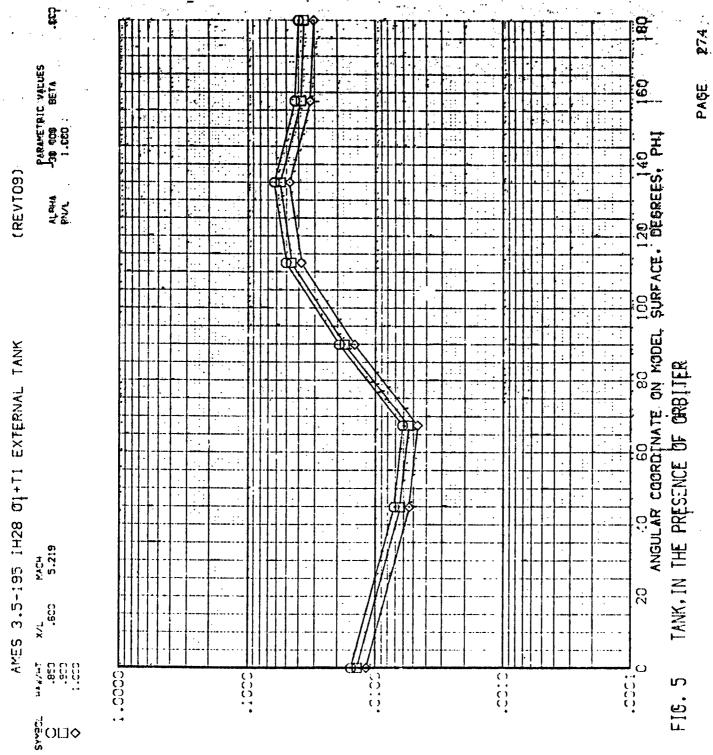
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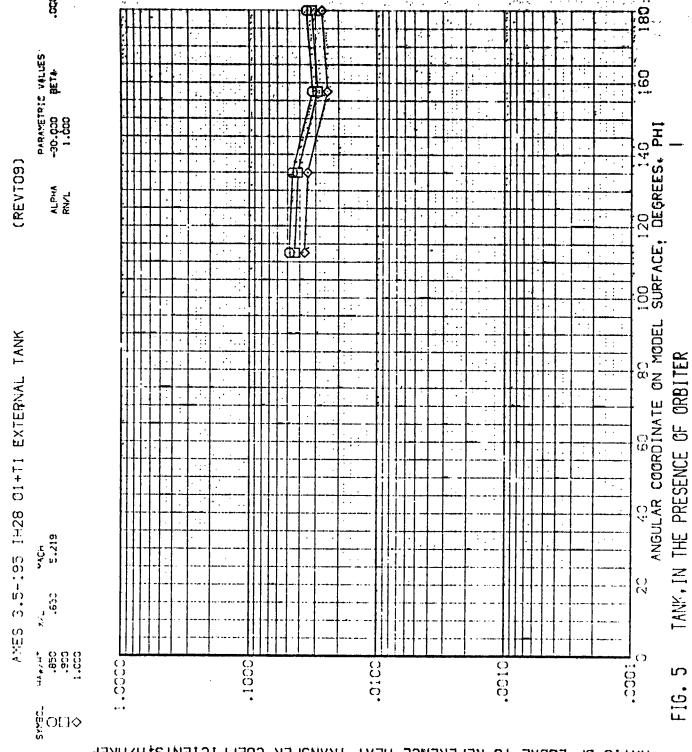
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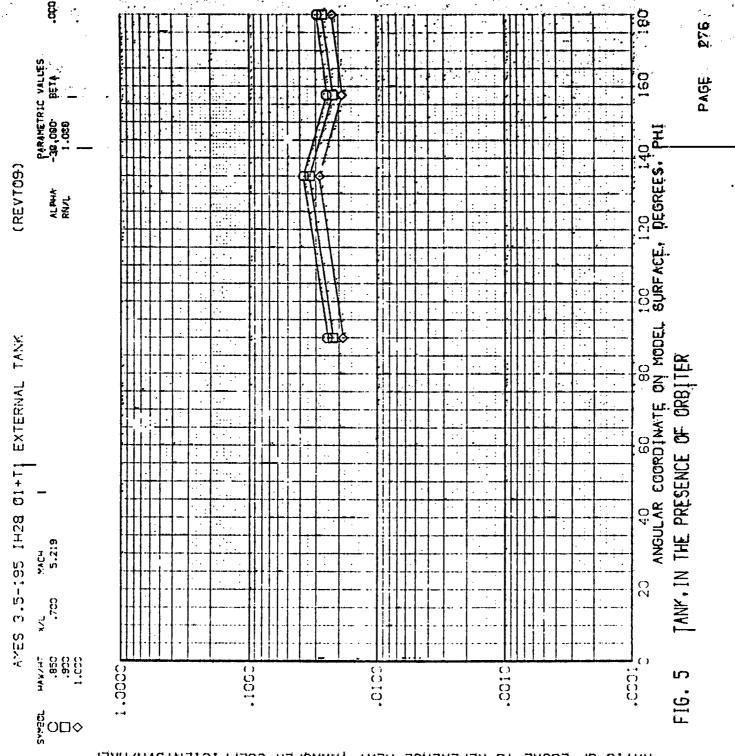
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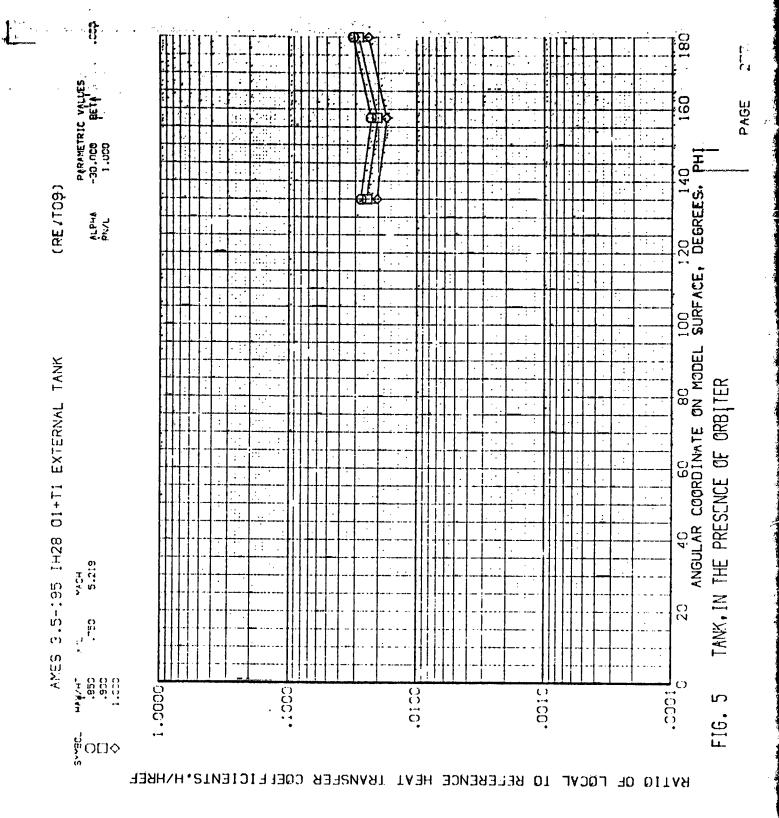
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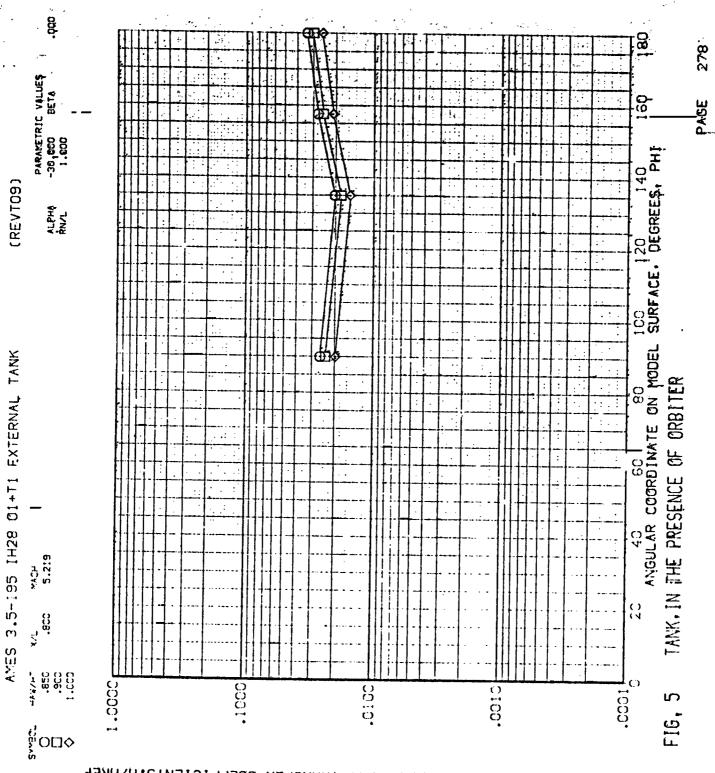


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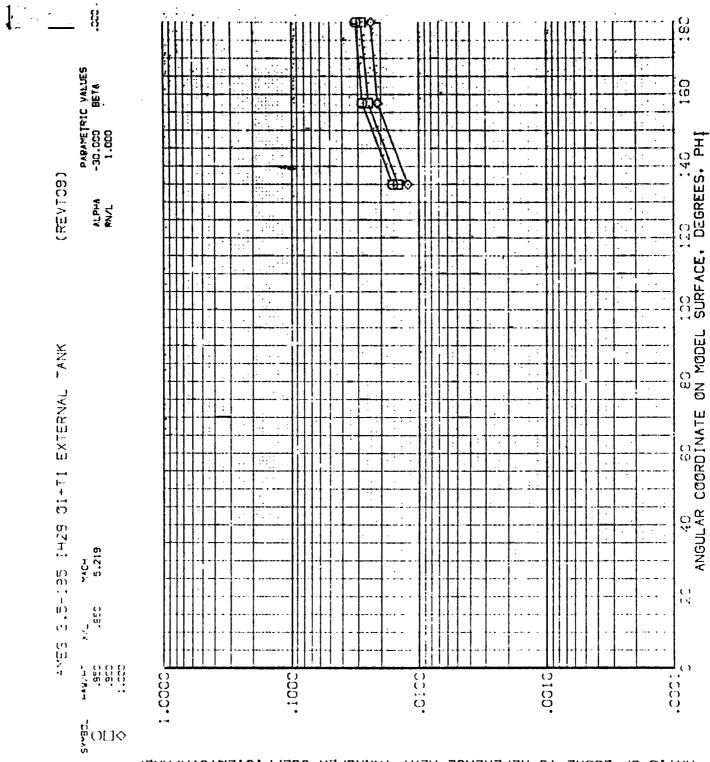




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TANK, IN THE PRESENCE OF ORBITER

FIG. 5



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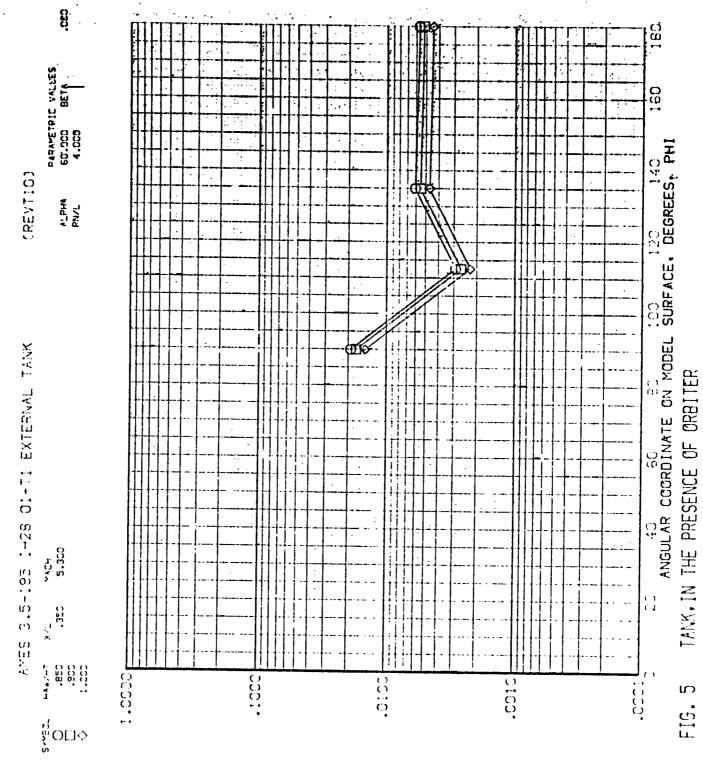
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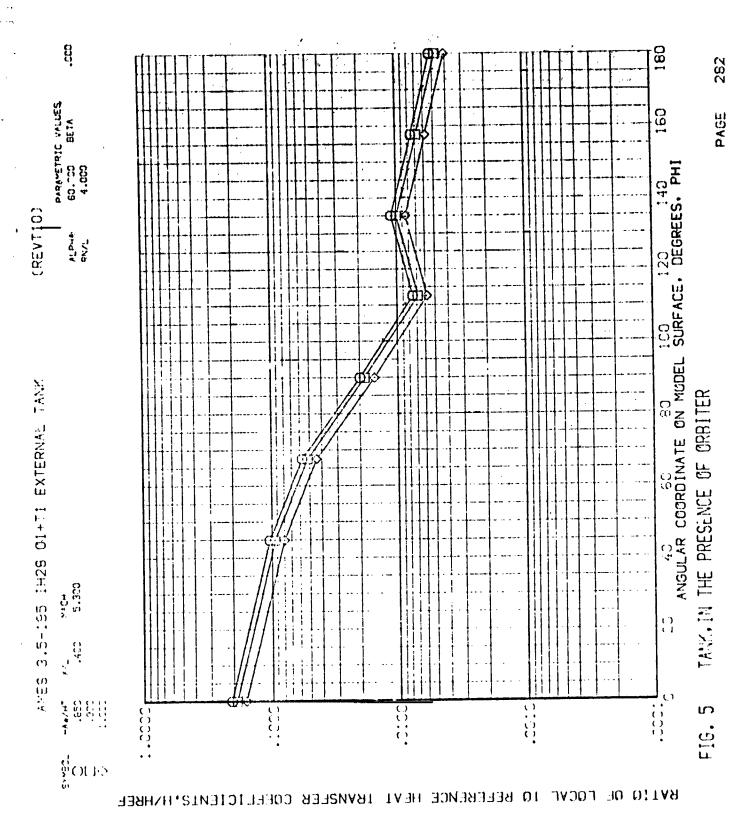
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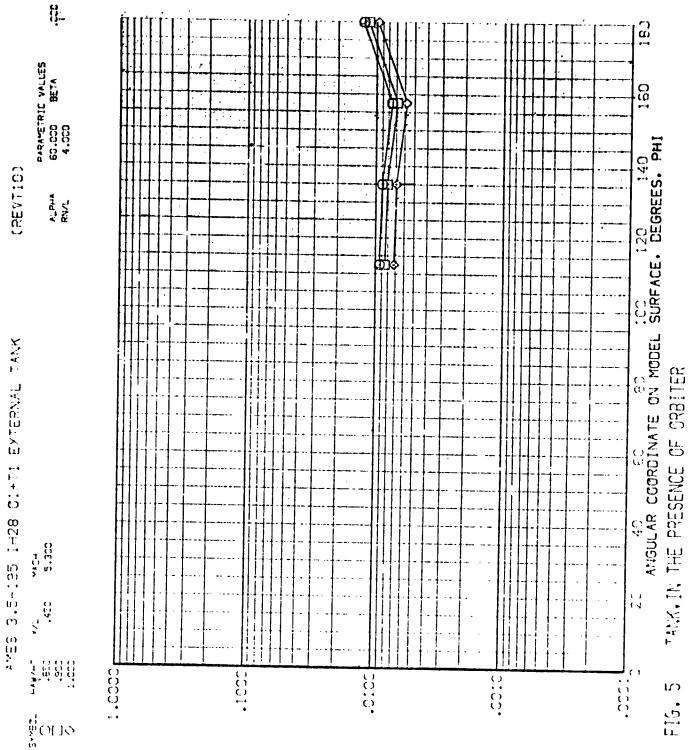
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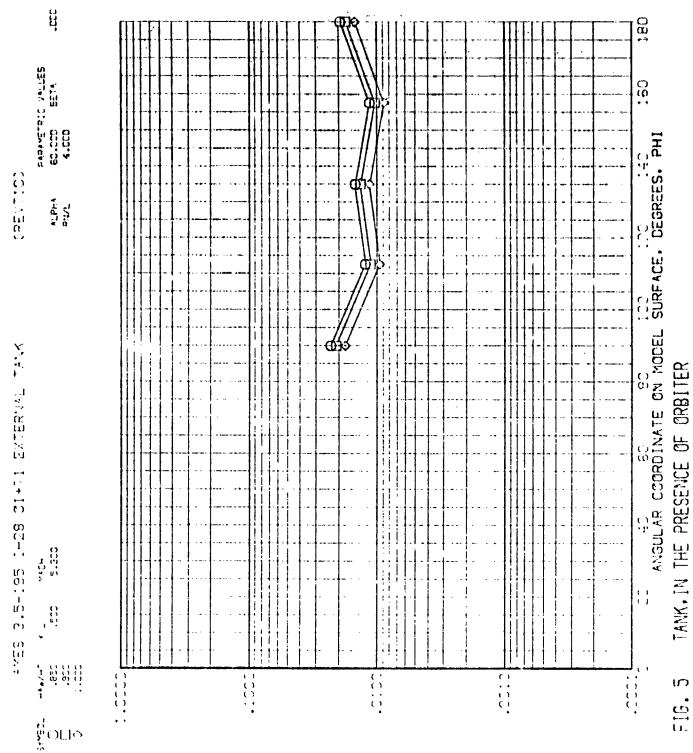




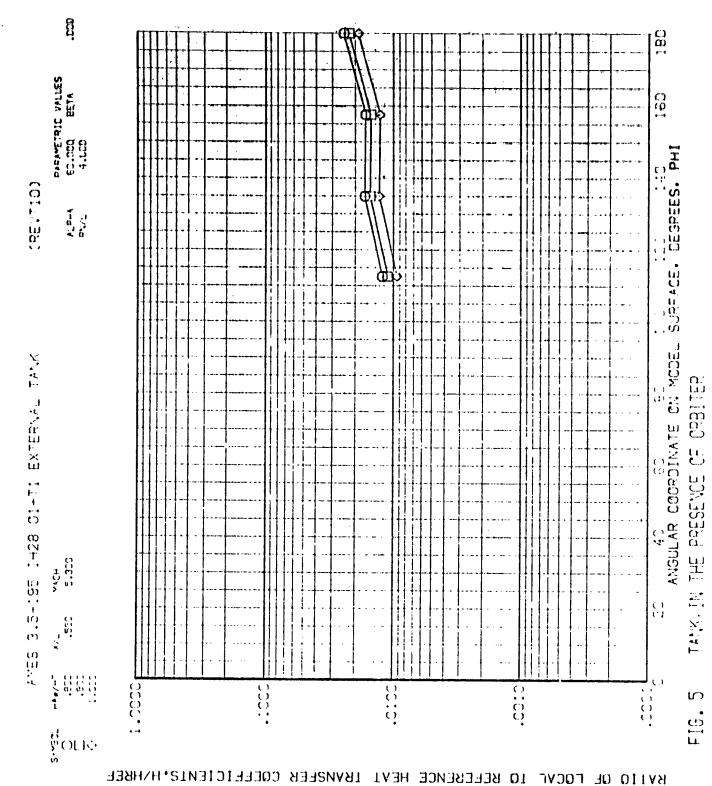
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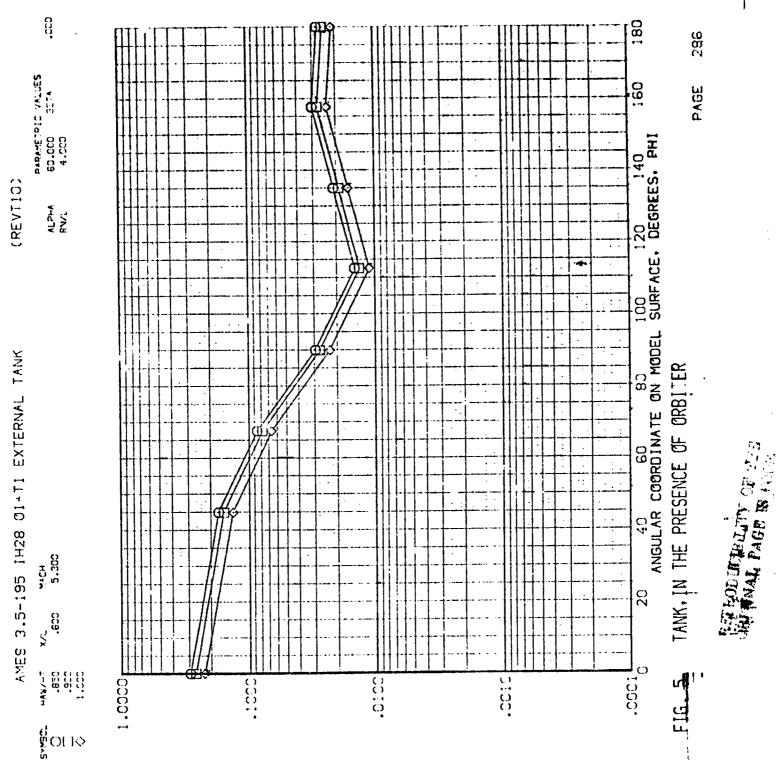
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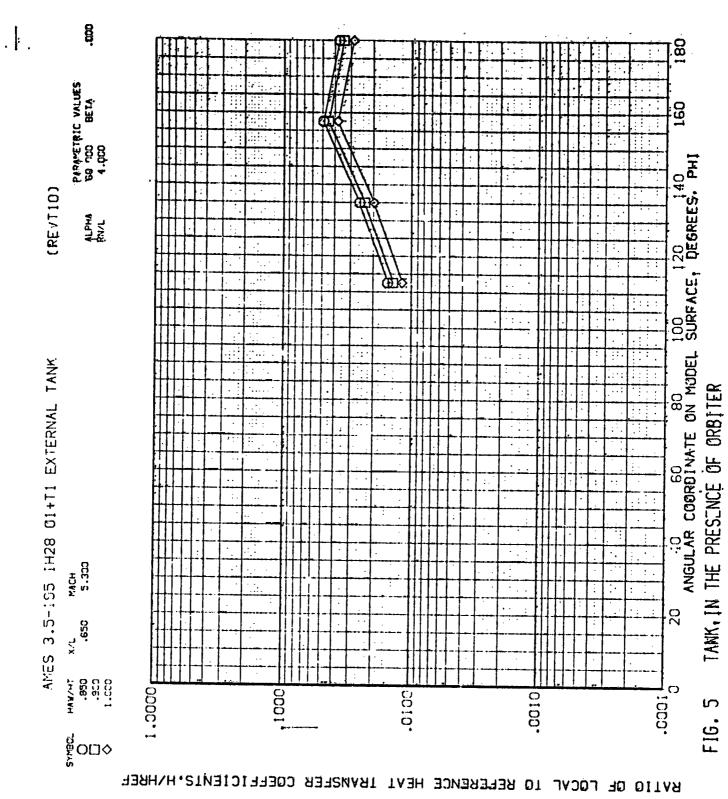
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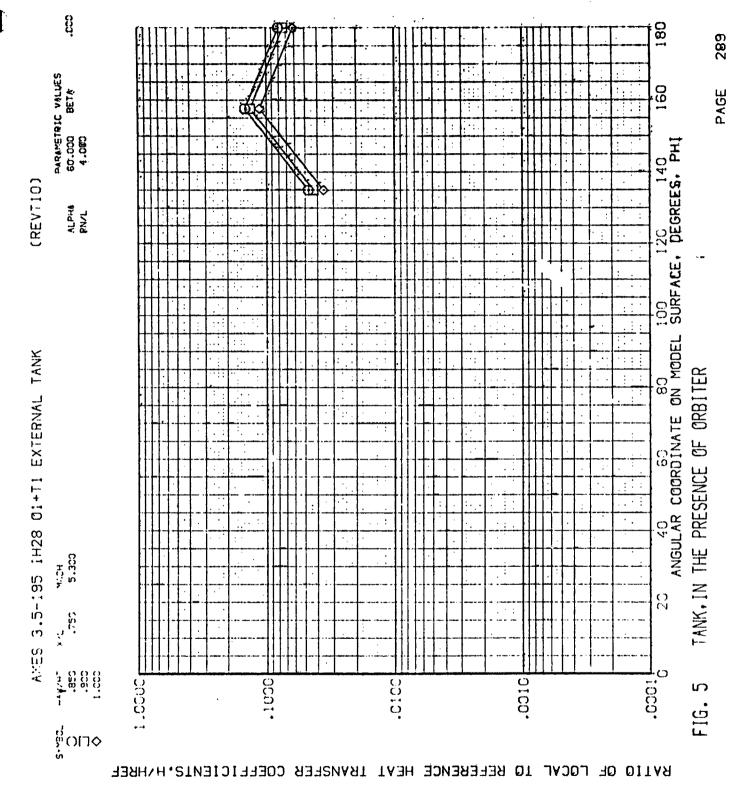
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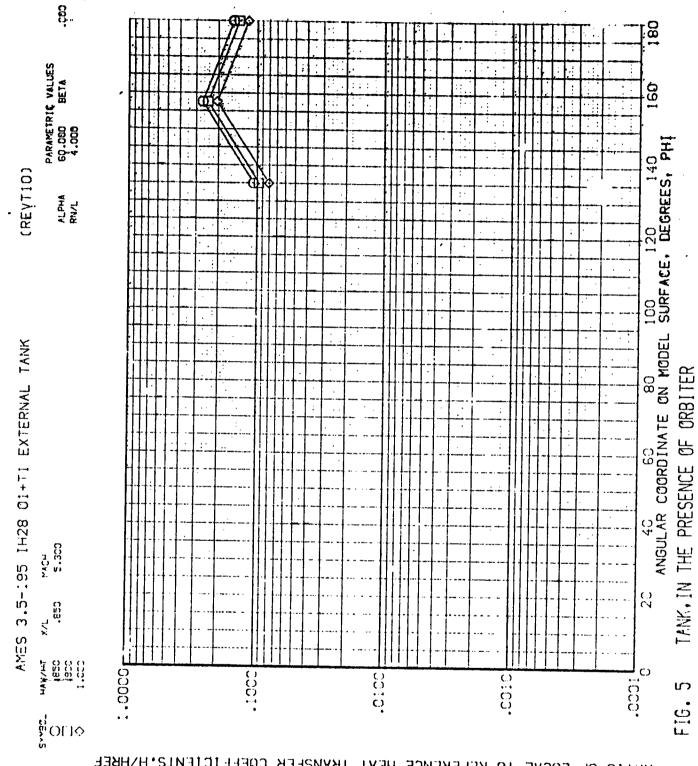
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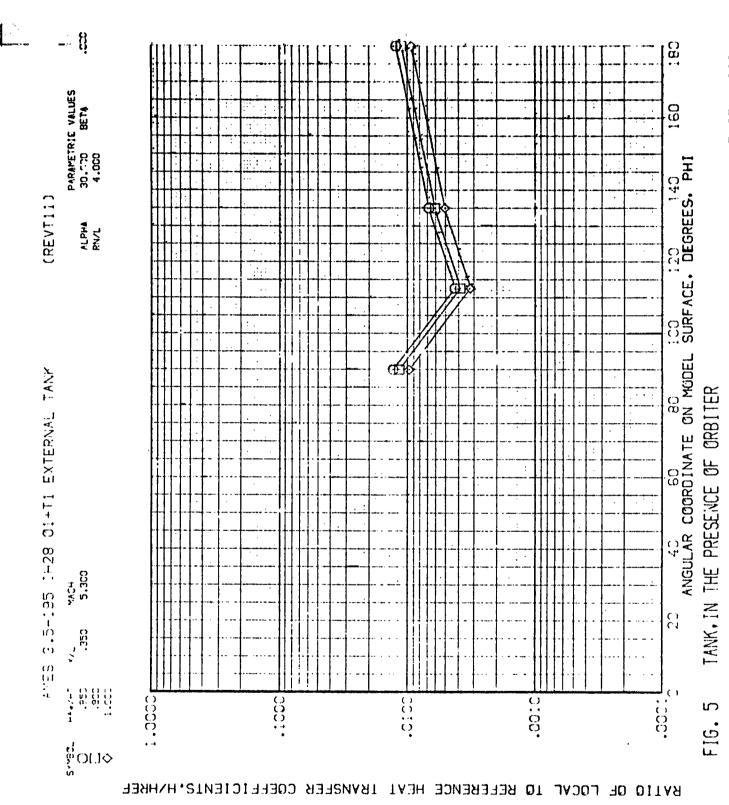
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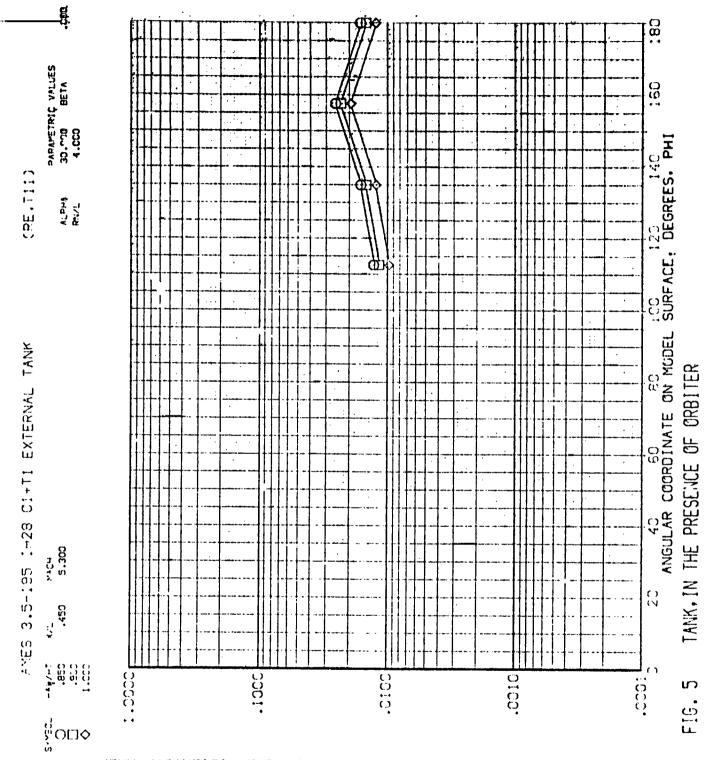
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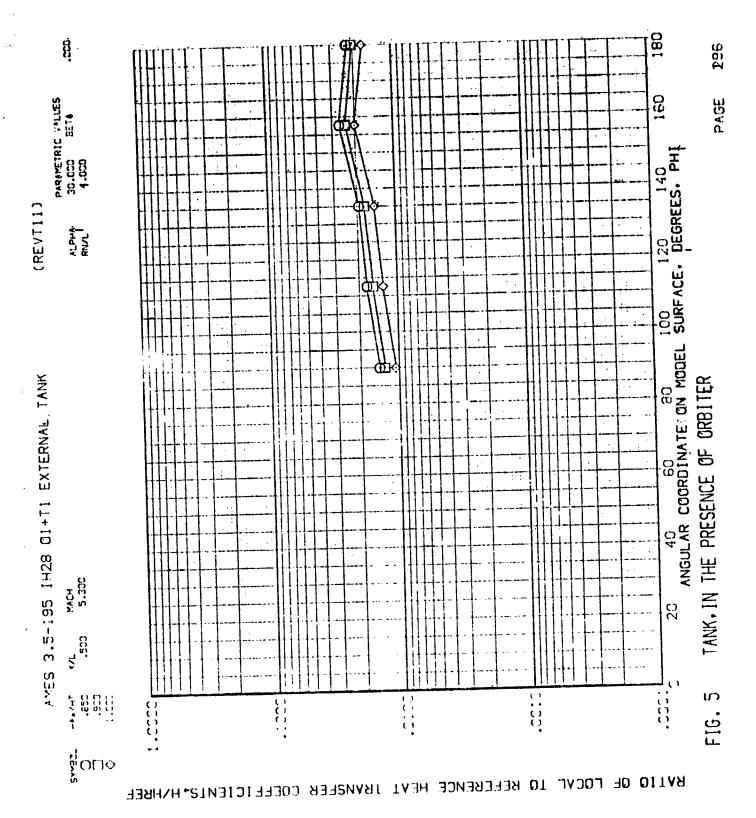
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FIG. 5

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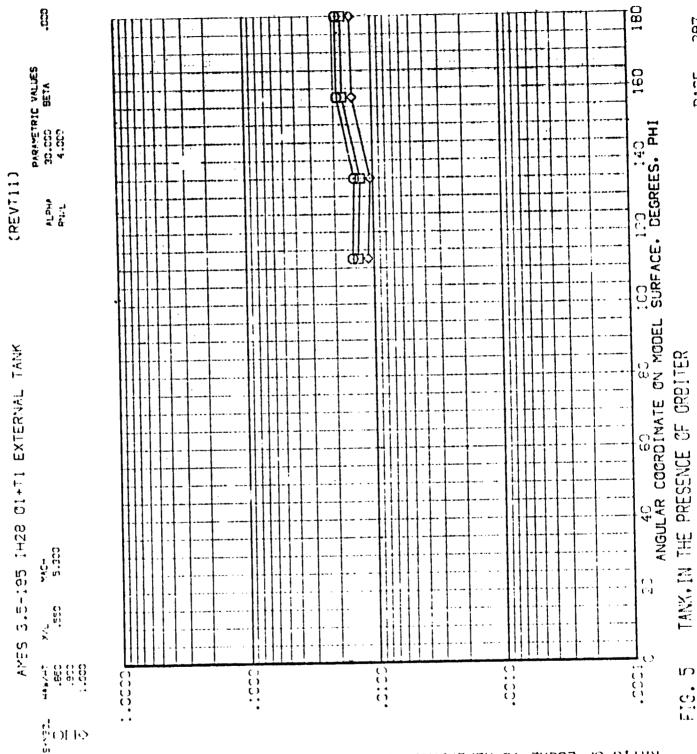


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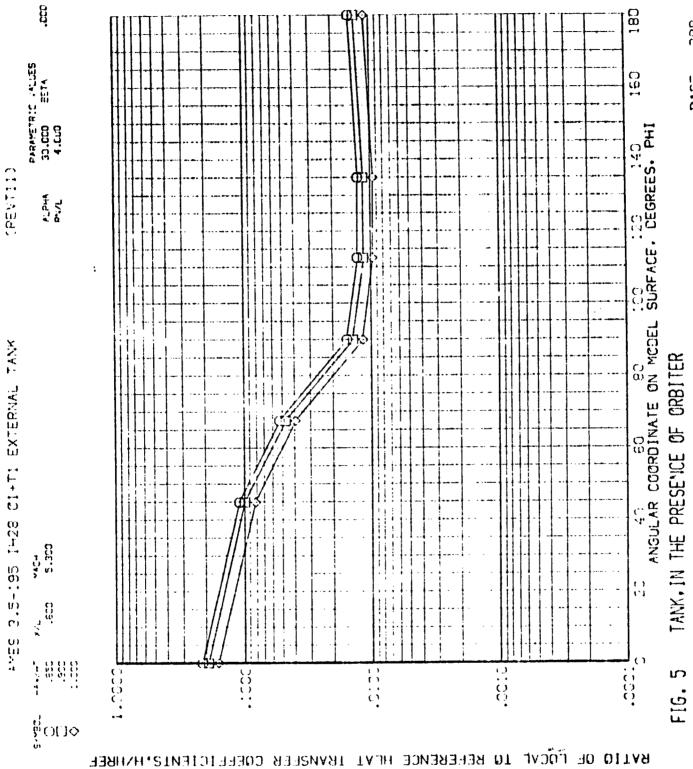


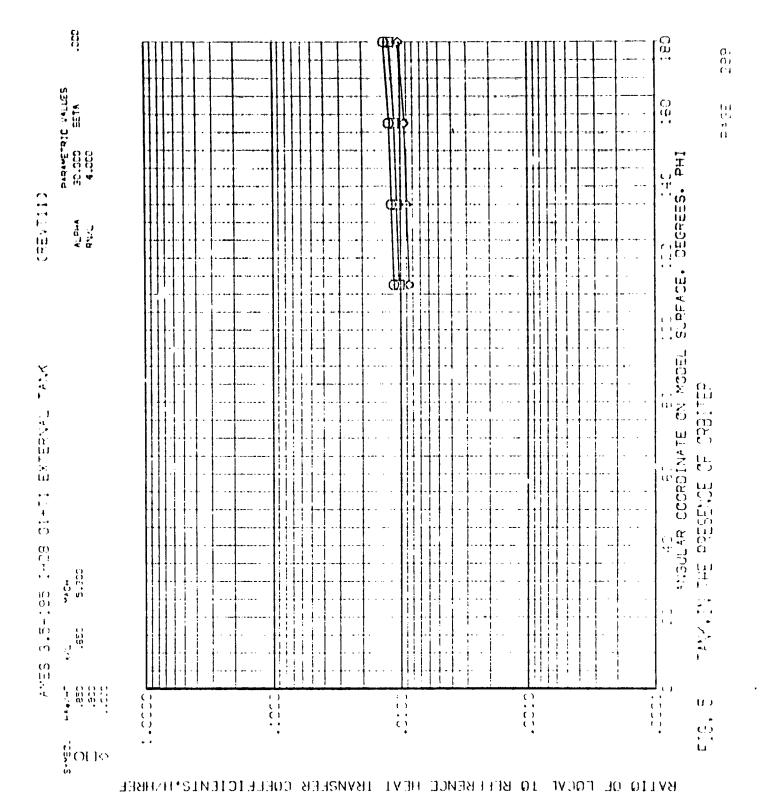
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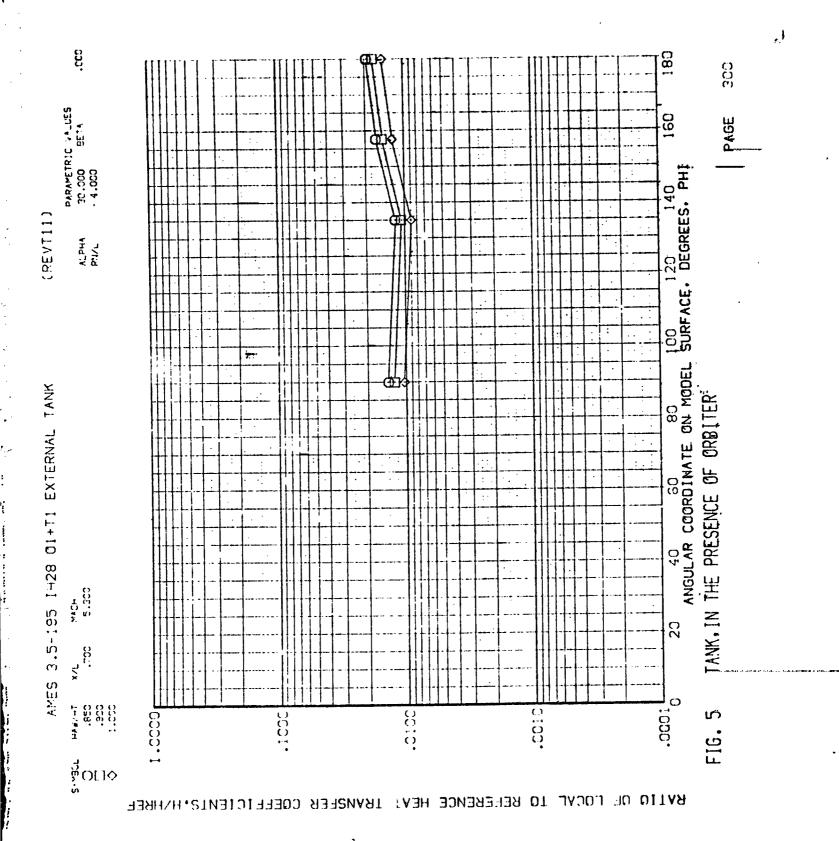
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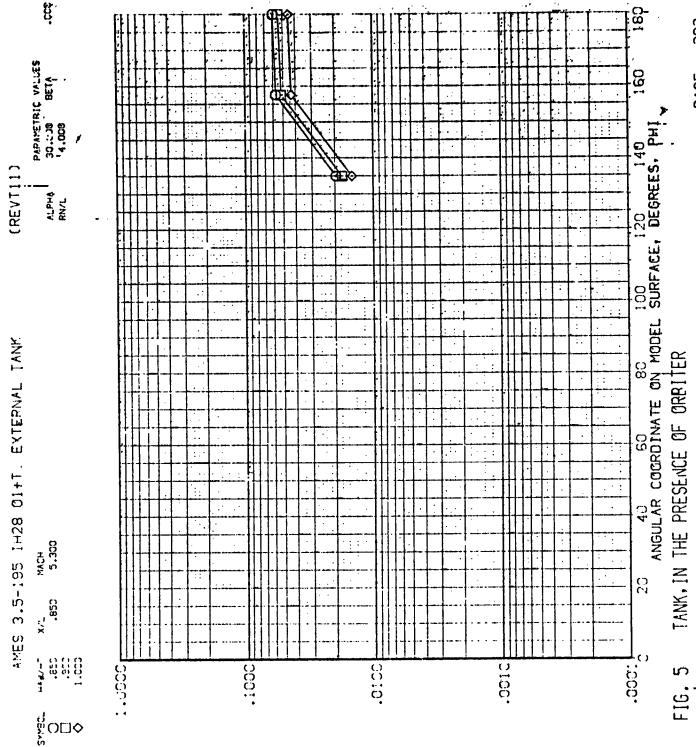






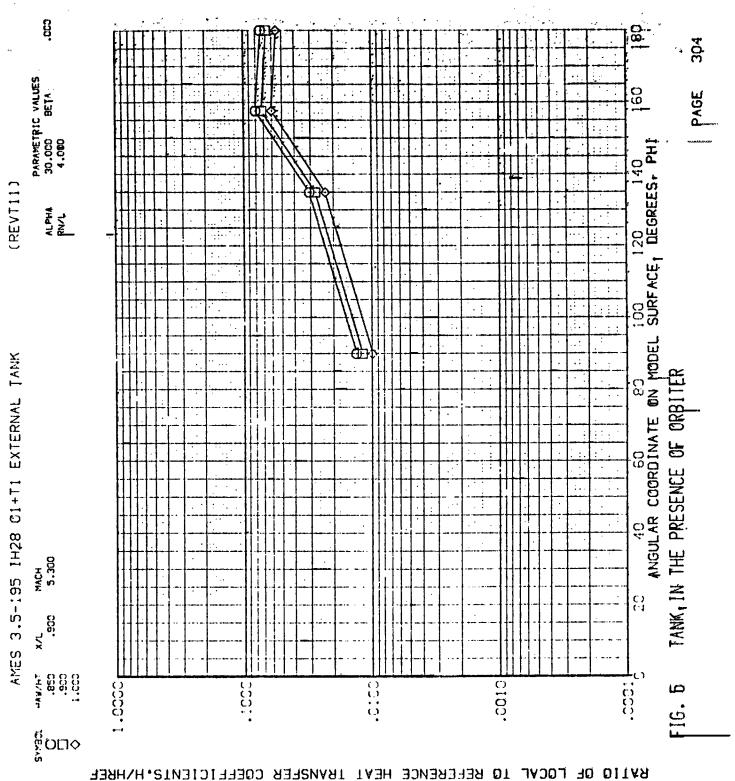
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FIG. 5

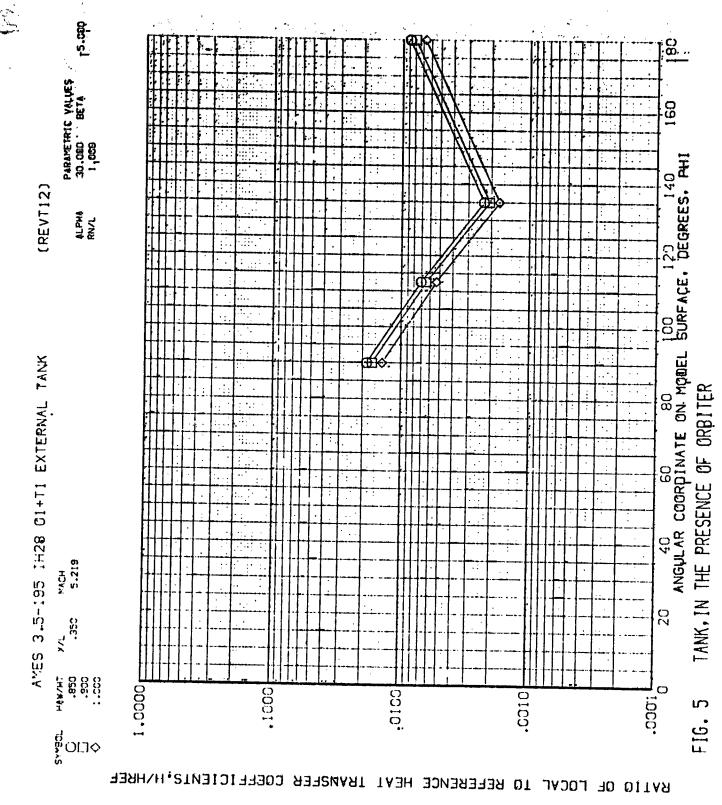


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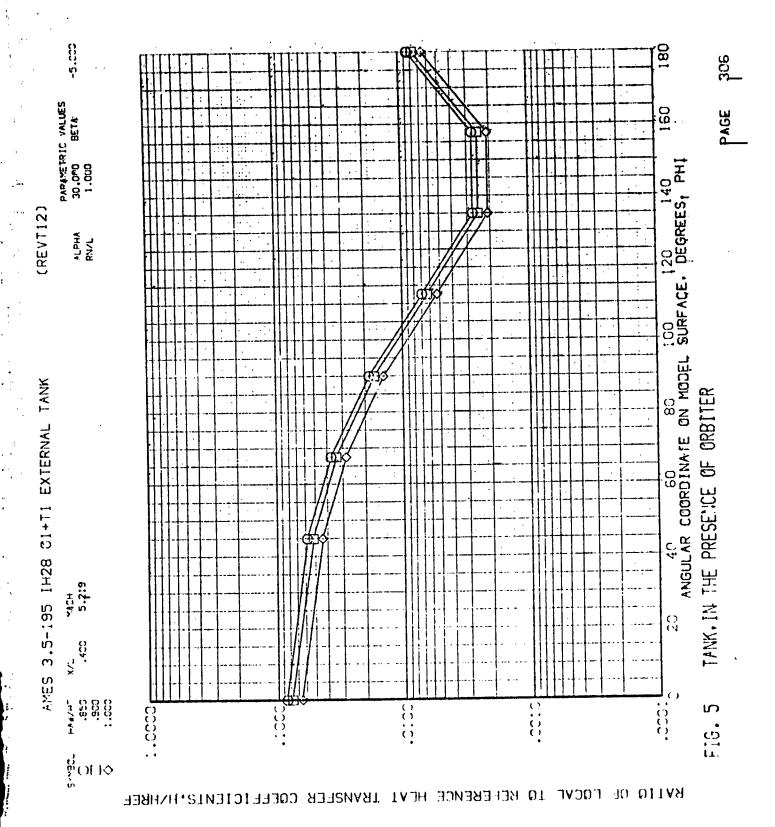
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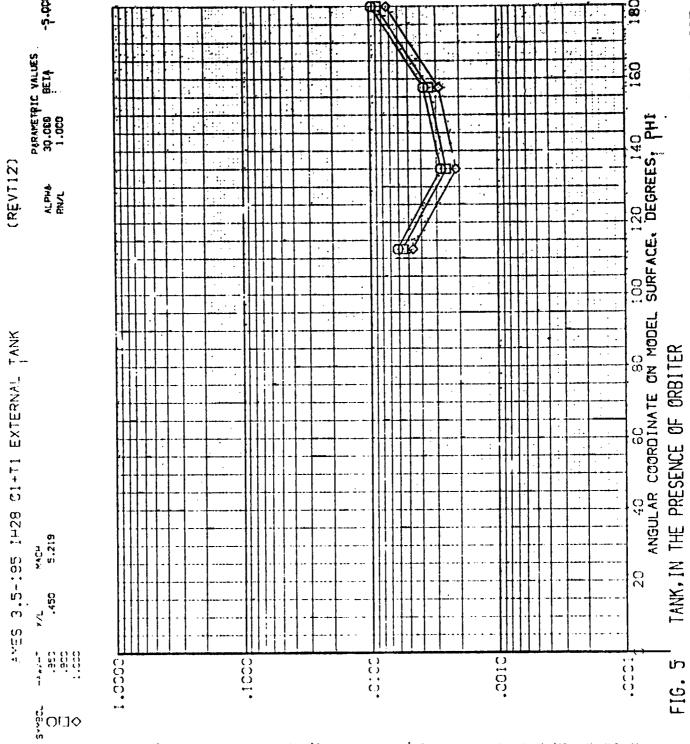


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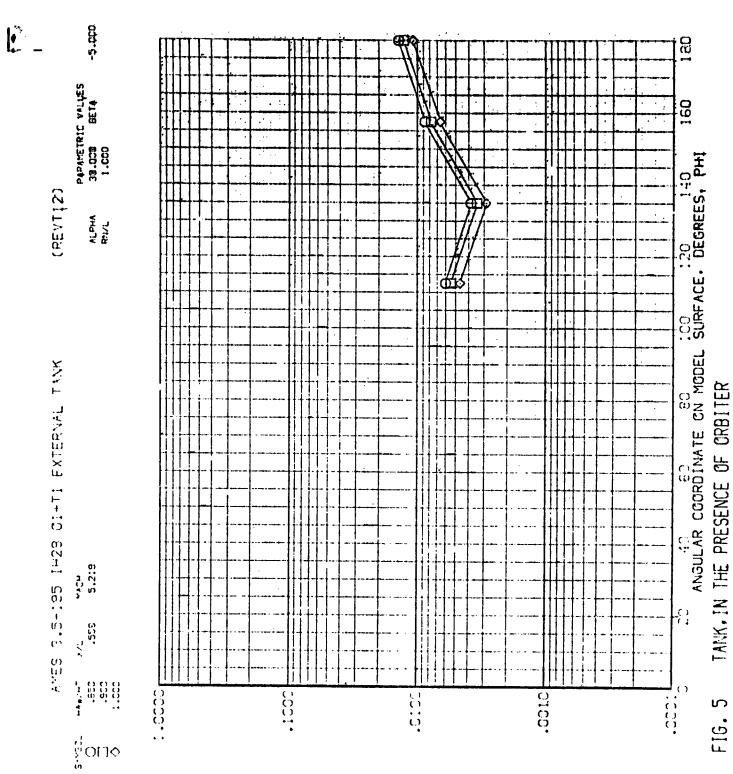
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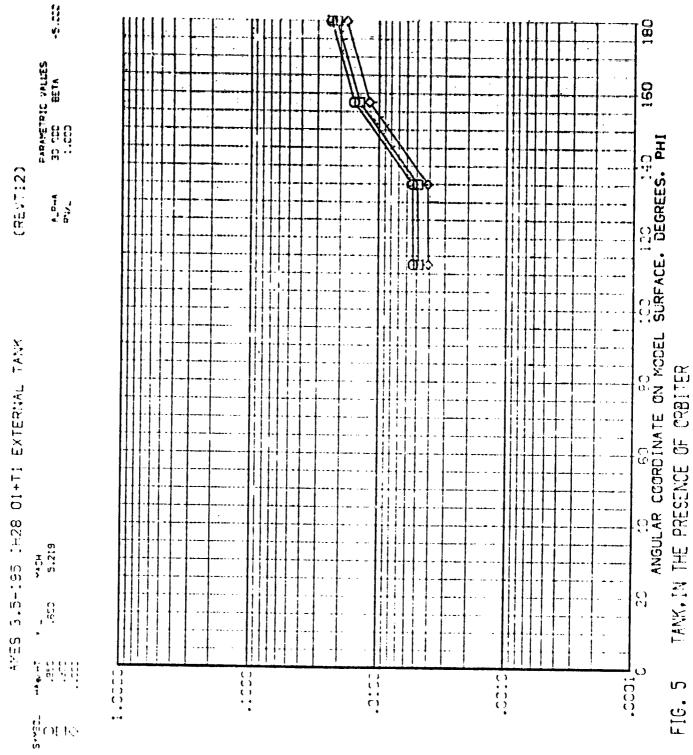
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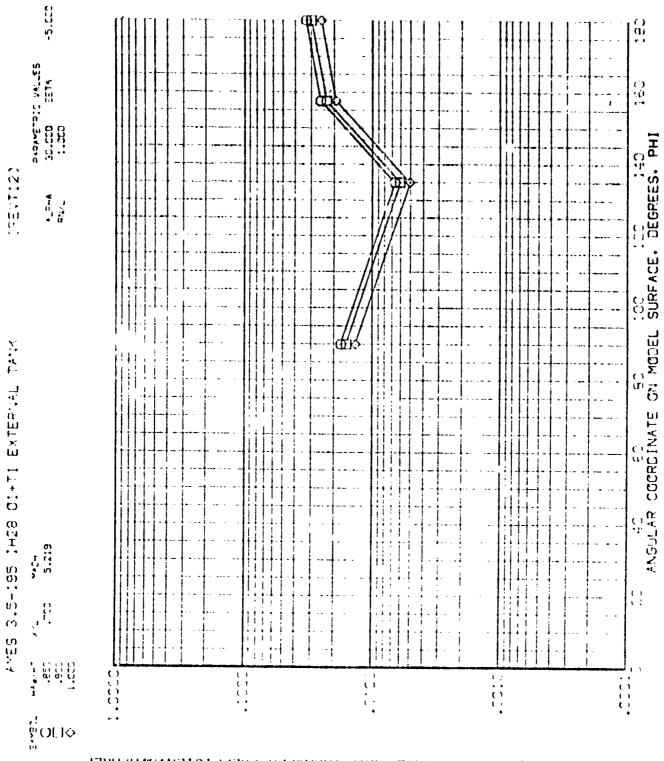
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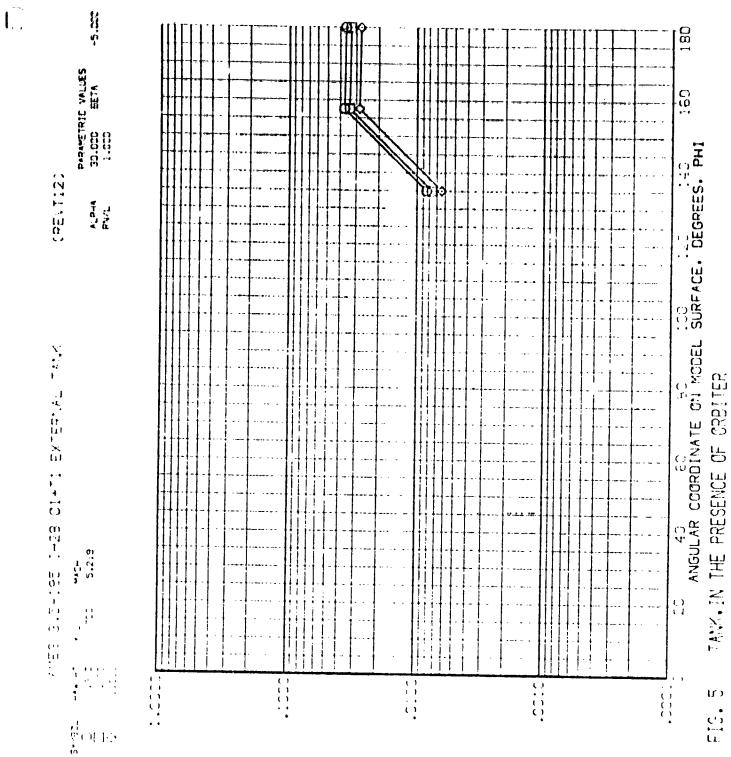
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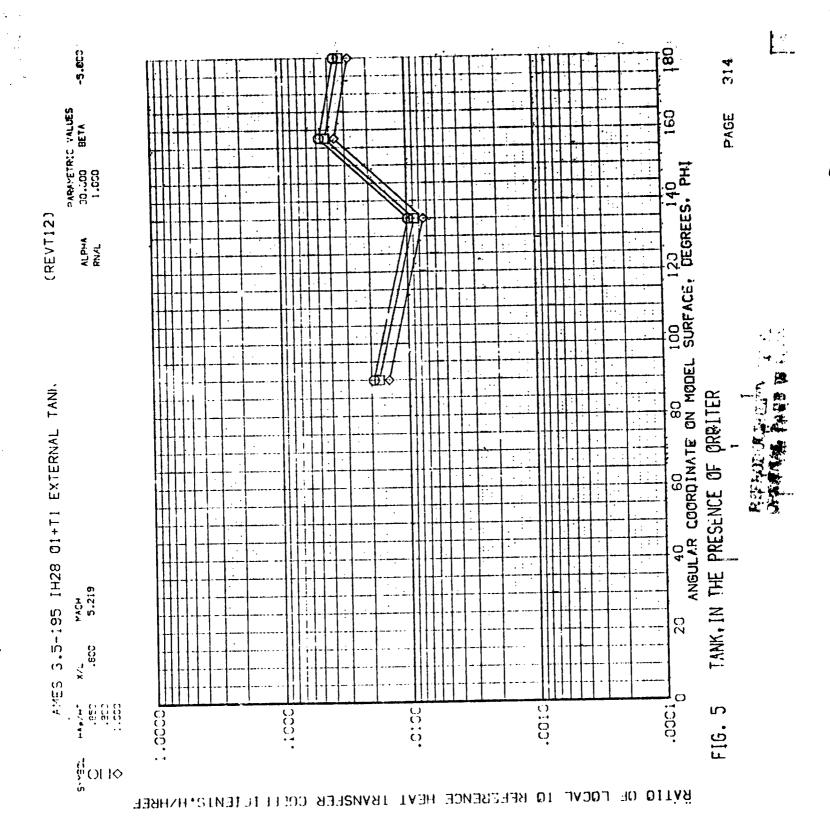
FIG. 5

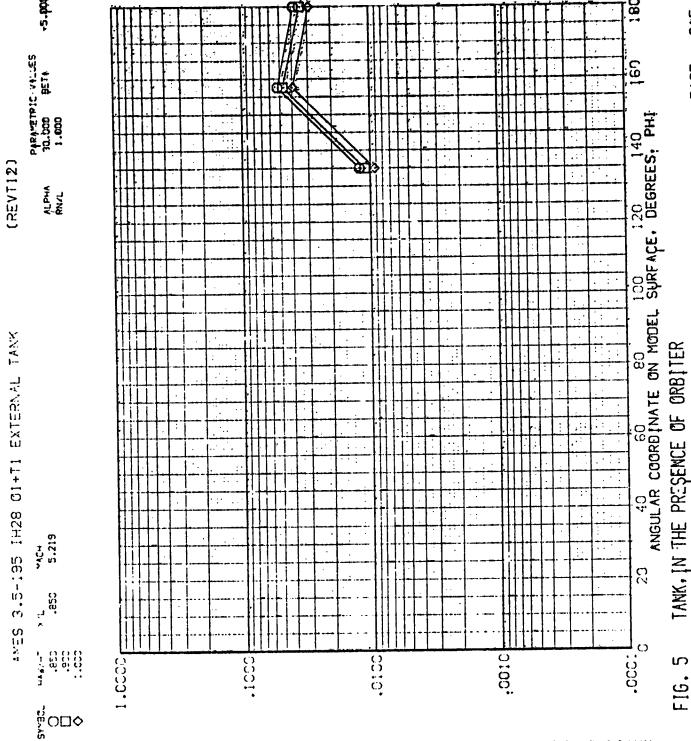


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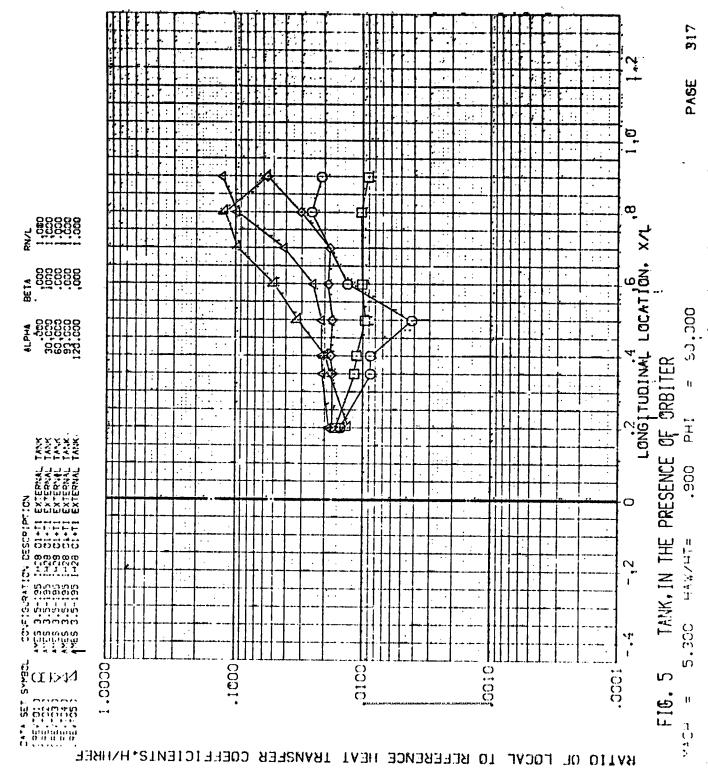


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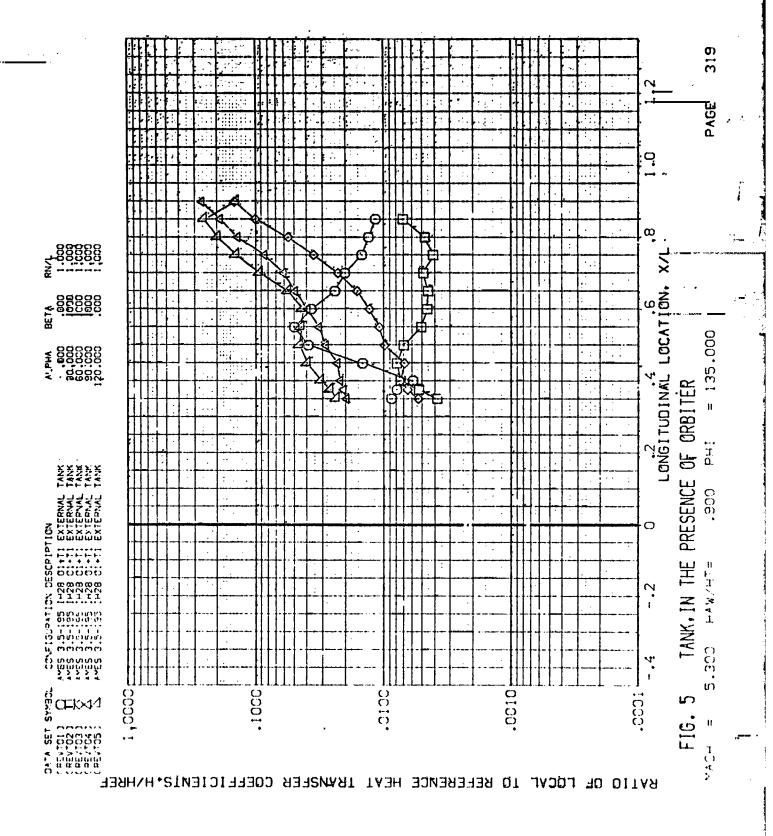
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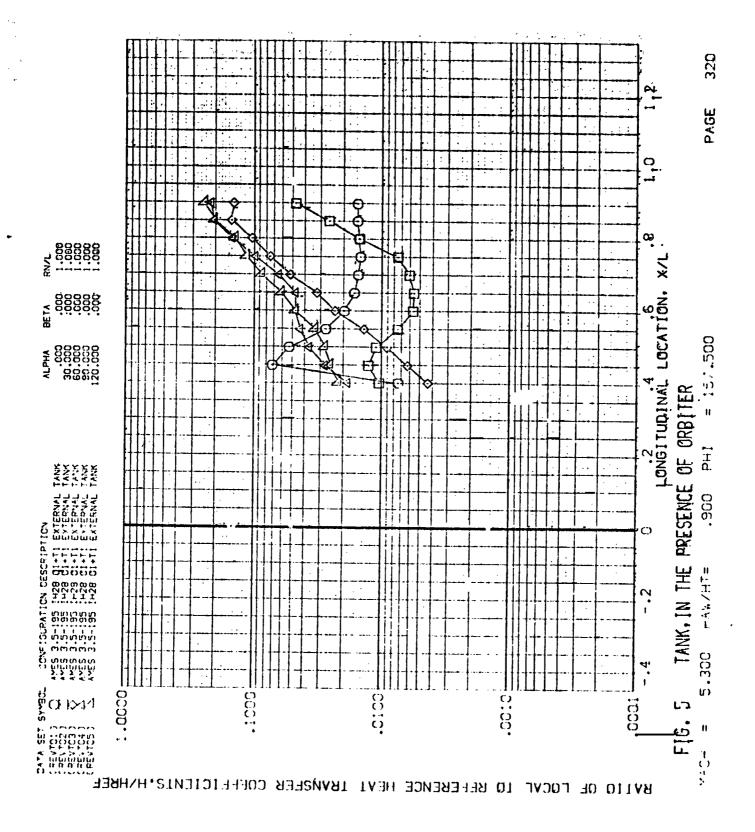
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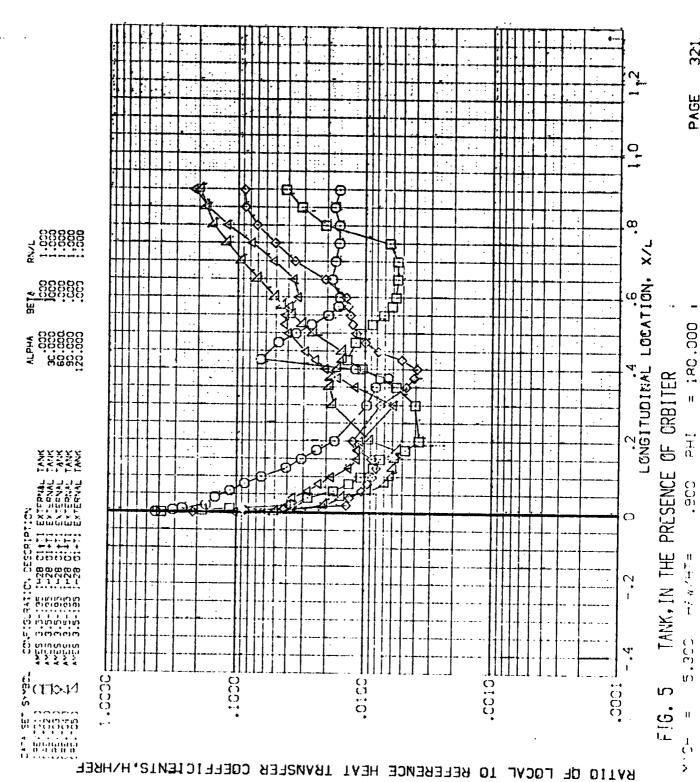
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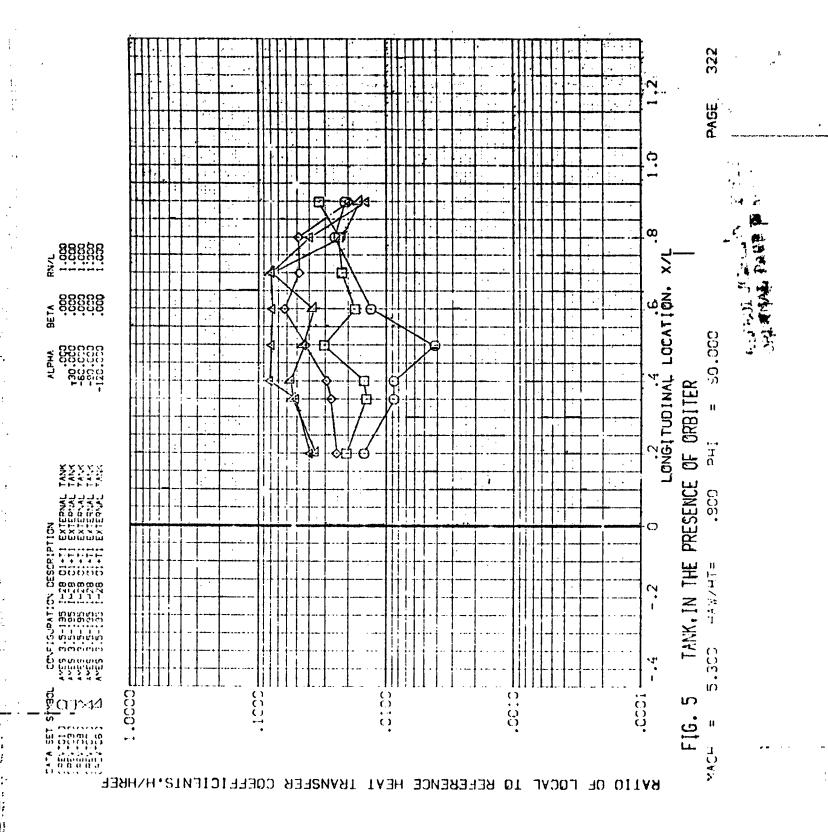




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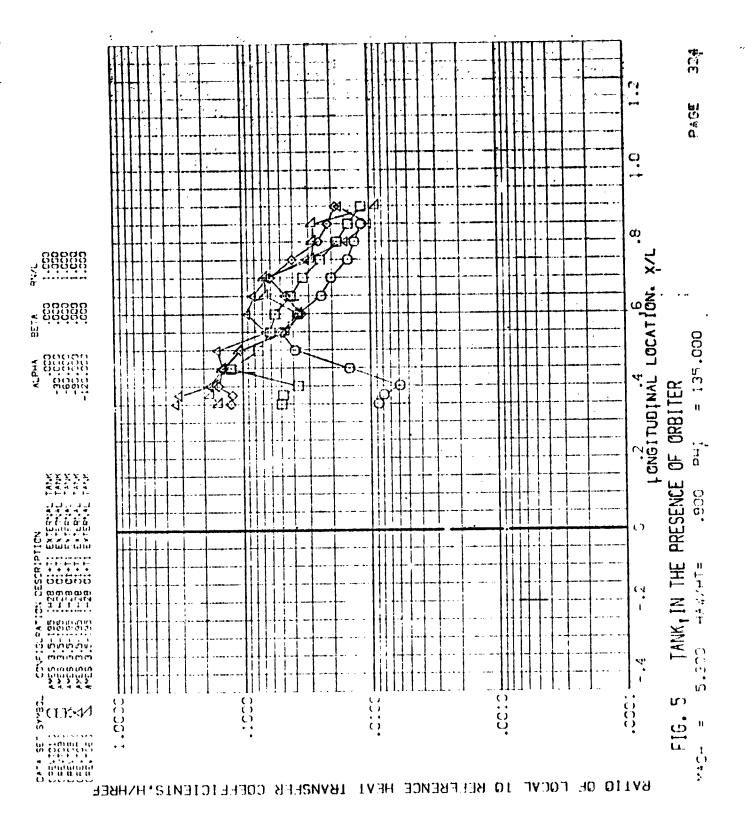
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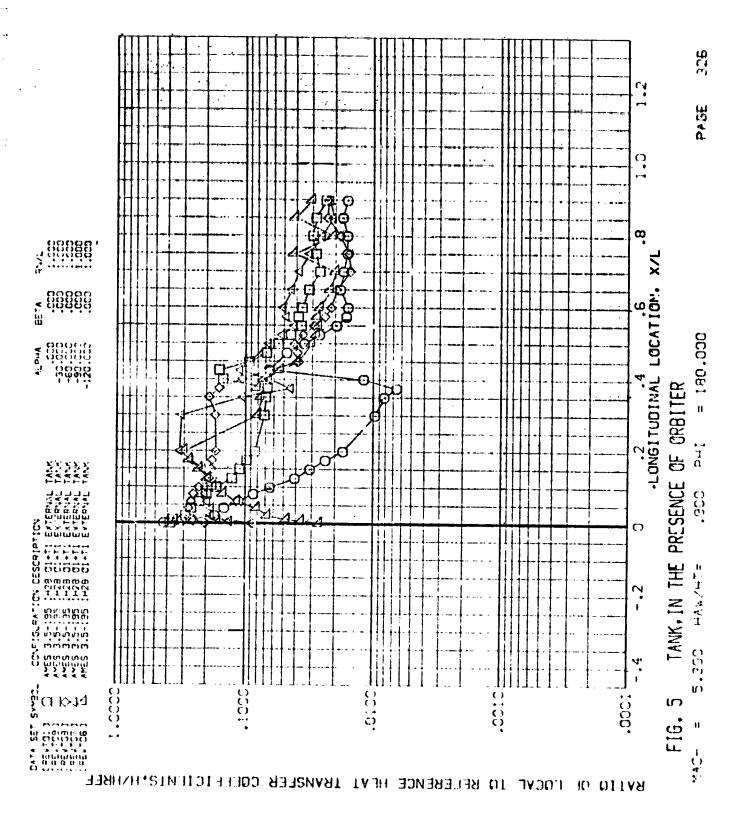
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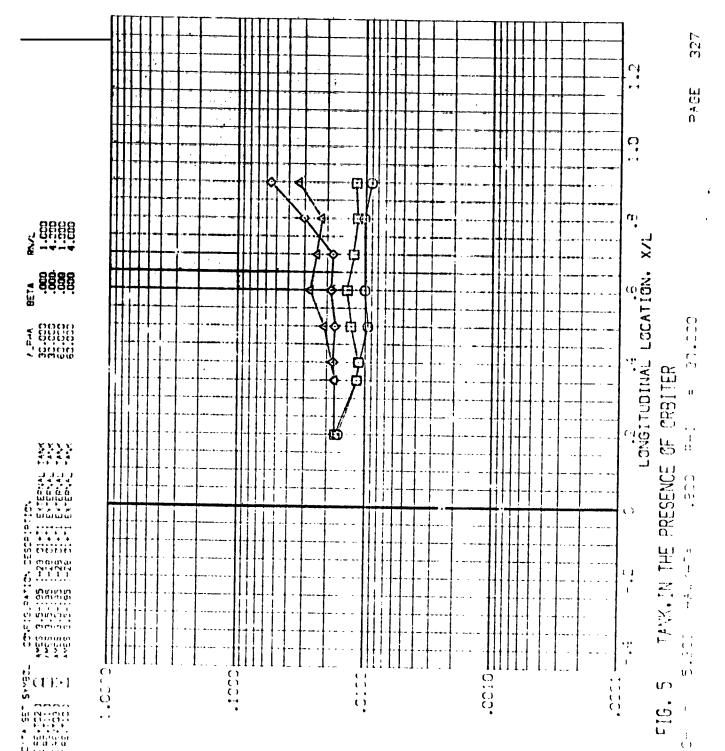
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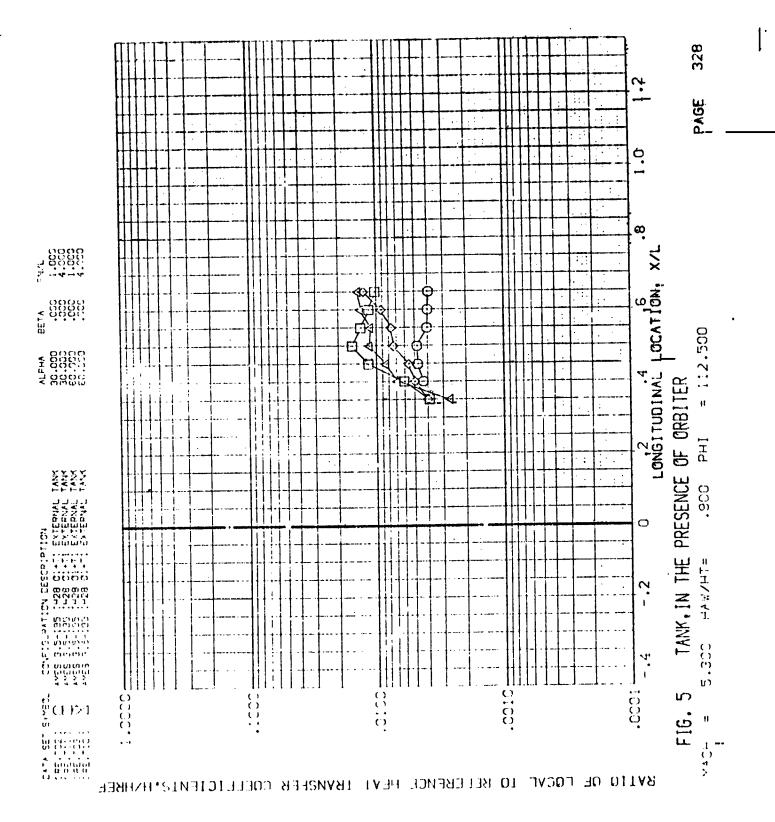


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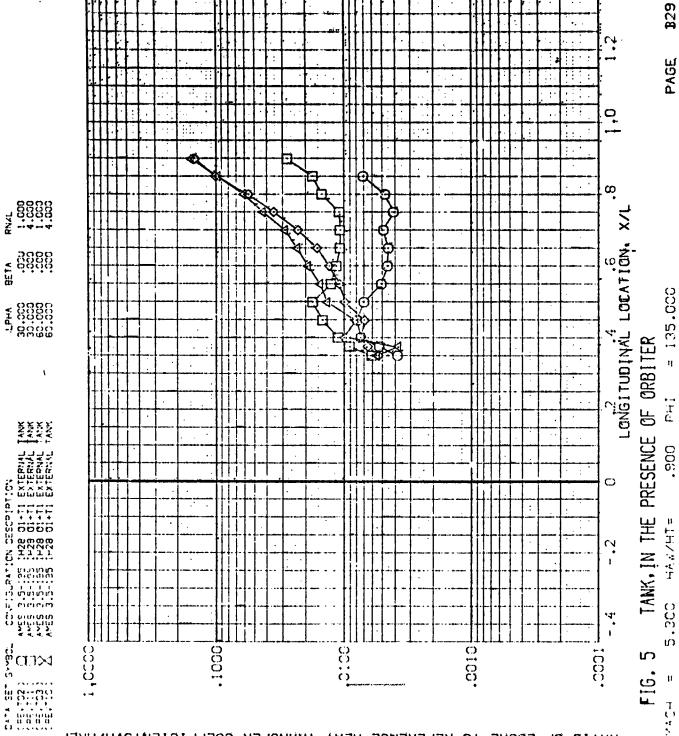
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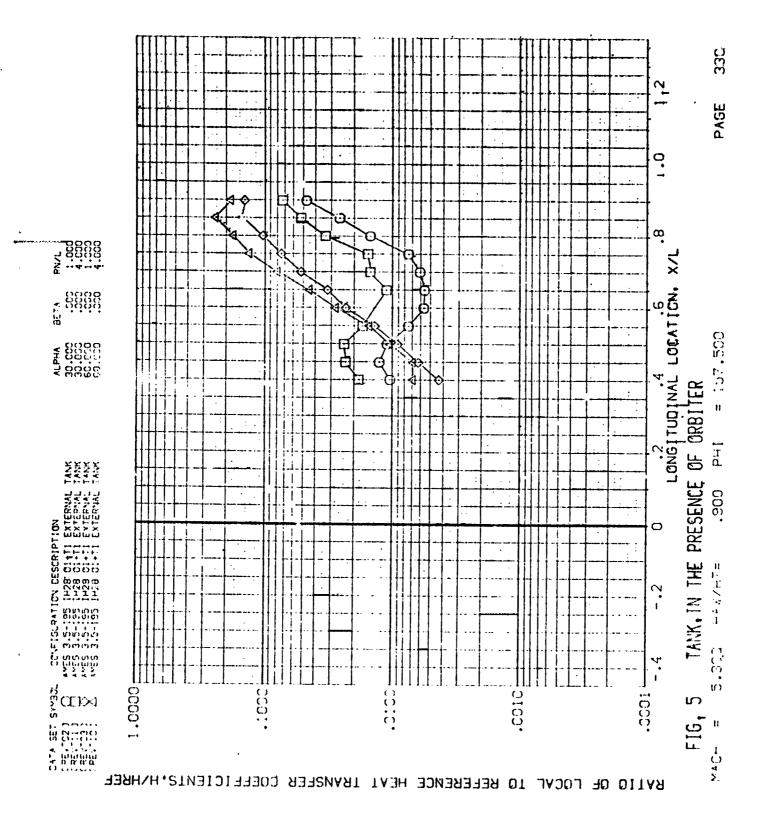
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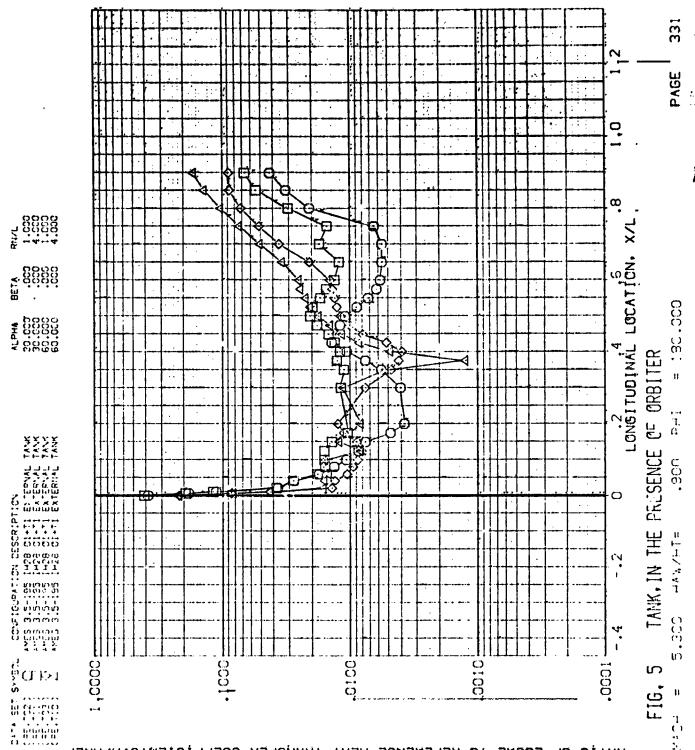
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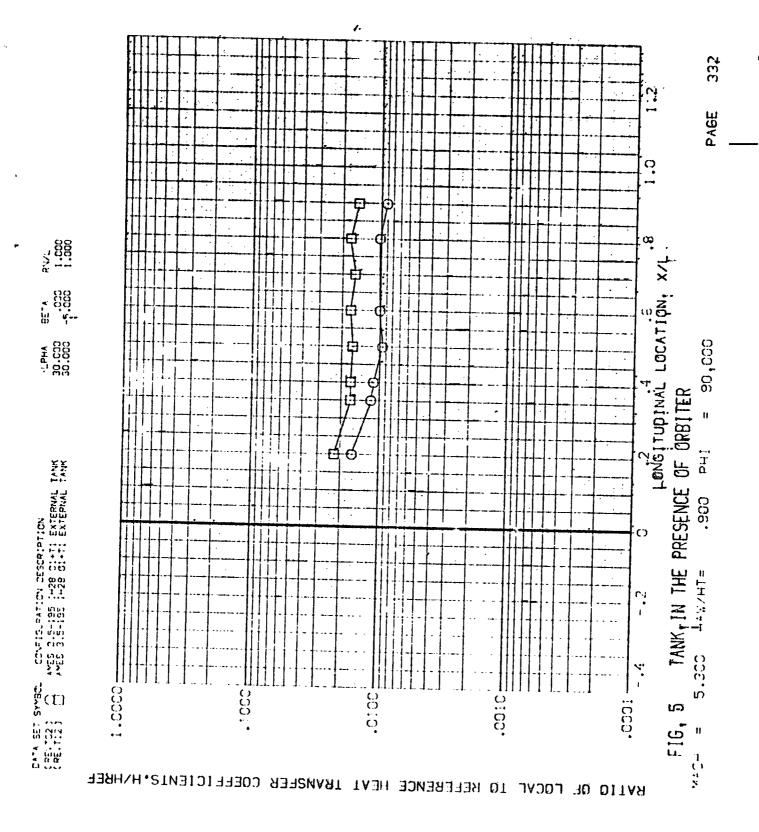
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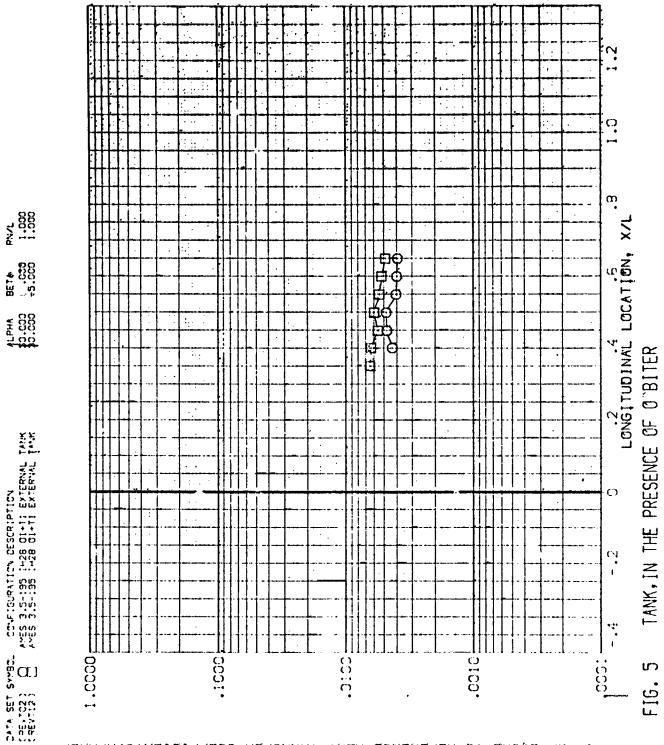
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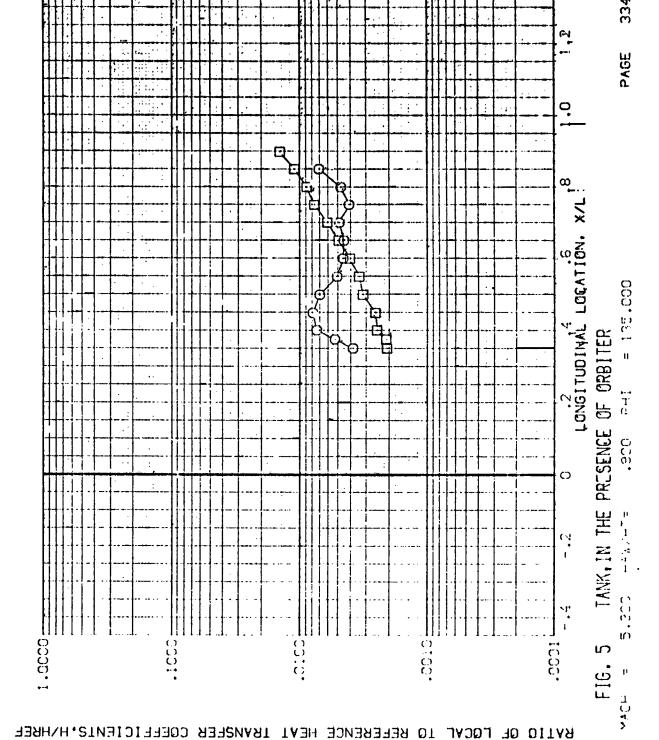
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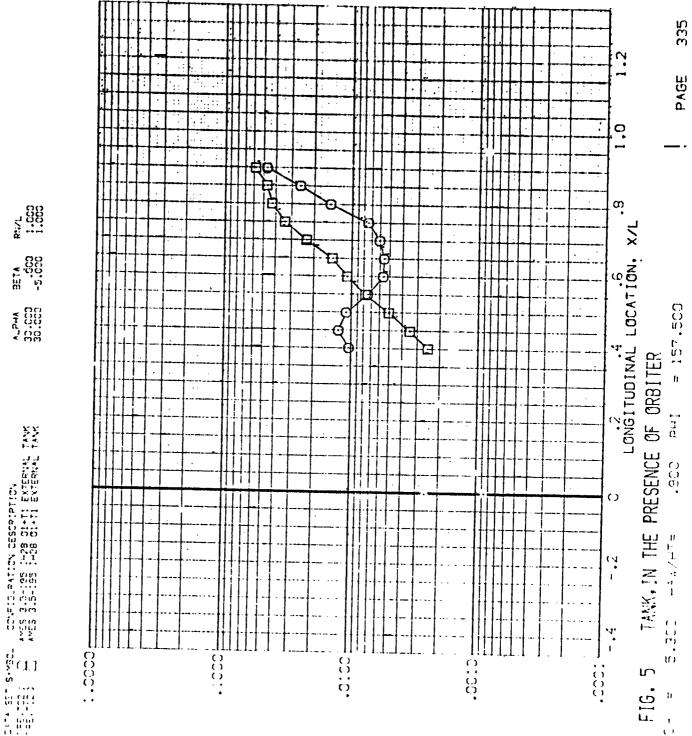
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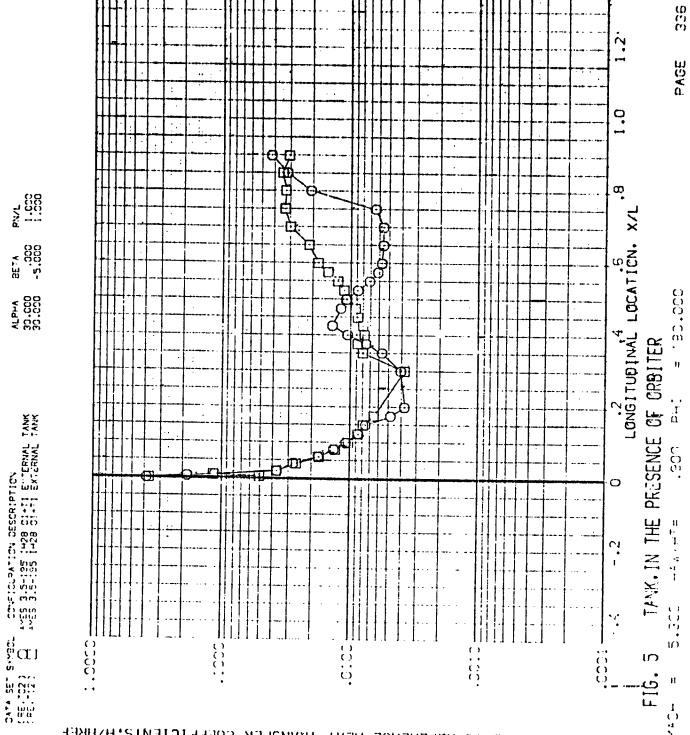
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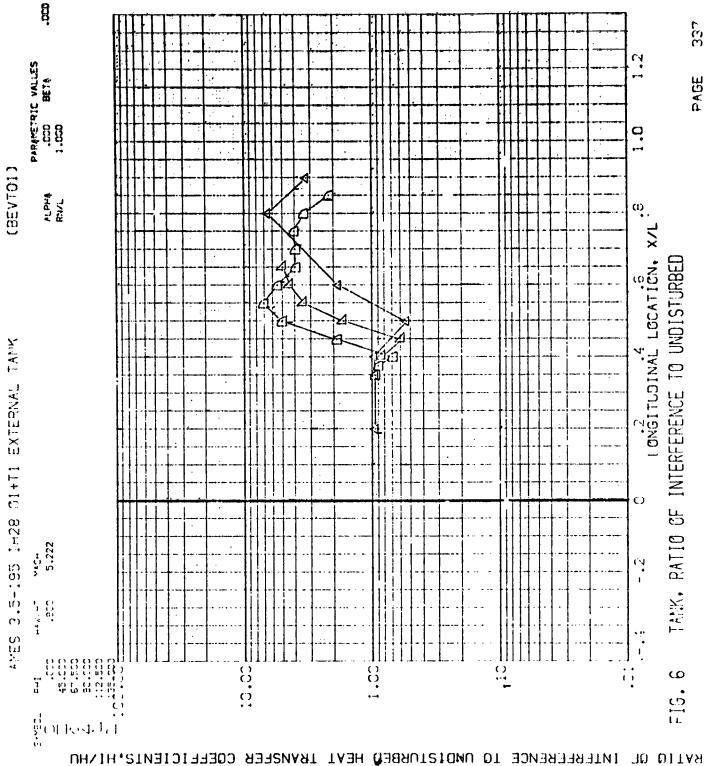
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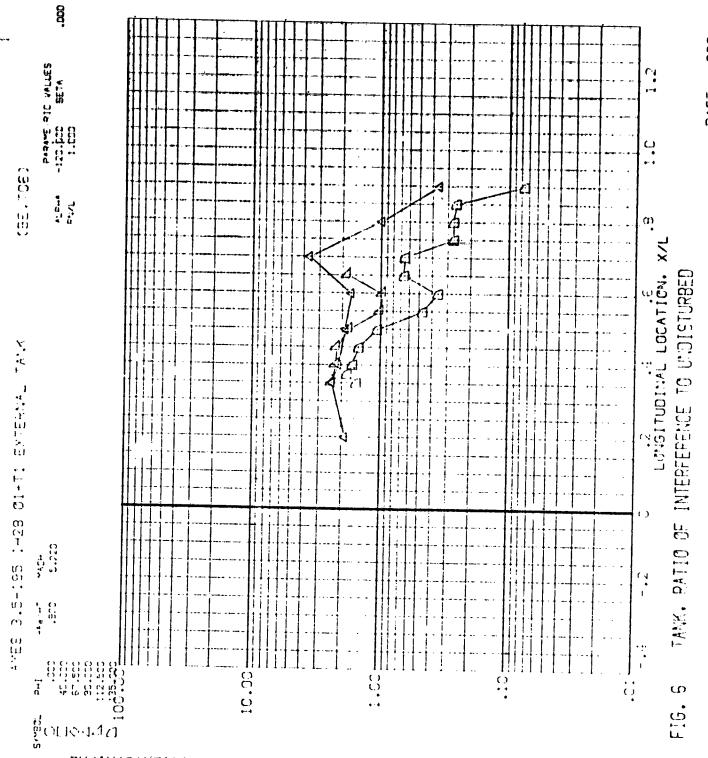
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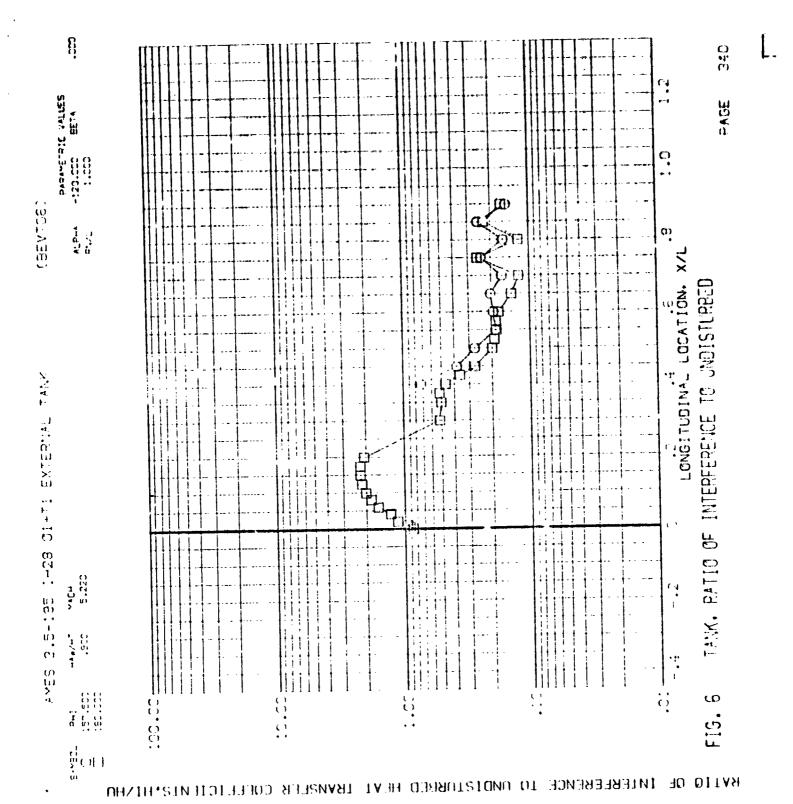
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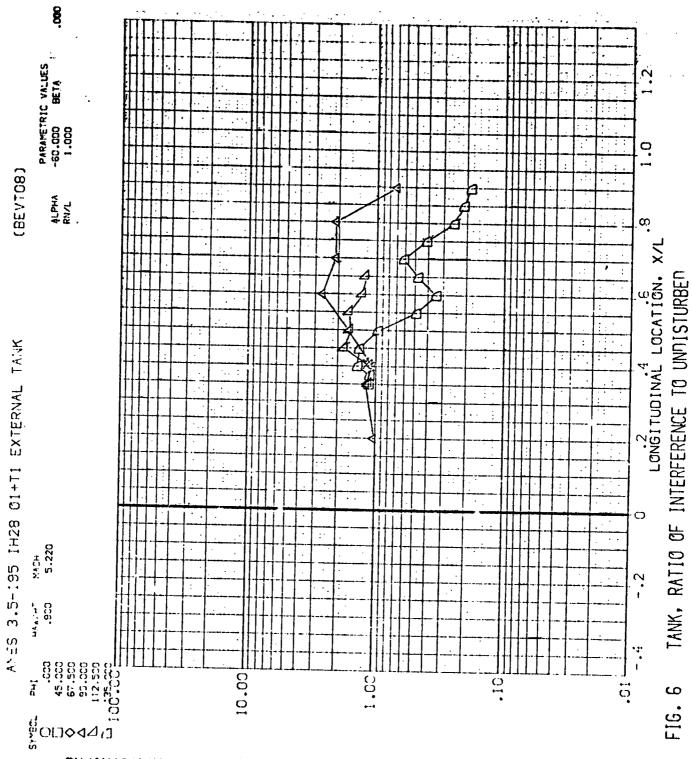


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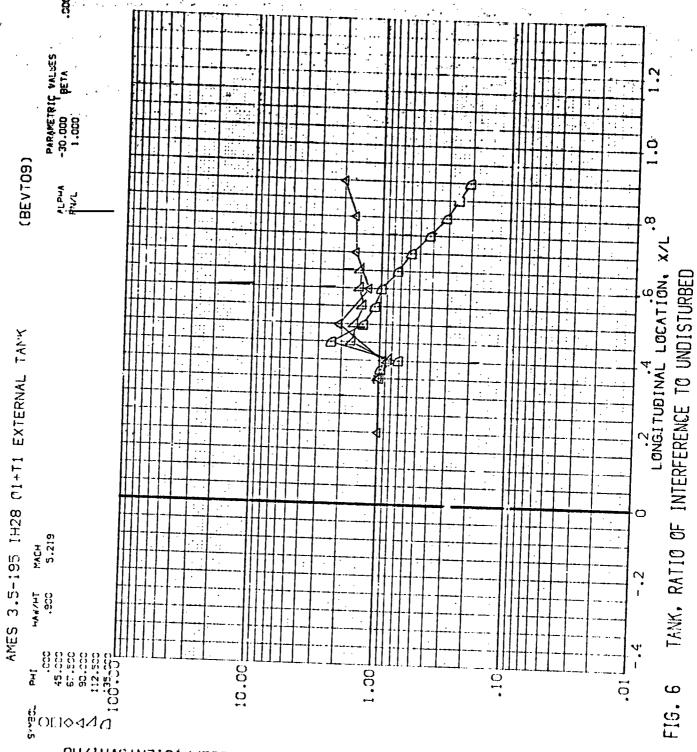
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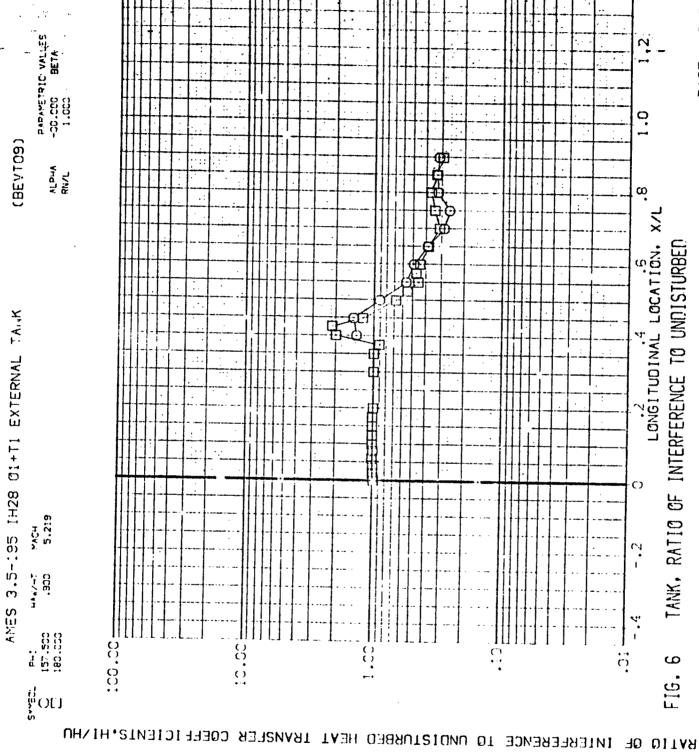


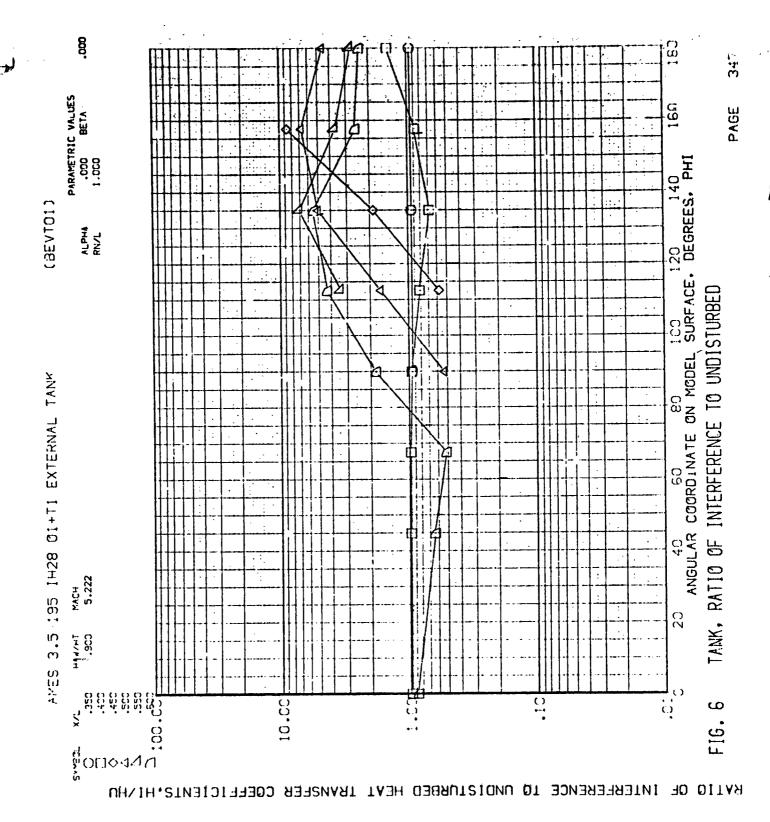
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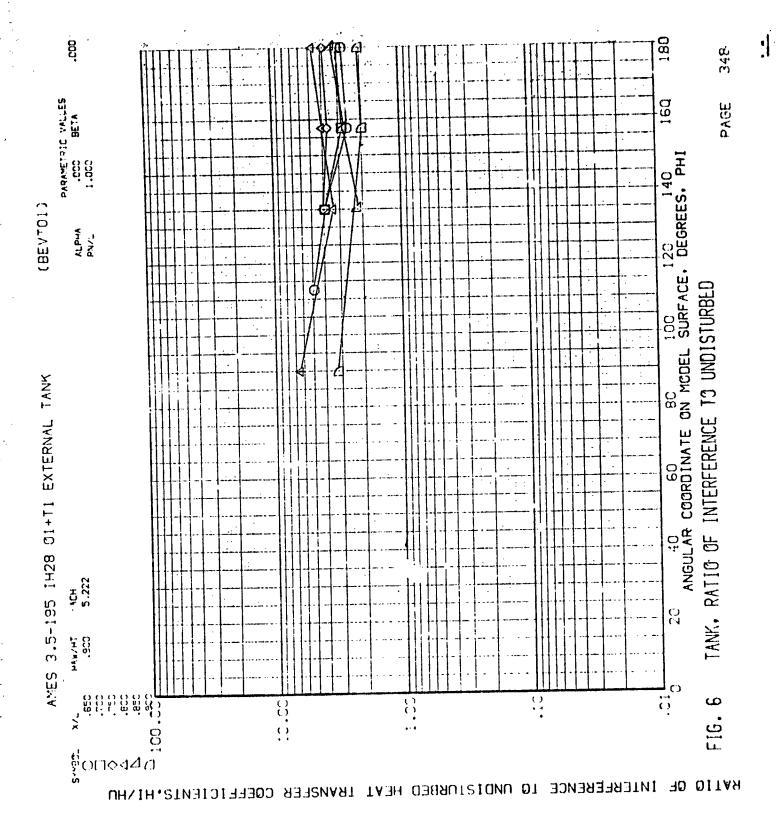
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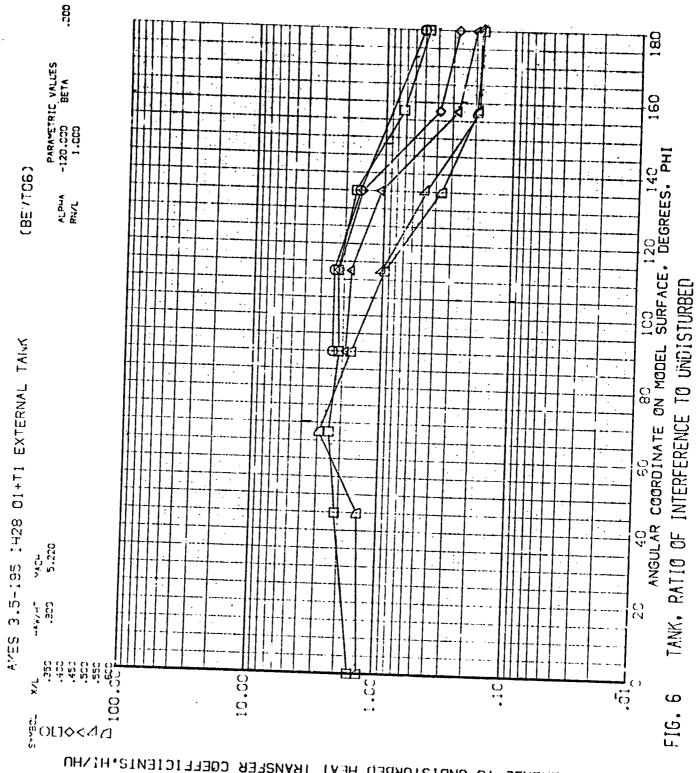
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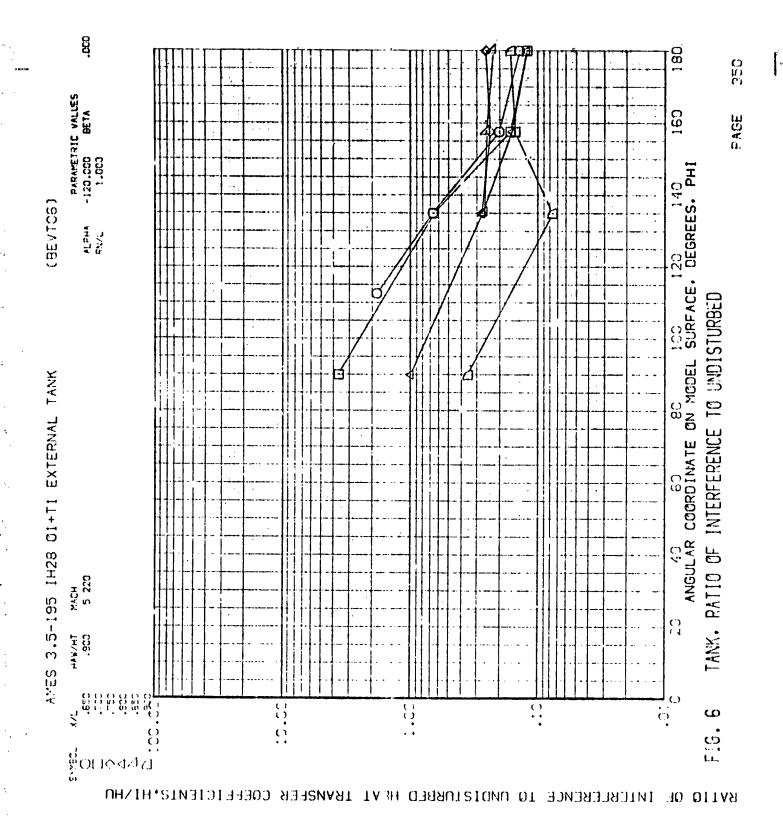


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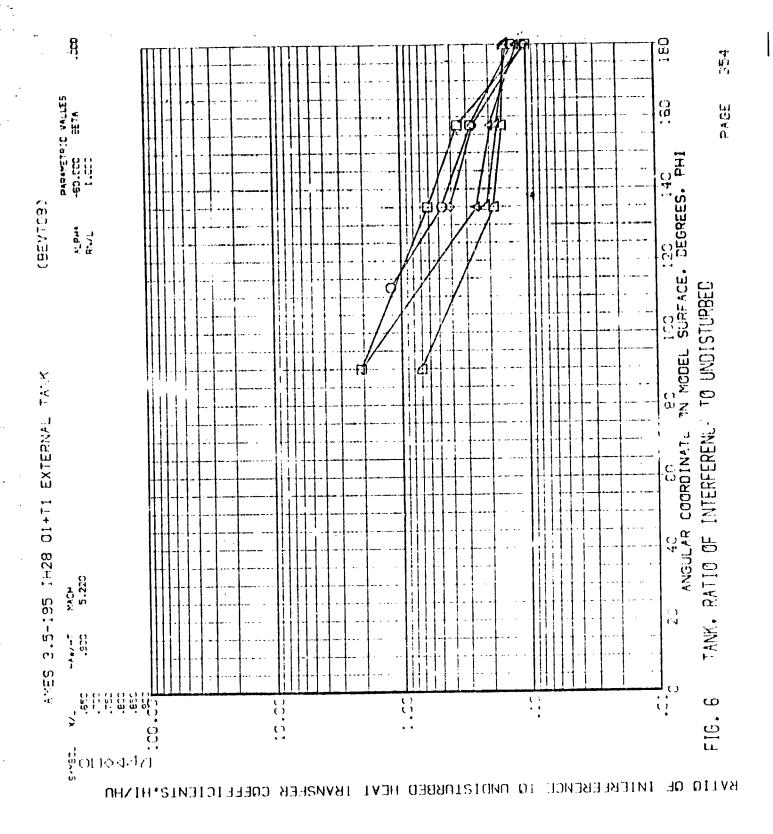
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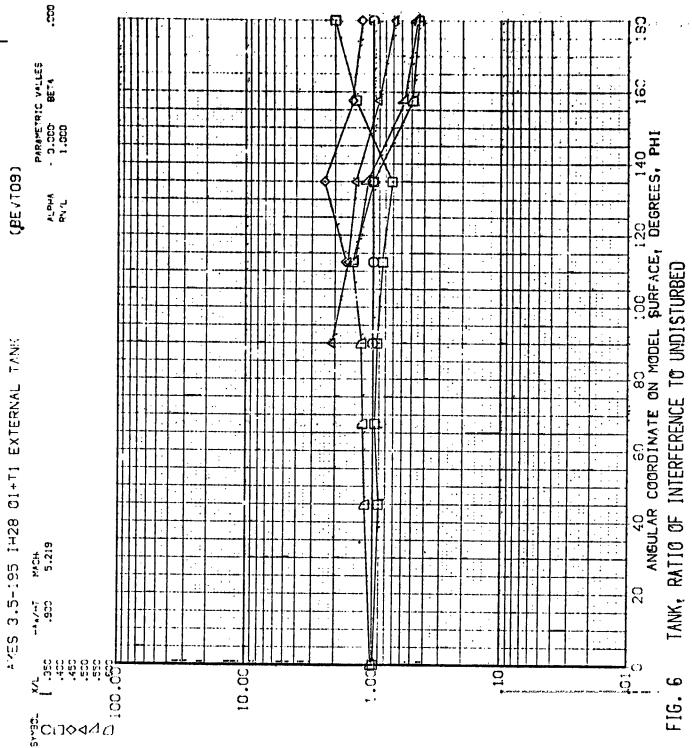
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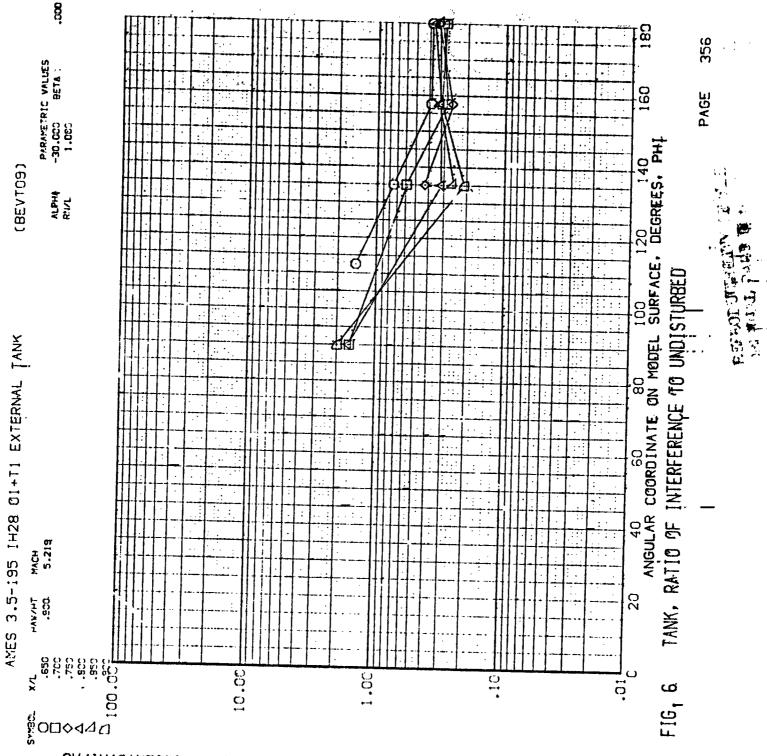
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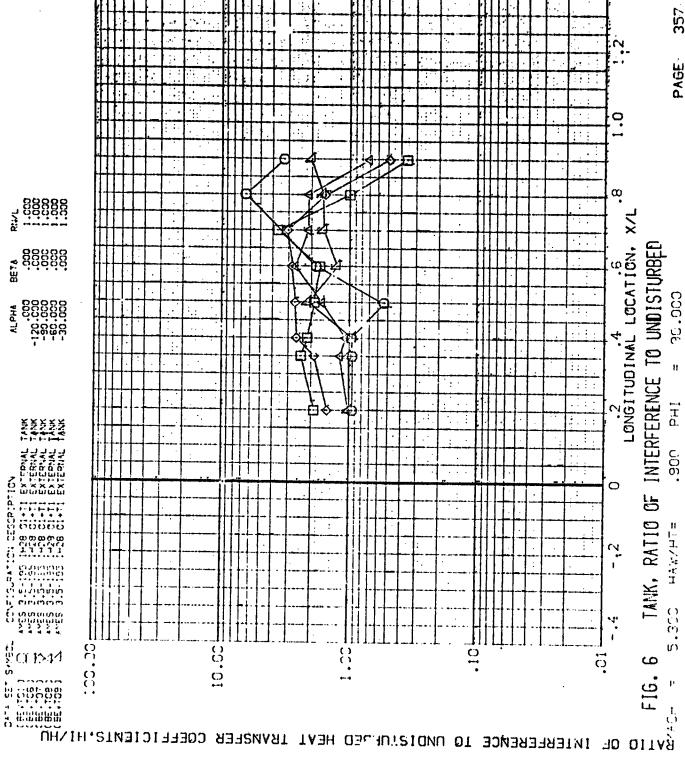


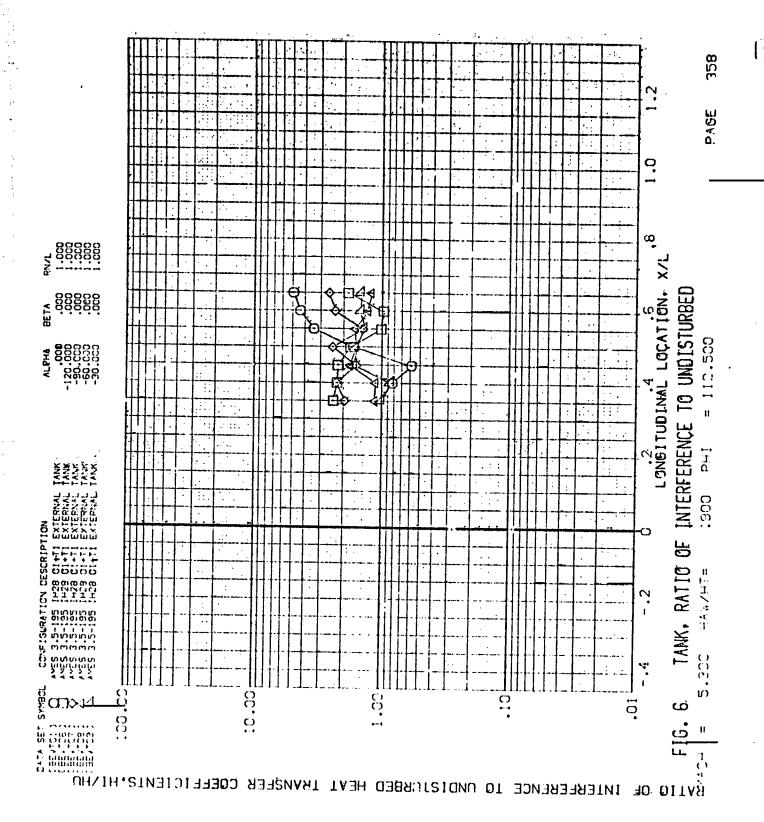


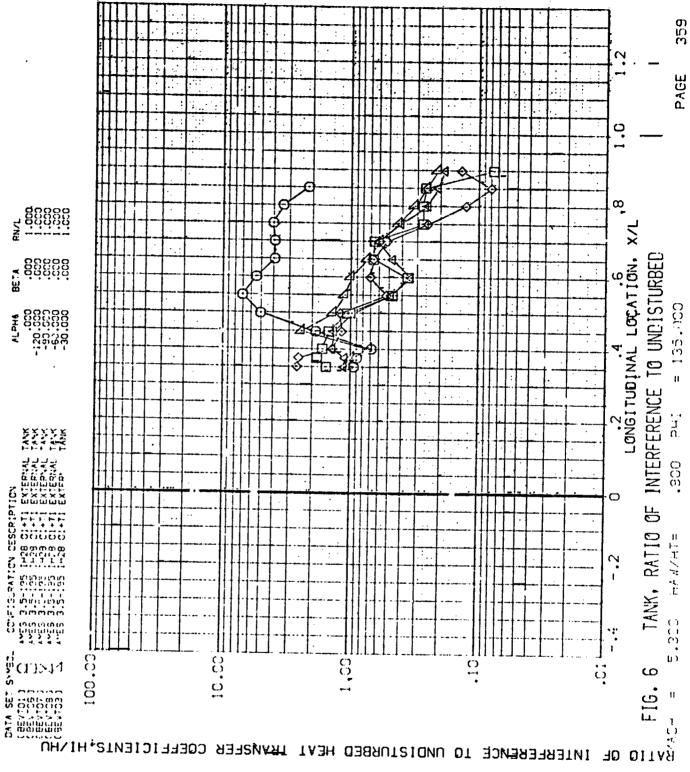
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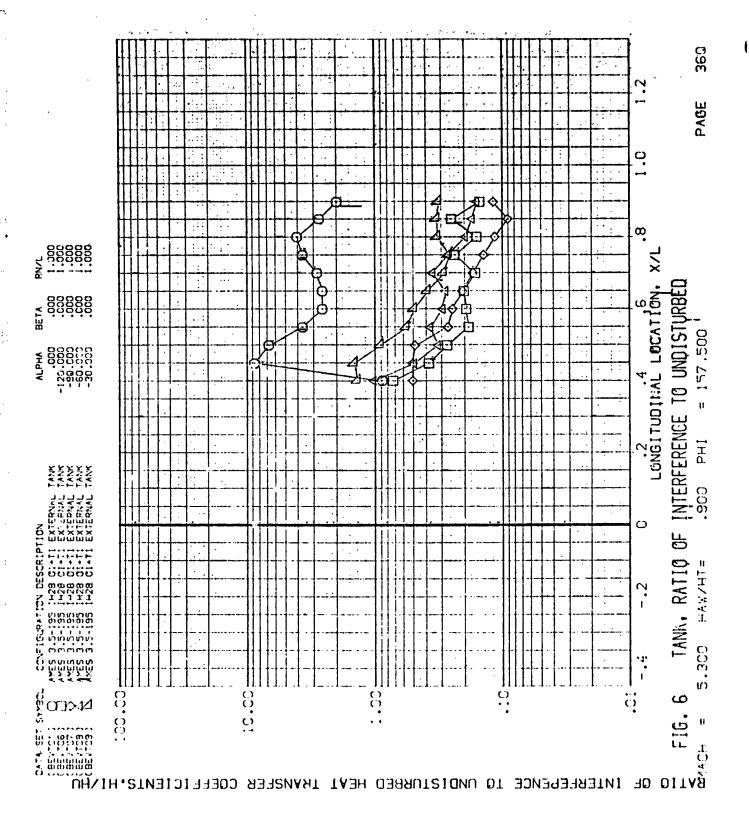


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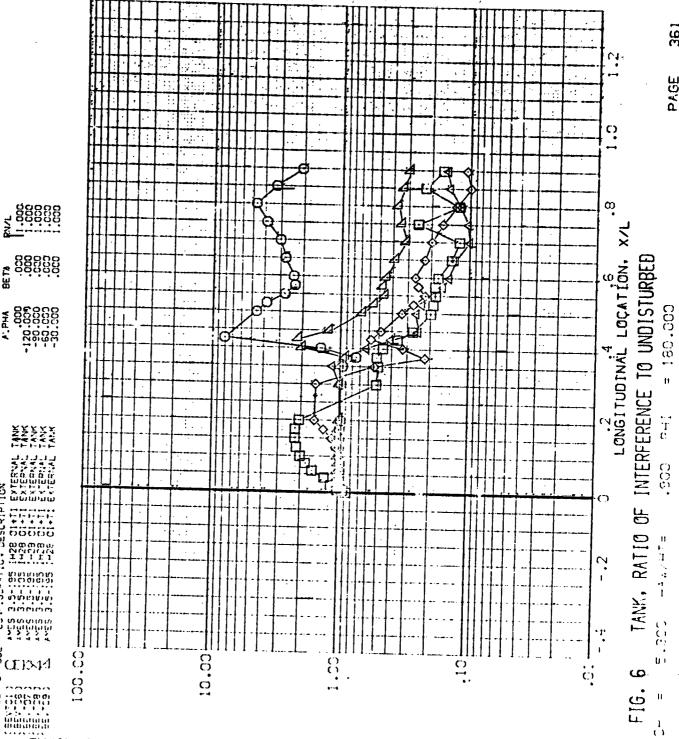








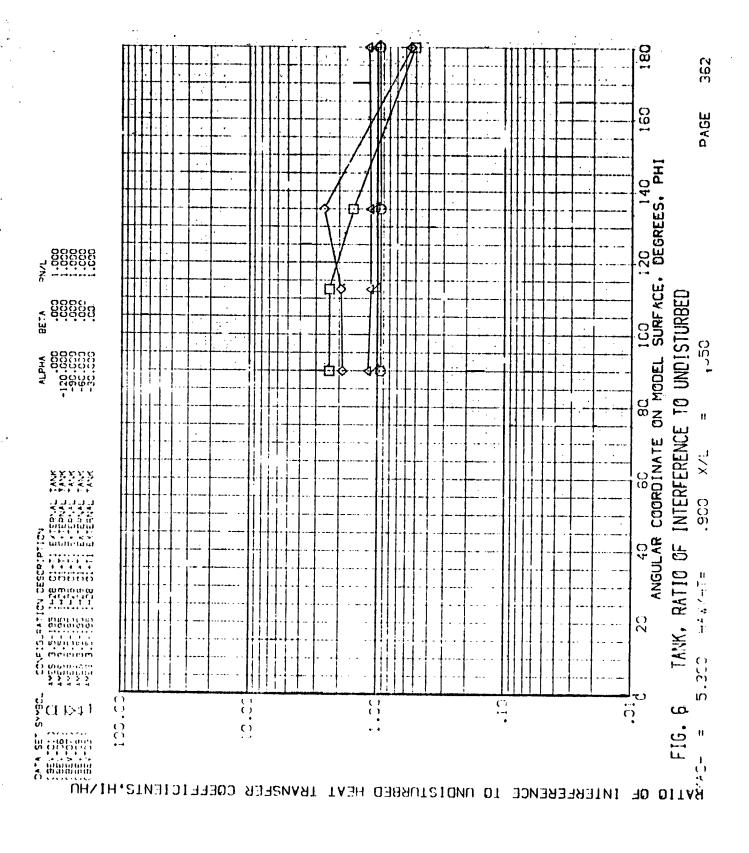
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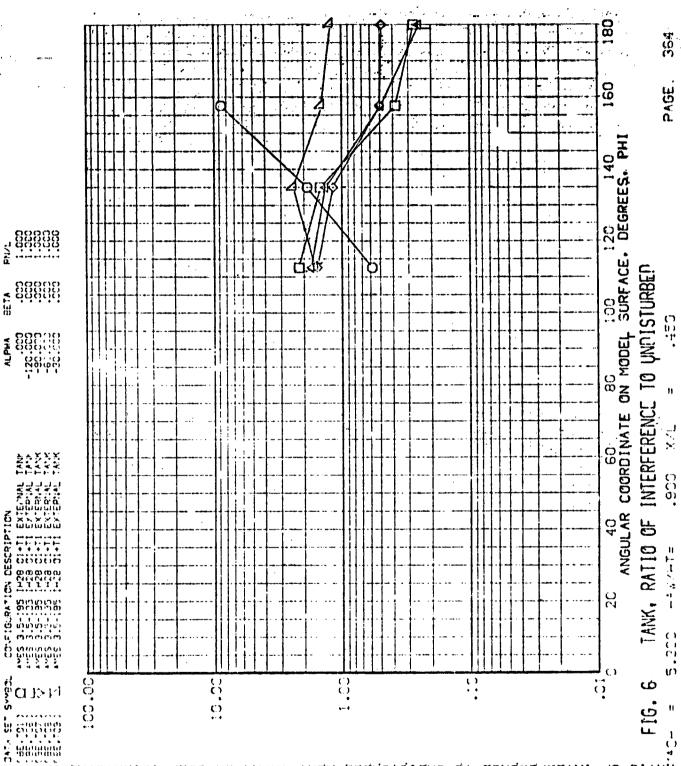


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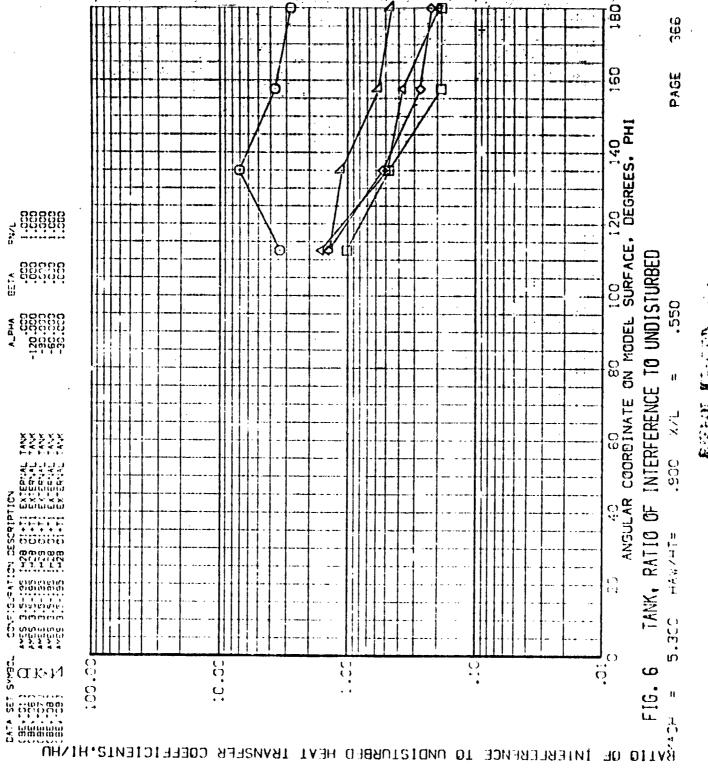
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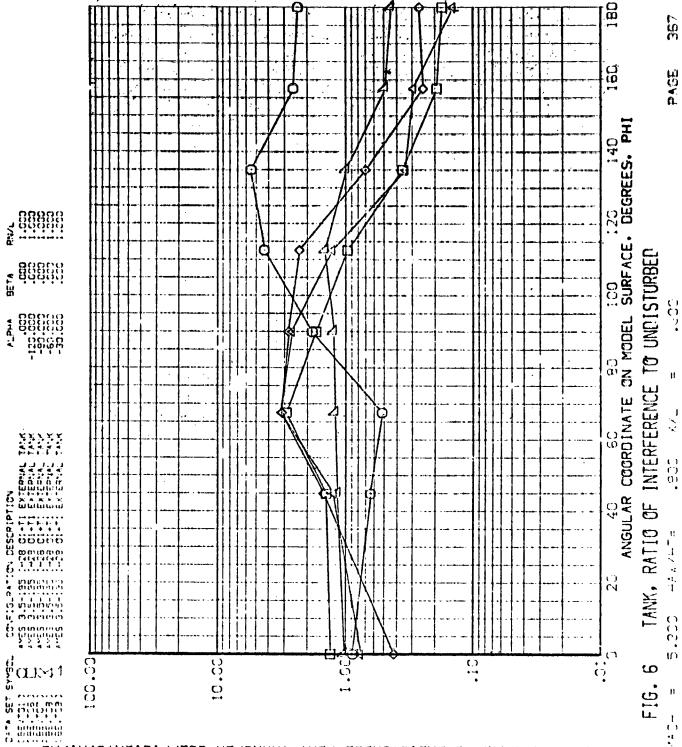




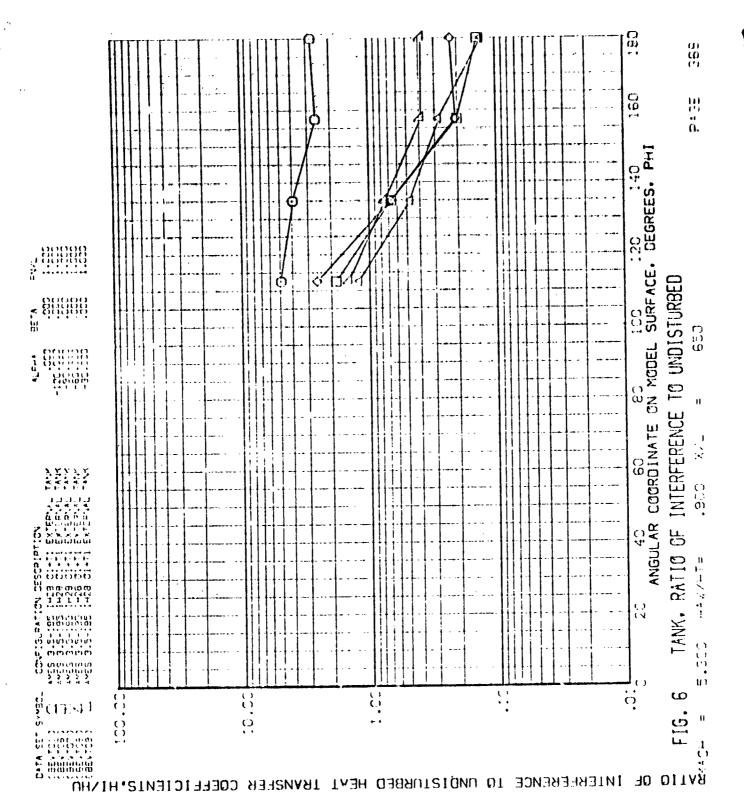
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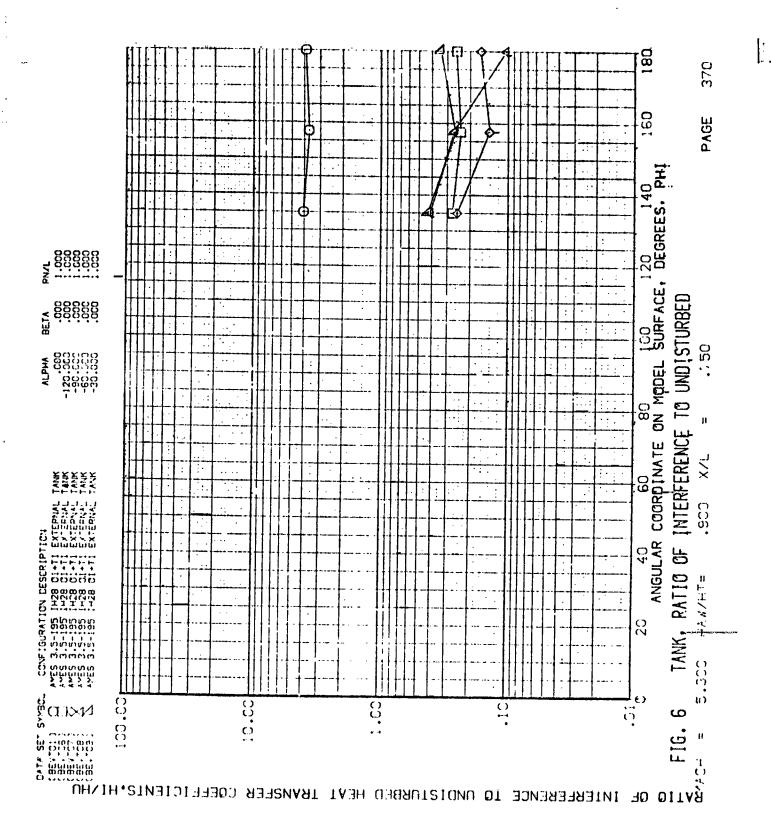
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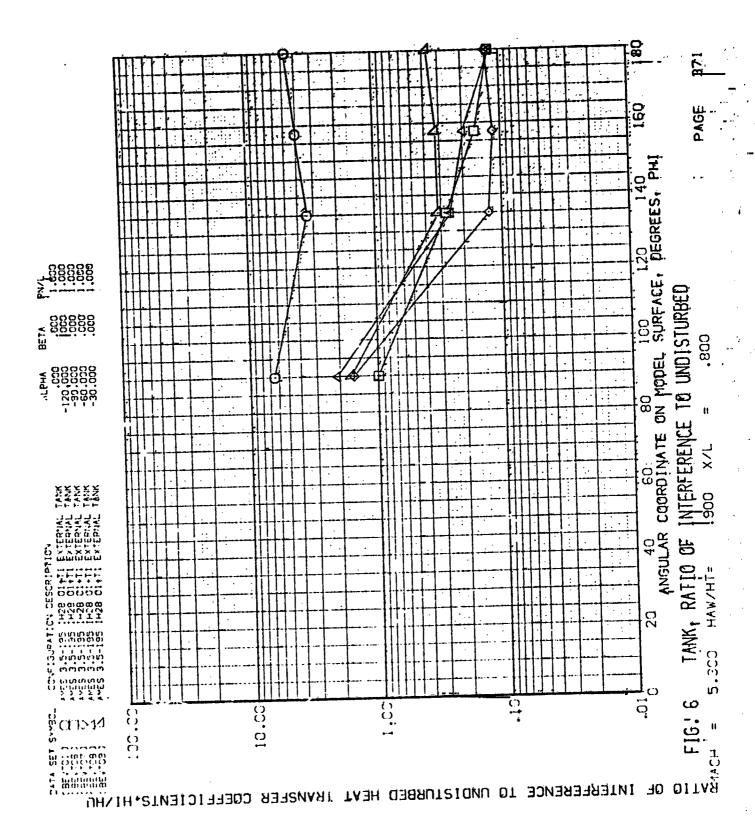




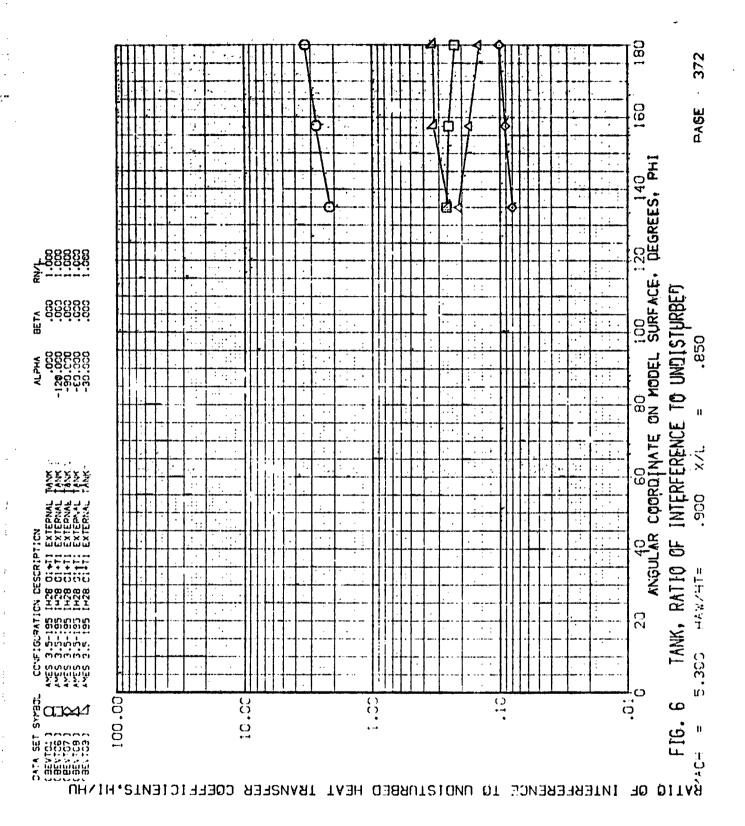
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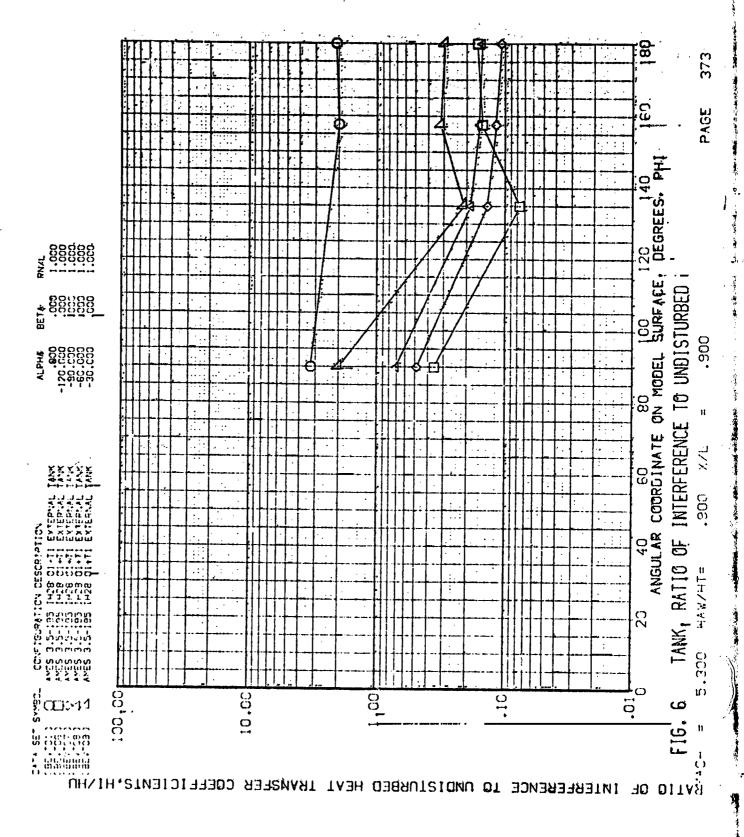






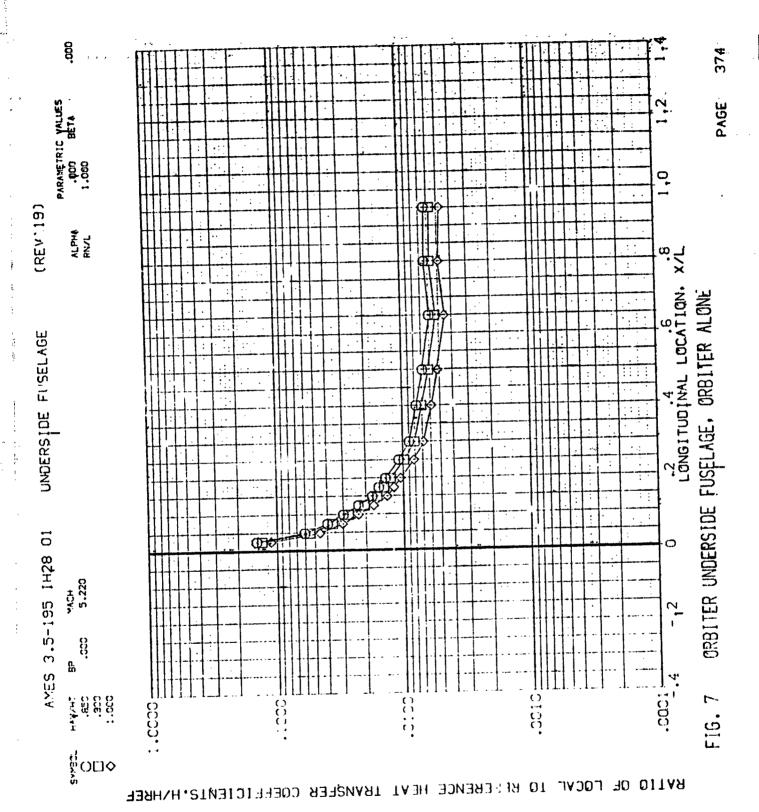
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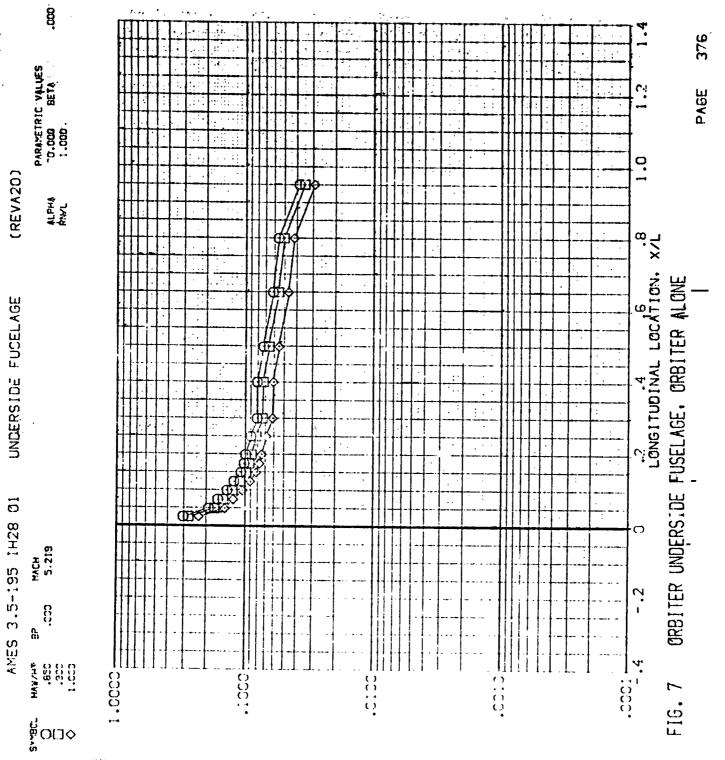
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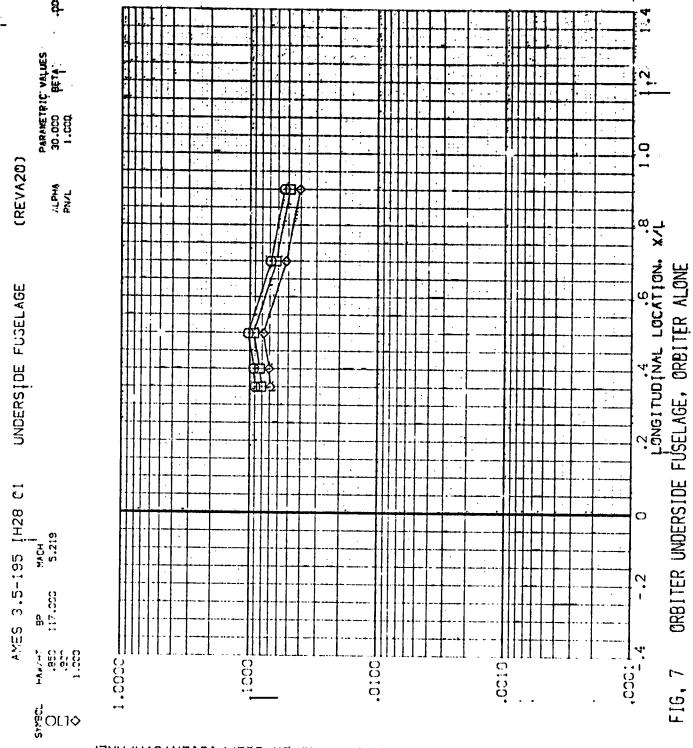


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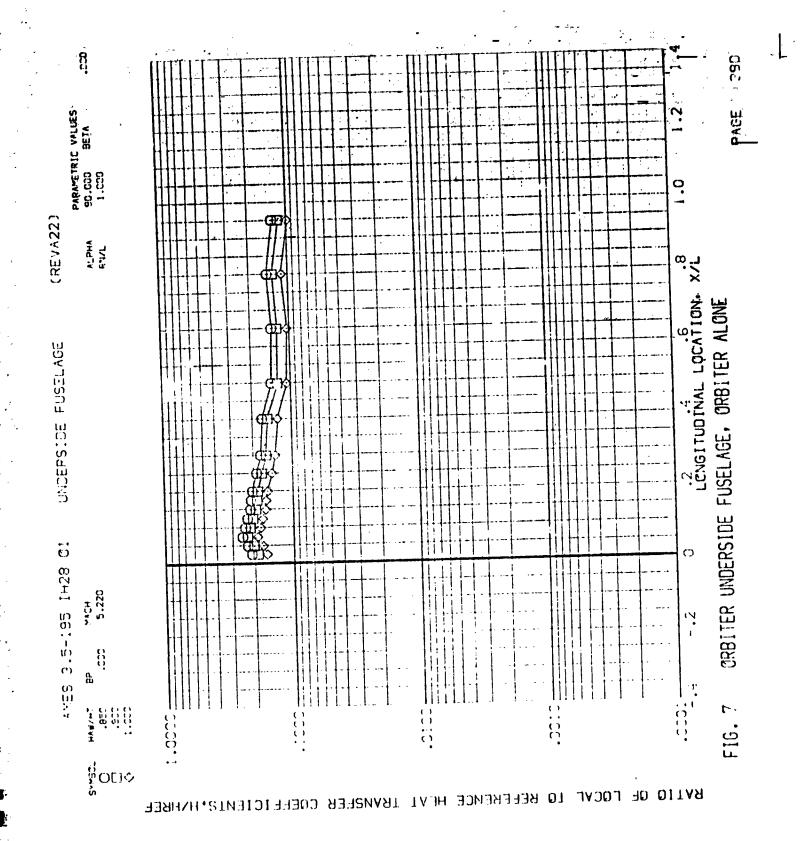
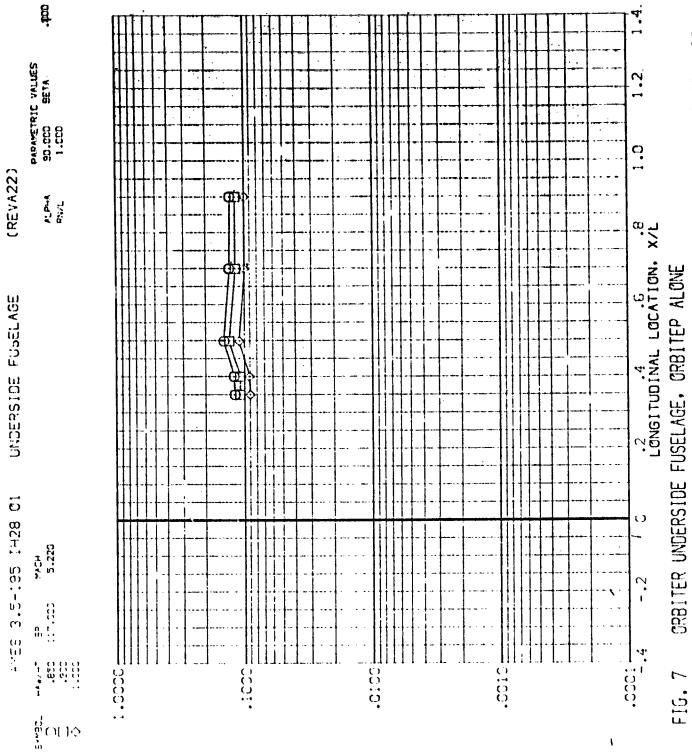


FIG. 7



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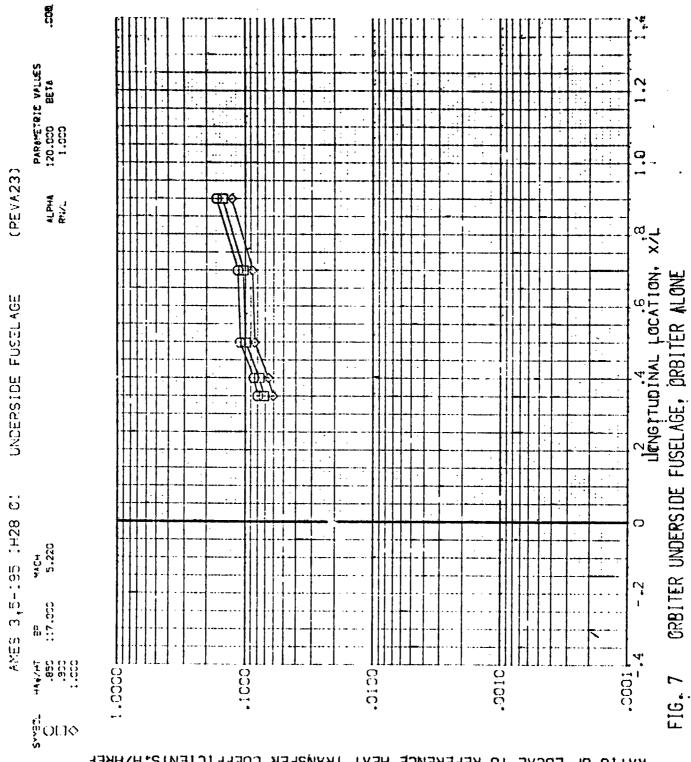
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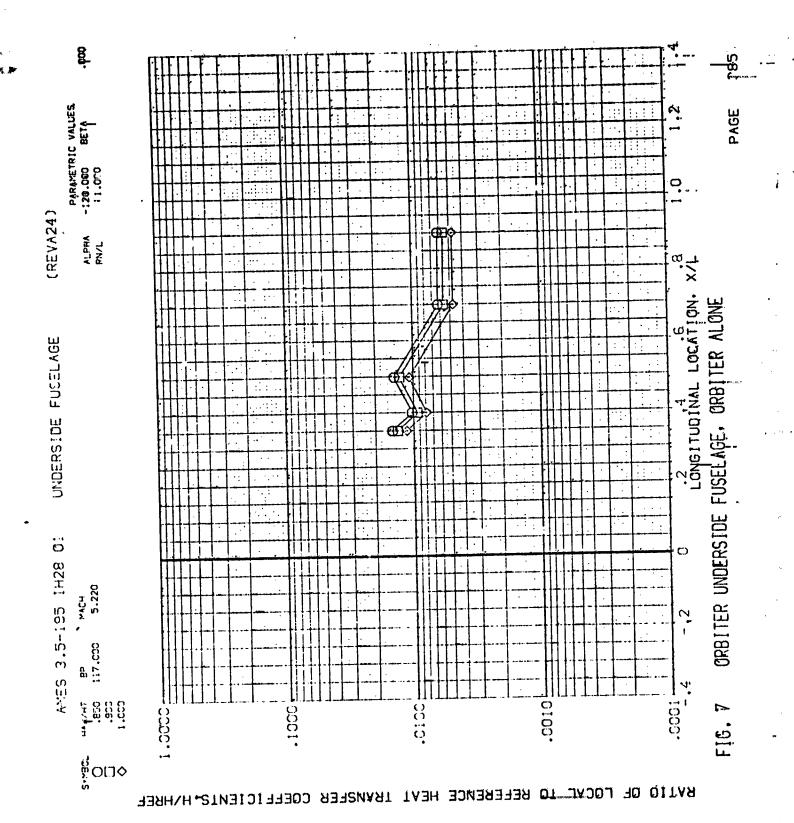
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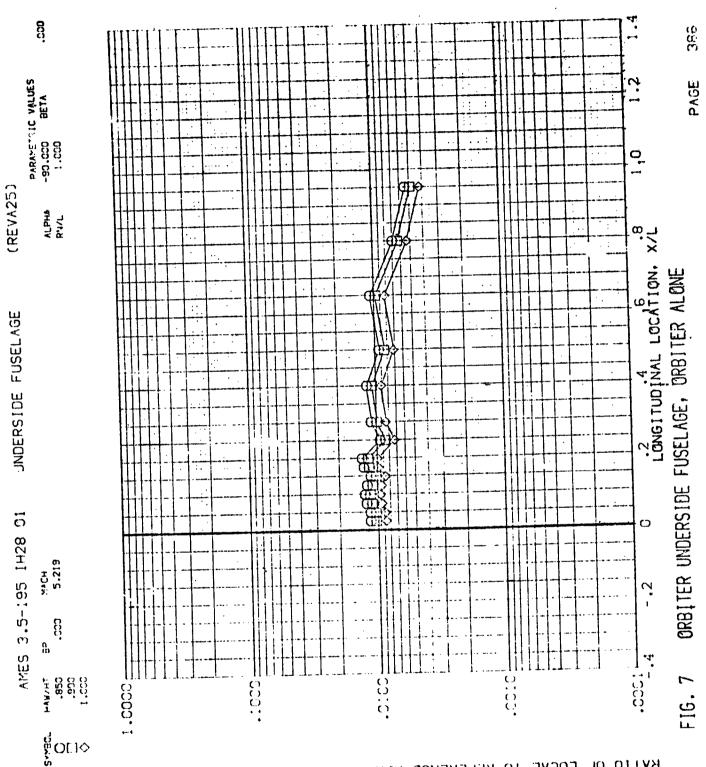
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FIG. 7

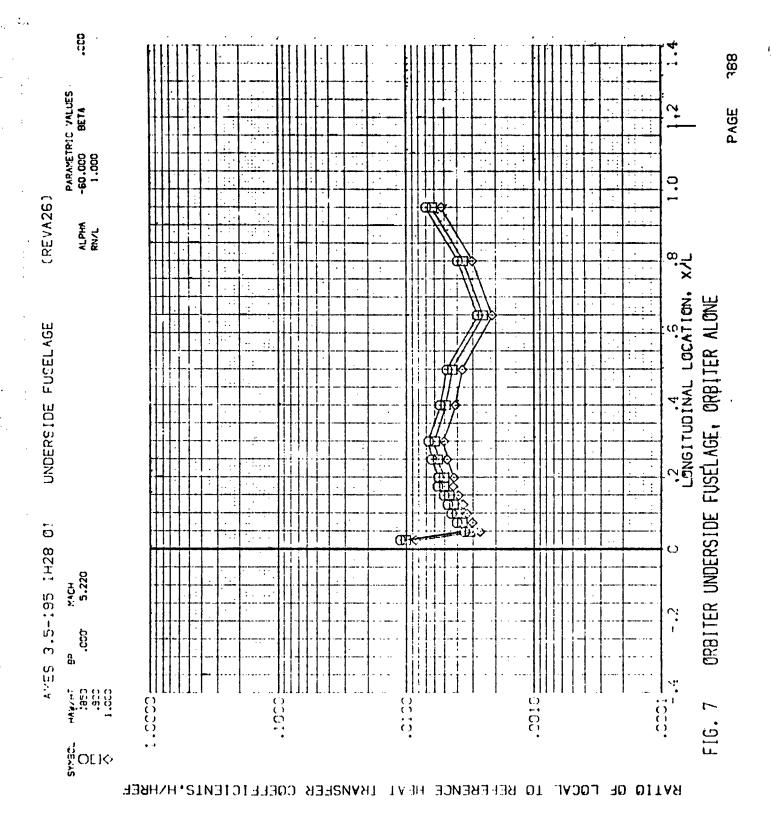


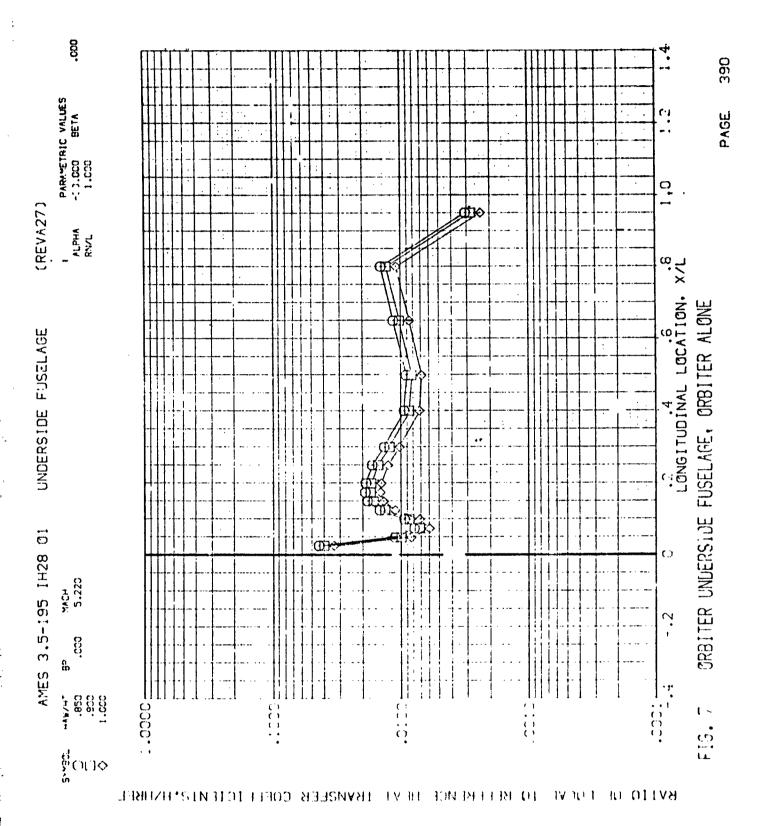
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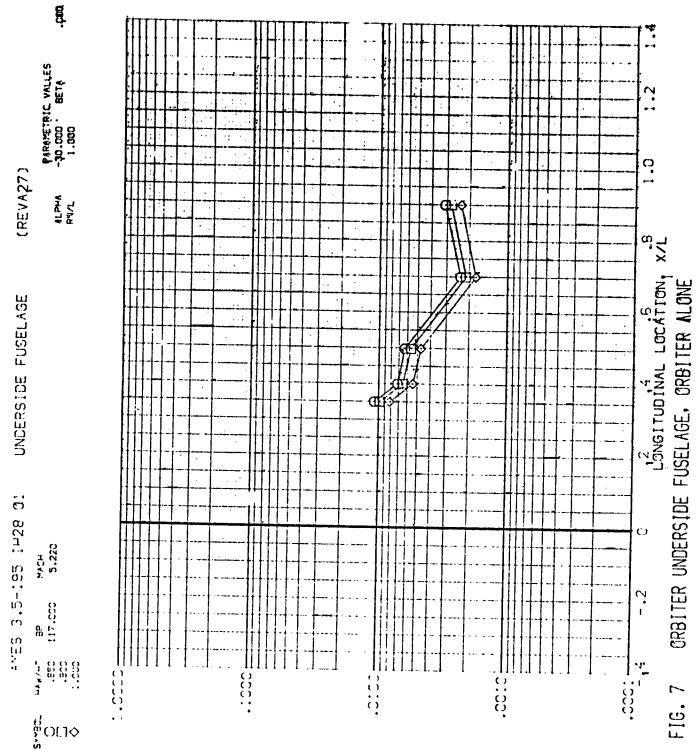




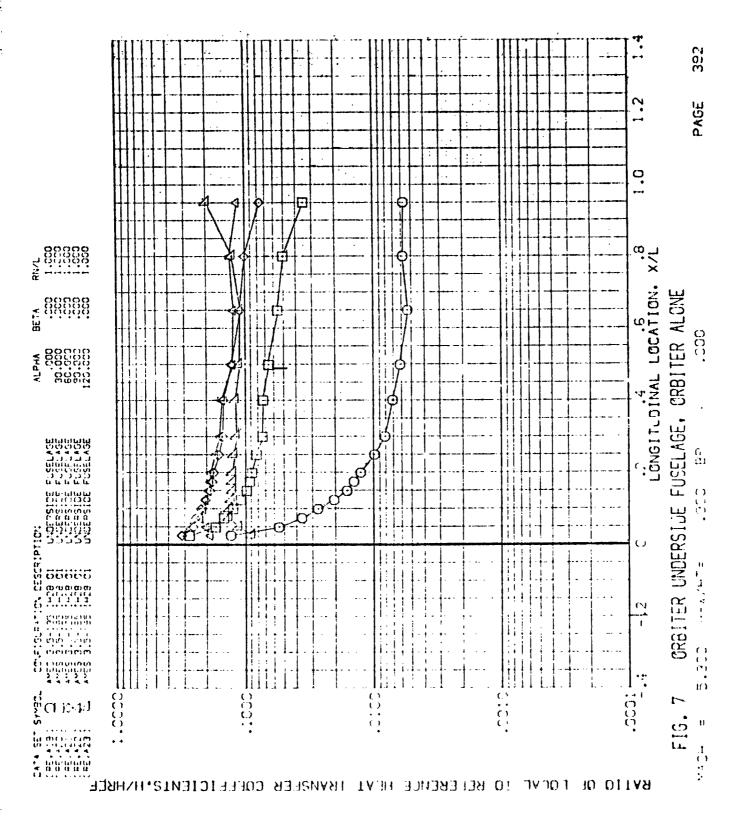
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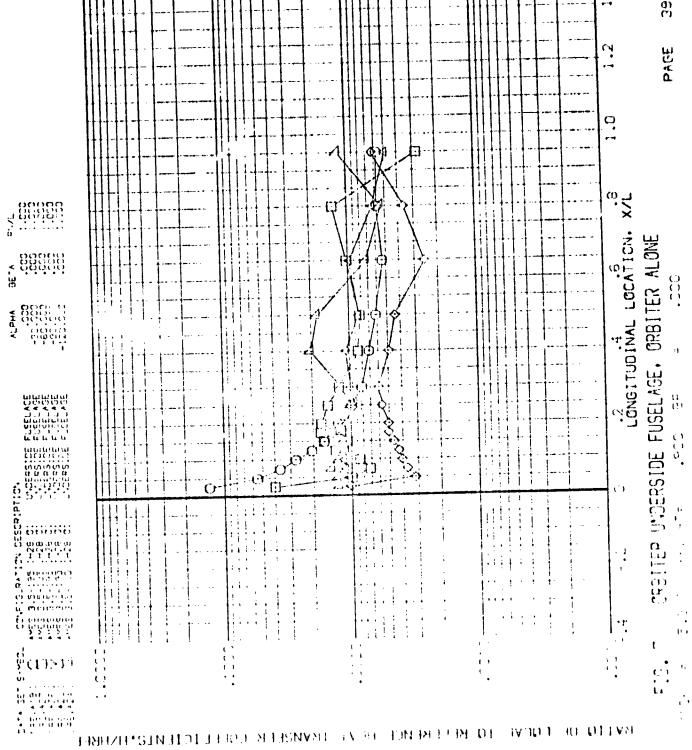


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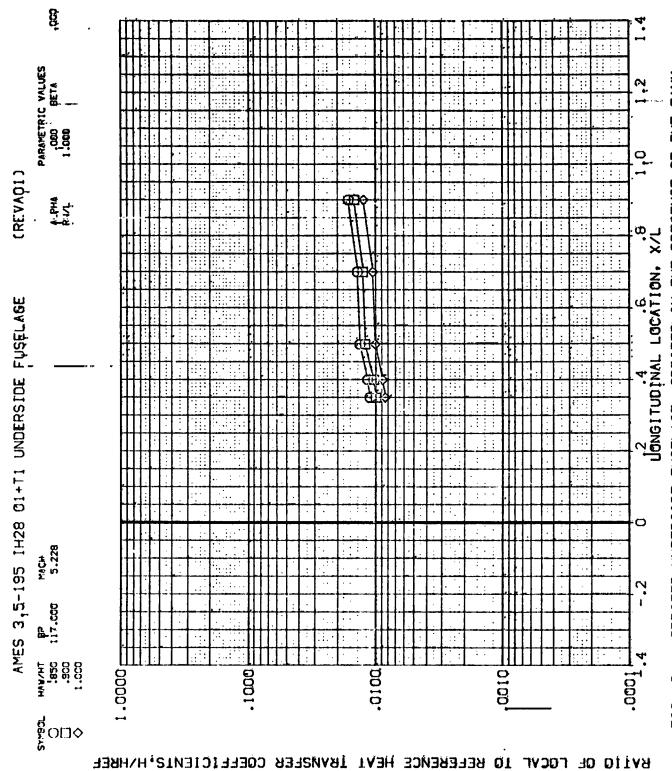
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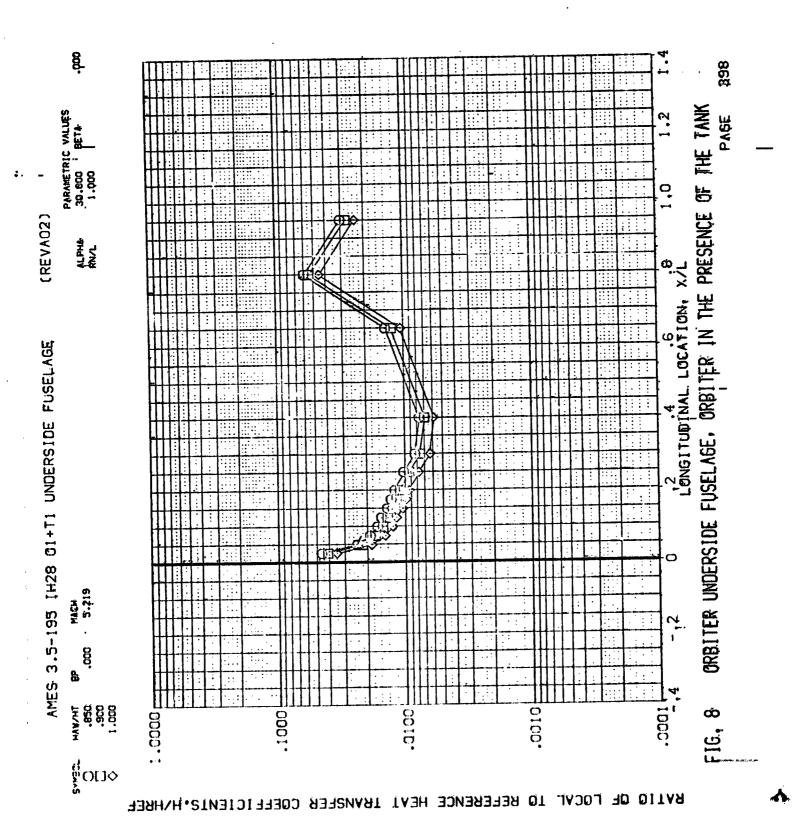
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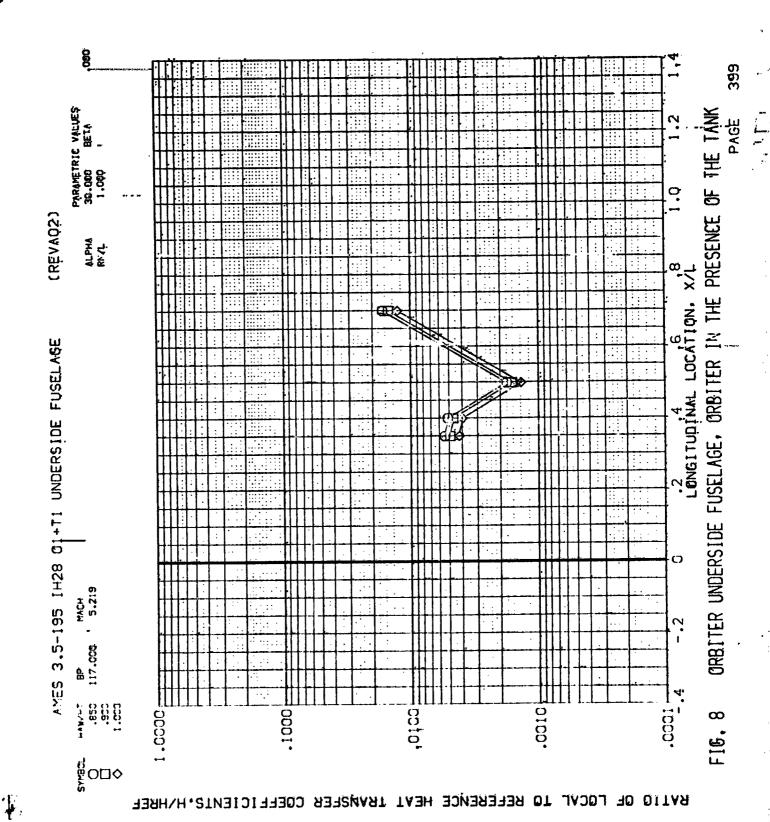
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ORBITER UNDERSIDE FUSELAGE, ORBITER IN THE PRESENCE OF THE TANK



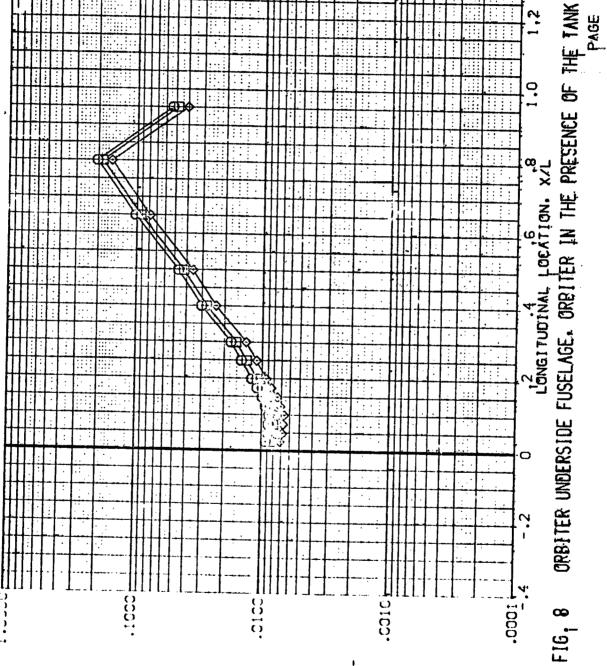
ORBITER UNDERSIDE FUSELAGE, ORBITER IN THE PRESENCE OF THE TANK FIG. 8

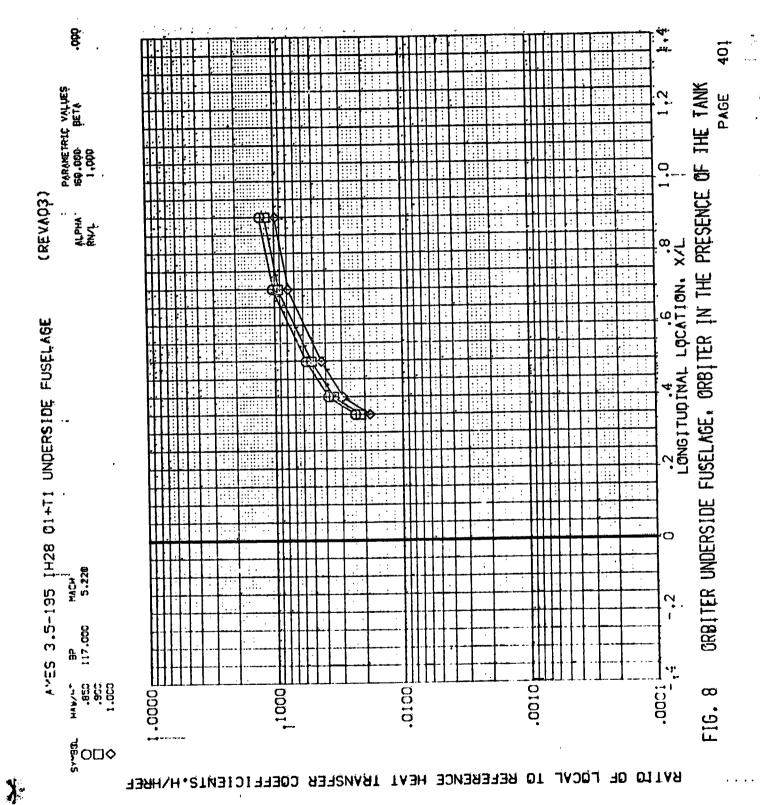


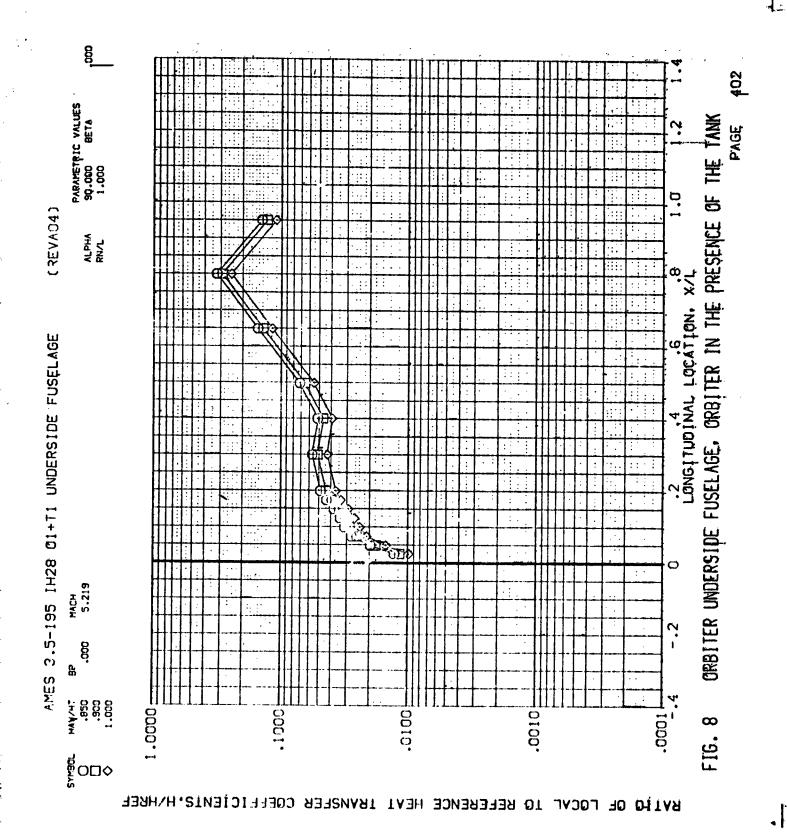


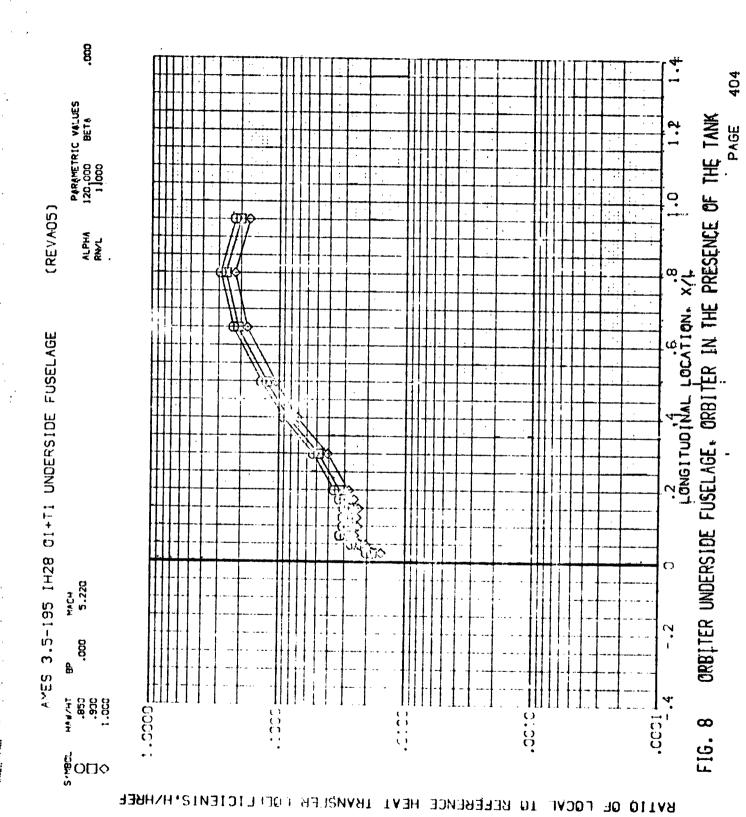
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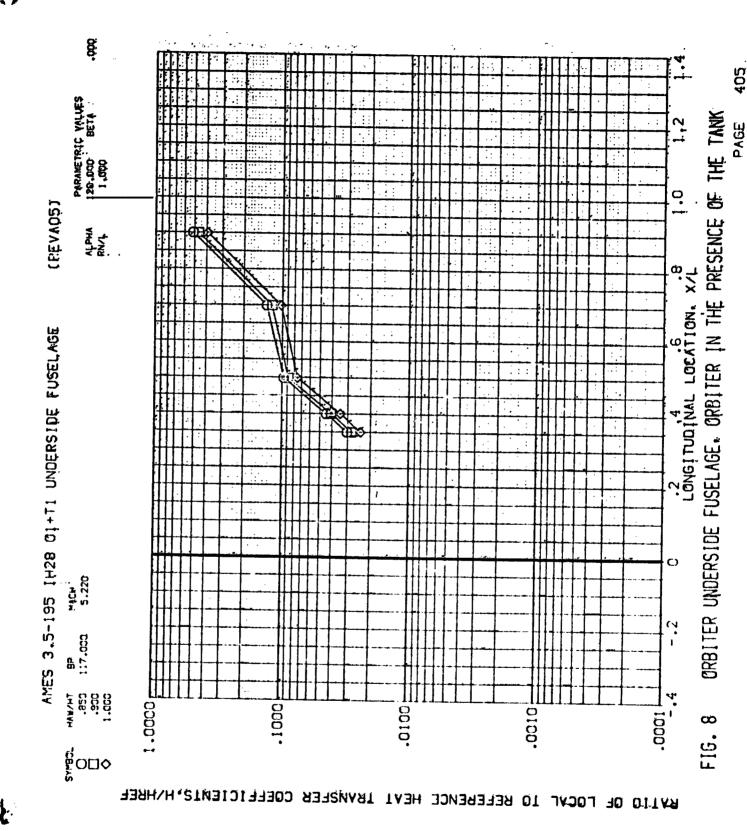
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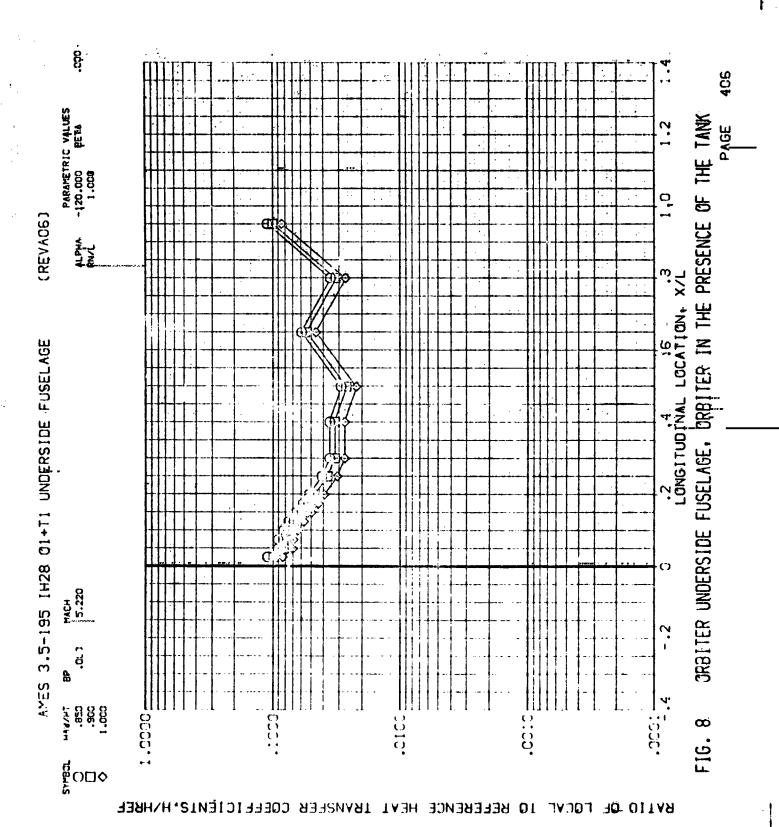


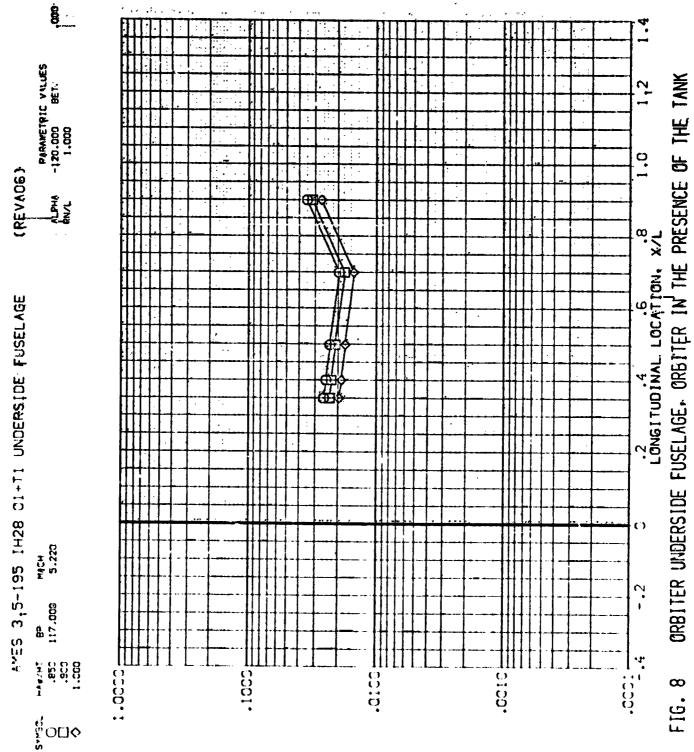












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FIG. 8

RATIO OF LOCAL TO REFERENCE HEAT TRANSFER COEFFICIENTS, HAHREF

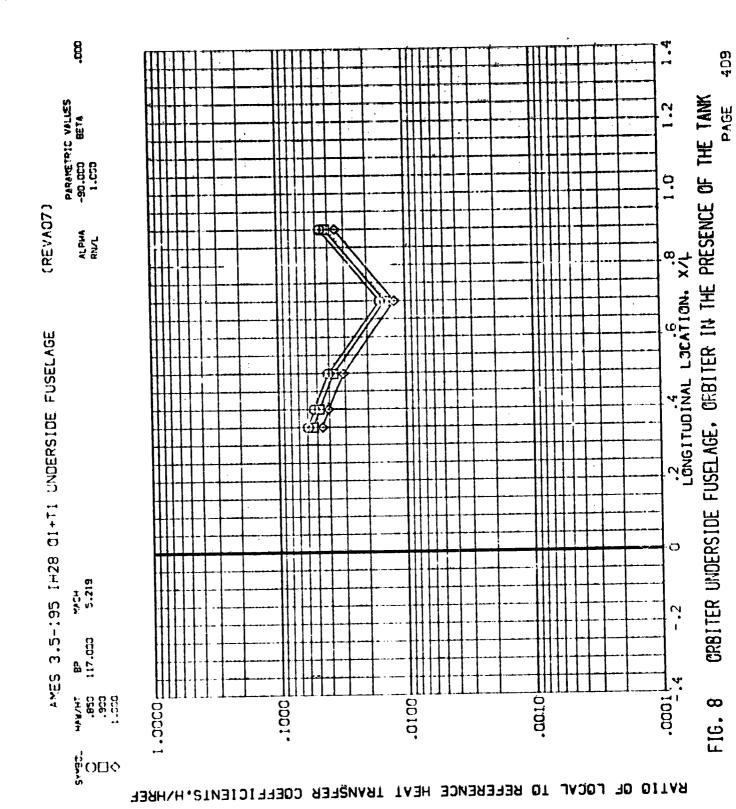
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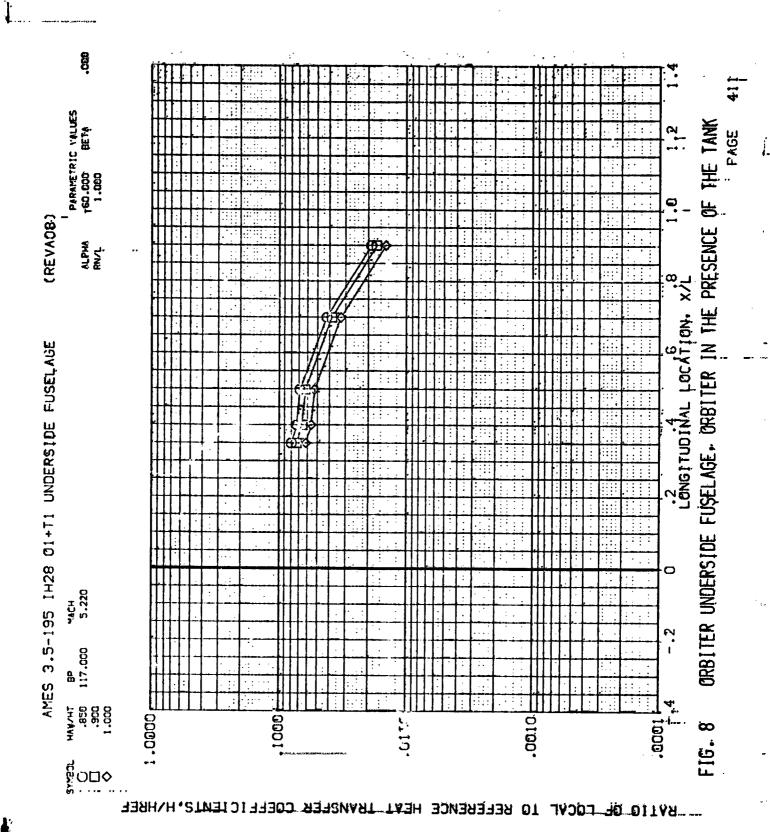
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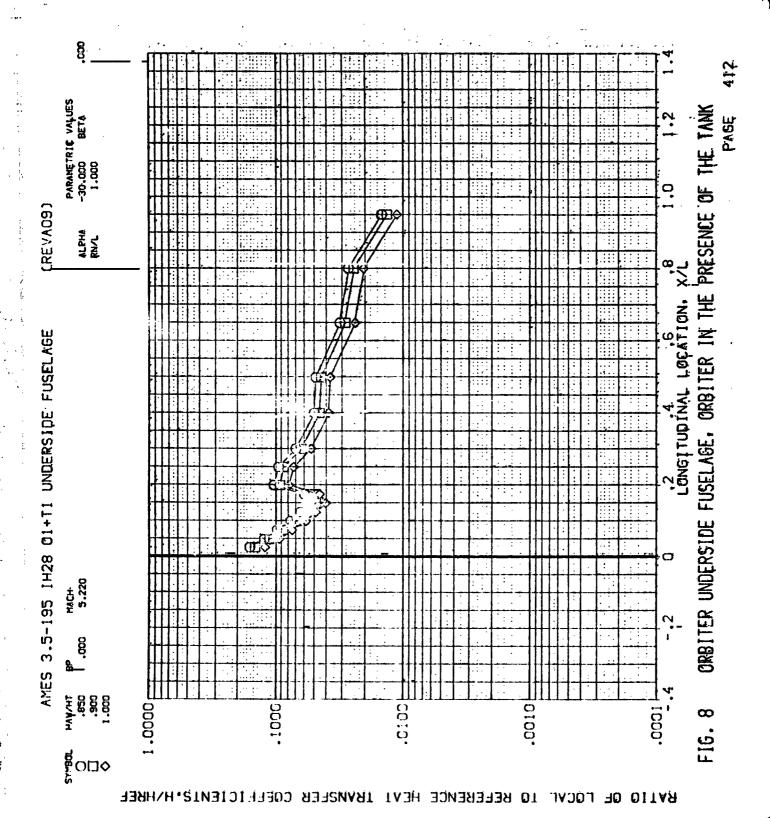
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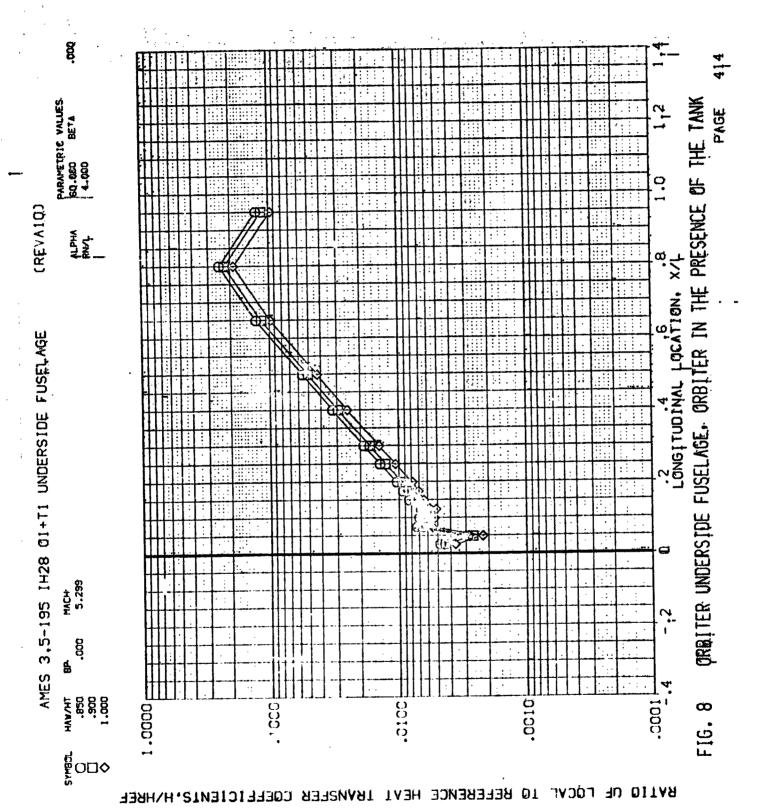
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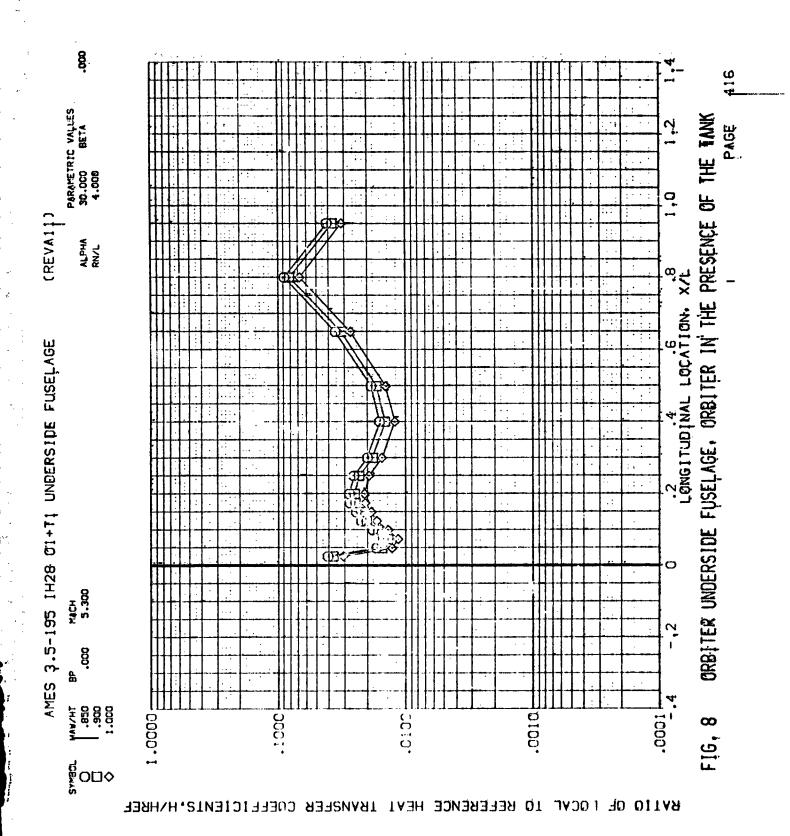
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FIG. 8







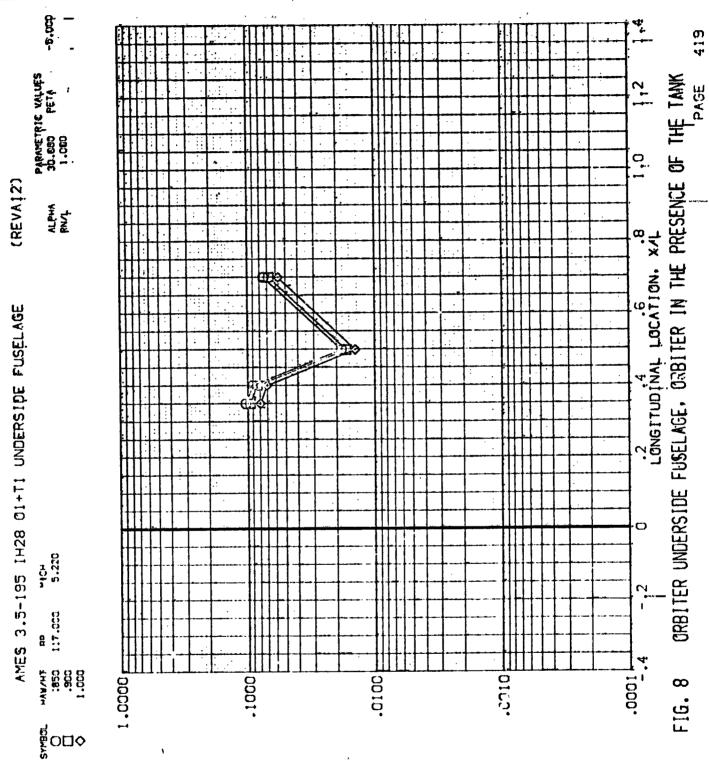


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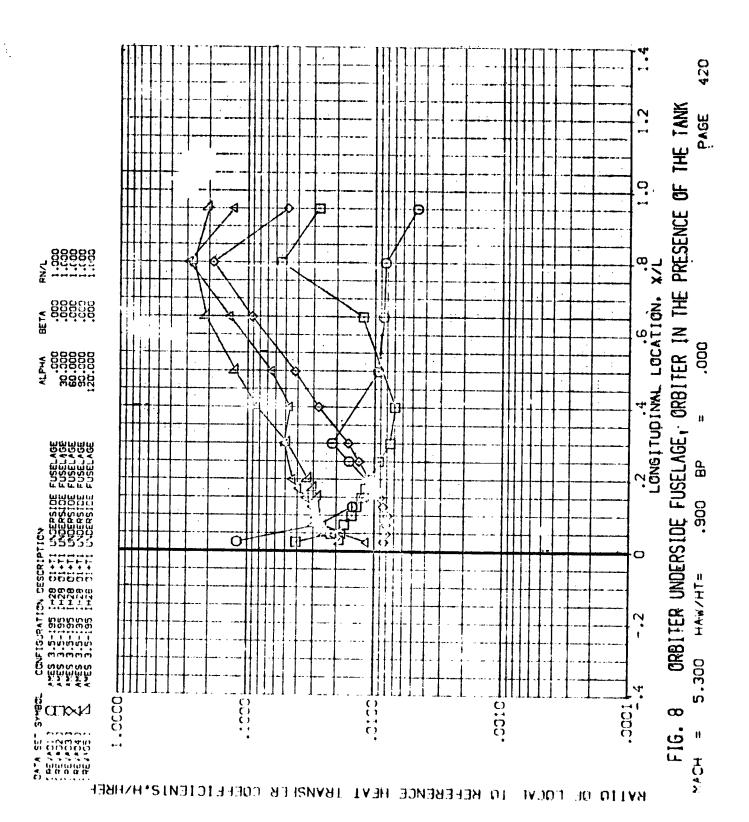
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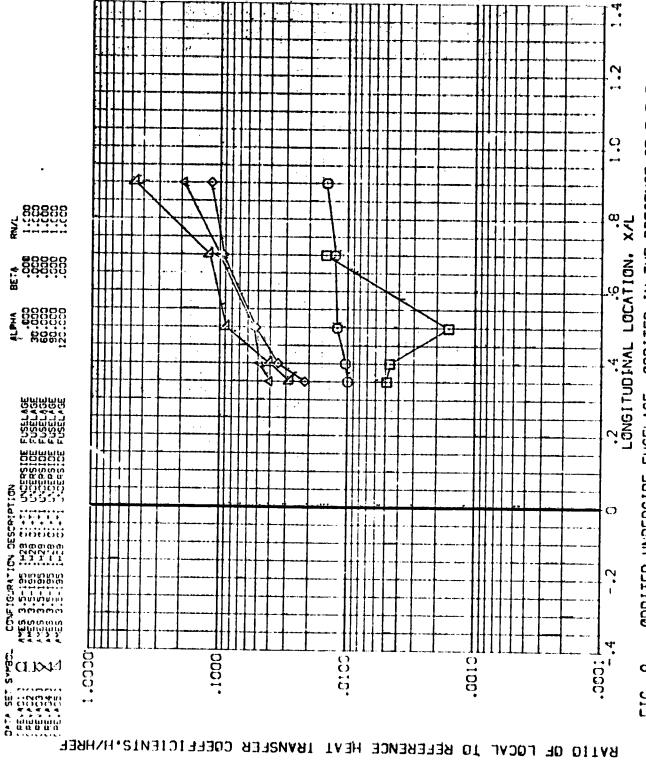
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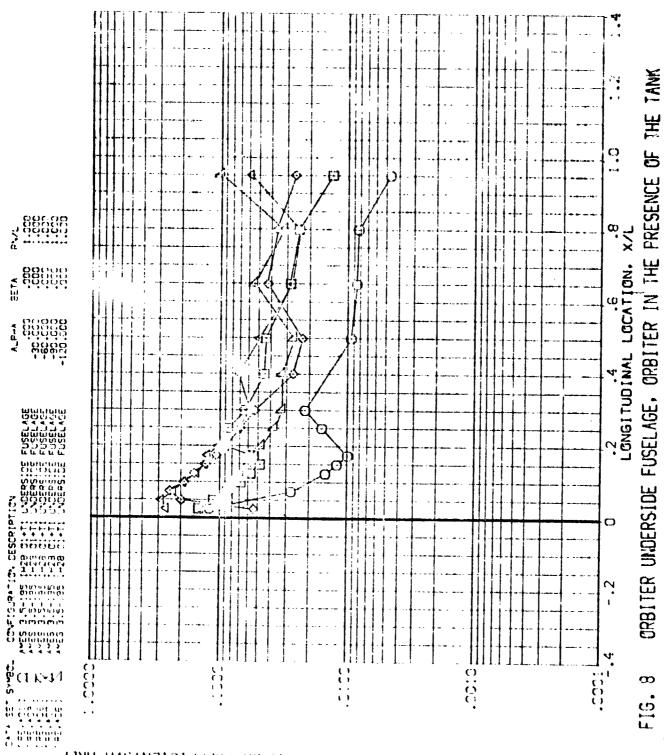
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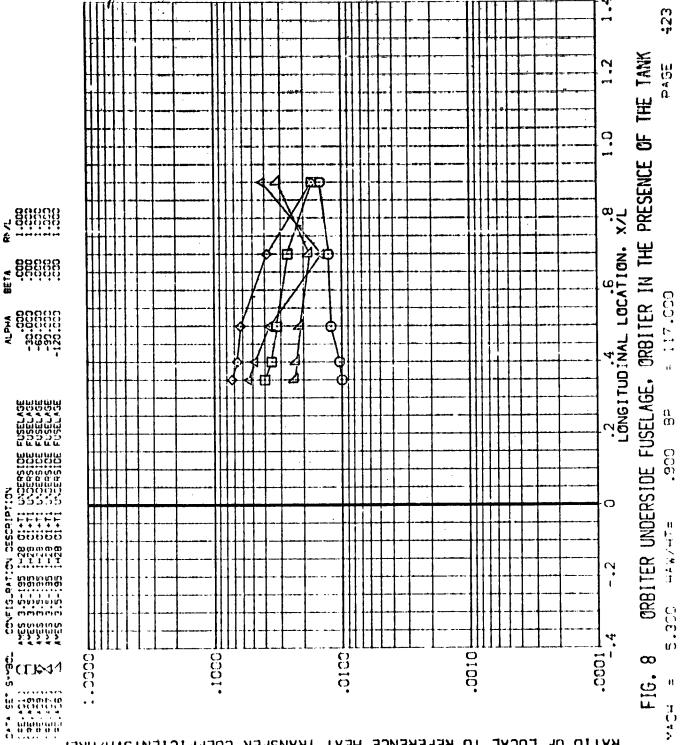
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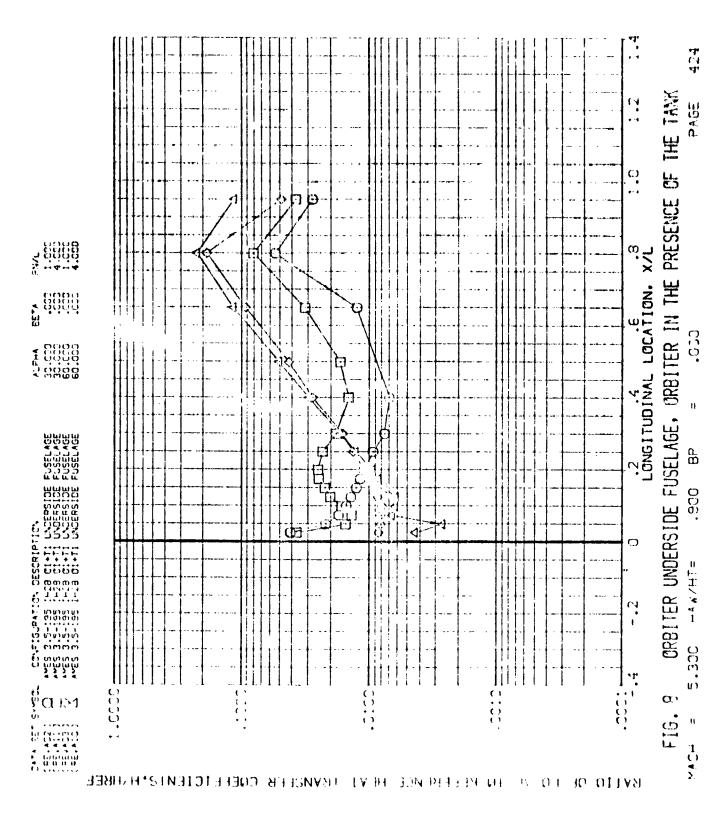


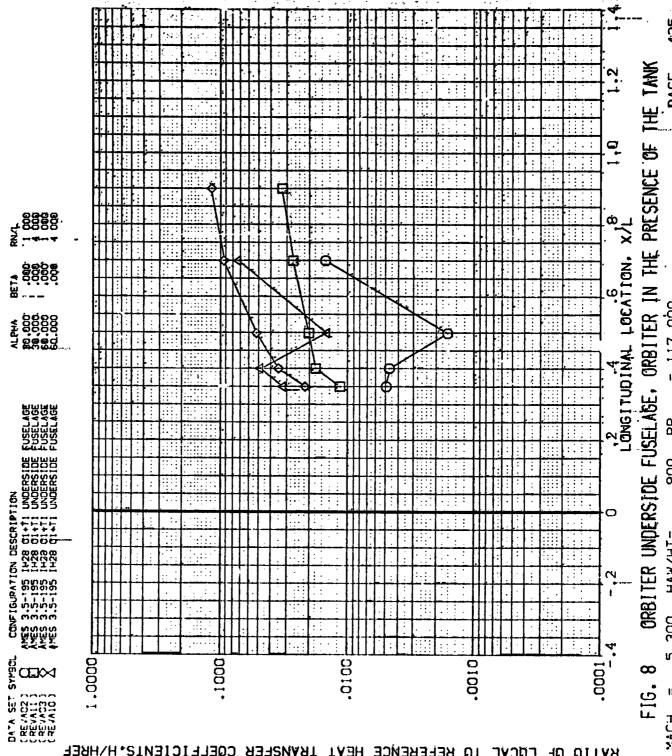
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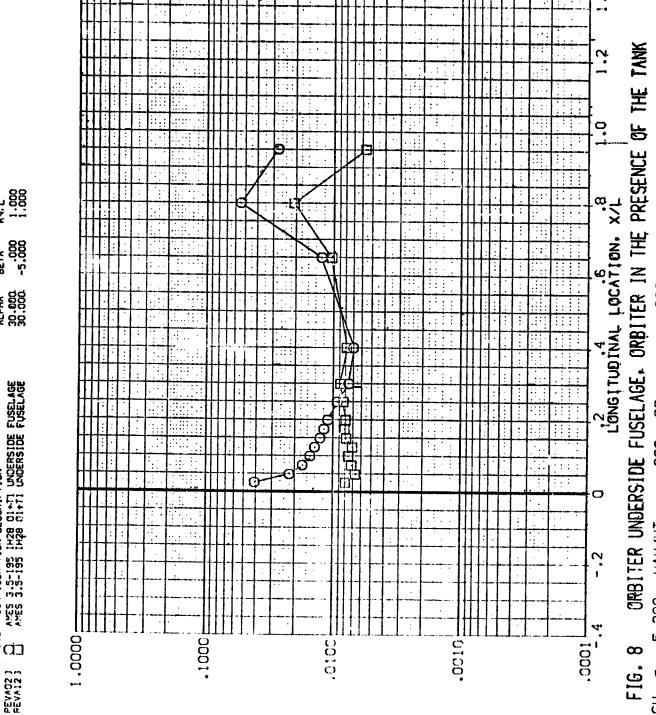
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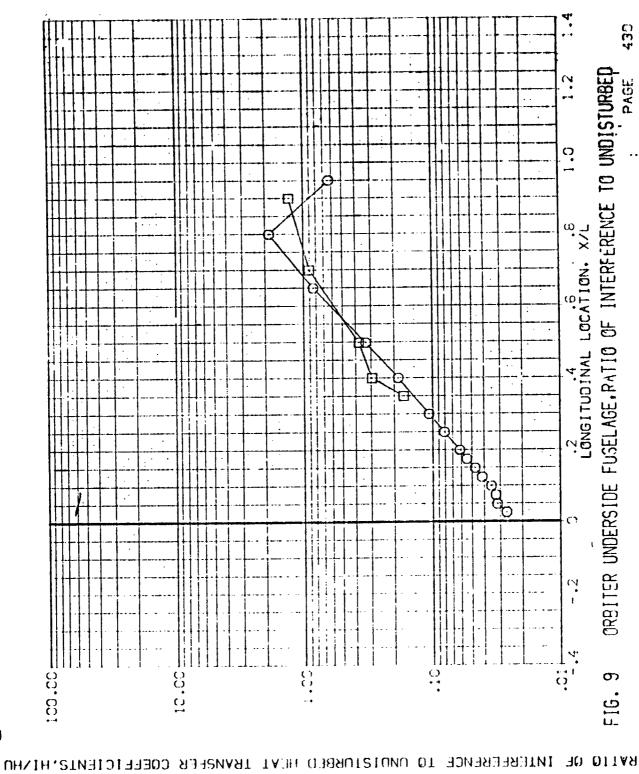
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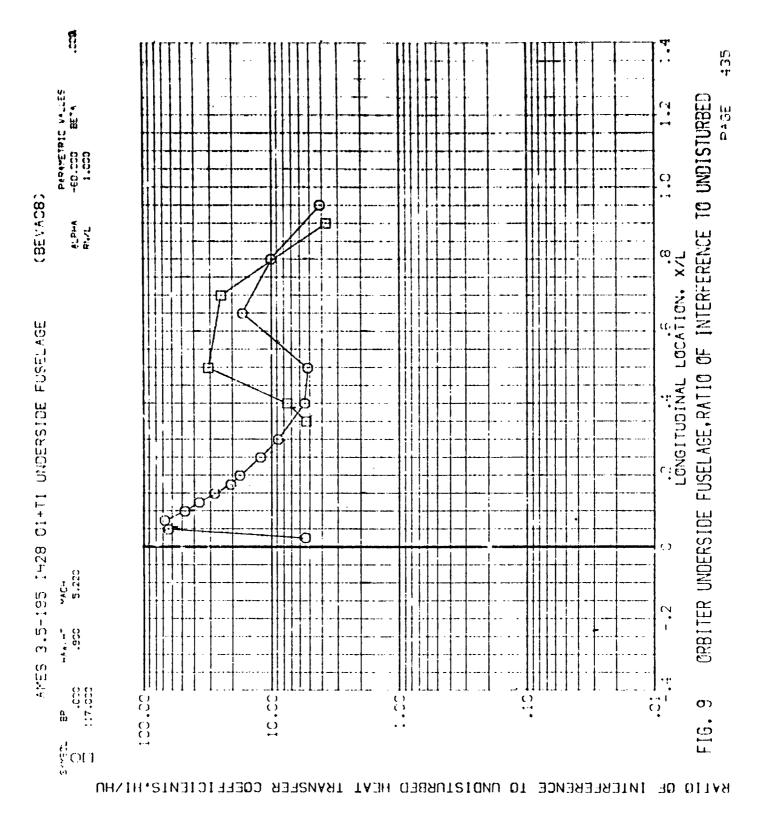
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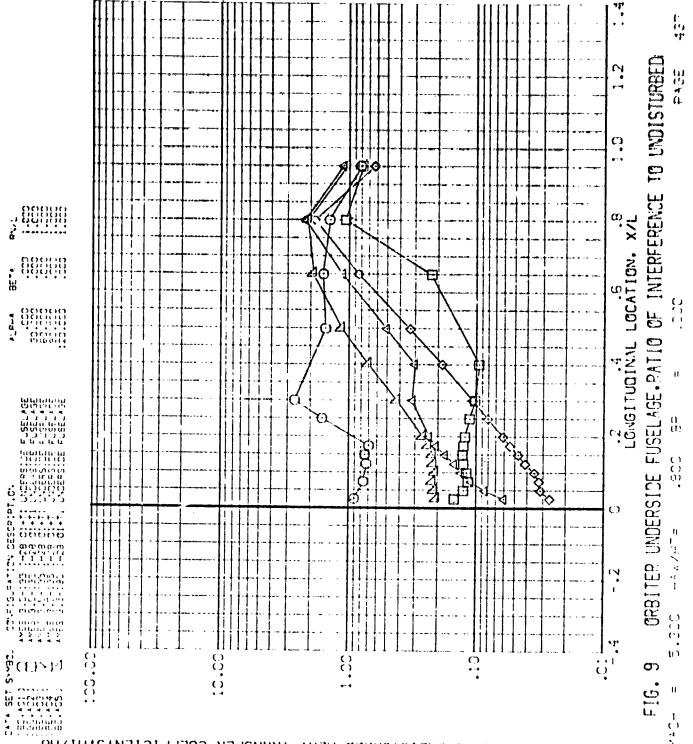
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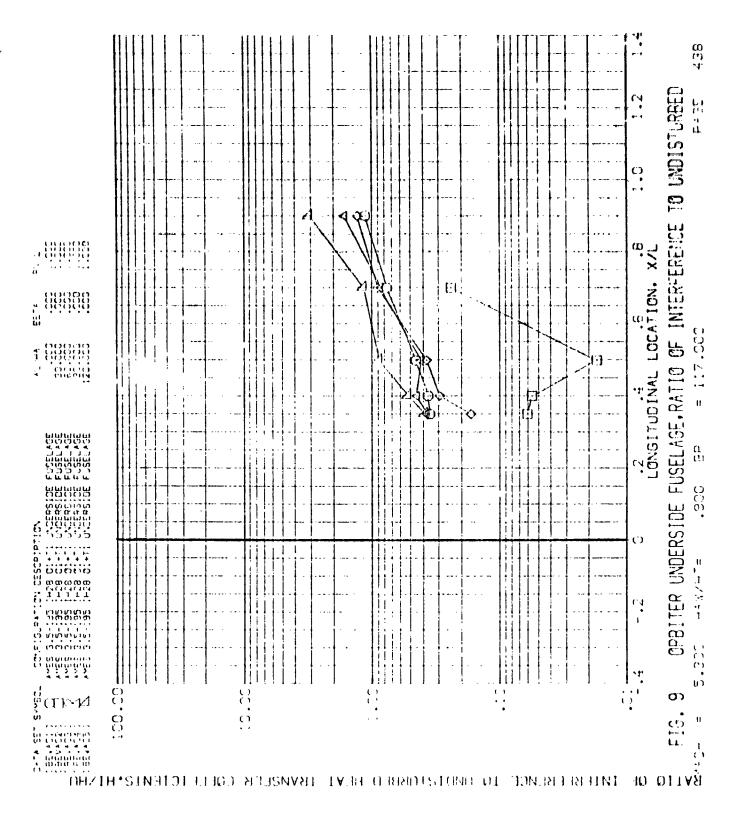
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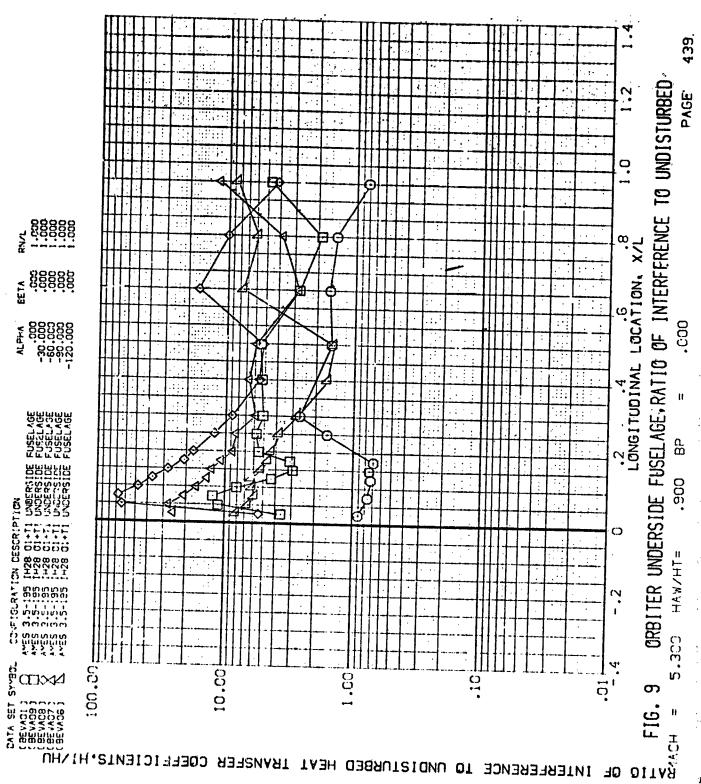
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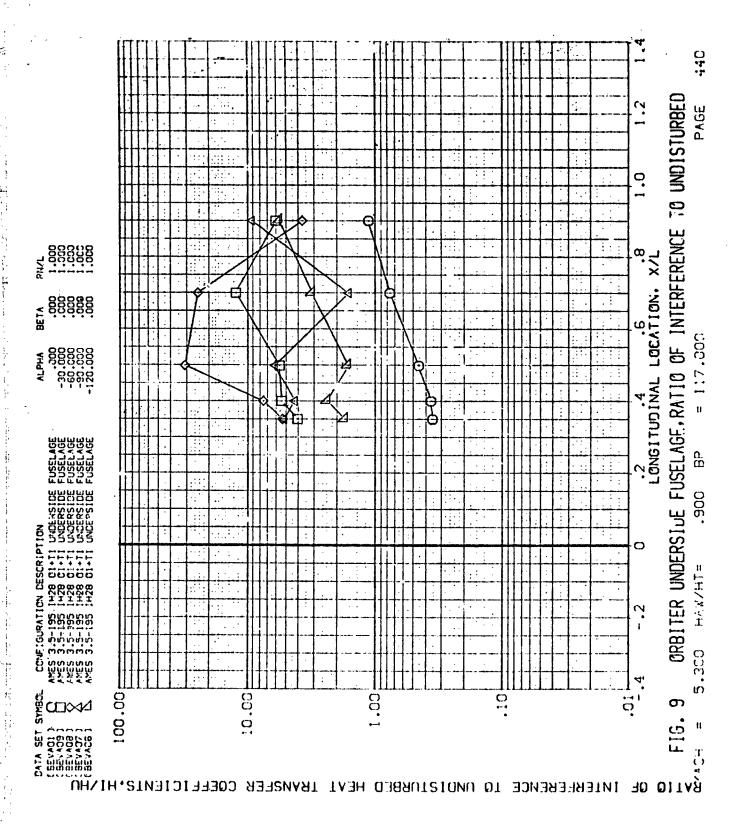


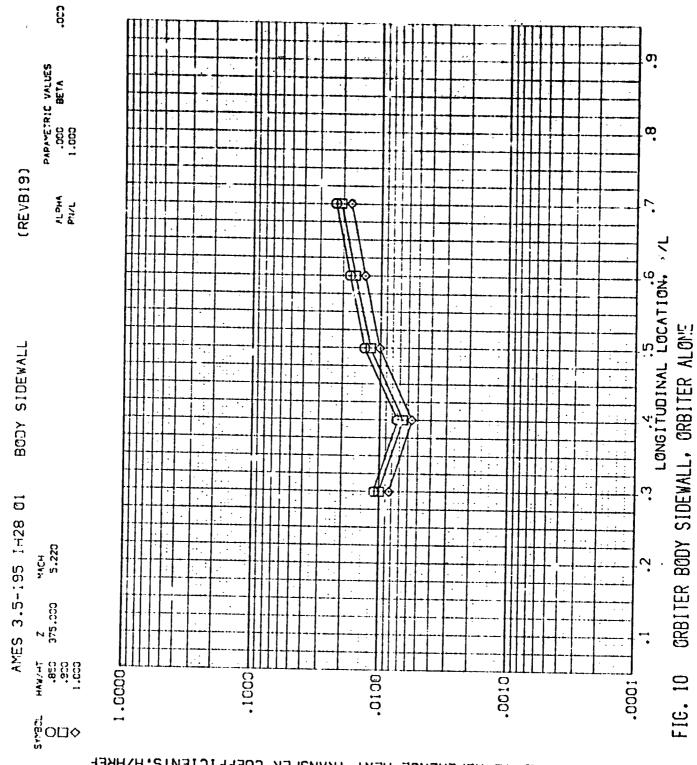
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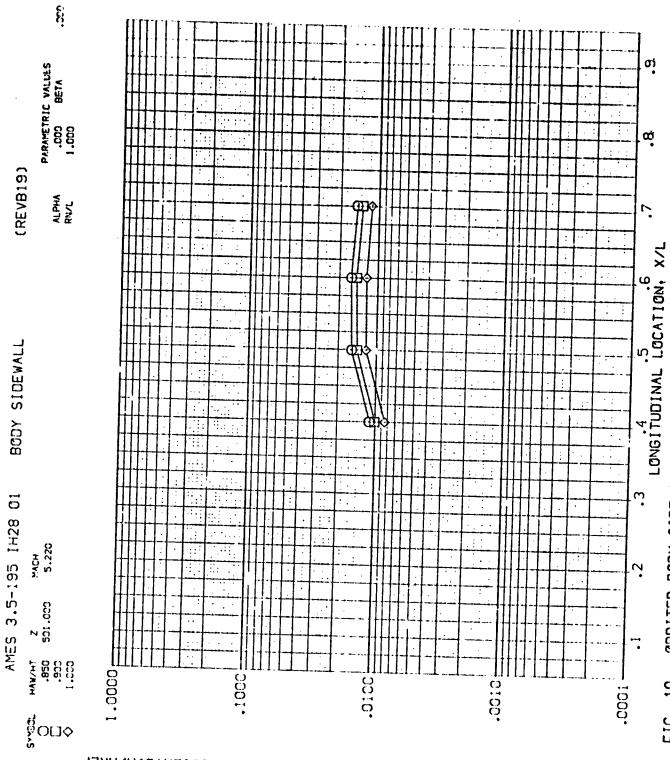


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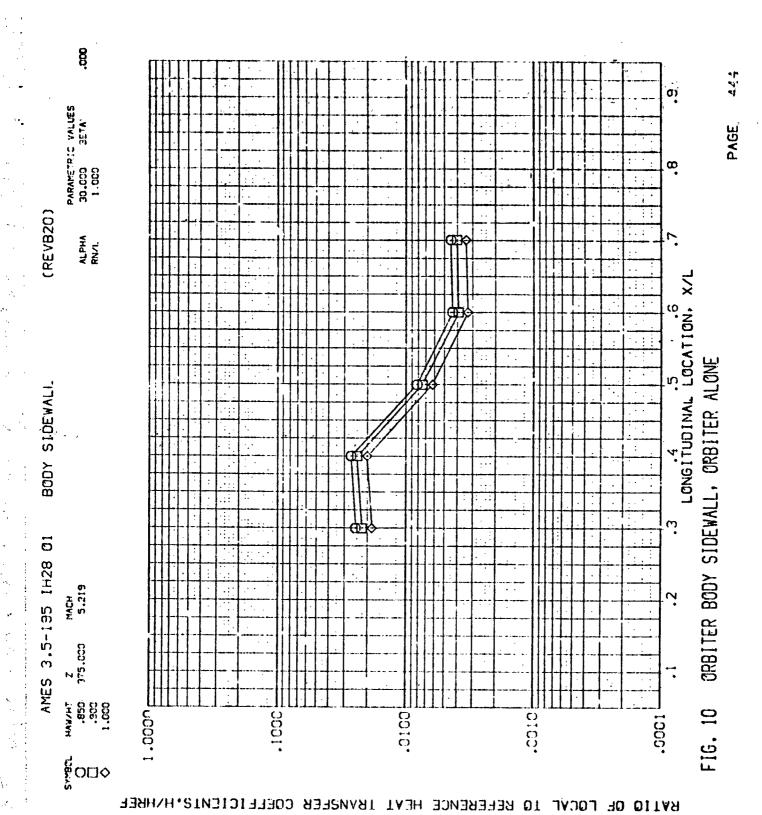
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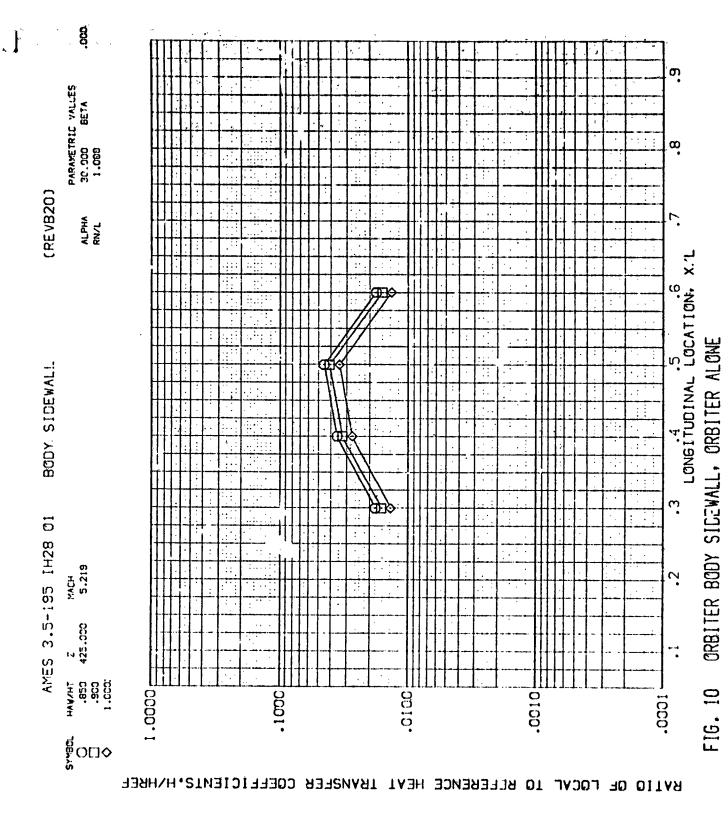
ORBITER BODY SIDEWALL, ORBITER ALONE

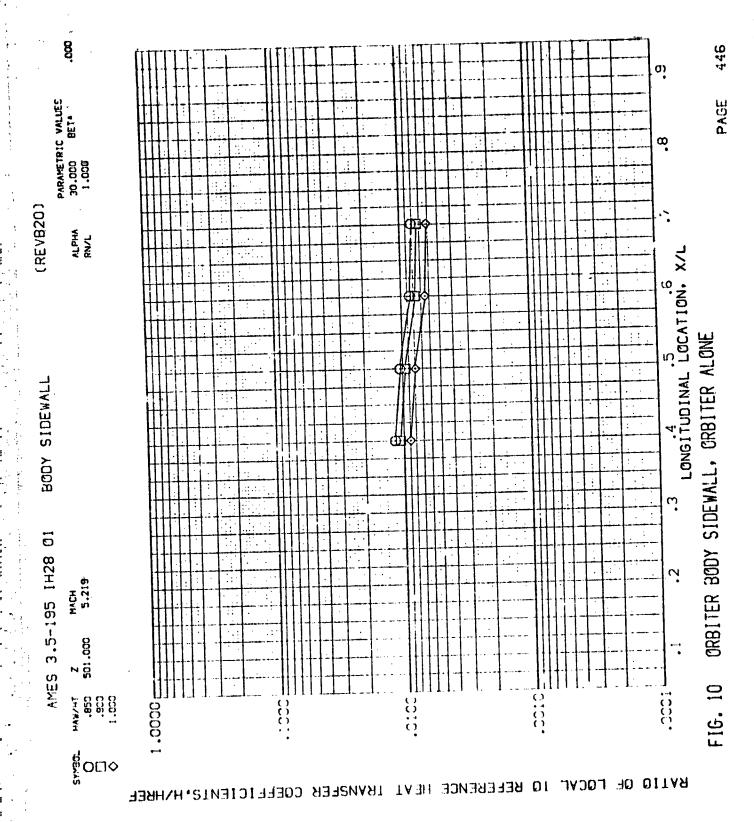
FIG. 10

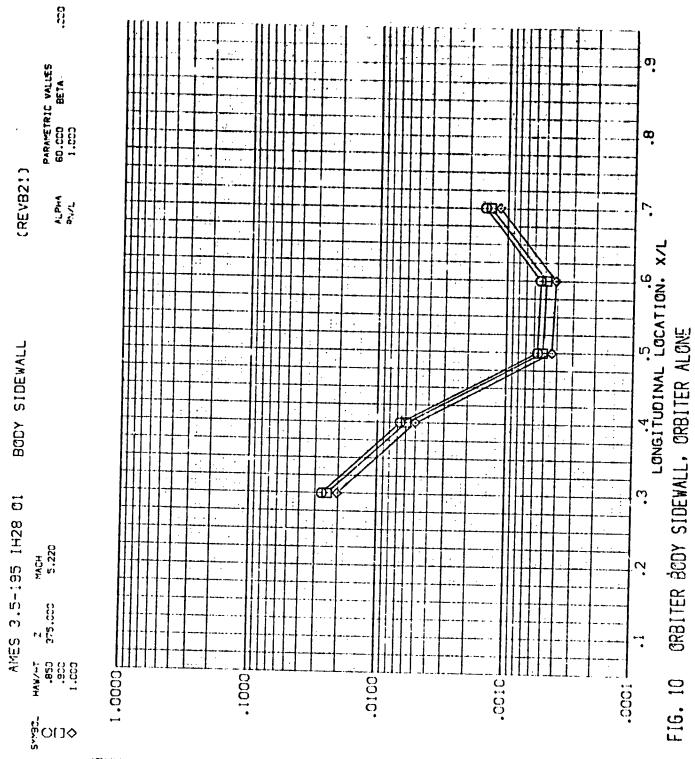


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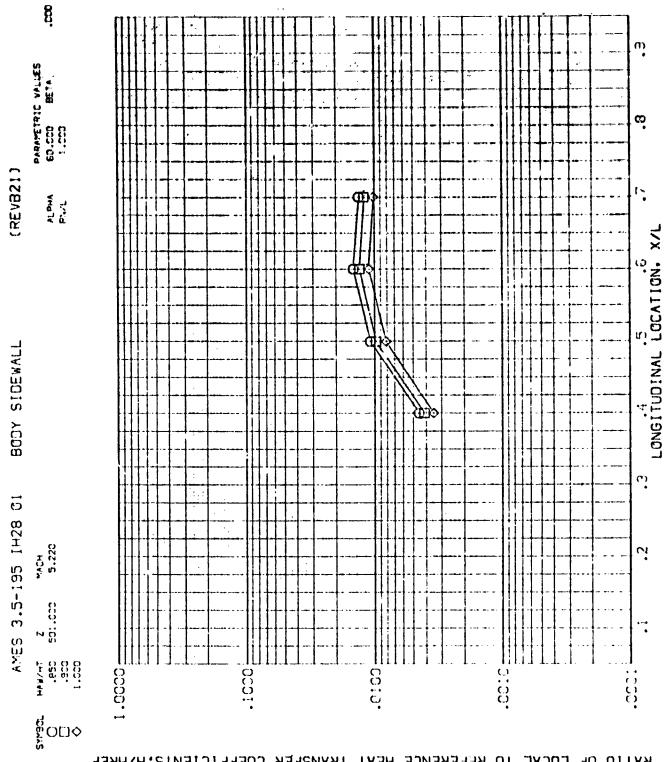


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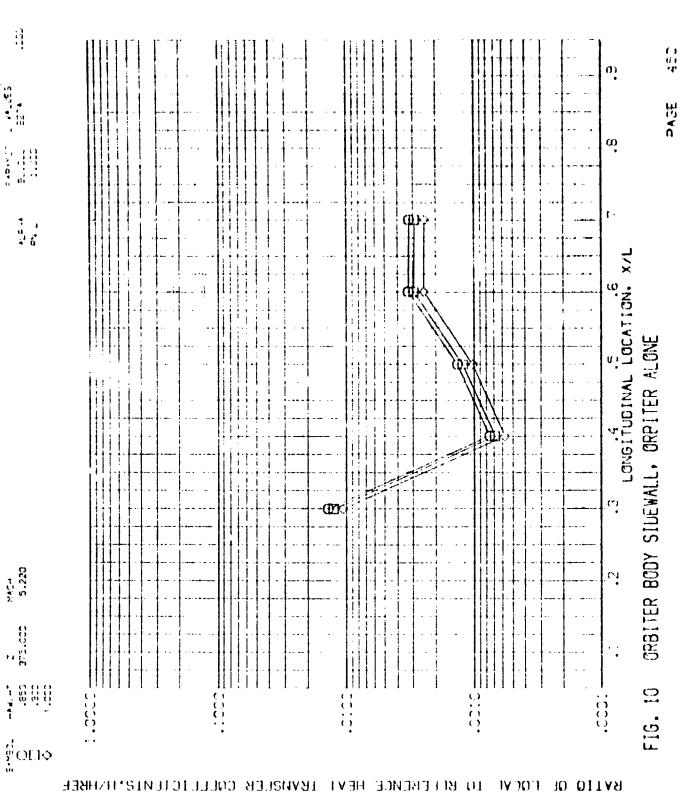
PAGE 448

GRBITER BODY SIDEWALL, ORBITER ALONE

FIG. 10

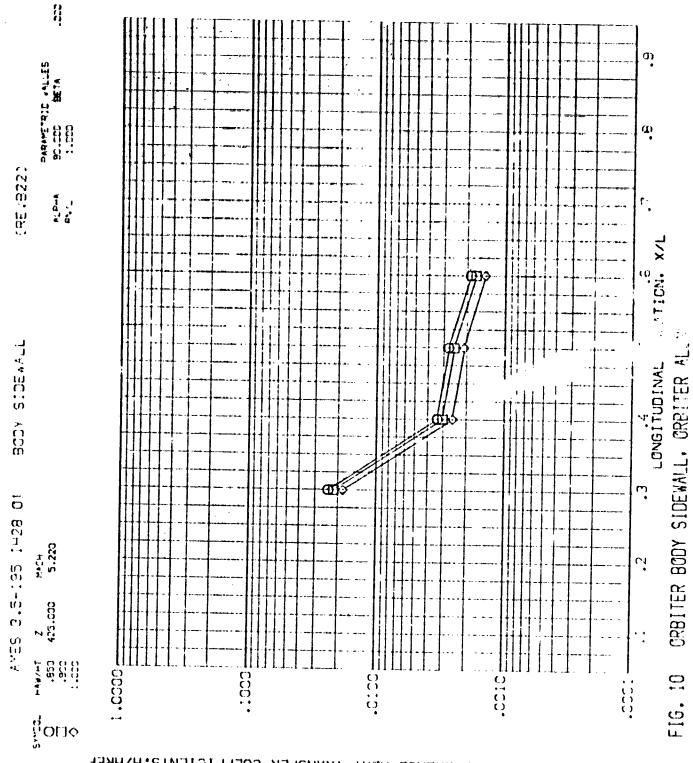


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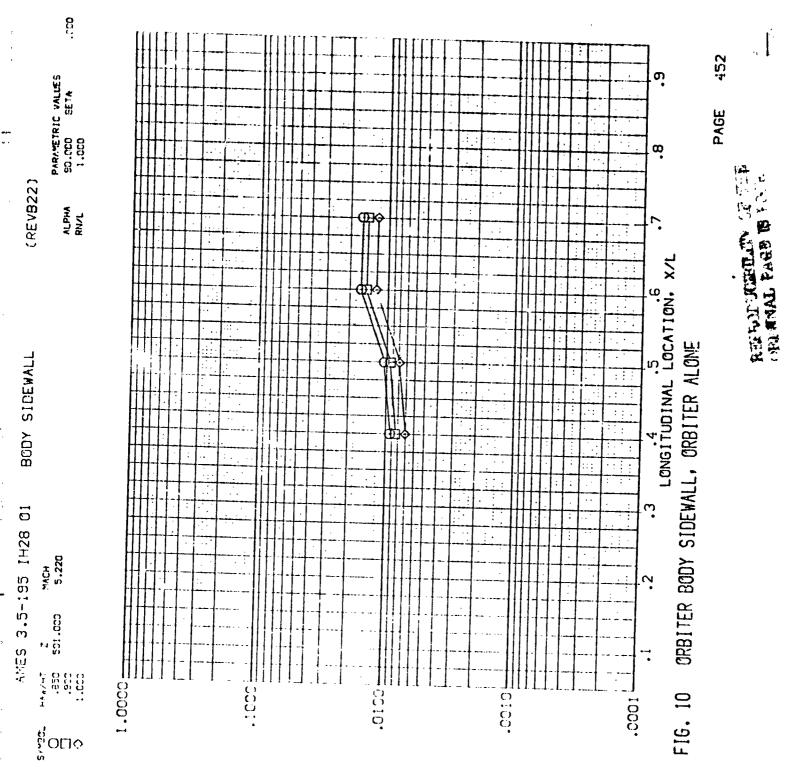


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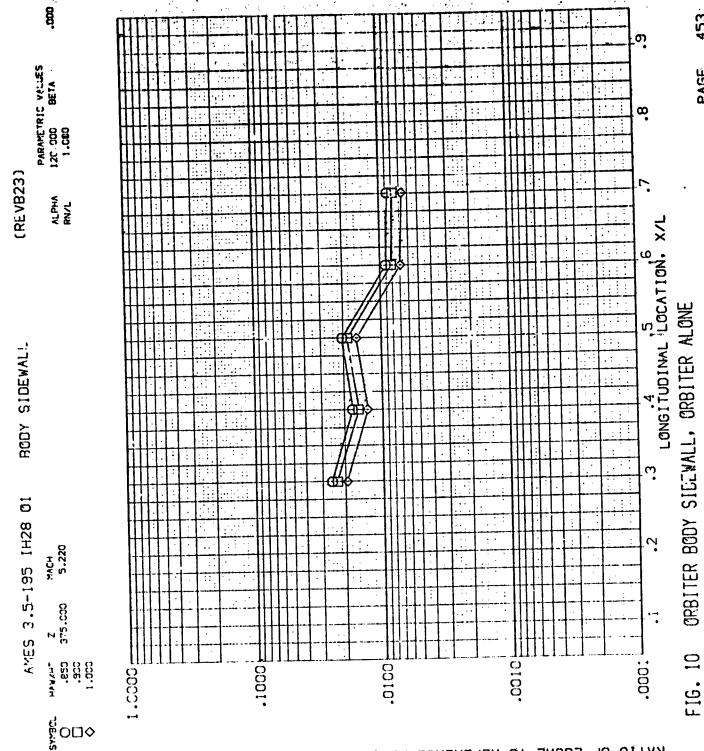
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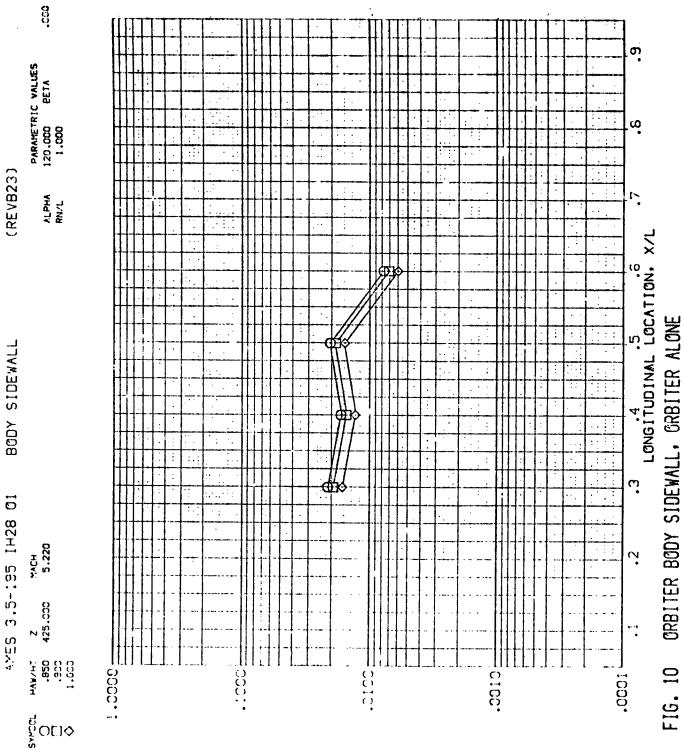
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PAGE ထ (REVB23) ALPHA RN/L LONGITUDINAL LOCATION X/L FIG. 10 ORBITER BODY SIJEWALL, ORBITER ALONE BCDY SIDEWALL AMES 3.5-195 IH28 01 : : : MACH 5.220 2 501.000 .850 .900 1.000 0100 .0010 .0001 Ş O□¢ RATIO OF LOCAL TO REFERENCE HEAT TRANSFER COEFFICIENTS, HVHREF

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ORBITER BODY SIDEWALL, ORBITER ALONE

FIG. 10

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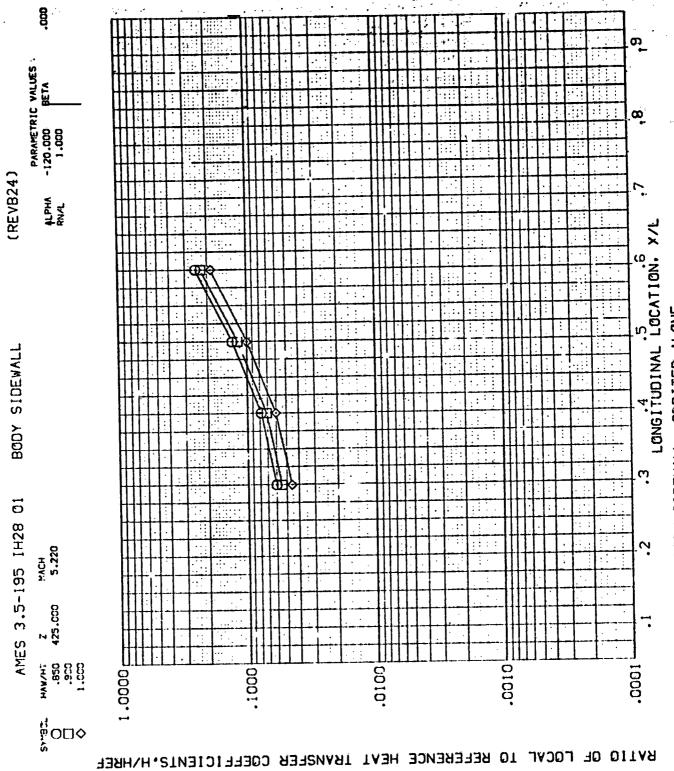
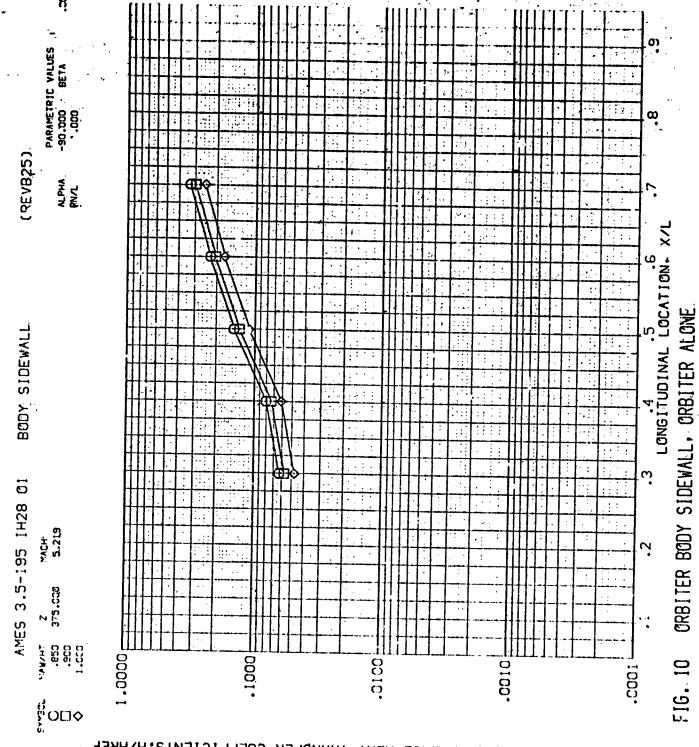
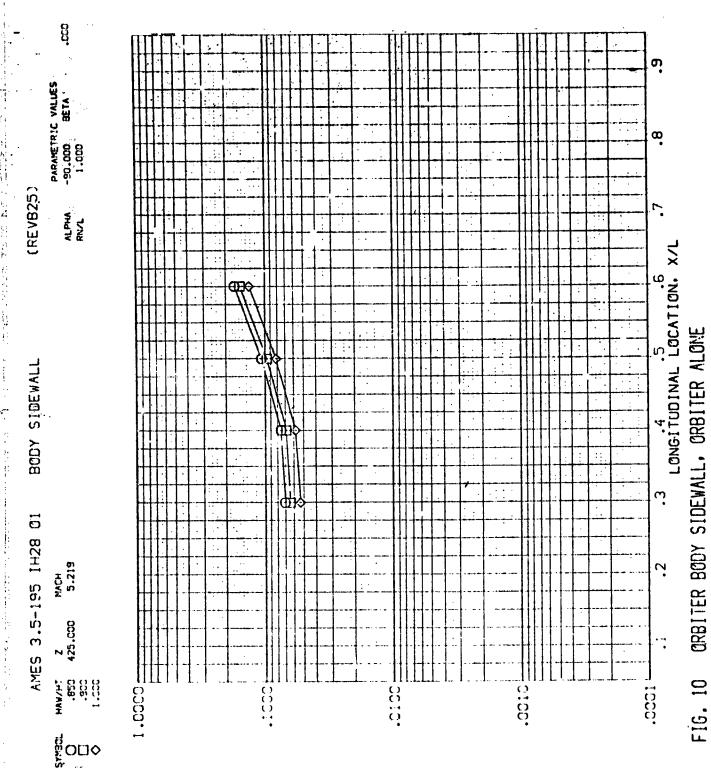


FIG. 10 SABITER BODY SIDEWALL, ORBITER ALONE

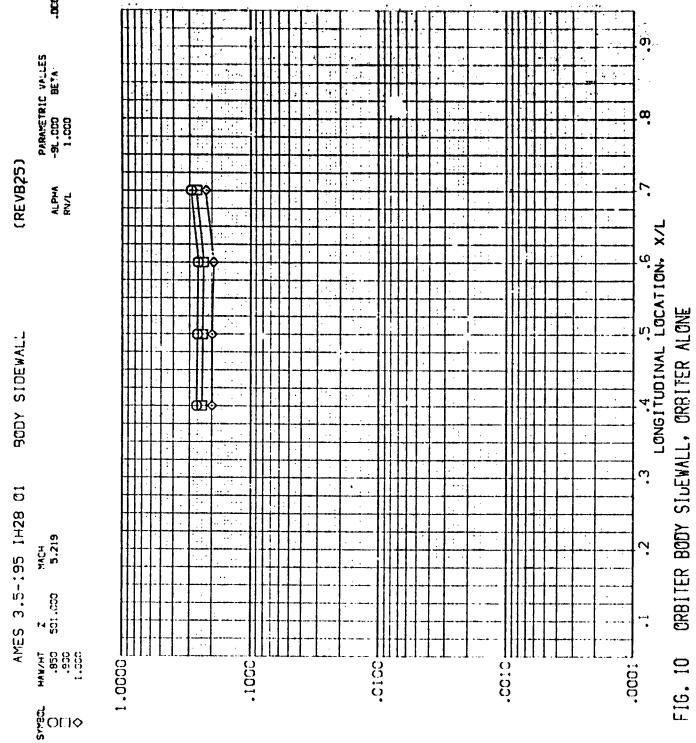
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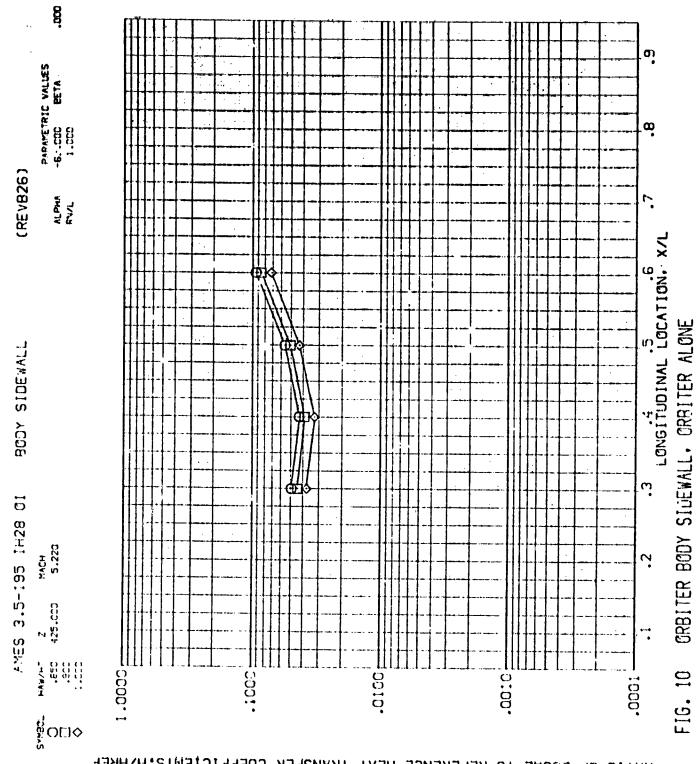
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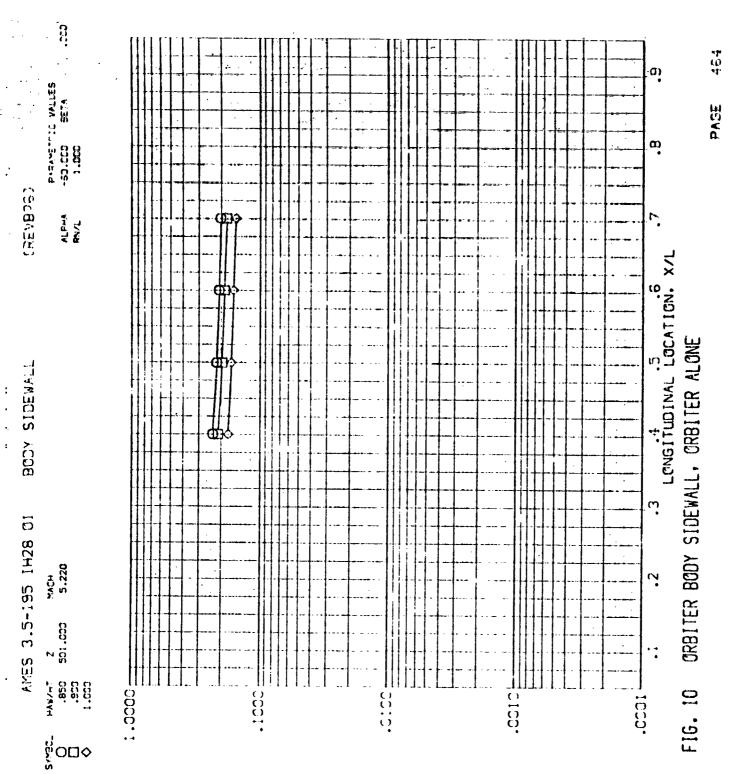
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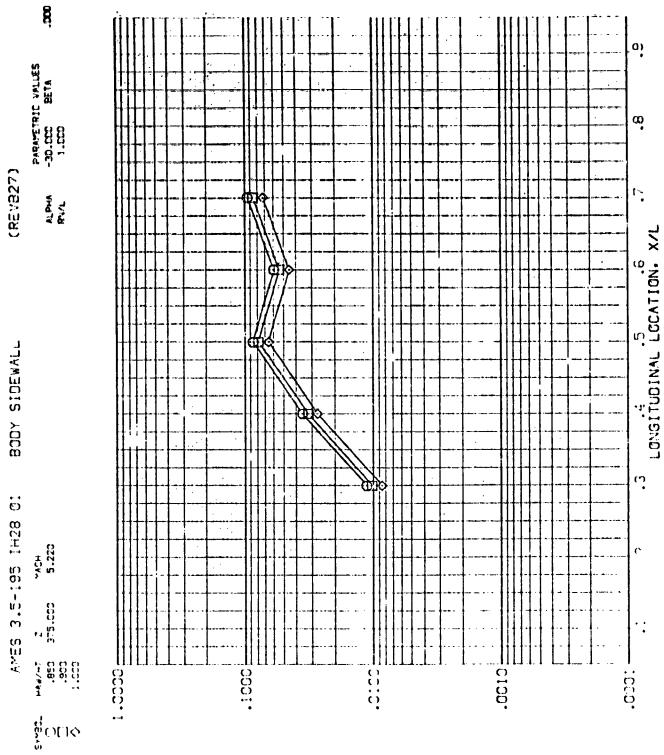


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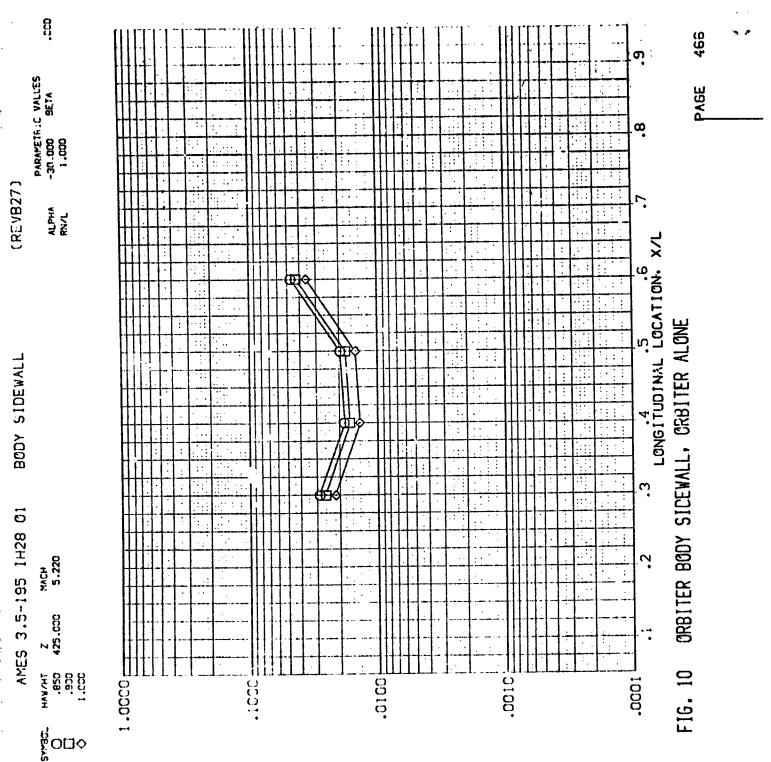
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FIG. 10 GRBITER BODY SIDEWALL, ORBITER ALCHE

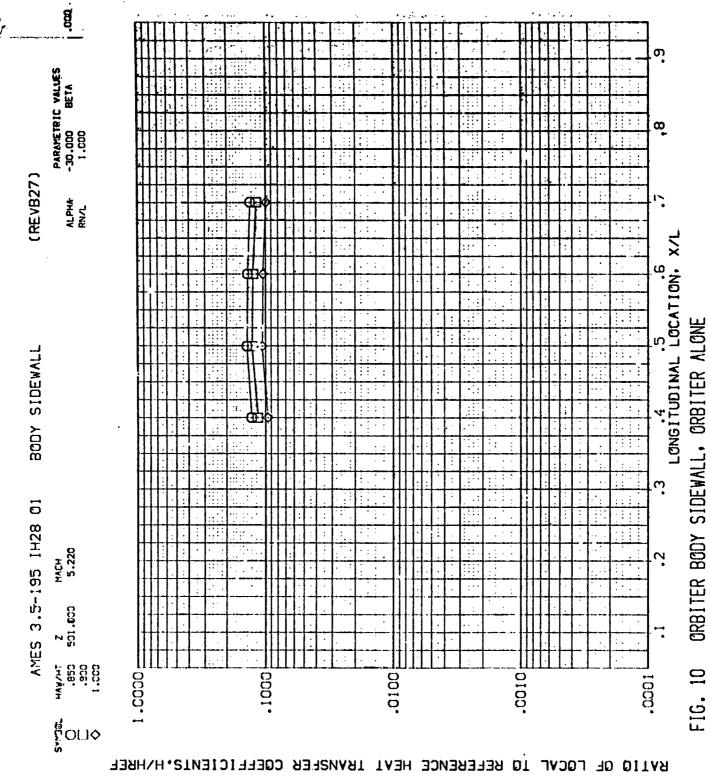


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420 440 460 VERTICAL LOCATION. Z ORBITER BODY SIDEWALL, ORBITER ALONE FIG. 10

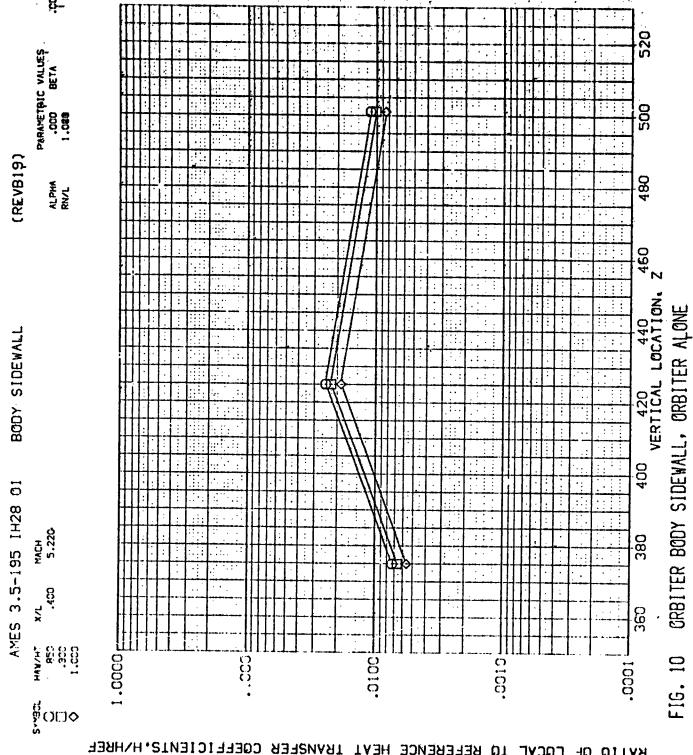
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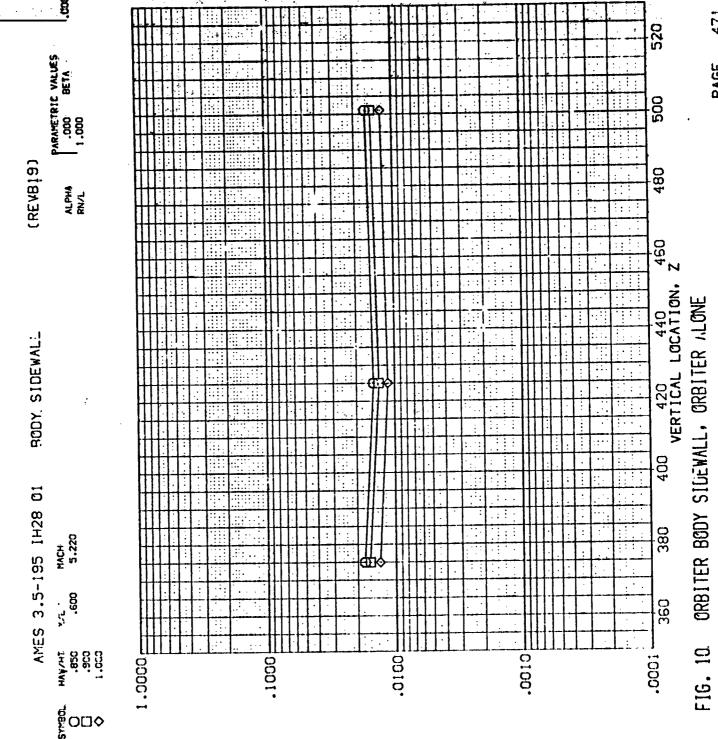
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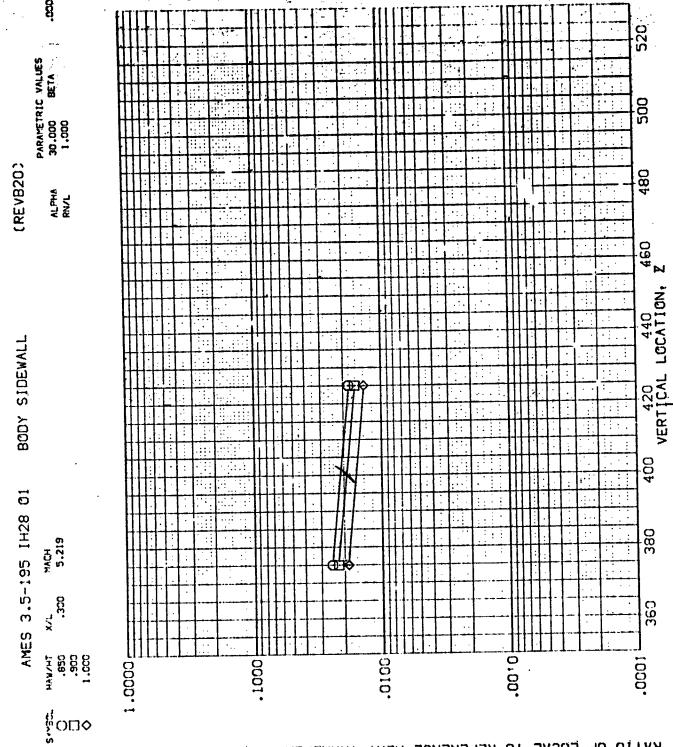
FIG. 10 ORBITER BODY SIDEWALL, ORBITER ALONE



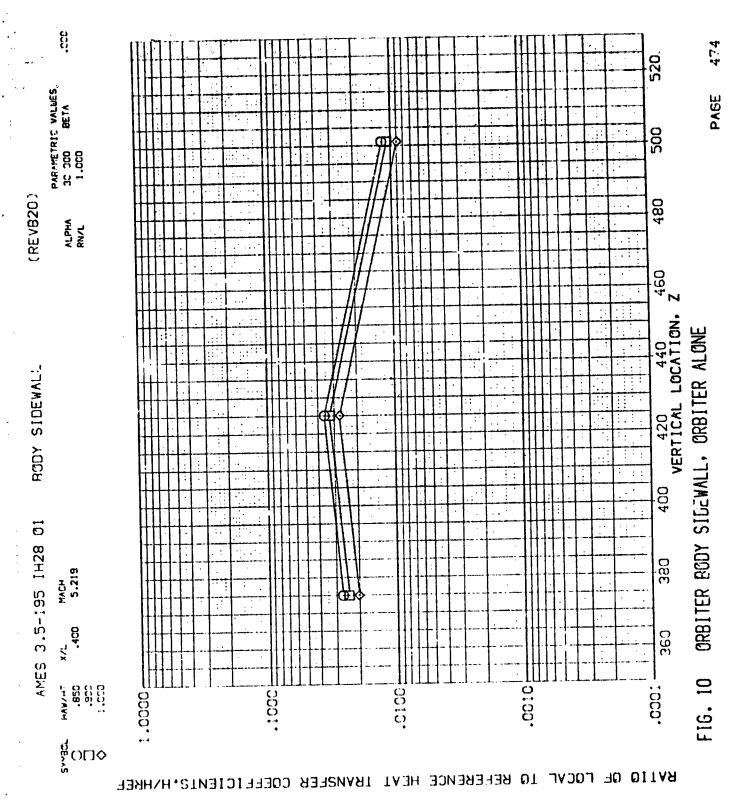
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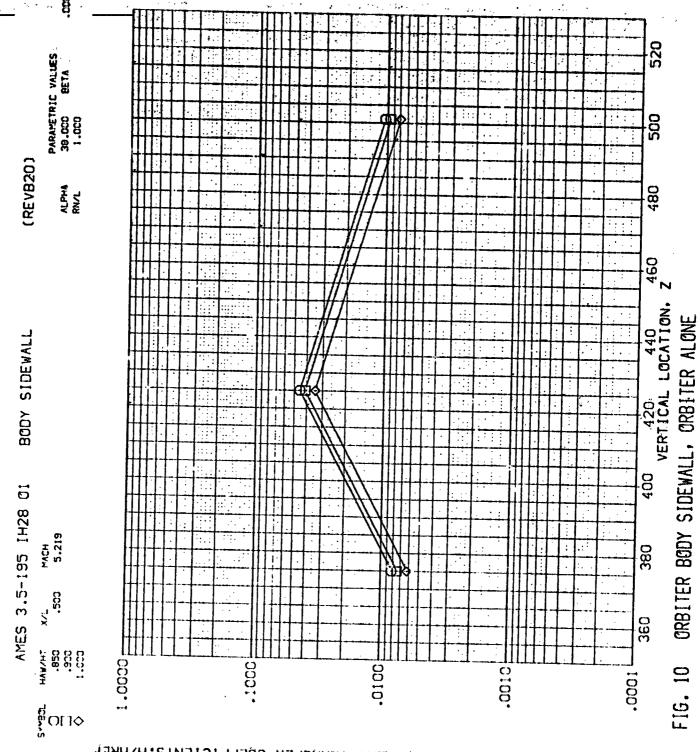
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FIG. 10 ORBITER BODY SIDEWALL, ORBITER ALONE

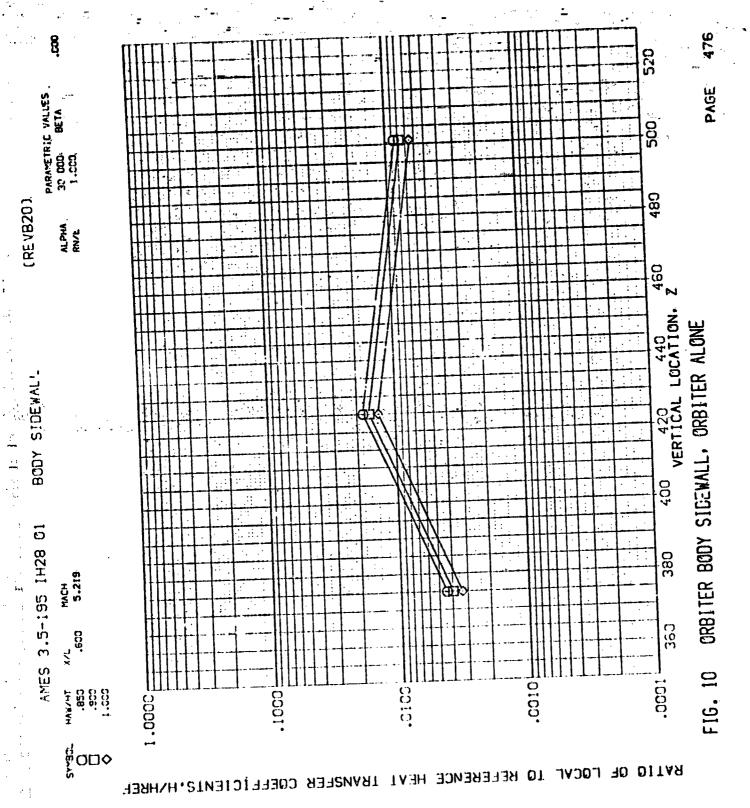


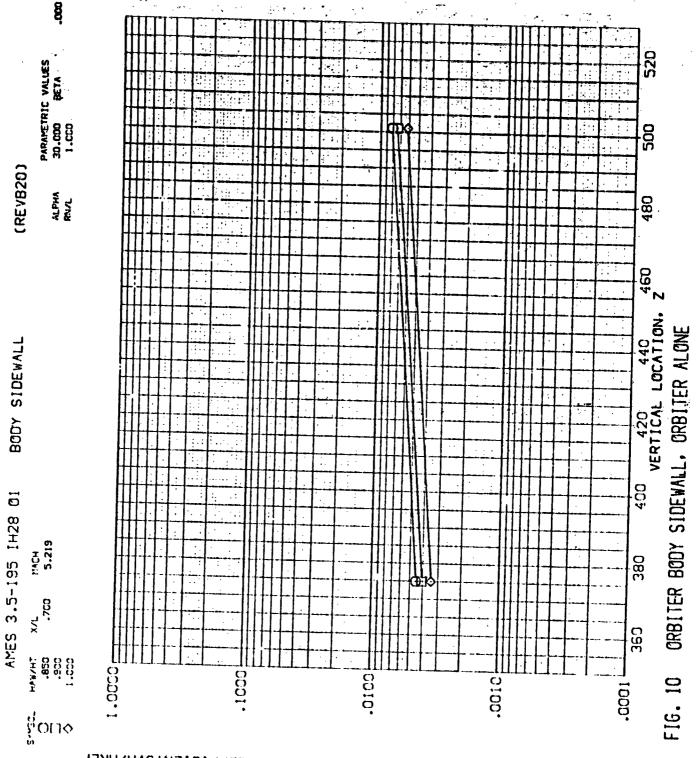
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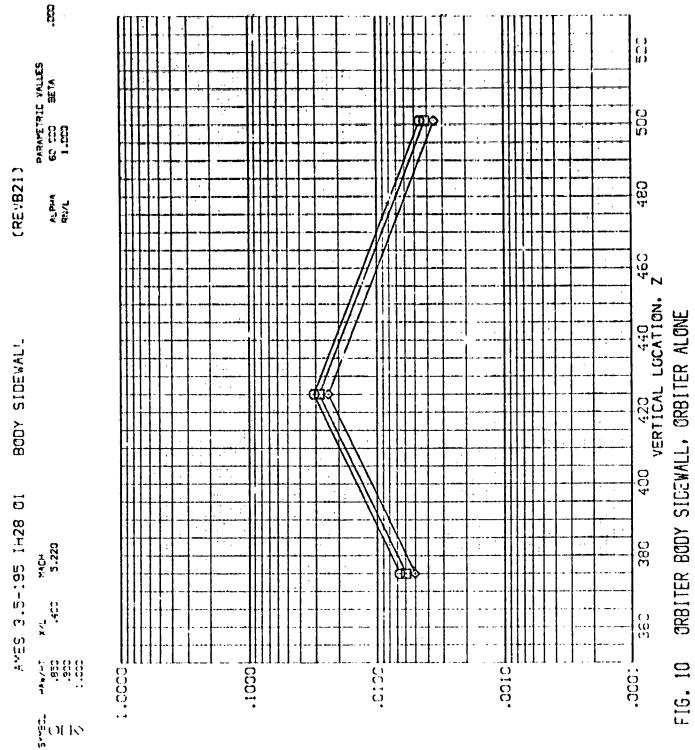




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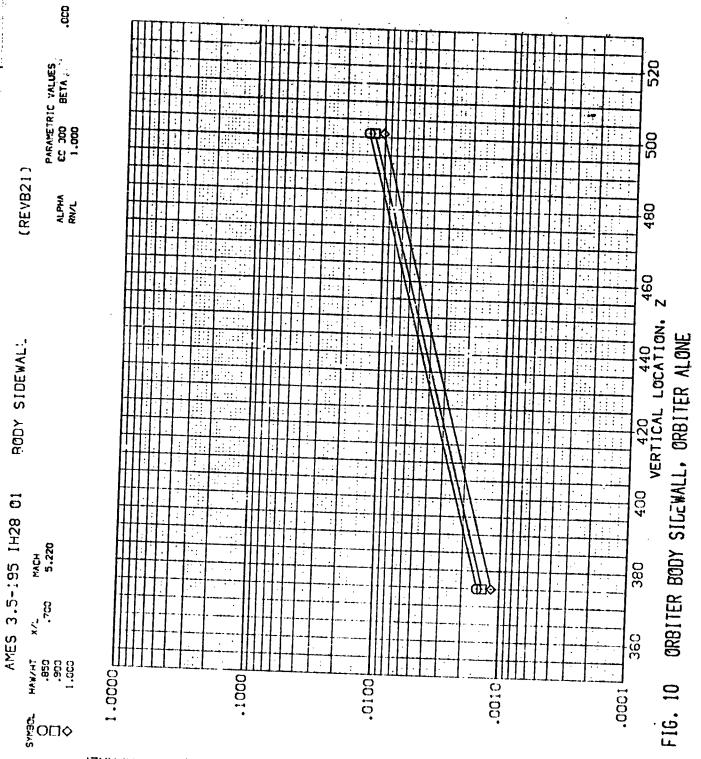
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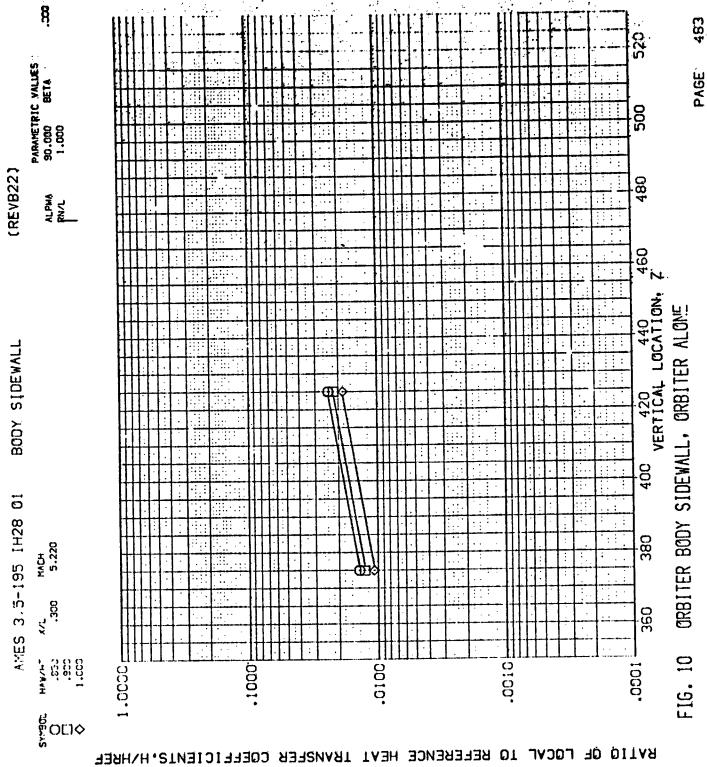
FIG. 10 ORBITER BODY SIDEWALL, ORBITER ALONE

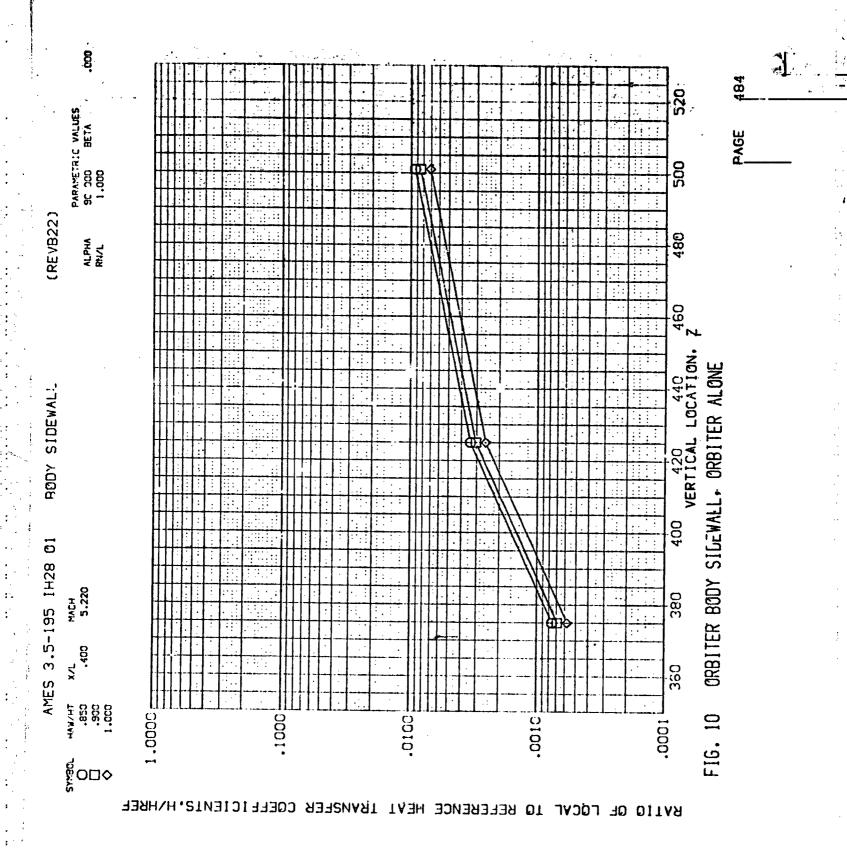
8 520 PARAMETRIC VALUES 60.000 BETA 1.000 500 (REVBZĮ) 480 ALPHA PN/L 450 VERTICAL LOCATION. FIG. 10 ORBITER BODY SIDEWALL, ORBITER ALONE BODY STDEWALL 400 AMES 3.5-195 1H28 01 мAСН 5.220 380 ×/L .630 380 нА**∦**/нт .850 .900 1.000 .0000; 10001 .0100 .0010 1000. Š O□◊ RATIO OF LOCAL TO REFERENCE HEAT TRANSFER COEFFICIENTS, HVHREF

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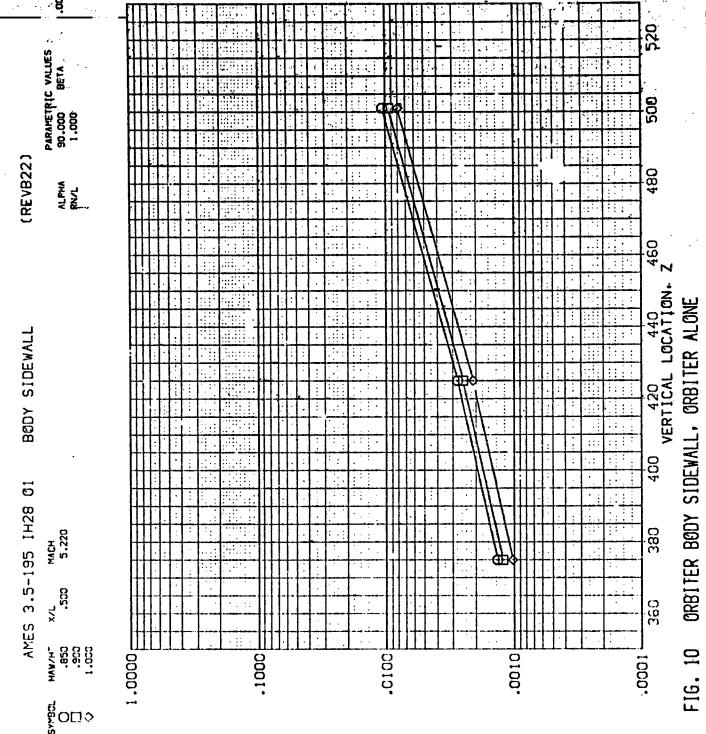


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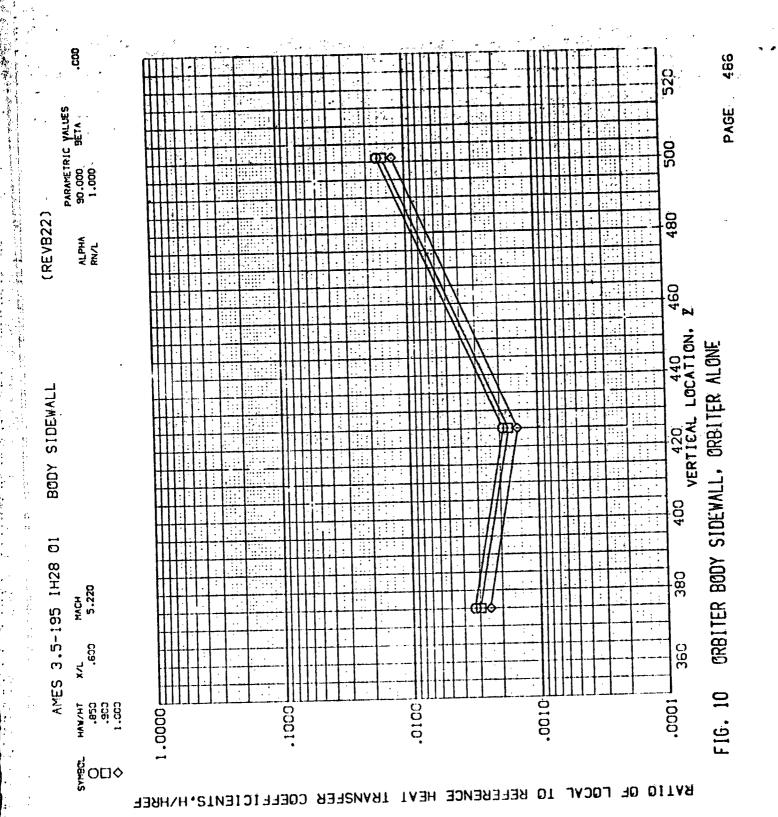


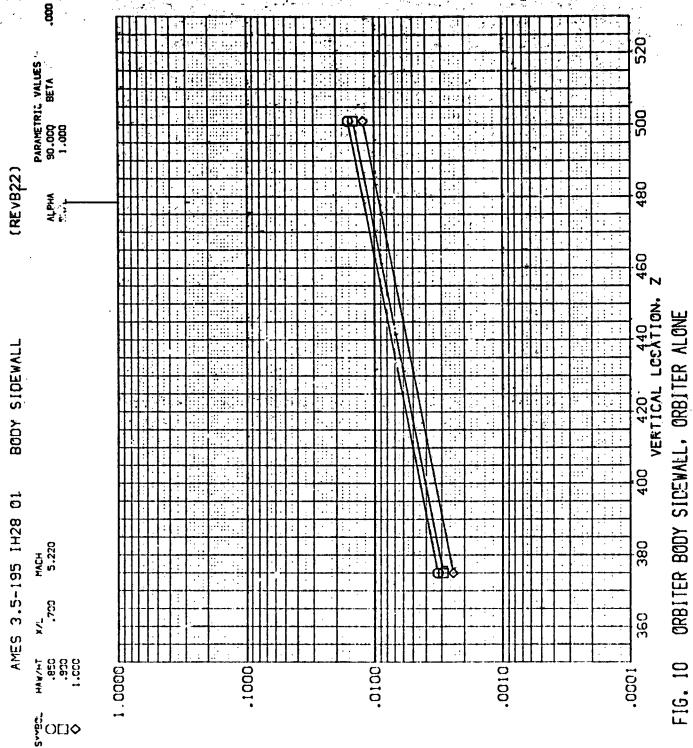


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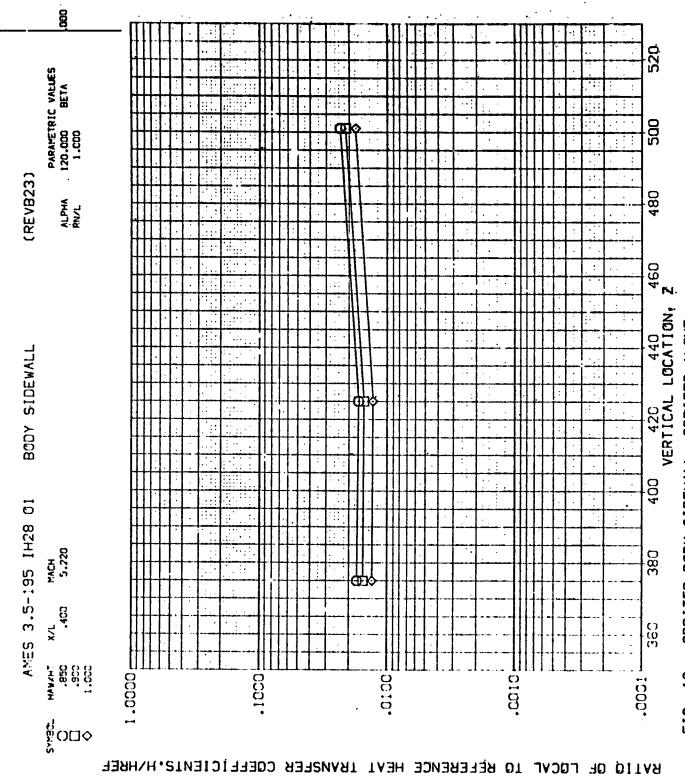


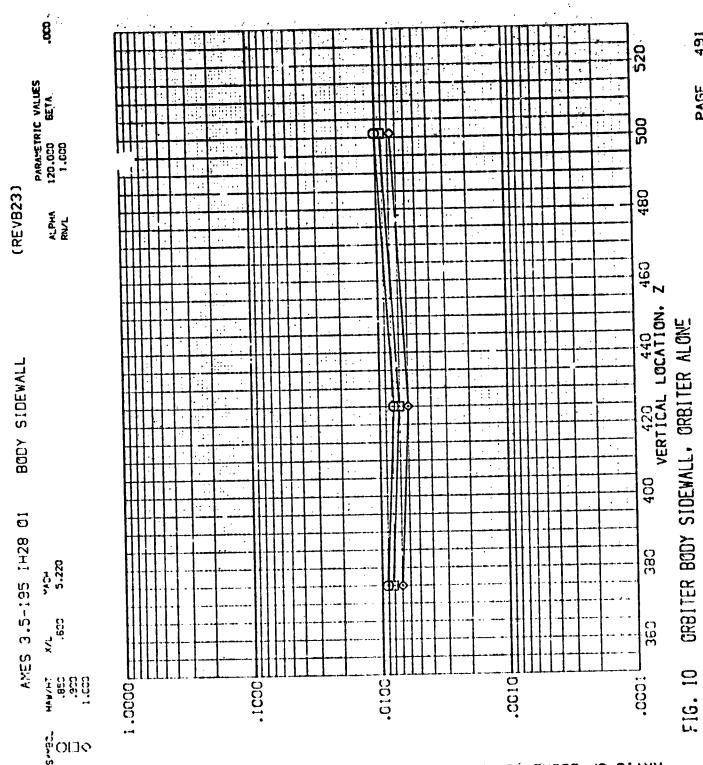
FIG. 10 ORBITER BODY SIDEWALL, ORBITER ALONE

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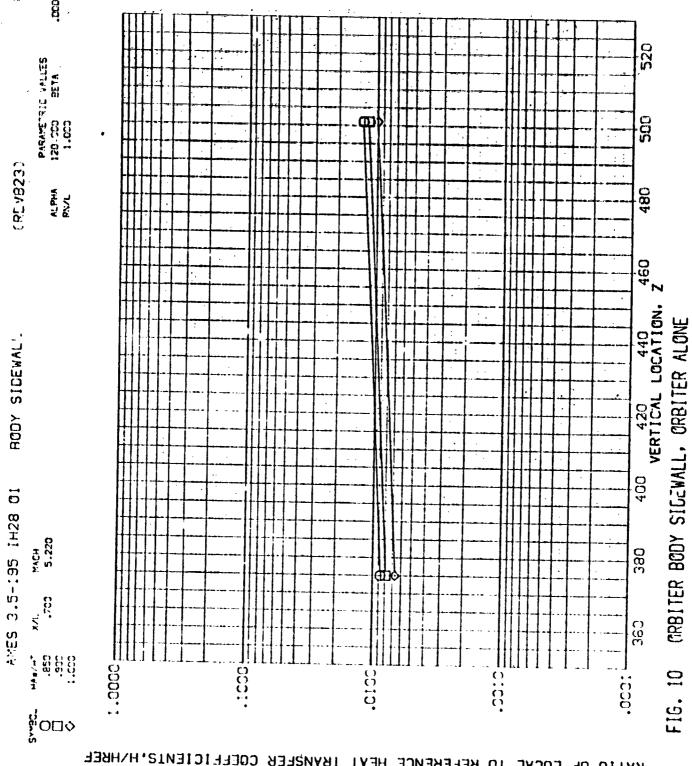
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FIG. 10 ORBITER BODY SIDEWALL, ORBITER ALONE

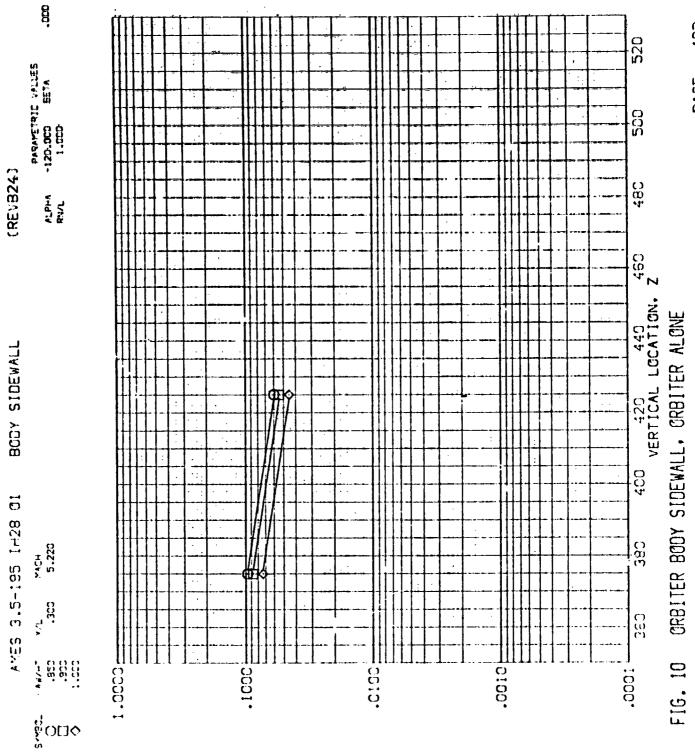


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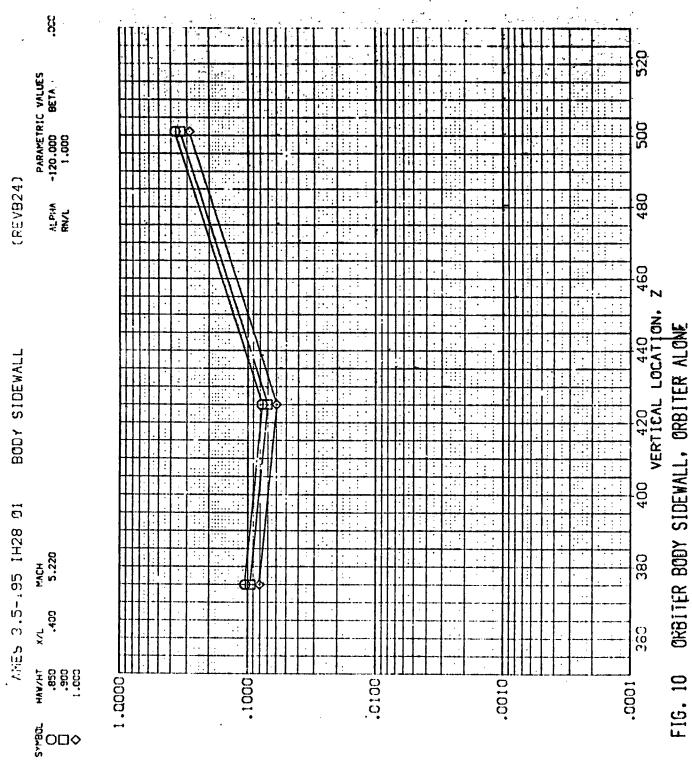
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FIG. 10

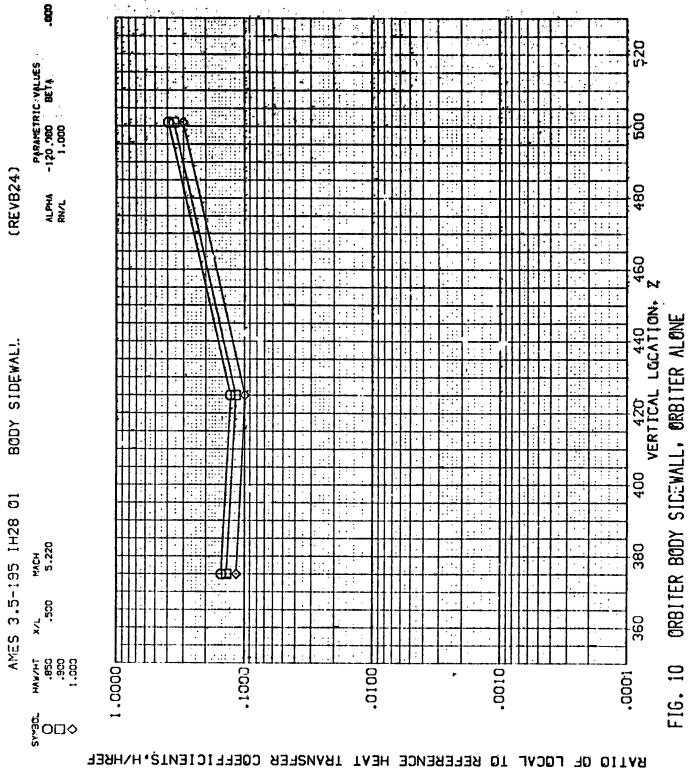


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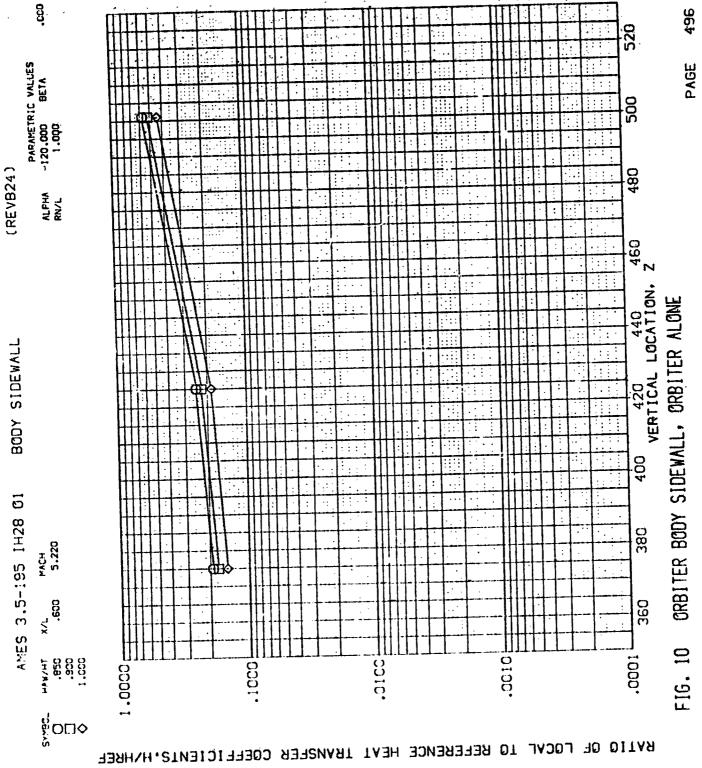
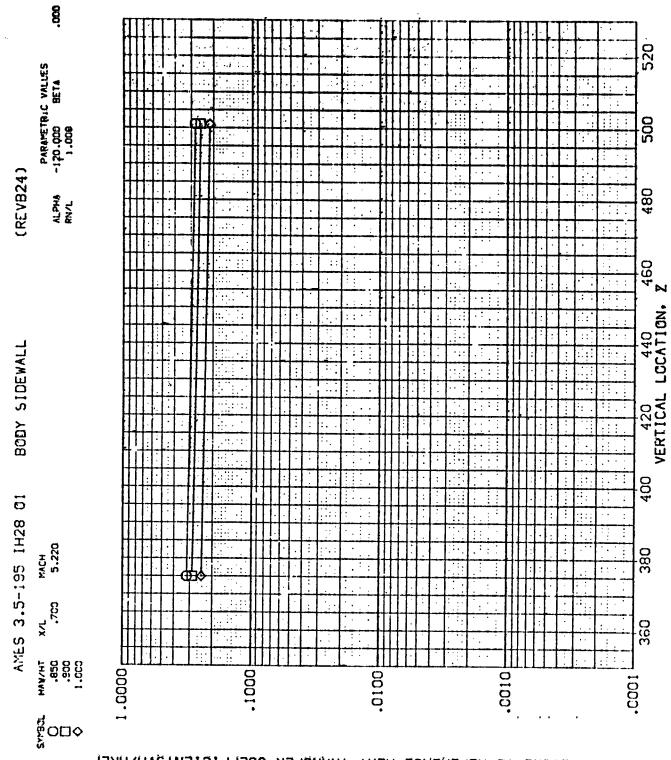


FIG. 10 ORBITER BODY SICEWALL, ORBITER ALGNE

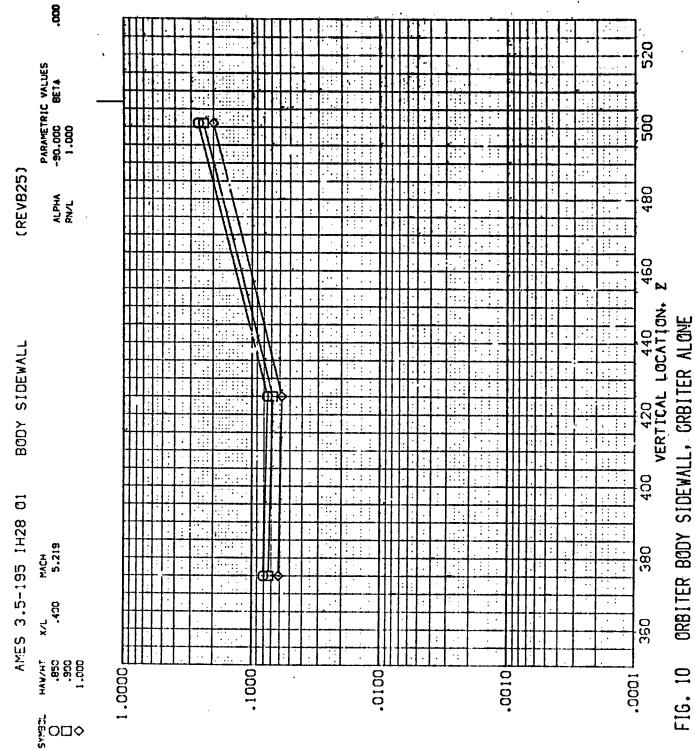


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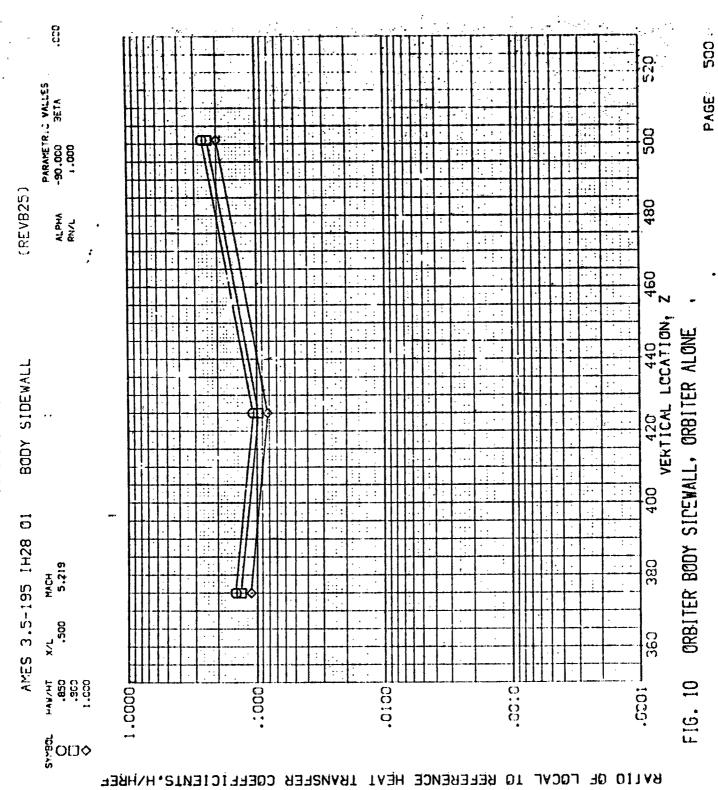
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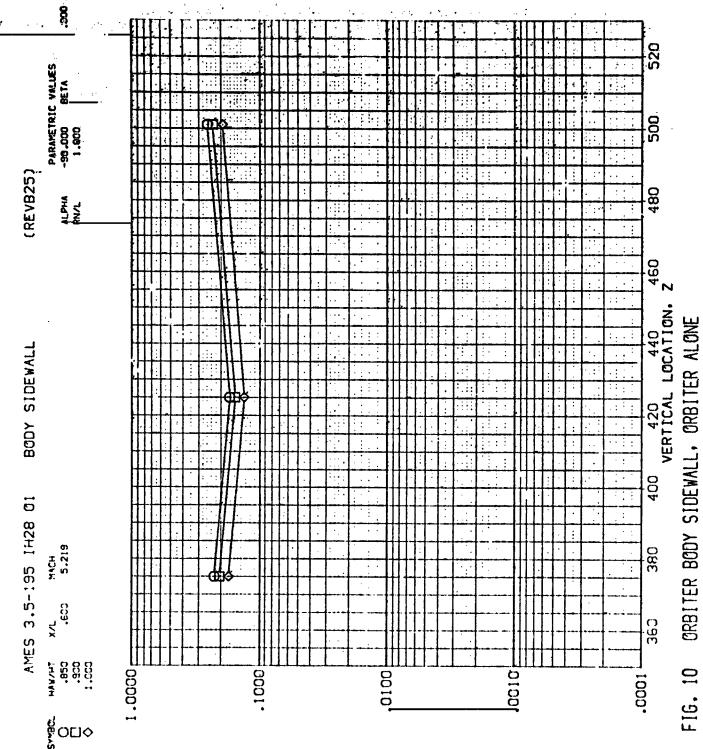


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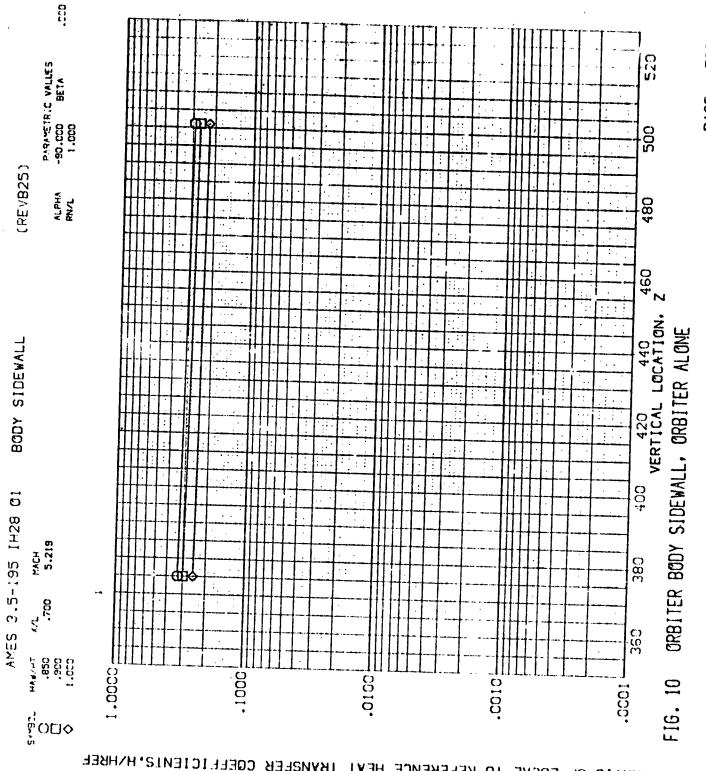
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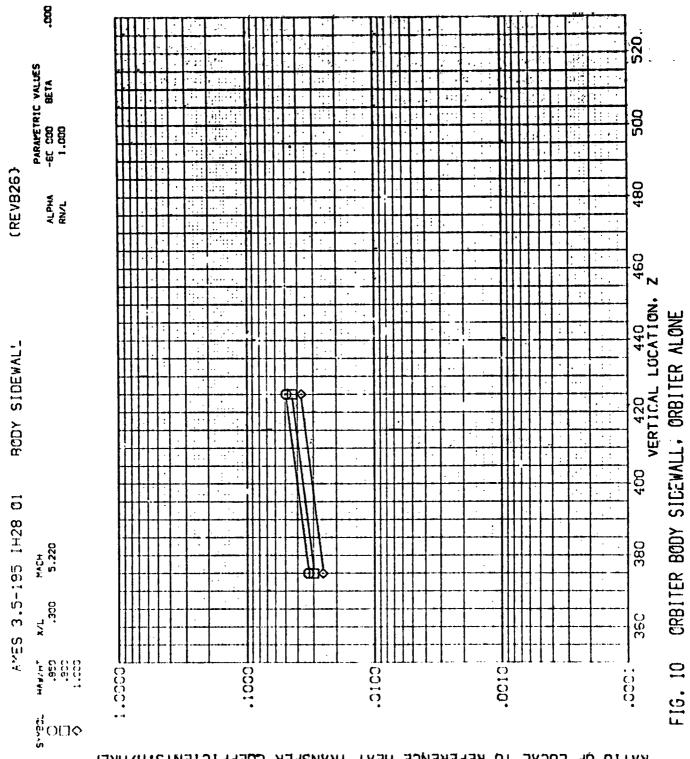


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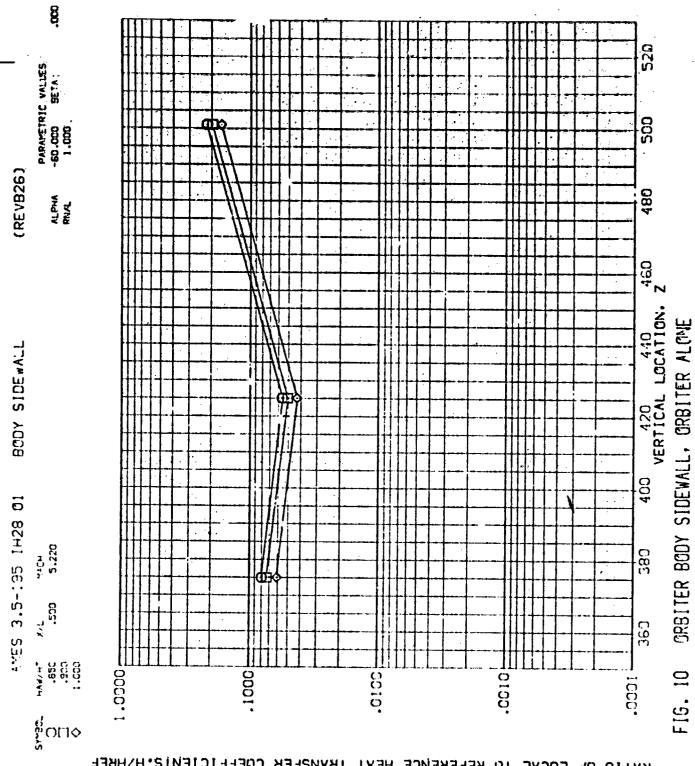
FIG. 10



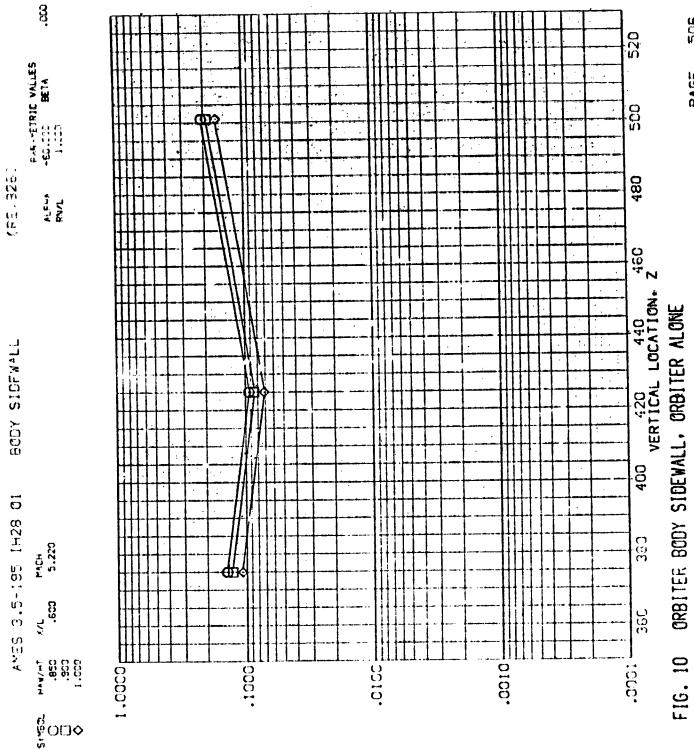
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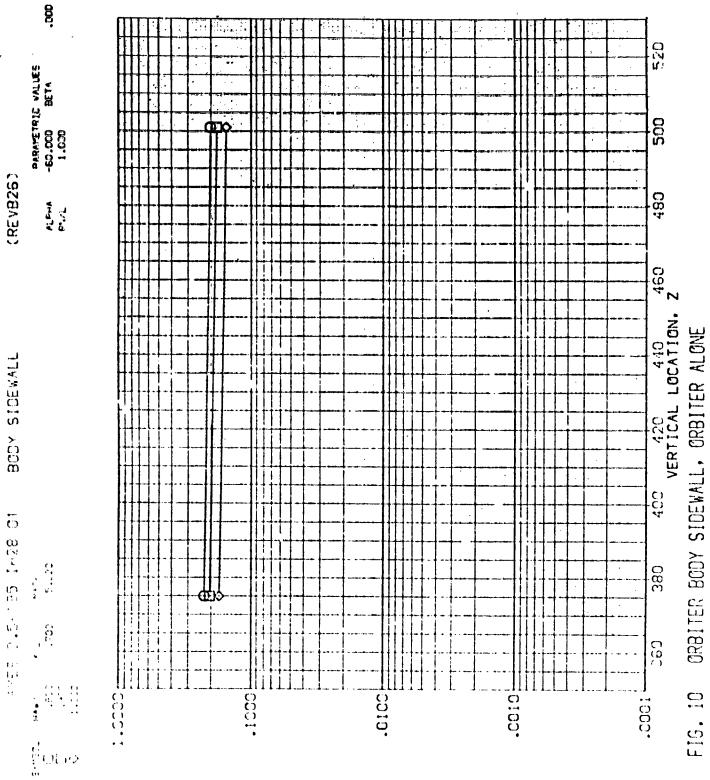
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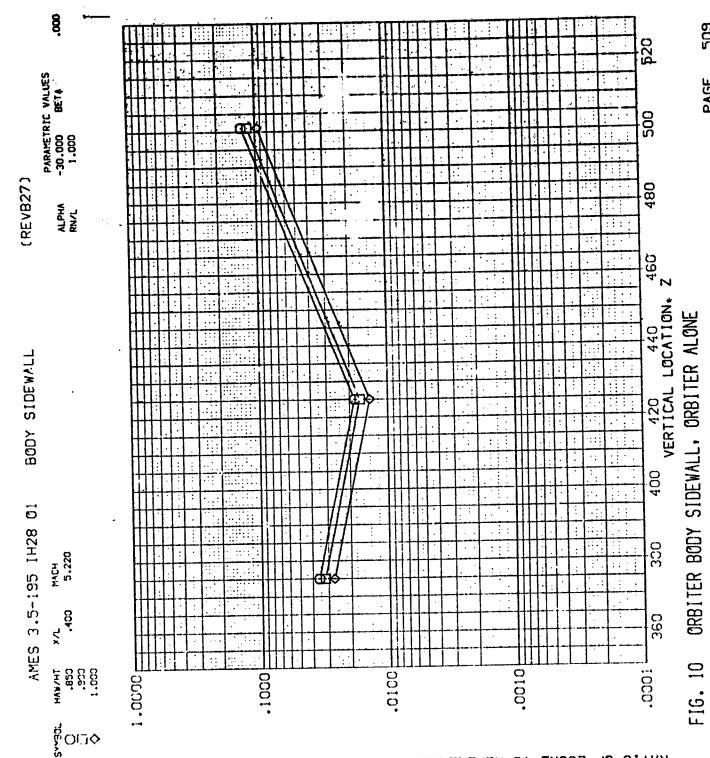


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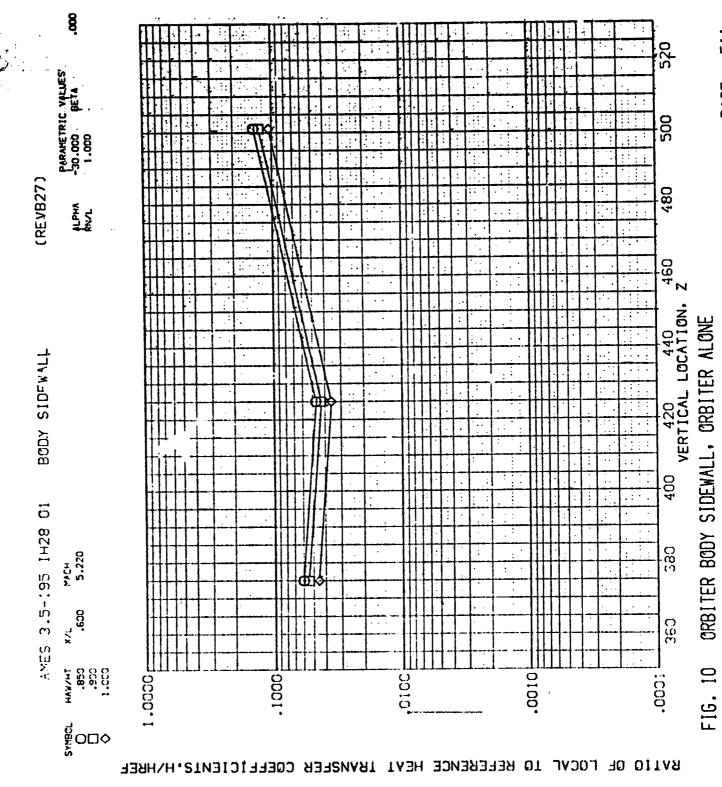
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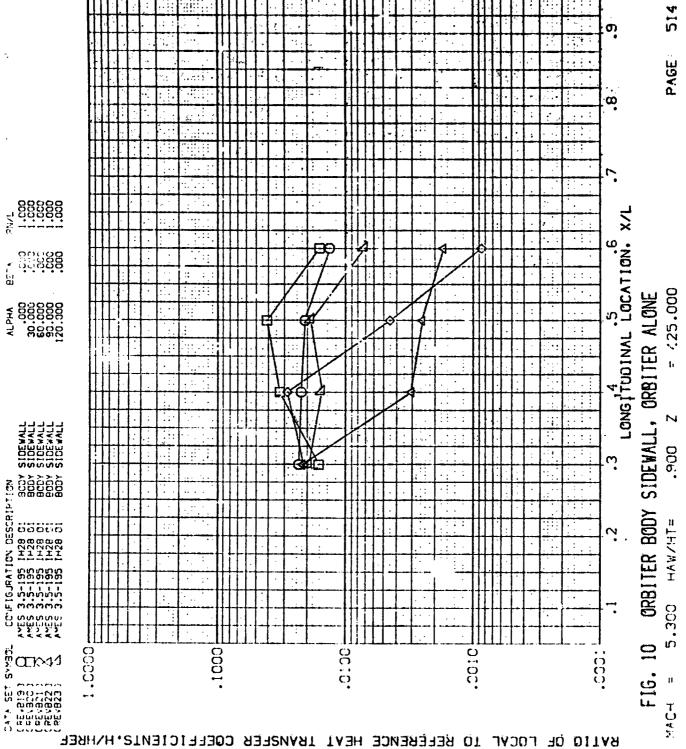


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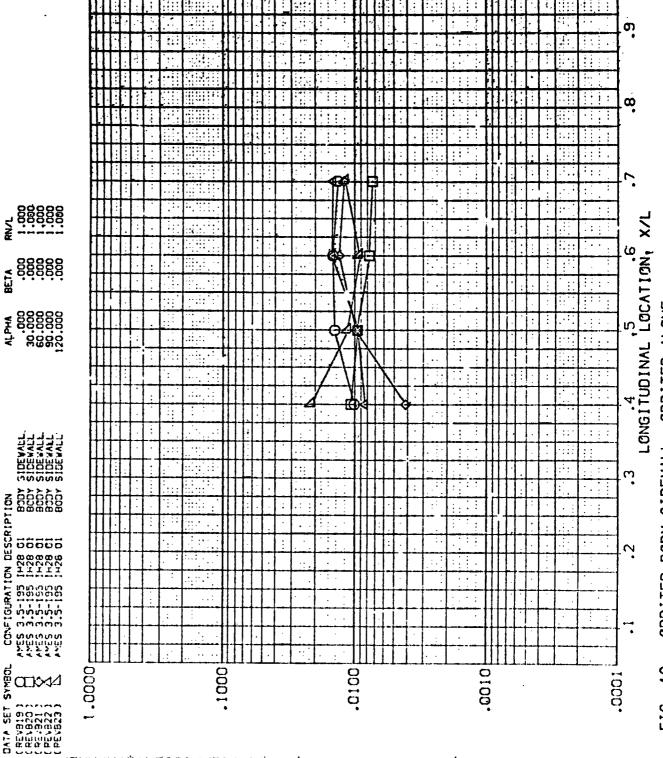
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375,000 FIG. 10 ORBITER BODY SIDEWALL, ORBITER ALONE

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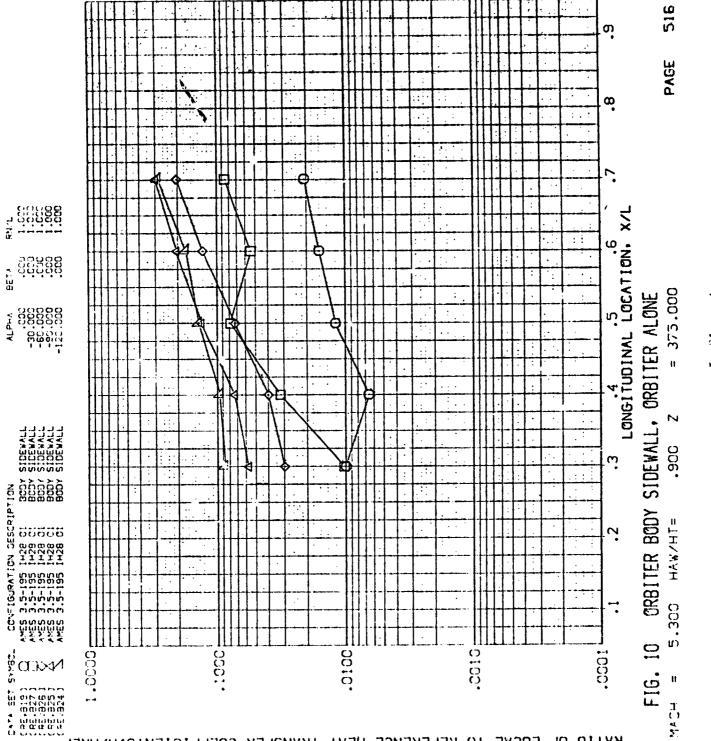


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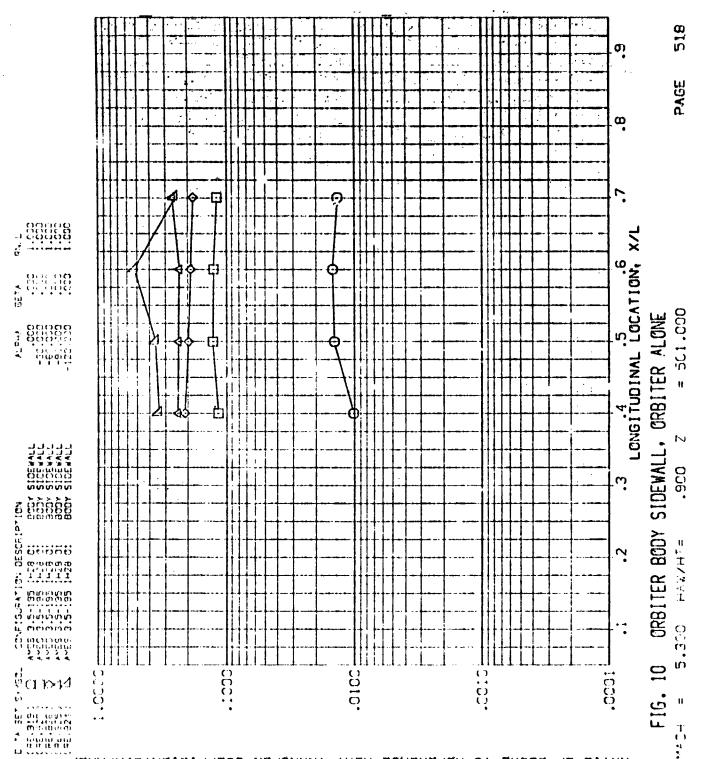


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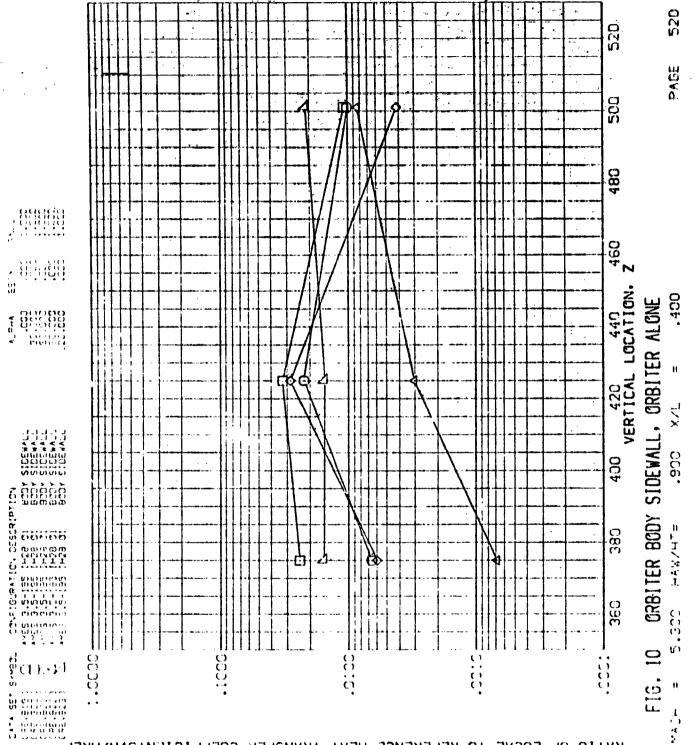


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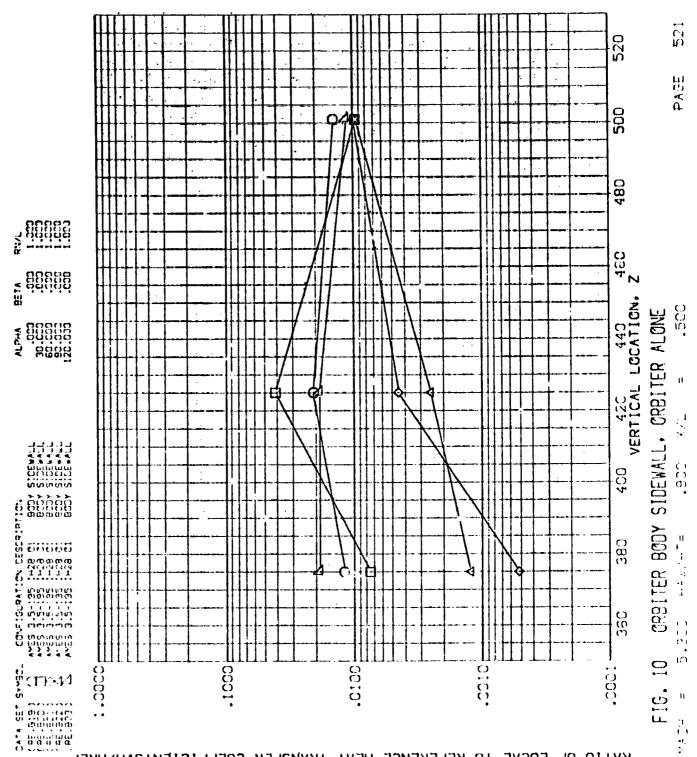
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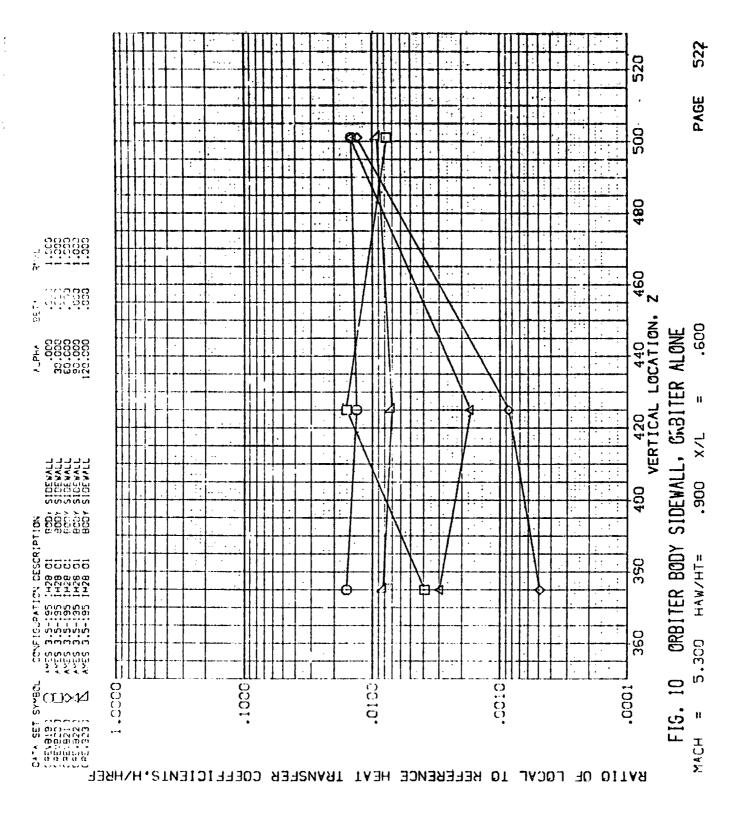


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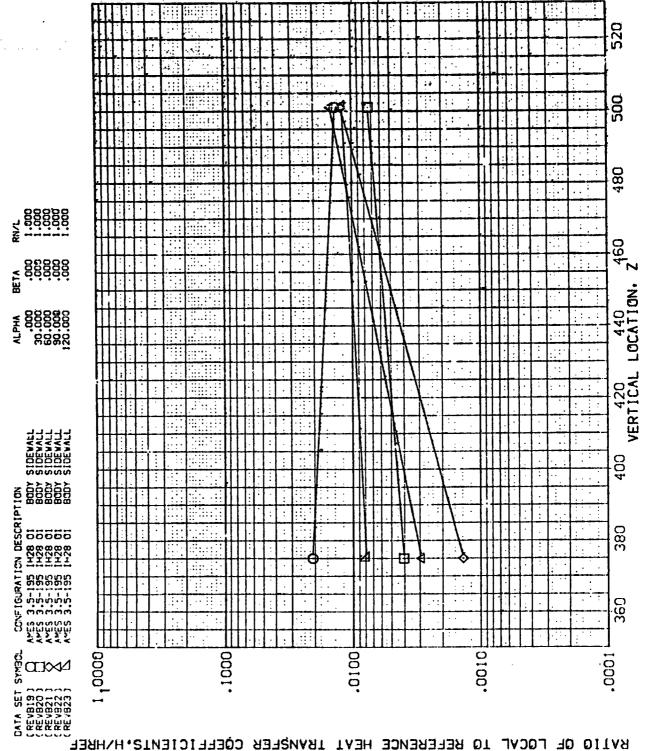
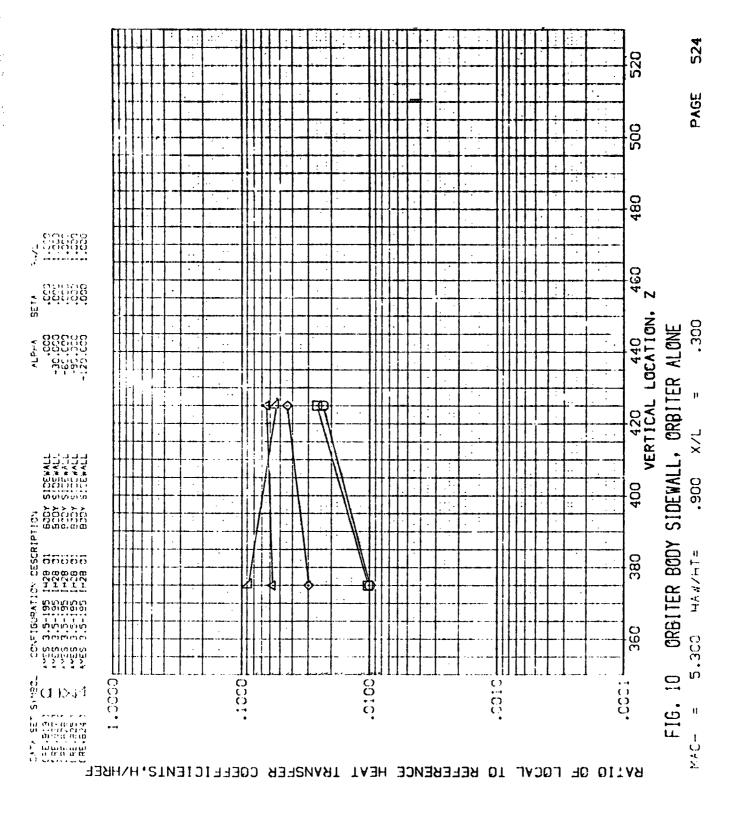


FIG. 10 ORBITER BODY SIDEWALL, ORBITER ALONE

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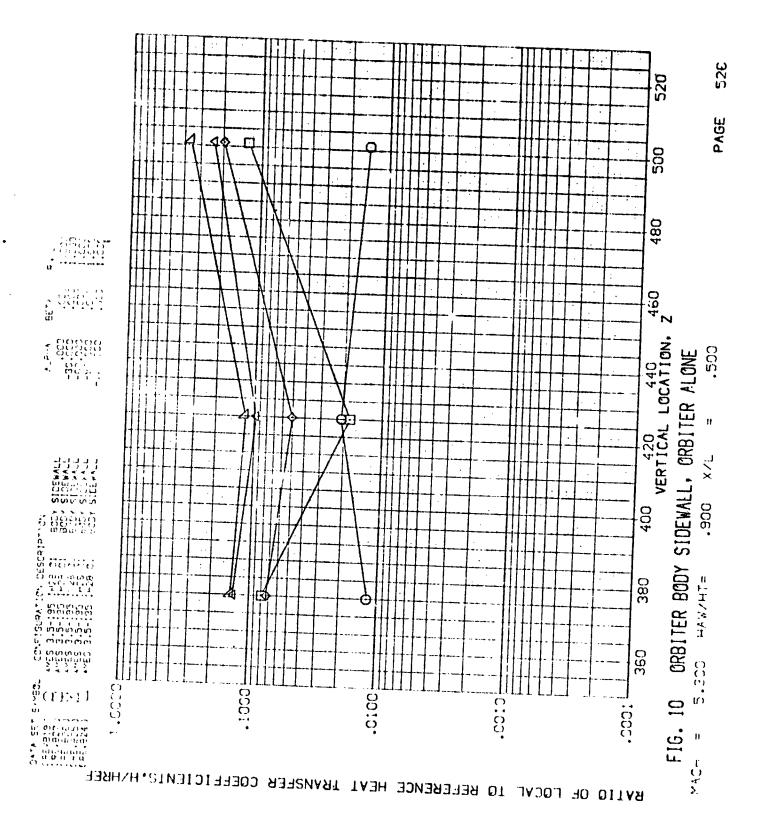
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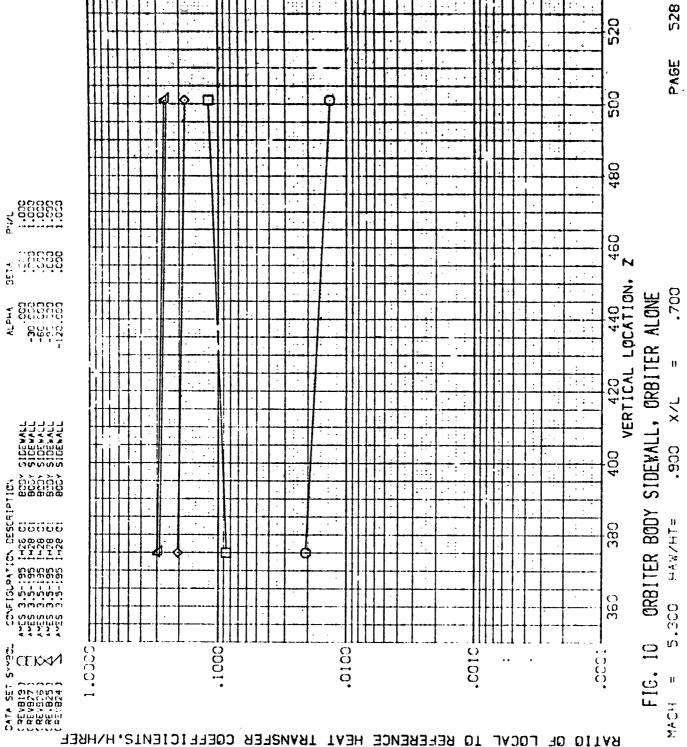
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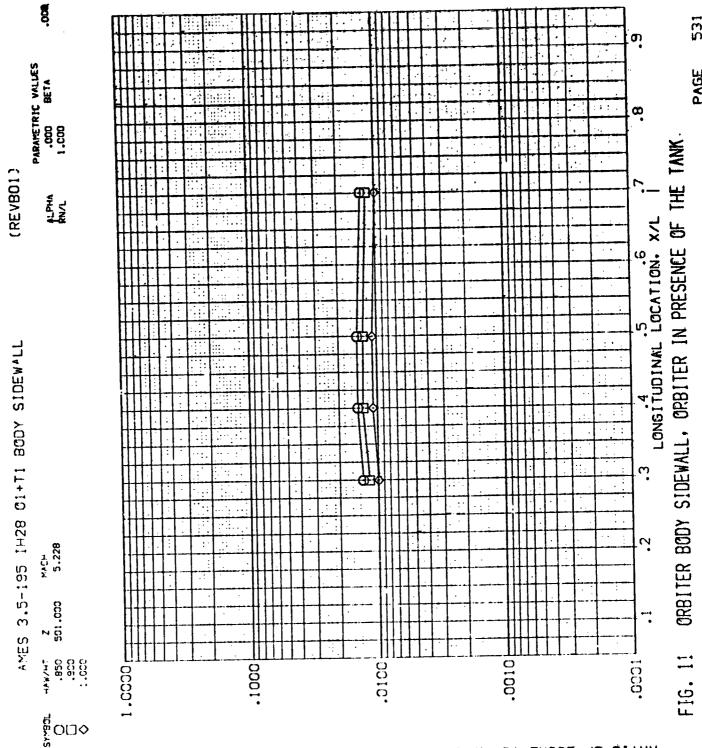
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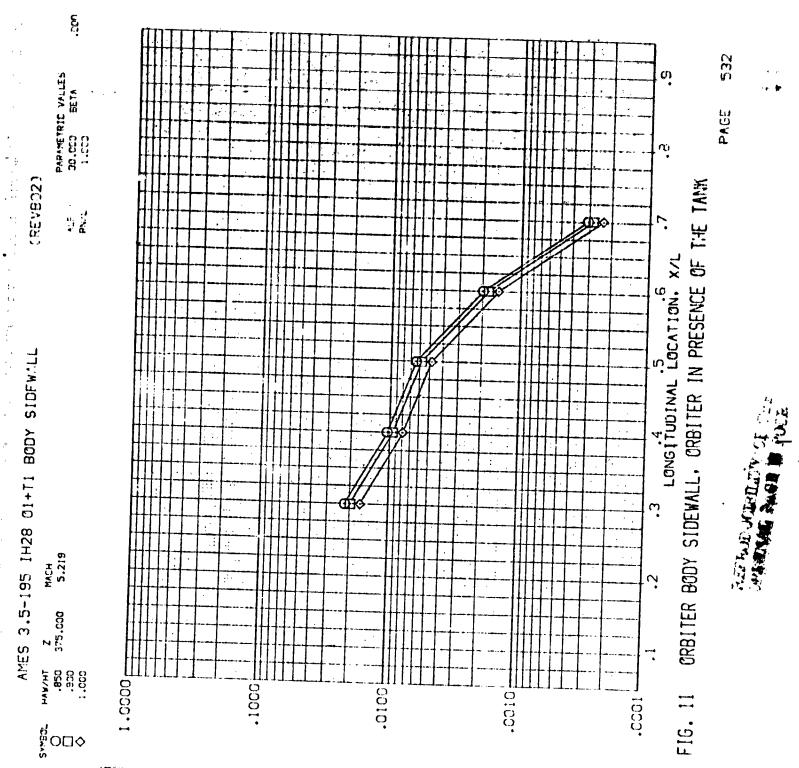
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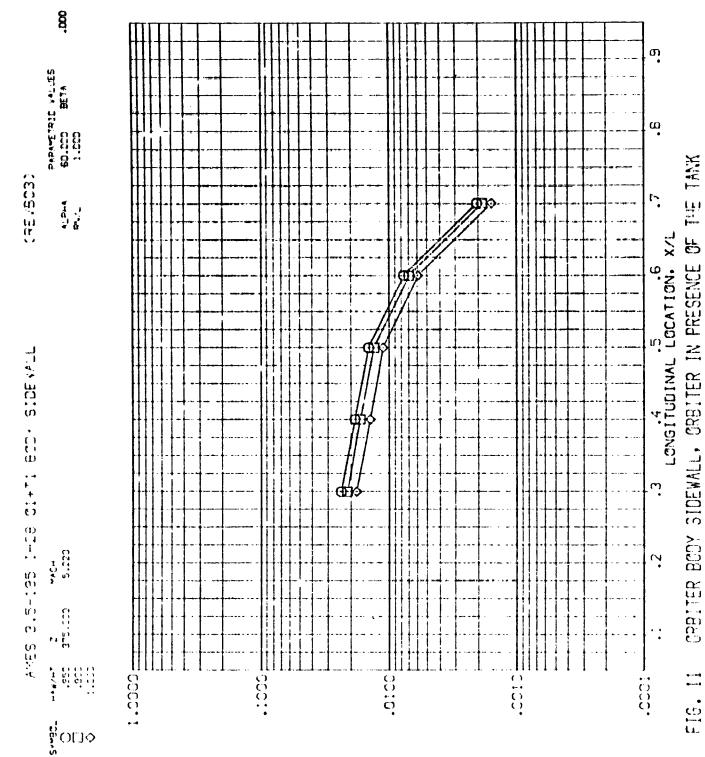
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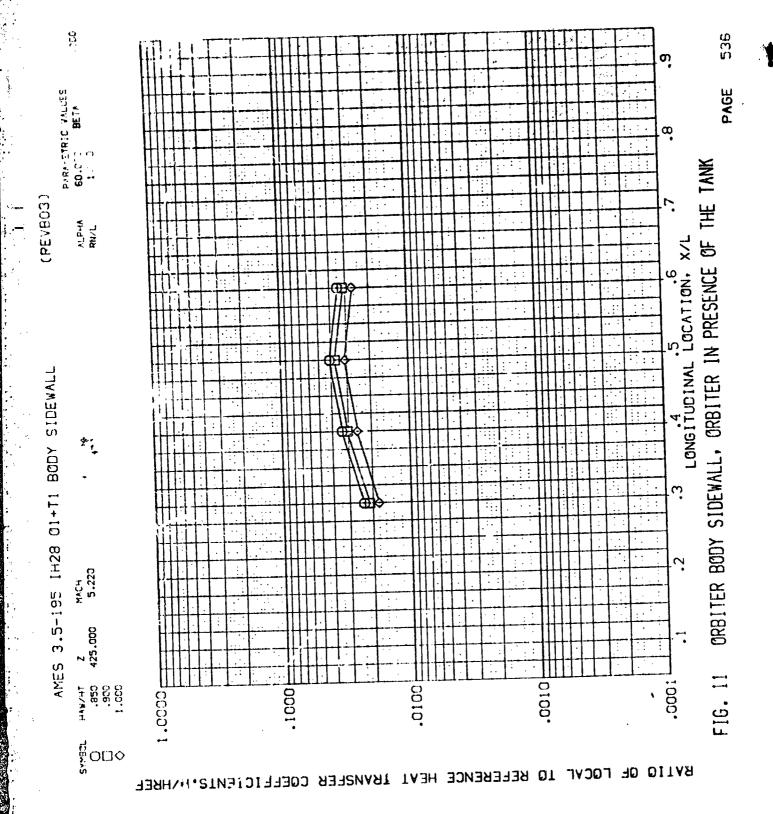
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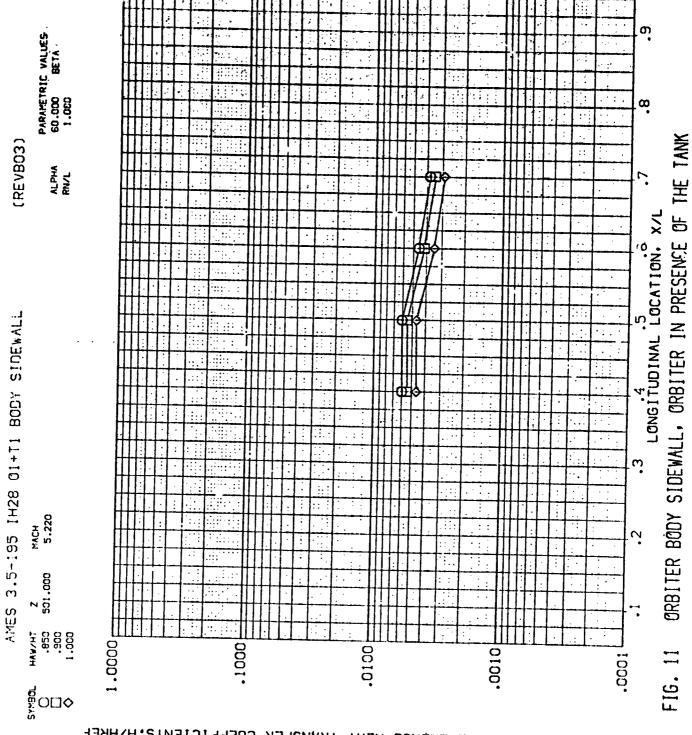
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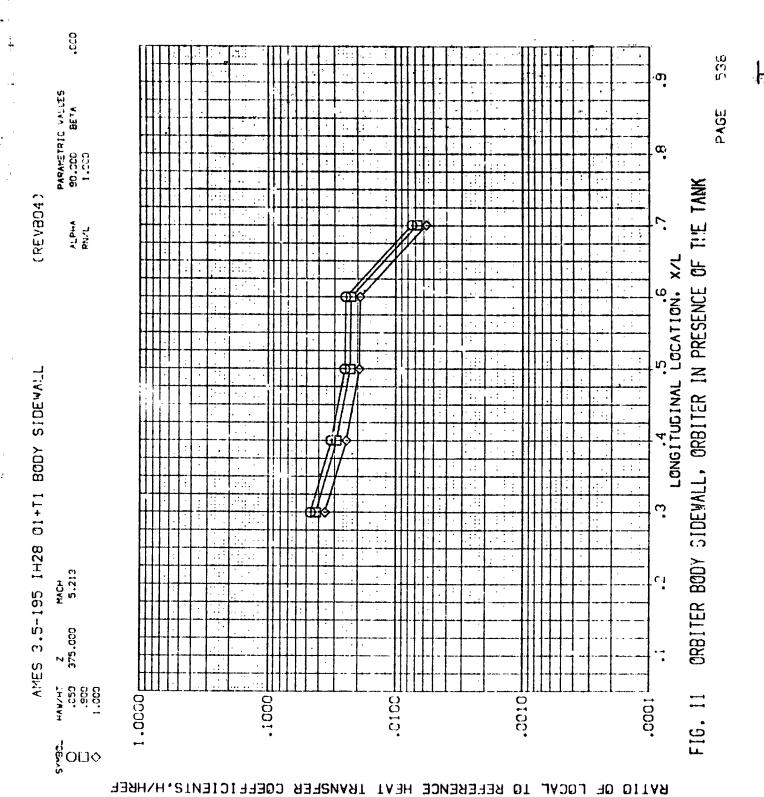
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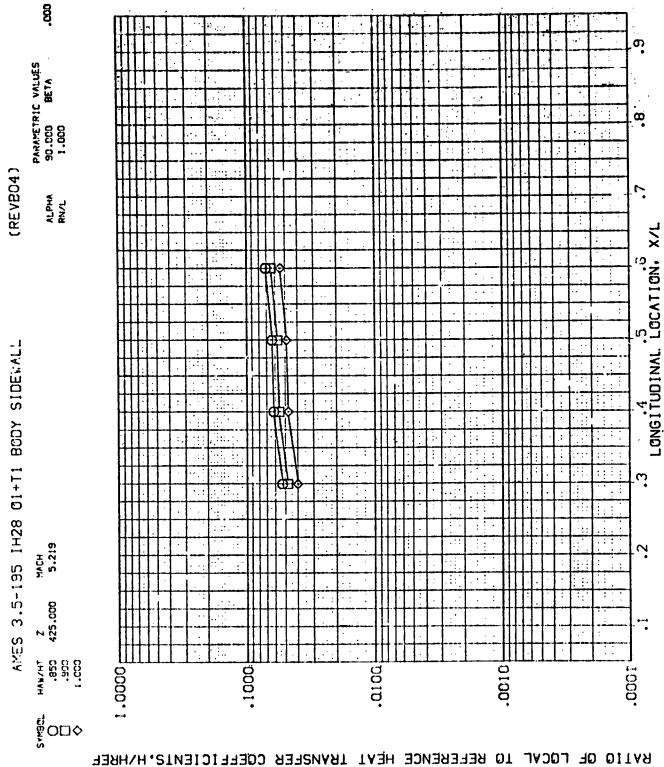


FIG. 11 ORBITER BODY SIDEWALL, ORBITER IN PRESENCE OF THE TANK

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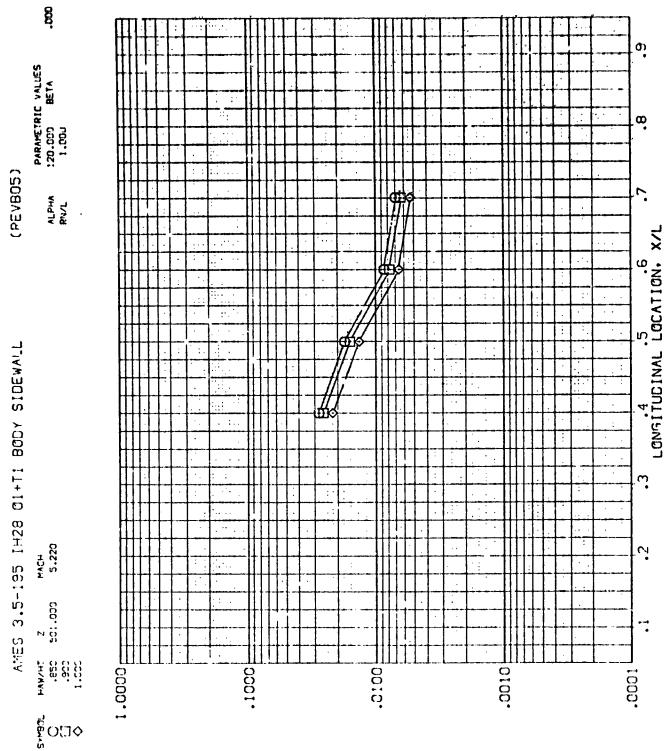
ຫຼ PAGE œ ORBITER BODY SIDEWALL, ORBITER IN PRESENCE OF THE TANK (REVBOS) ALPH. AMES 3.5-105 IH28 01+T1 BODY SIDEWALL 7 ₩ACH 5.220 2 425.000 FIG. 11 10100. .0001 0010. 10001 § dO∏♦

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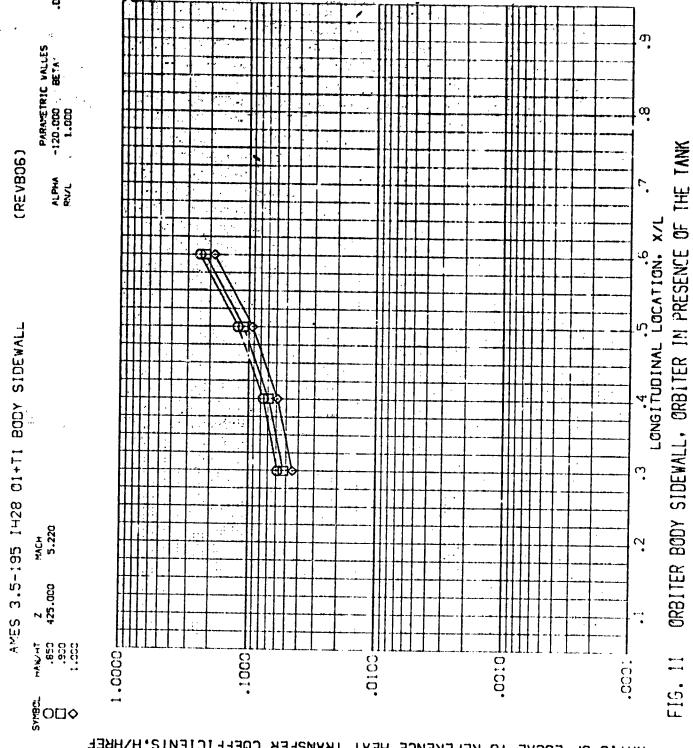
FIG. 11, ORBITER BODY SIDEWALL, ORBITER IN PRESENCE OF THE TANK



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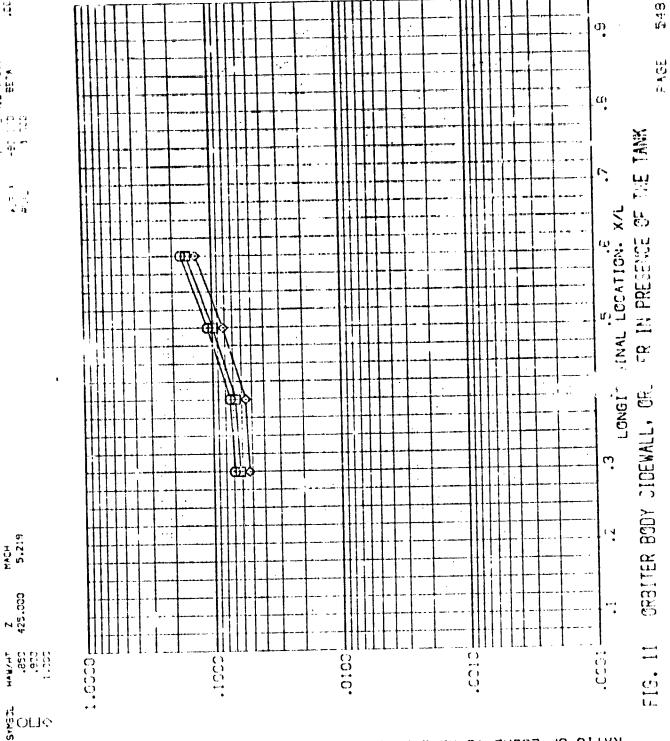
ORBITER BODY SIDEWALL, ORBITER IN PRESENCE OF T'E TANK FIG. 11

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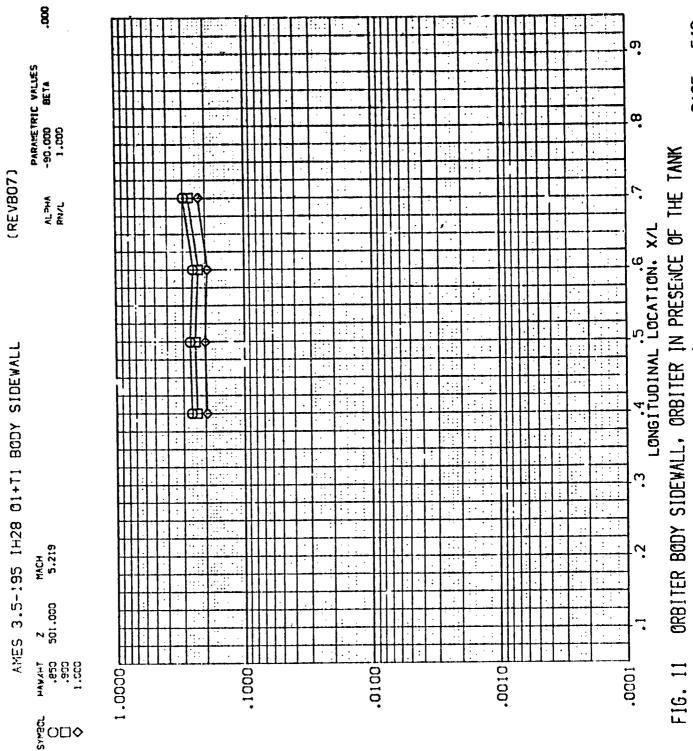
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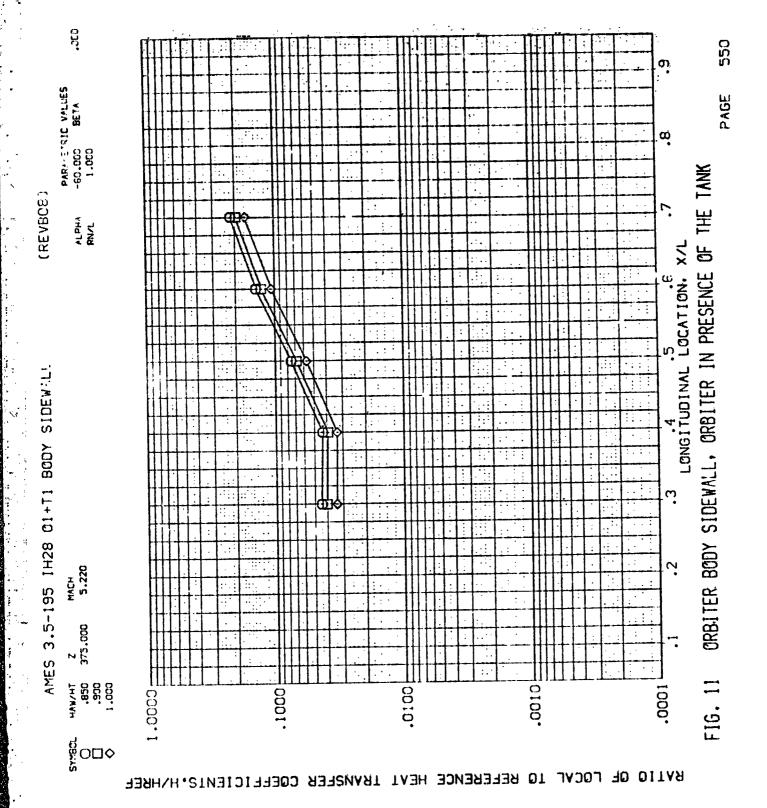
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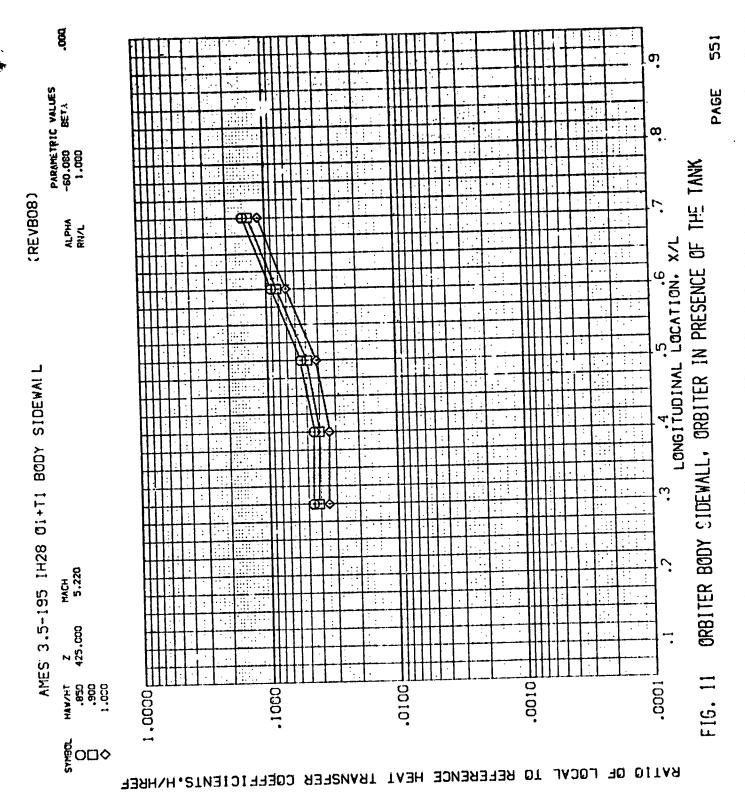
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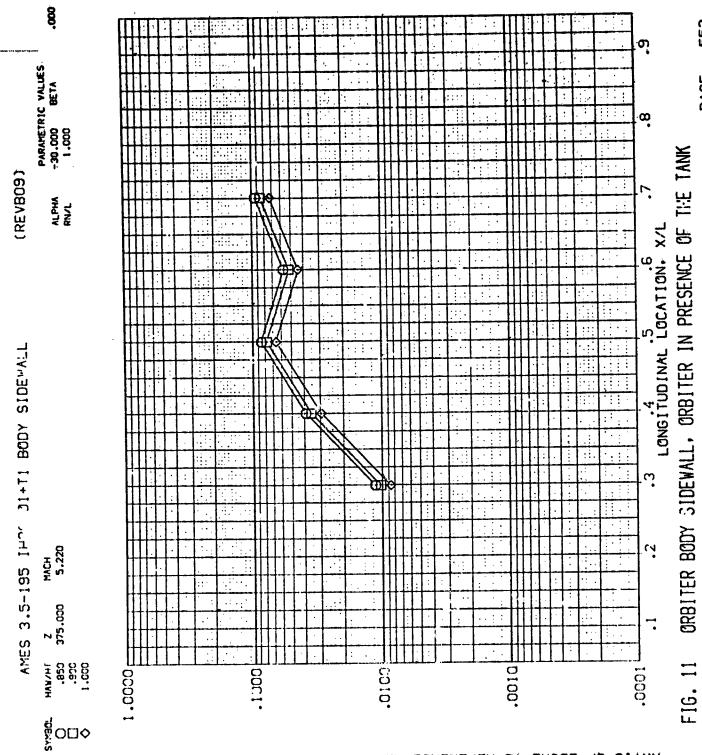
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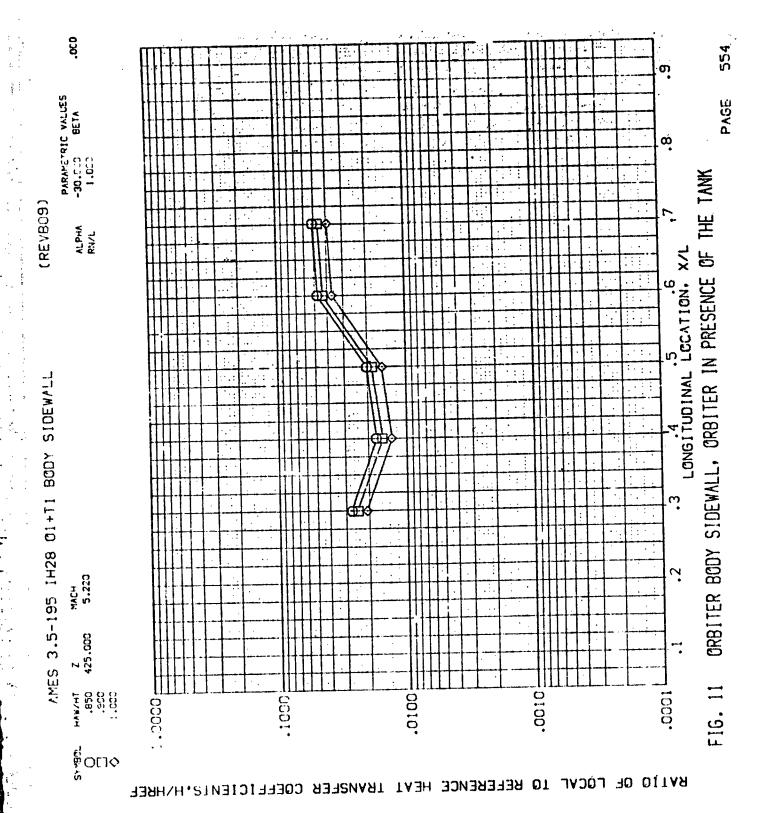
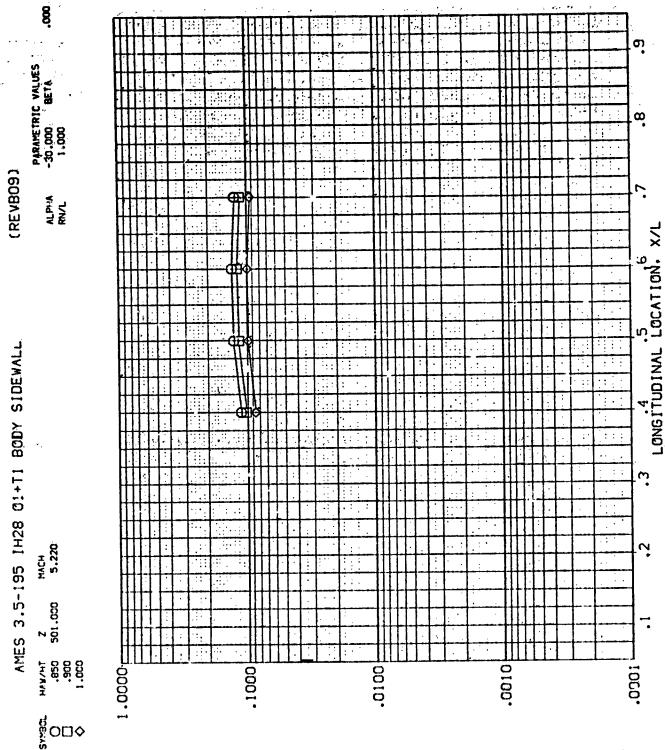
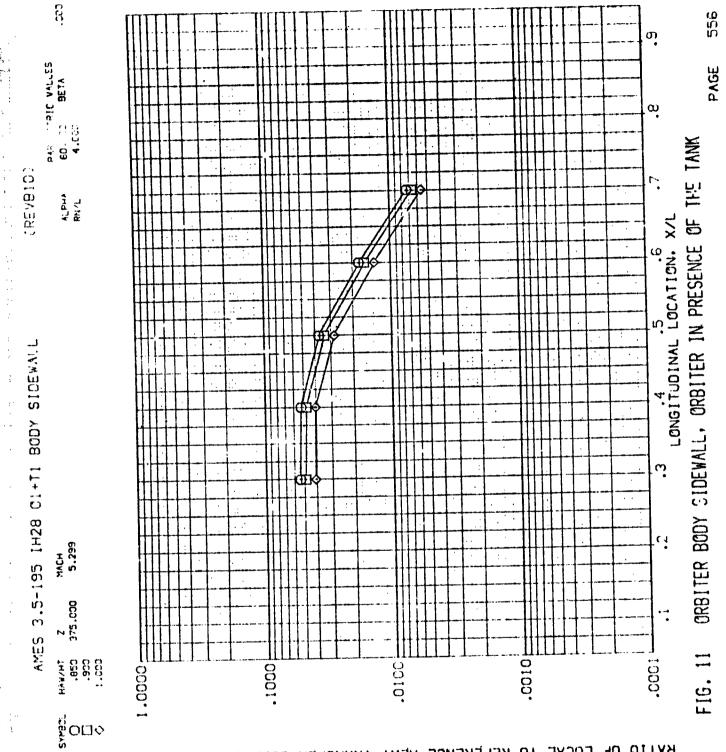


FIG. 11 ORBITER BODY SIDEWALL, ORBITER IN PRESENCE OF THE TANK



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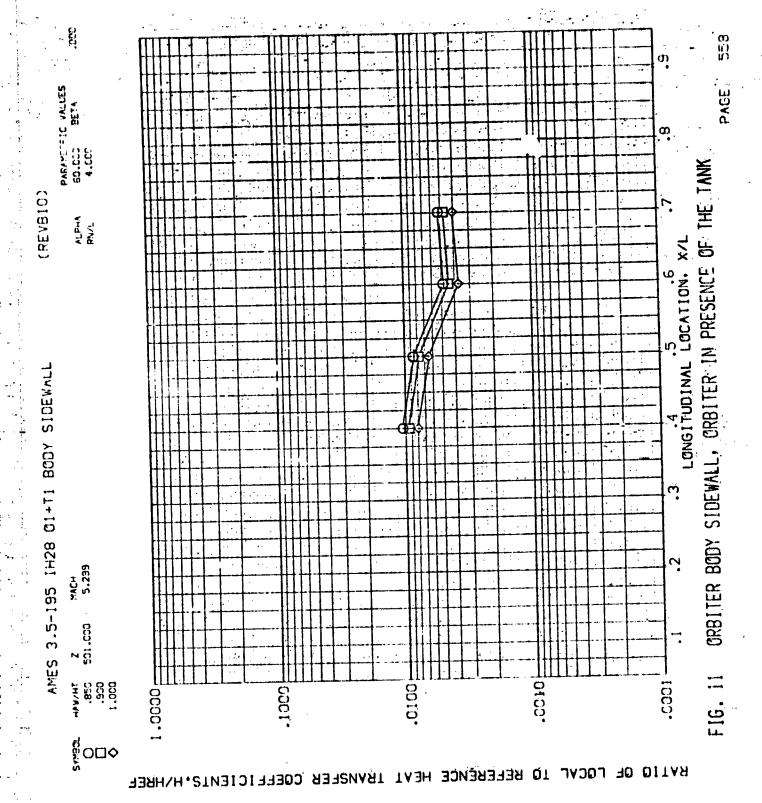


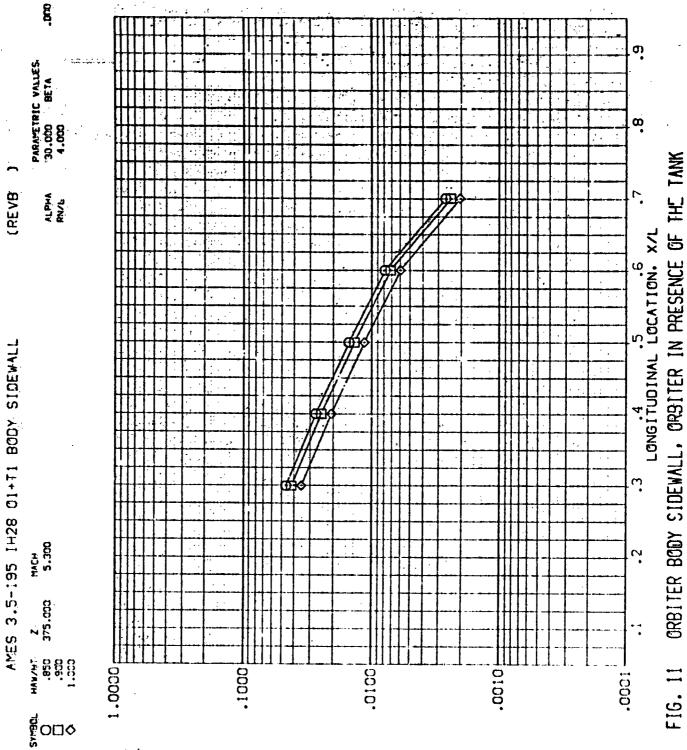
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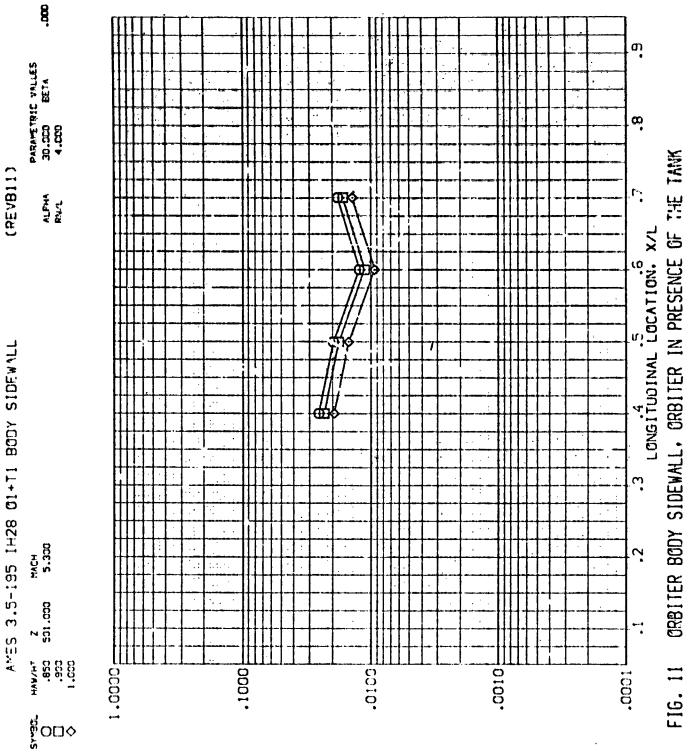
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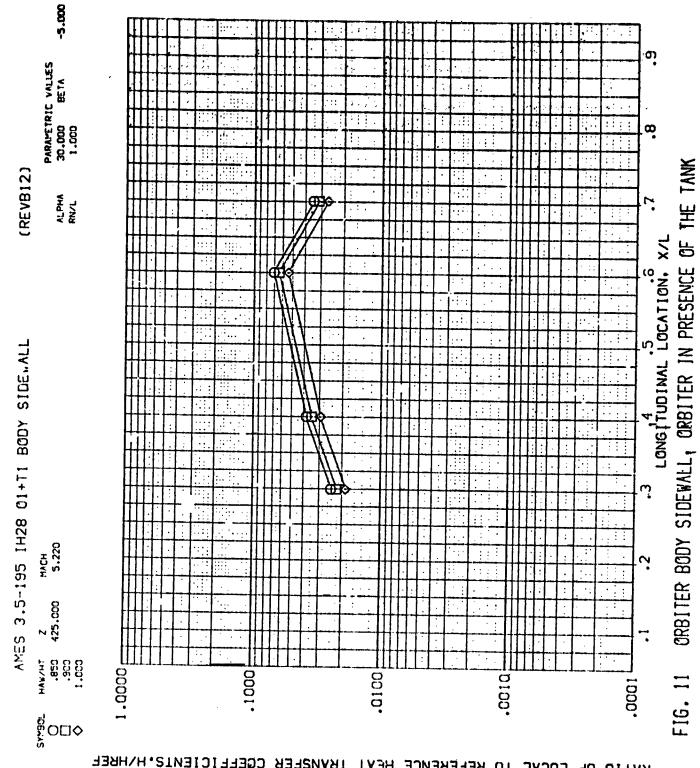
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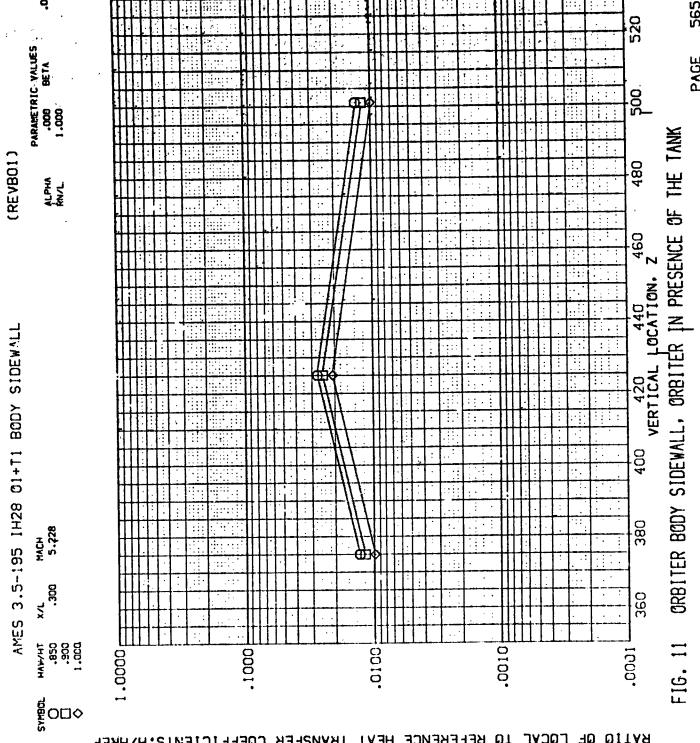
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FIG. 11 ORBITER BODY SIDEWALL, ORBITER IN PRESENCE OF THE TANK

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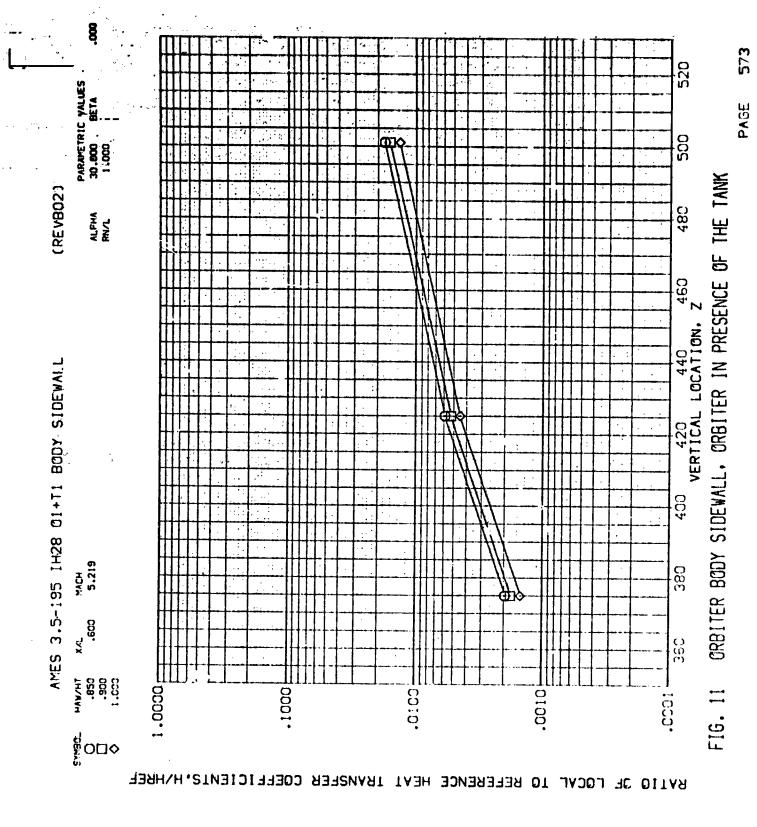
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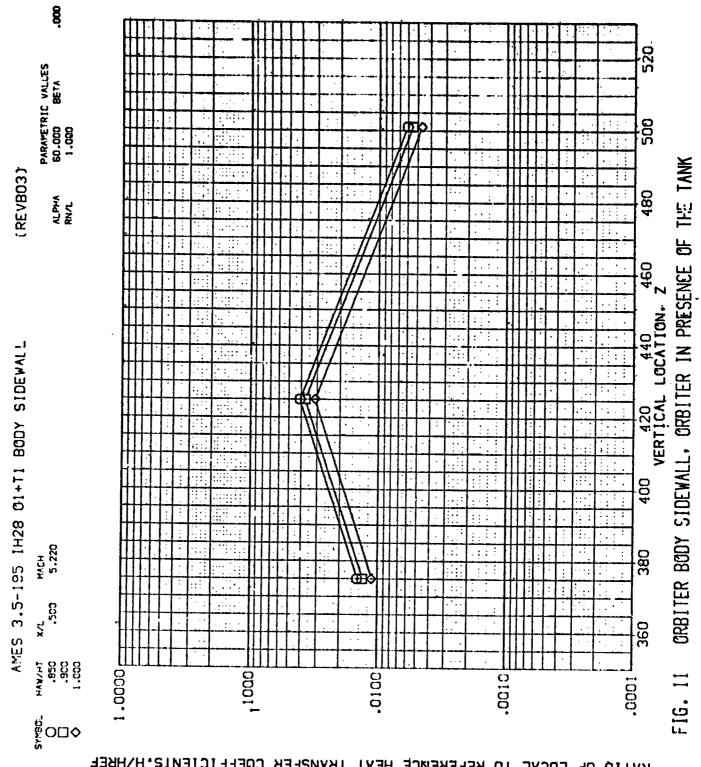
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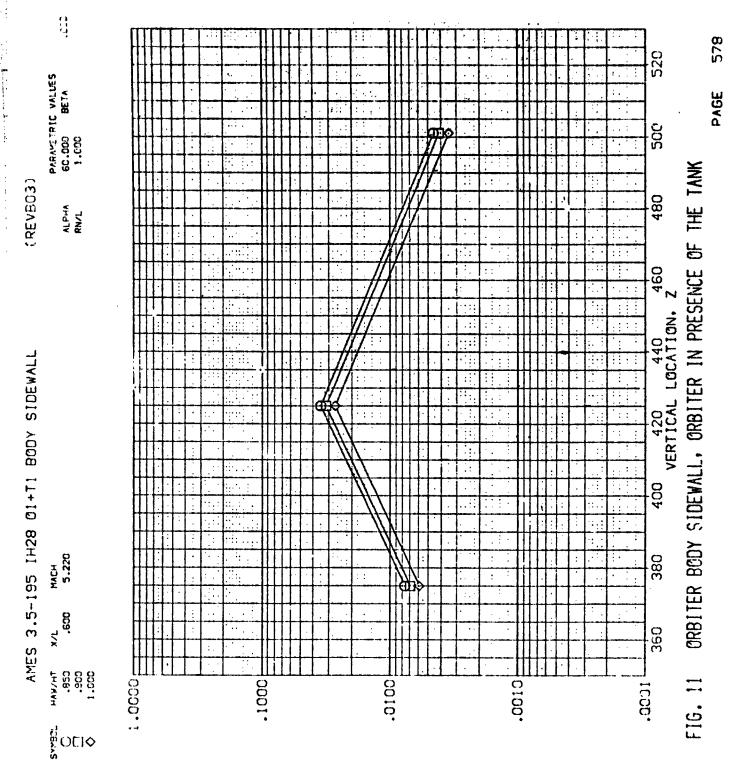


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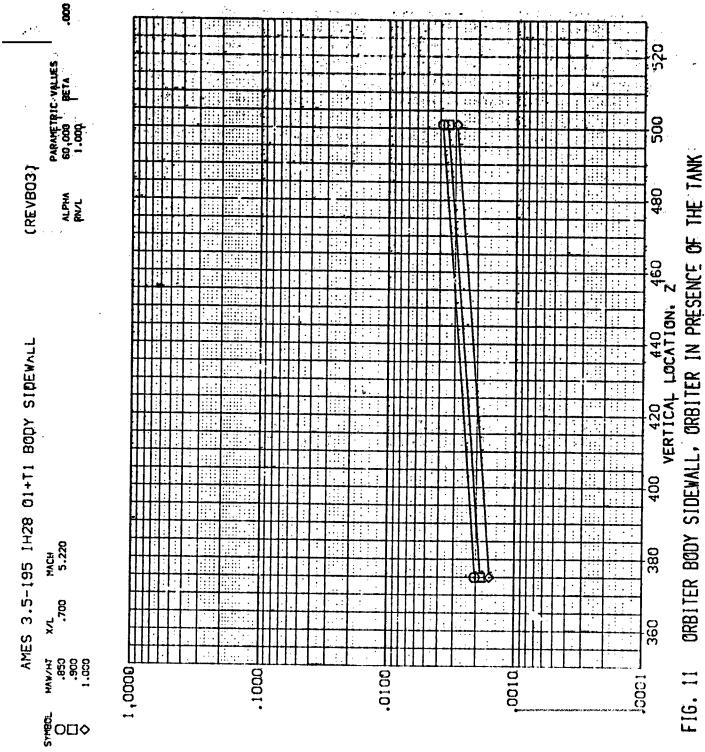
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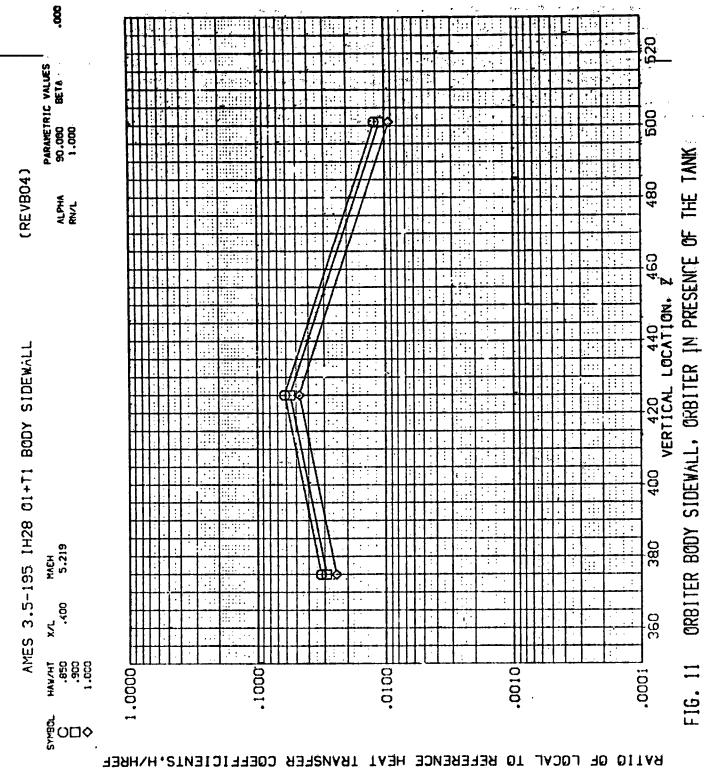
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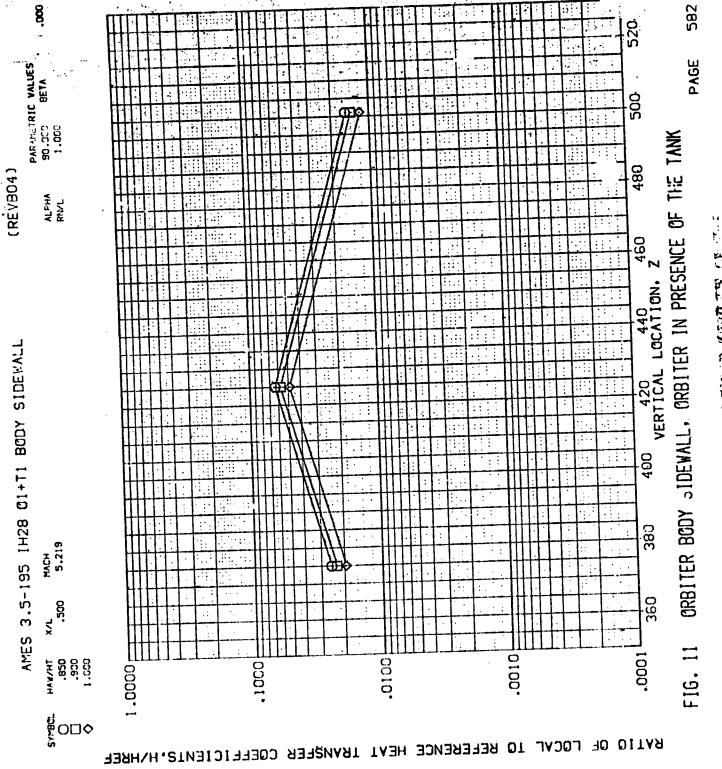
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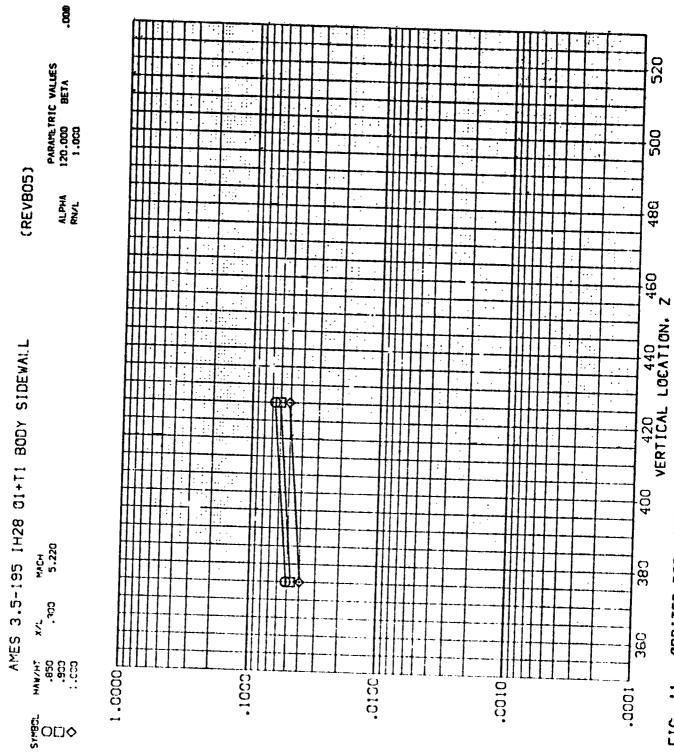
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ORBITER BODY SIDEWALL, ORBITER IN PRESENCE OF THE TANK FIG. 11

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ORBITER BODY SIDEWALL, ORBITER IN PRESENCE OF THE TANK

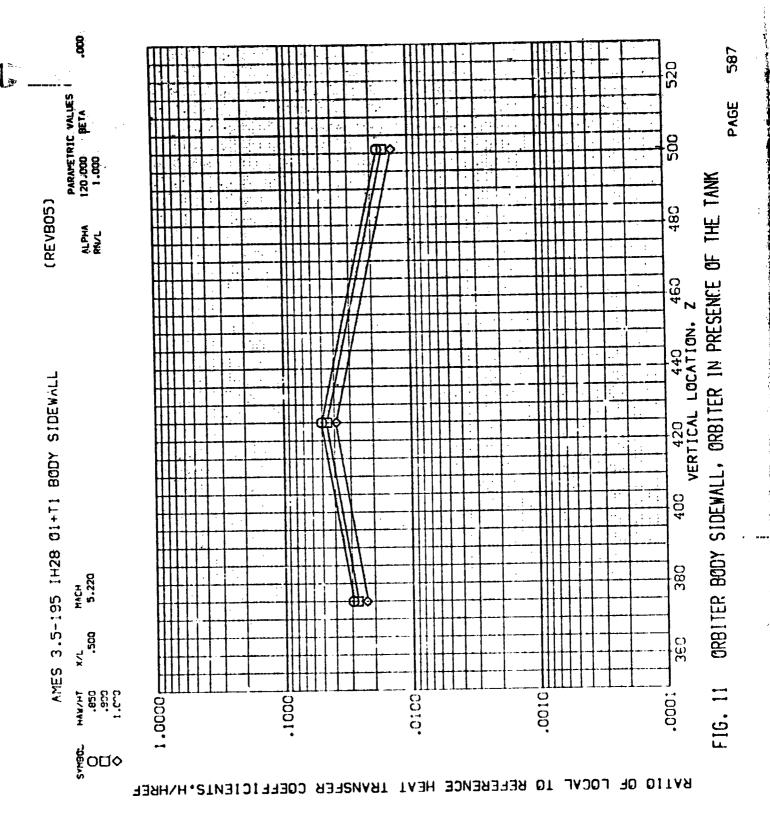


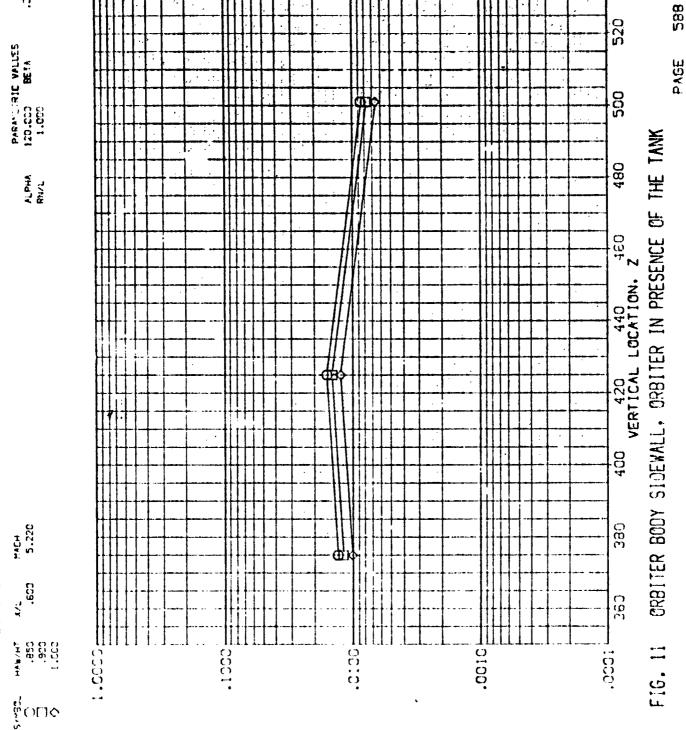
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FIG. 11

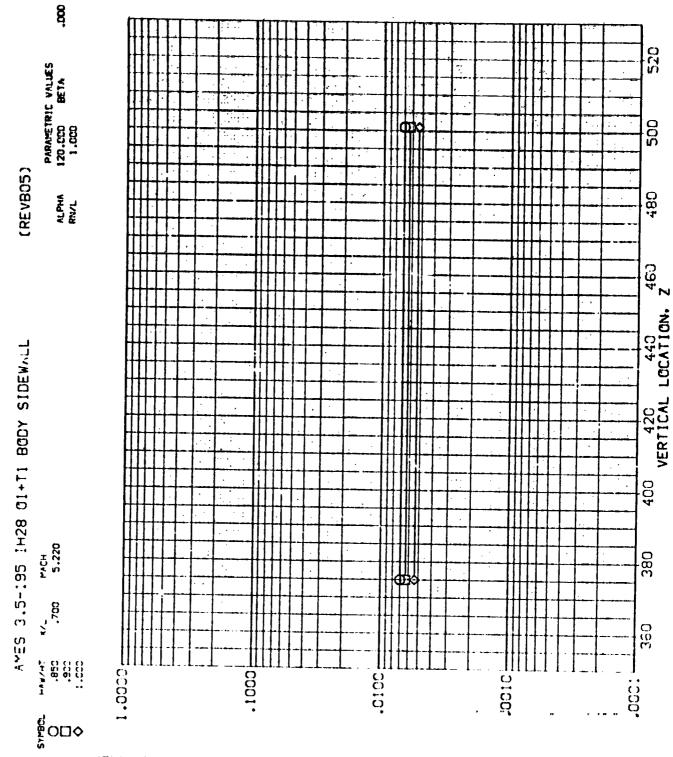
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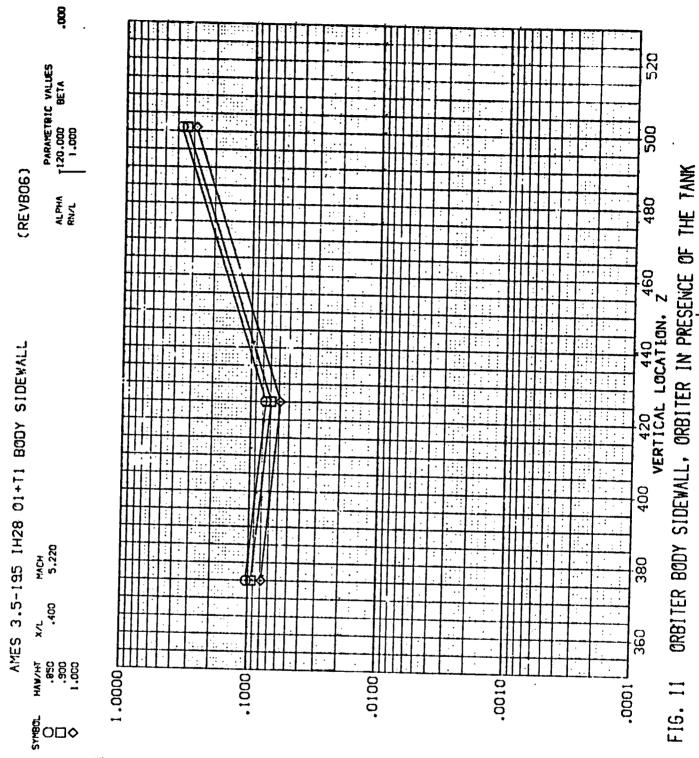


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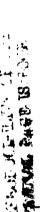
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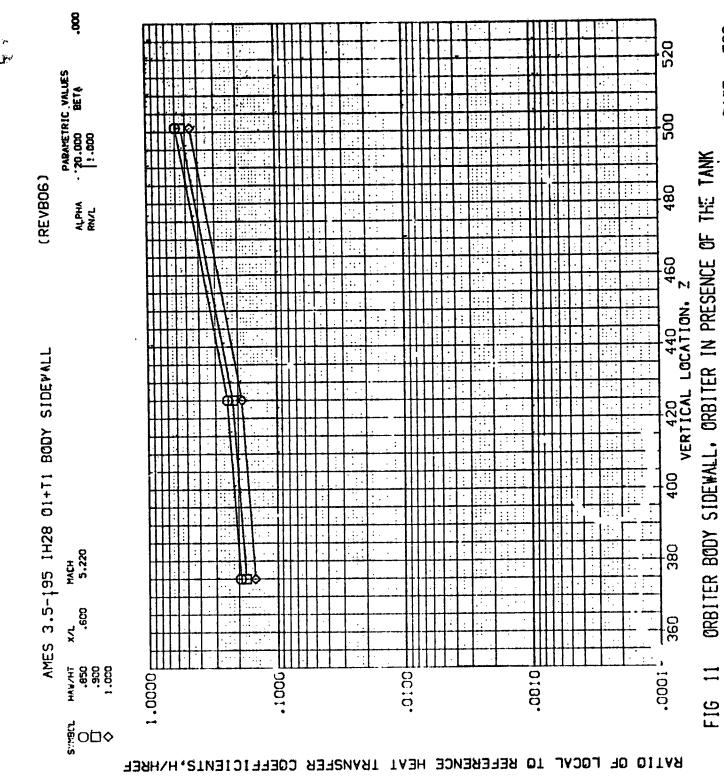
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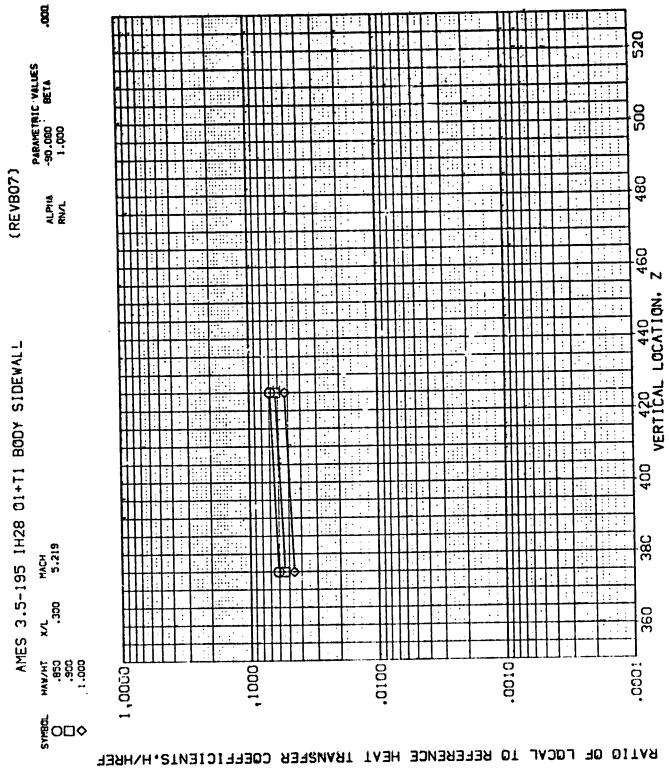
FIG. 11 GRBITER BODY SIDEWALL, ORBITER IN PRESENCE OF THE TANK



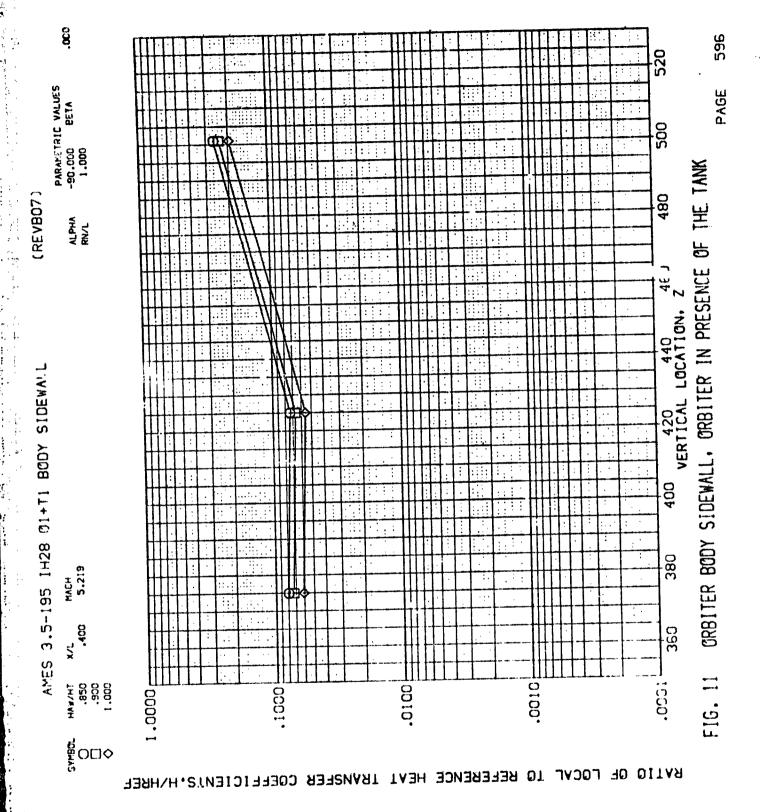
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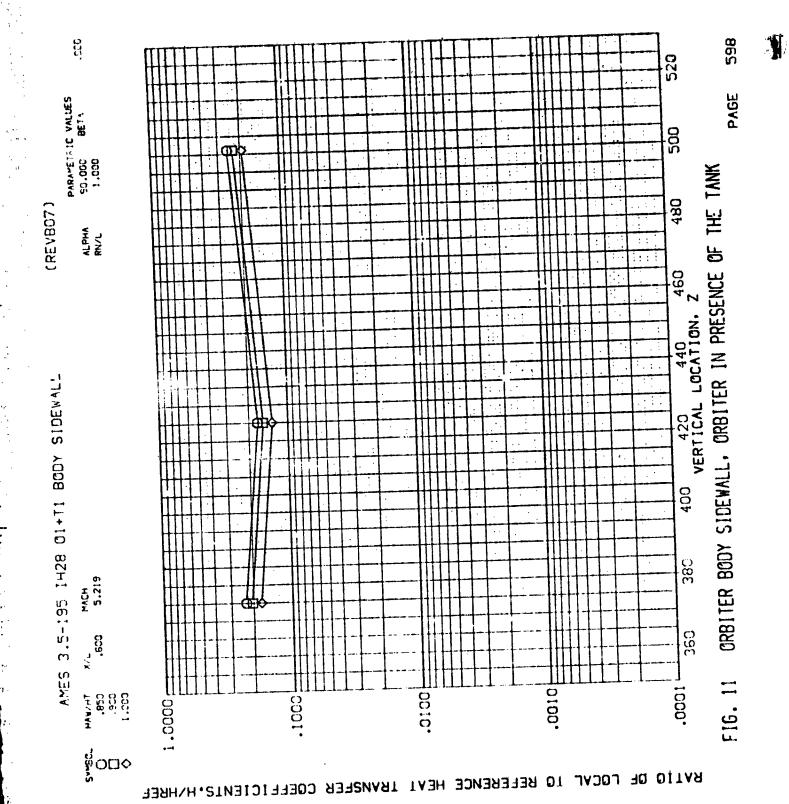


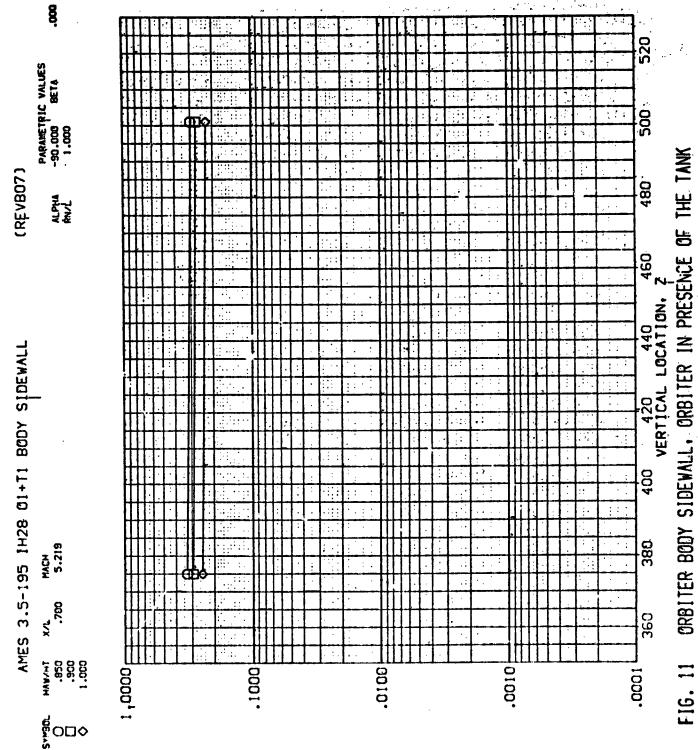


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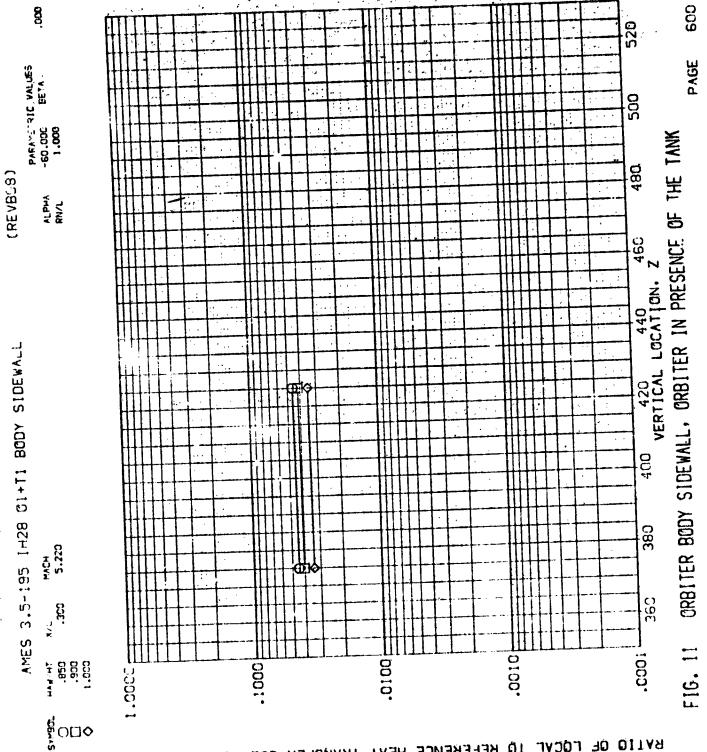


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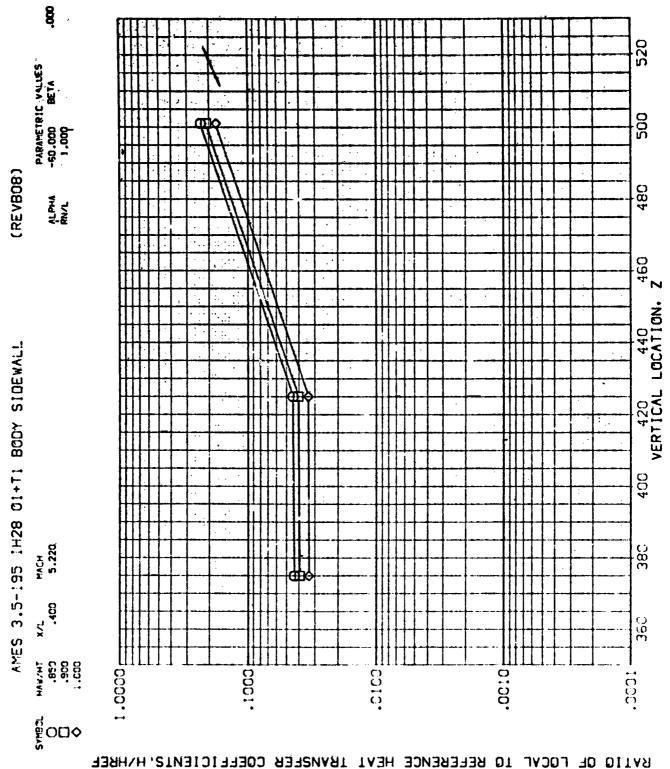




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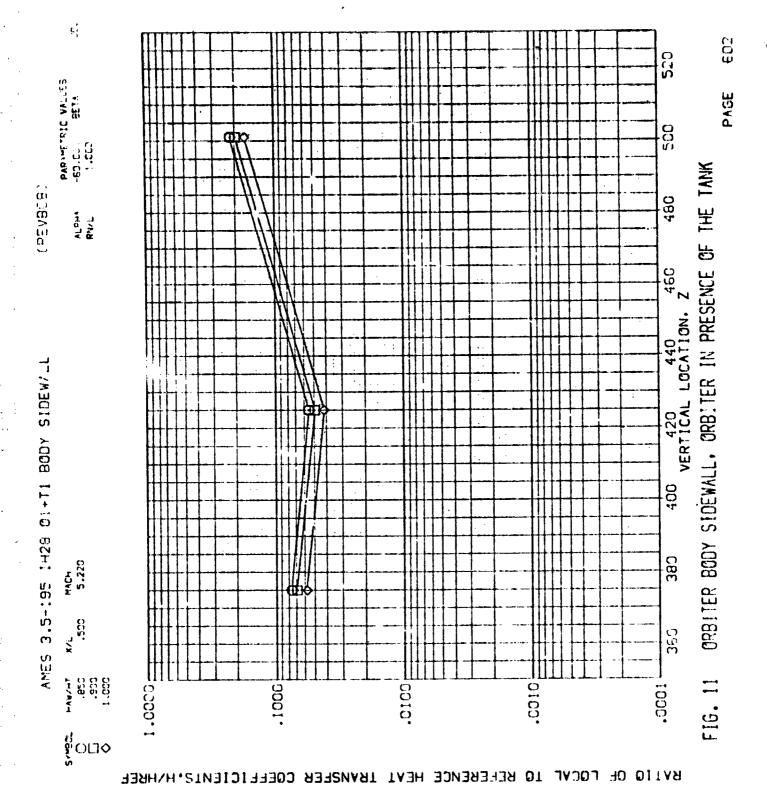
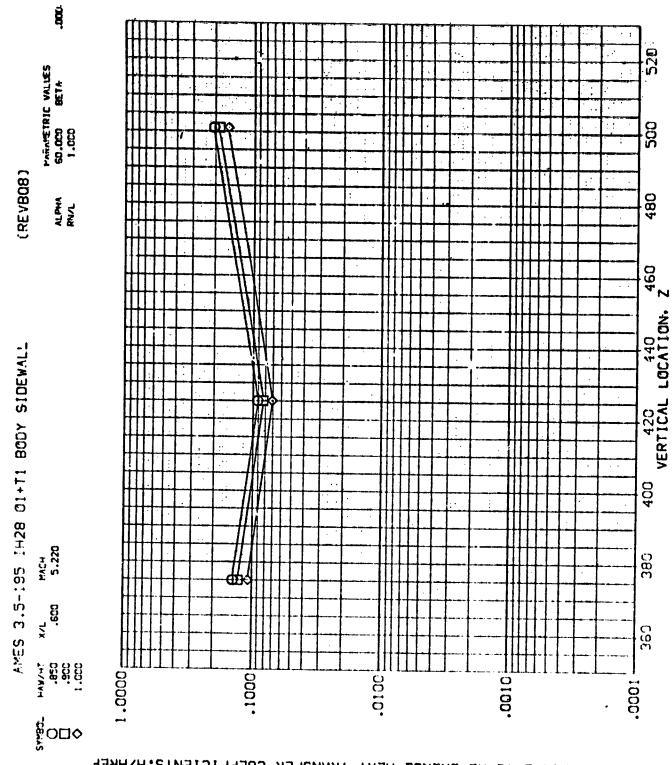


FIG. 11 ORBITER BODY SIDEWALL, ORBITER IN PRESENCE OF THE TANK

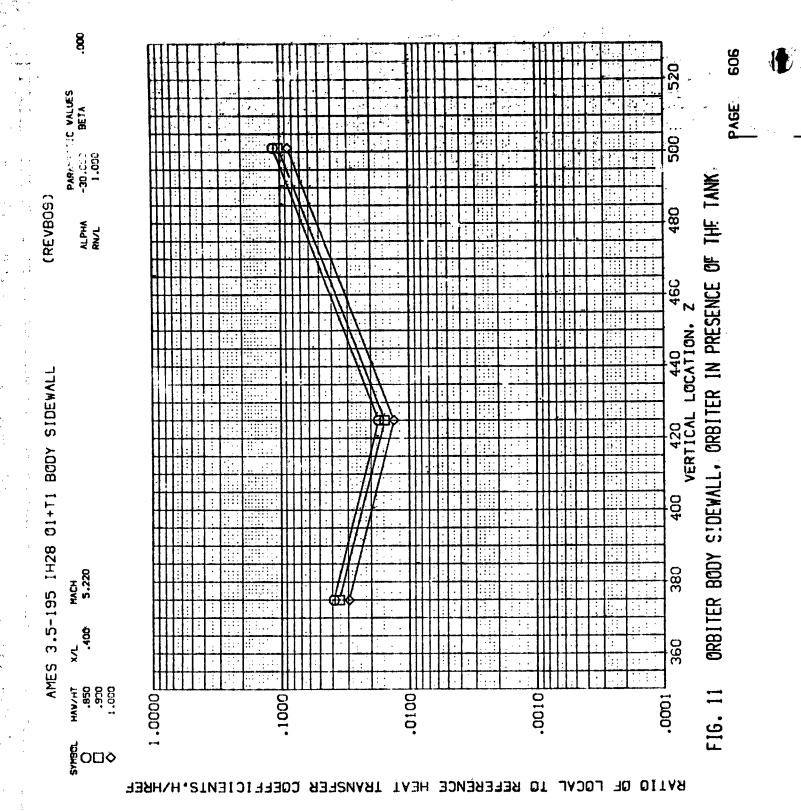


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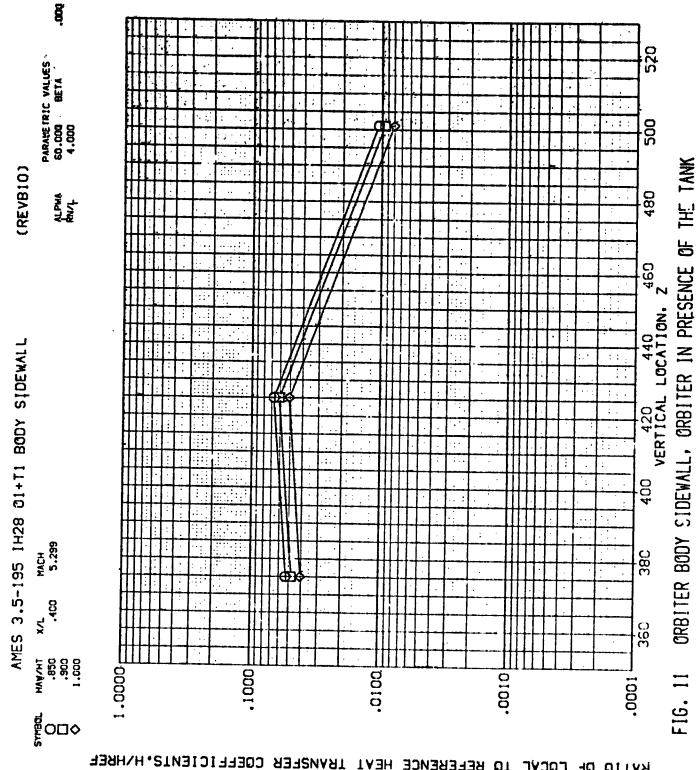
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ORBITER BODY SIDEWALL, ORBITER IN PRESENCE OF THE TANK

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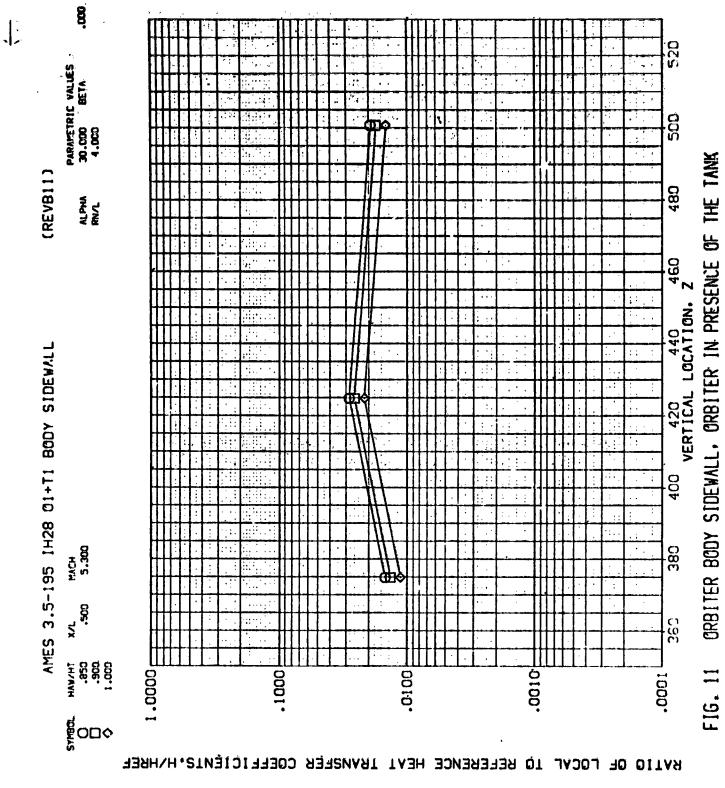
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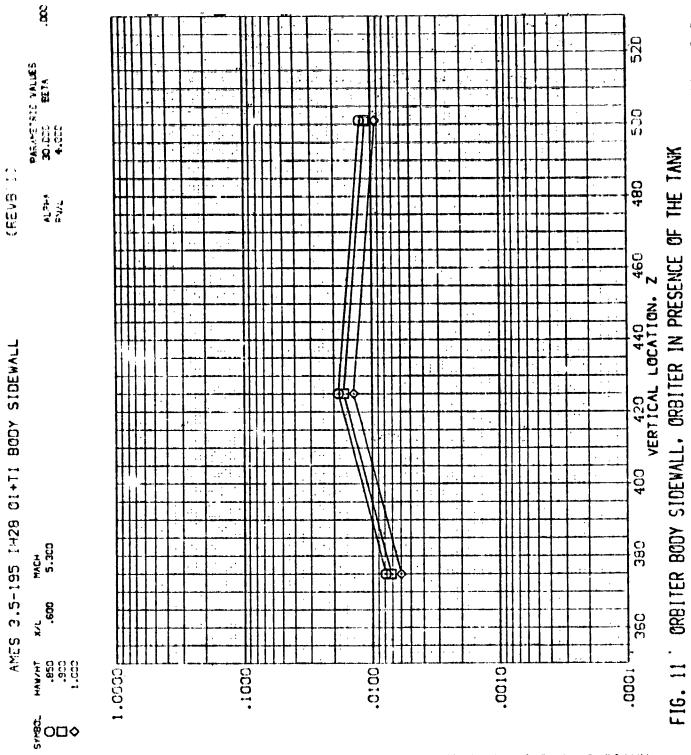
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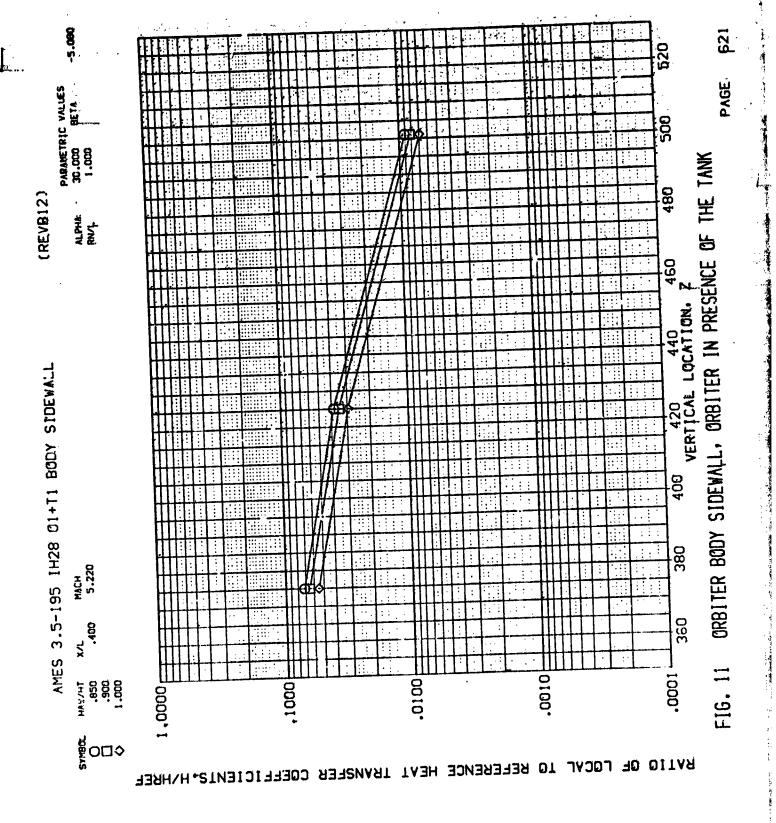
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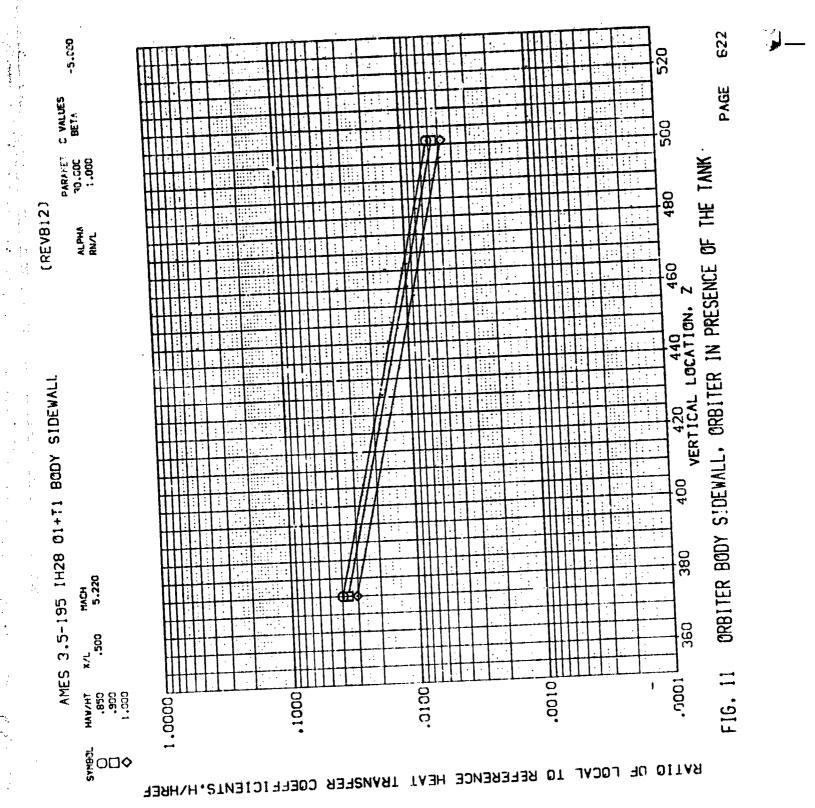


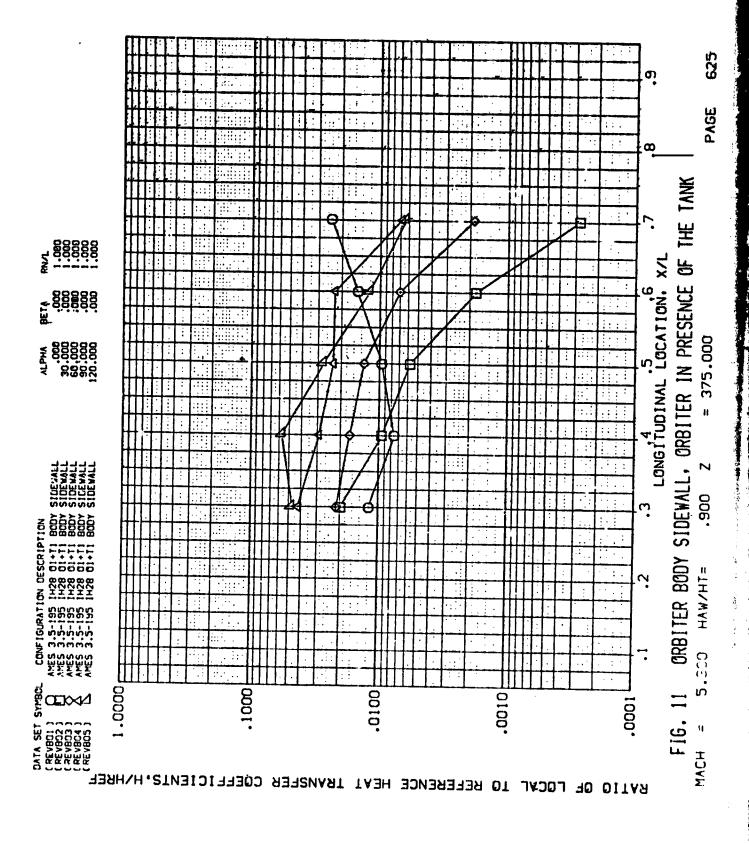


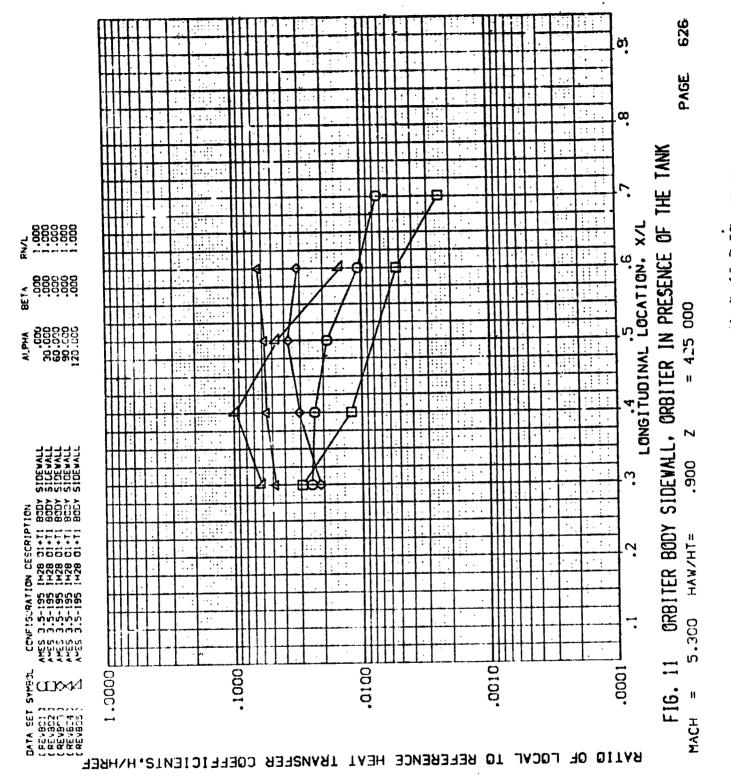
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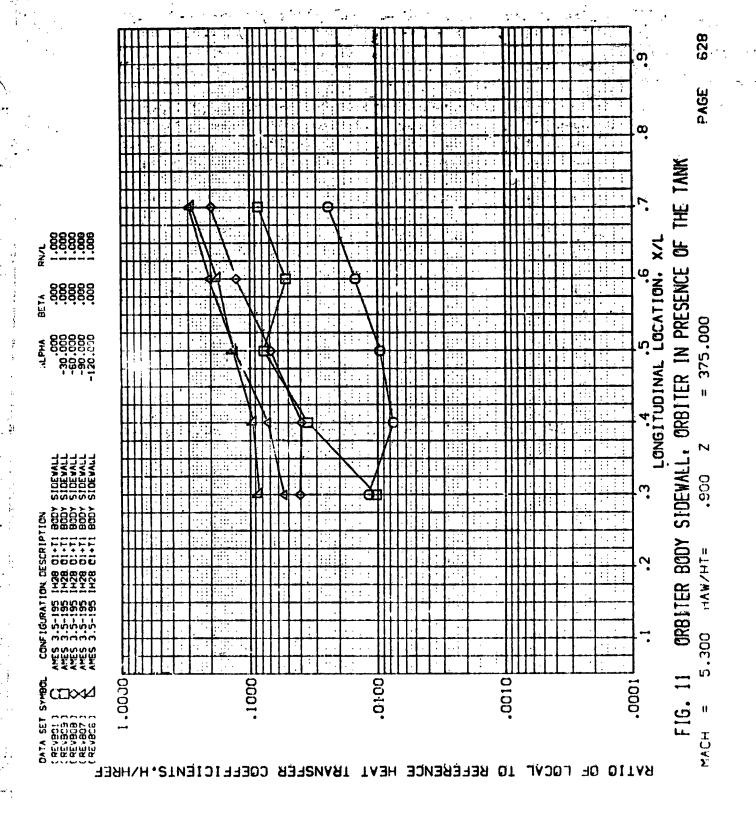


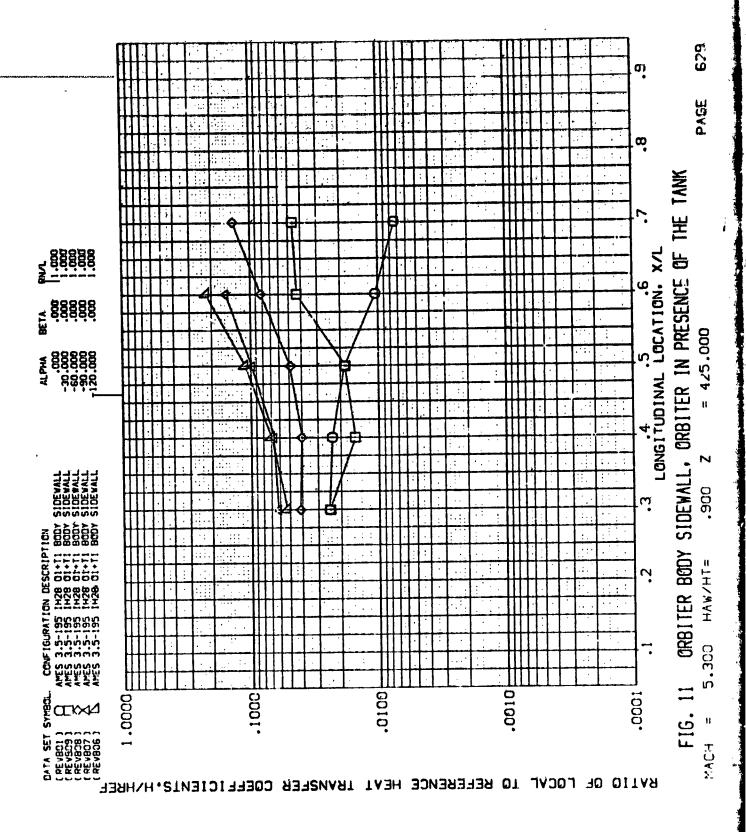






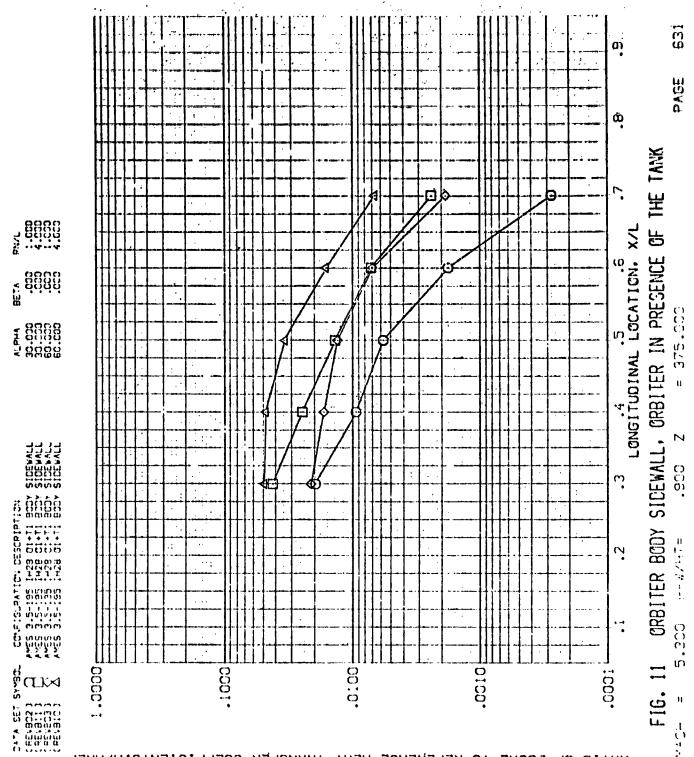
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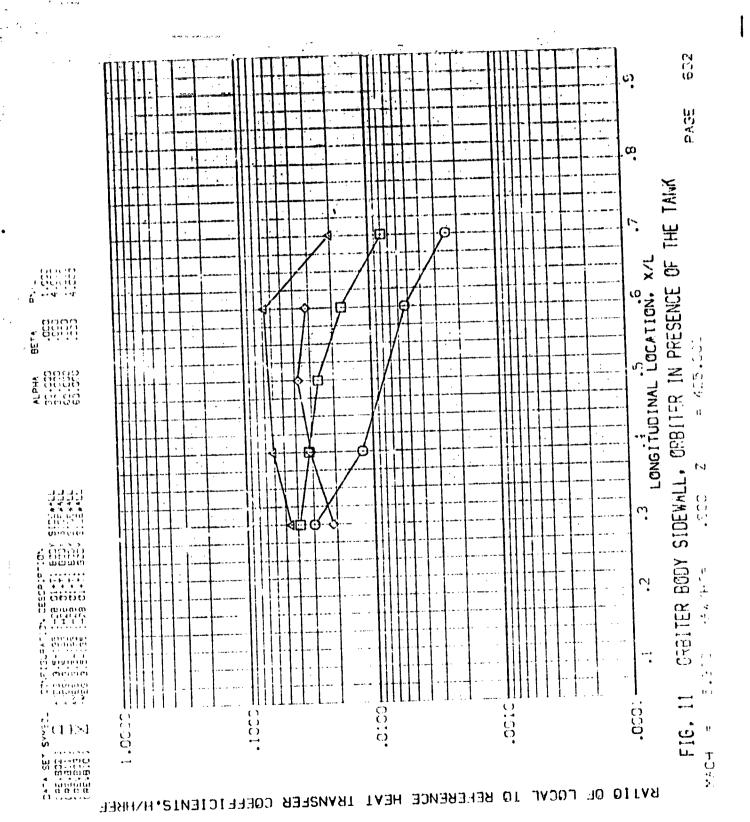


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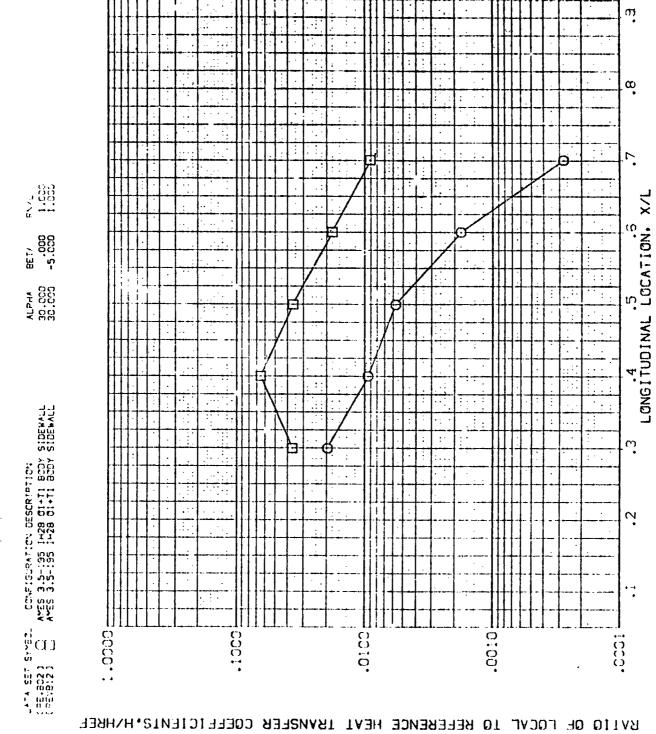


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633 ຫ 114.1 . : : PAGE ω FIG. 11 GRBITER BODY SIDEWALL, ORBITER IN PRESENCE OF THE TANK . : • • LONGITUDINAL LOCATION. X/L 501.000 : ALPHA 30.000 50.000 50.000 50.000 50.000 ---. . . ::. <u>.:</u>: . 900. : លីសីសីស • : : : 5005 5005 : : : : ::: ក្រុក ក្កក្រុក ក្រុក ក្ .0010H 0010 10001 .0001 រ្តុំ ពេល 1.0000 11 w come w

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CRBITER BODY SIDEWALL, ORBITER IN PRESENCE OF THE TANK FIG. 11

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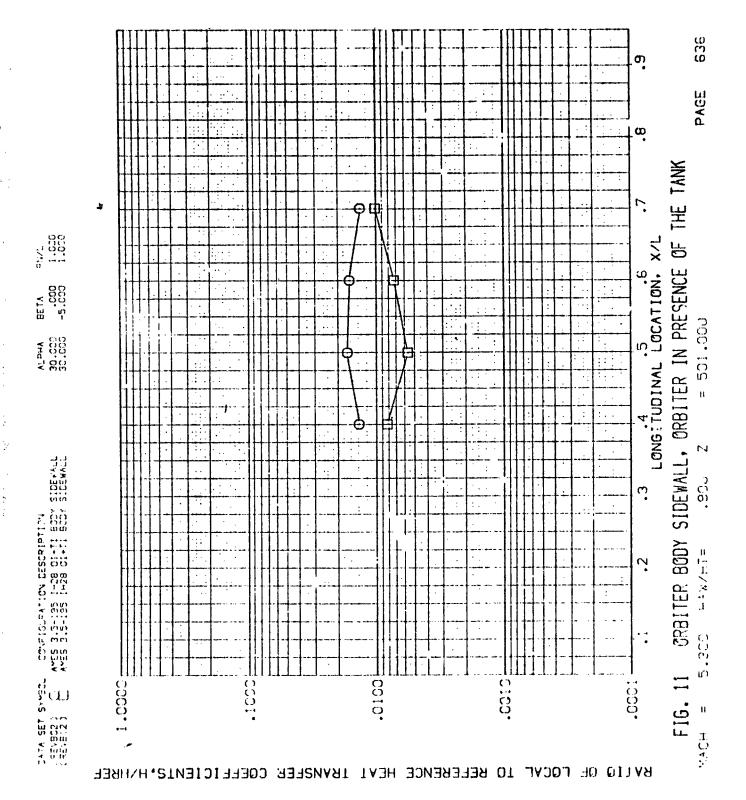
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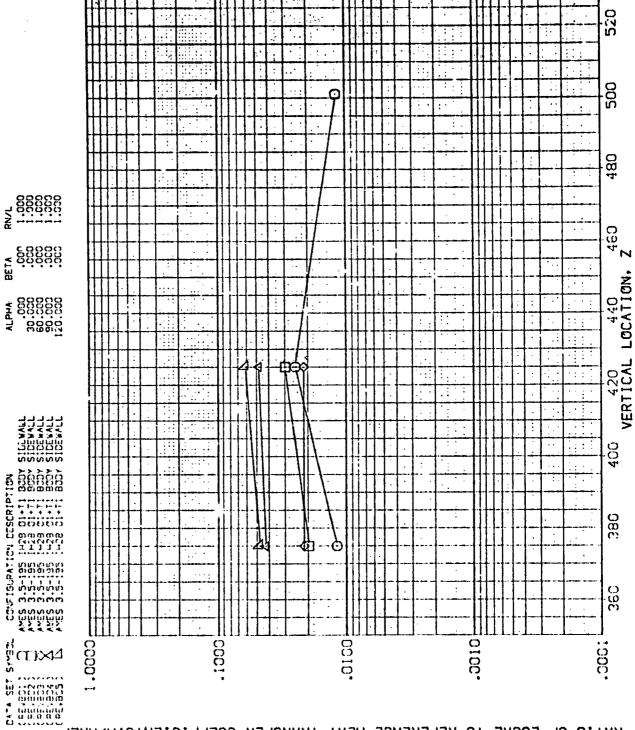
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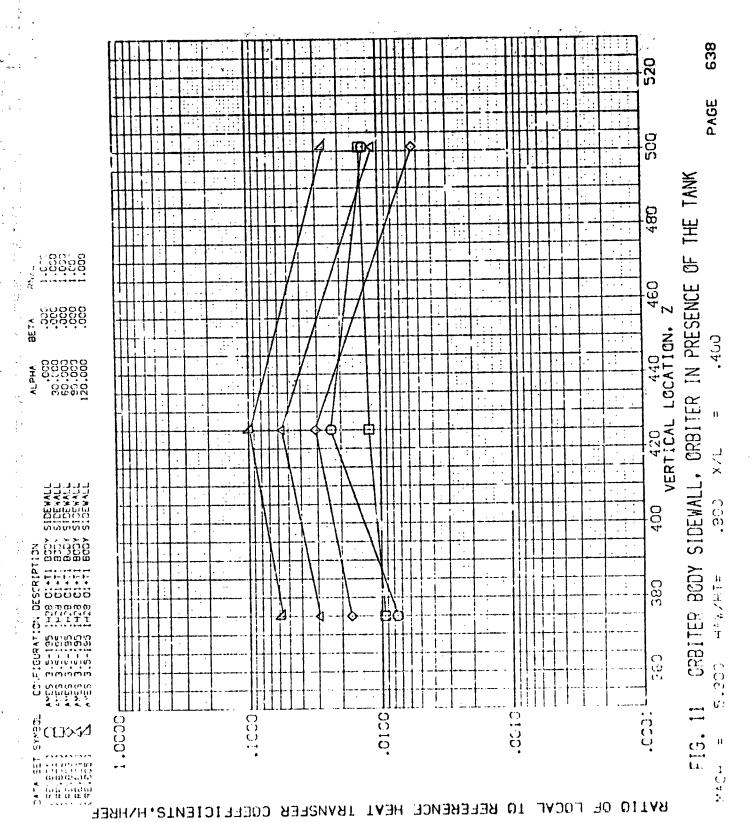
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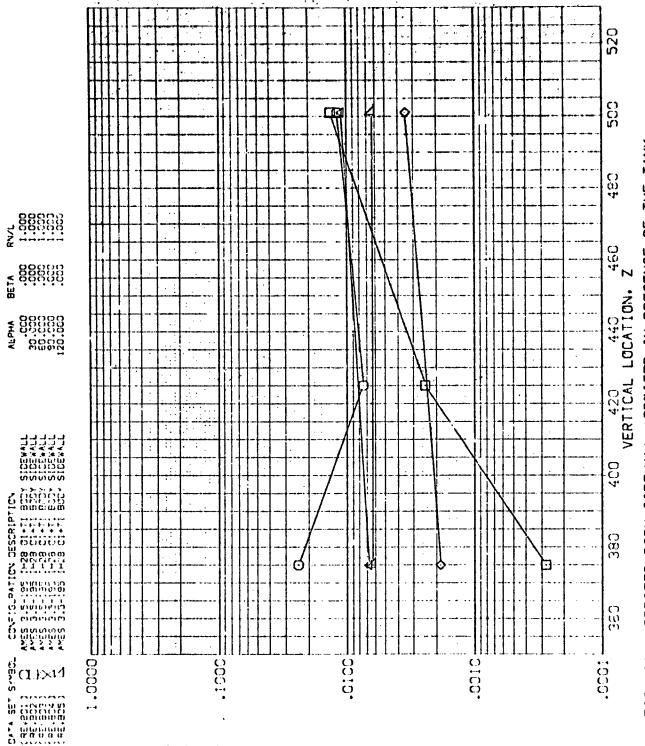
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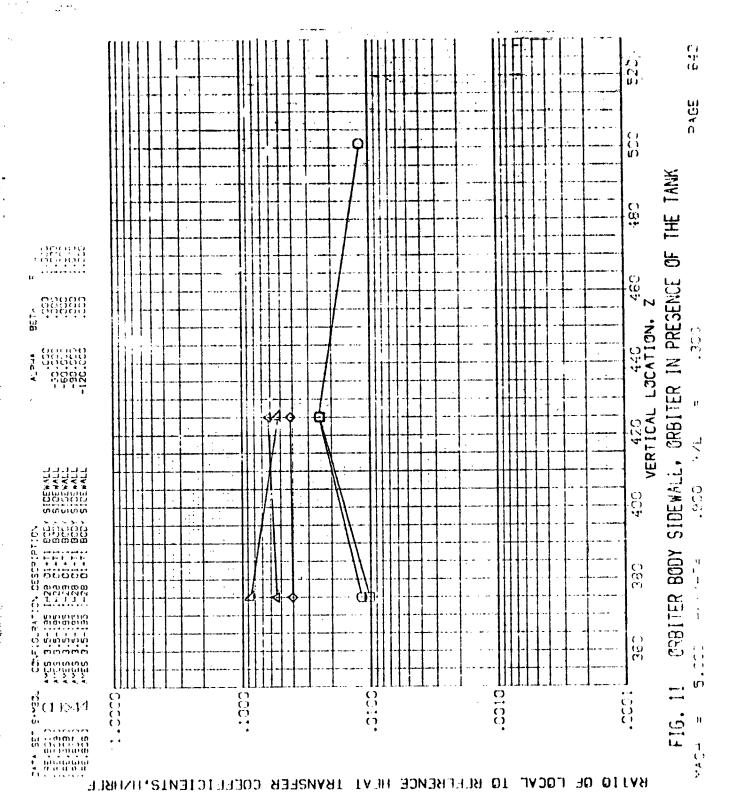


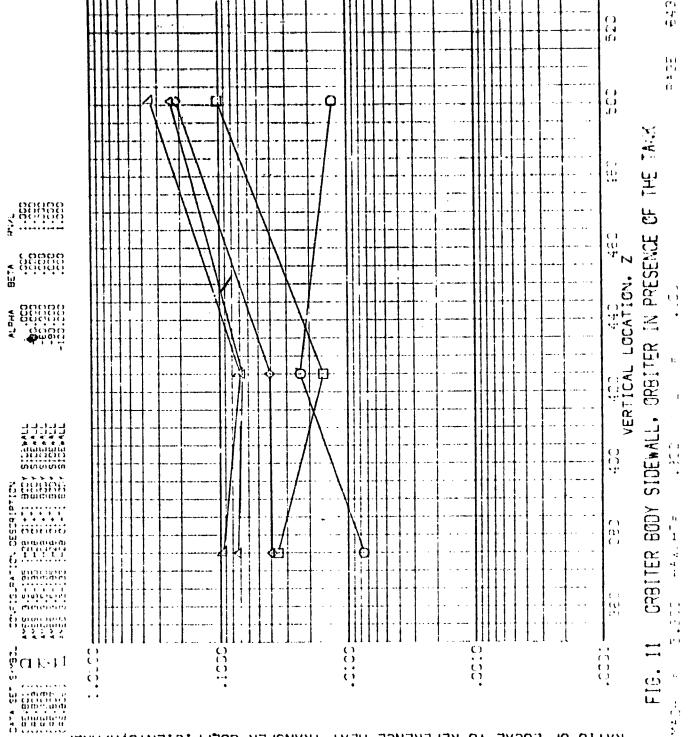
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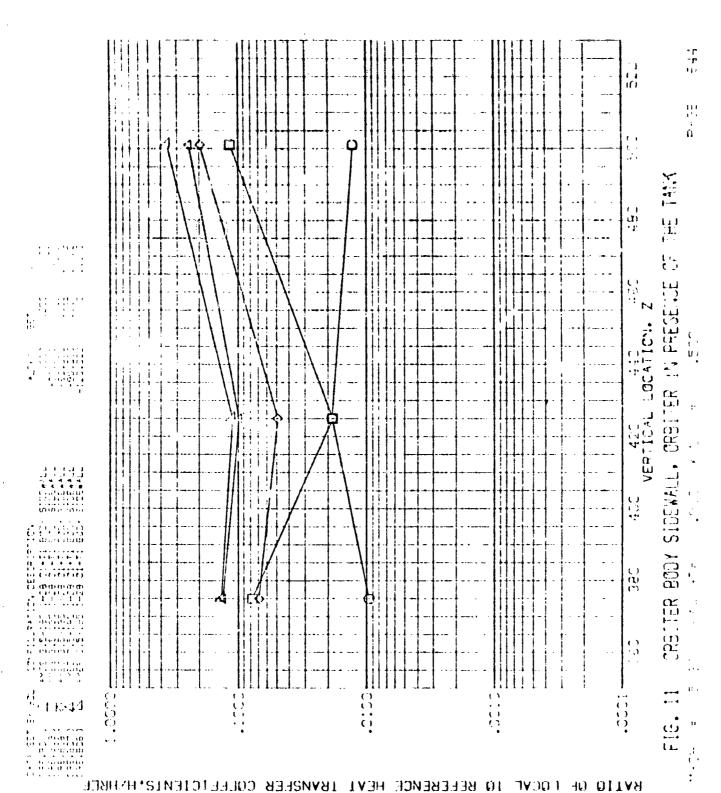


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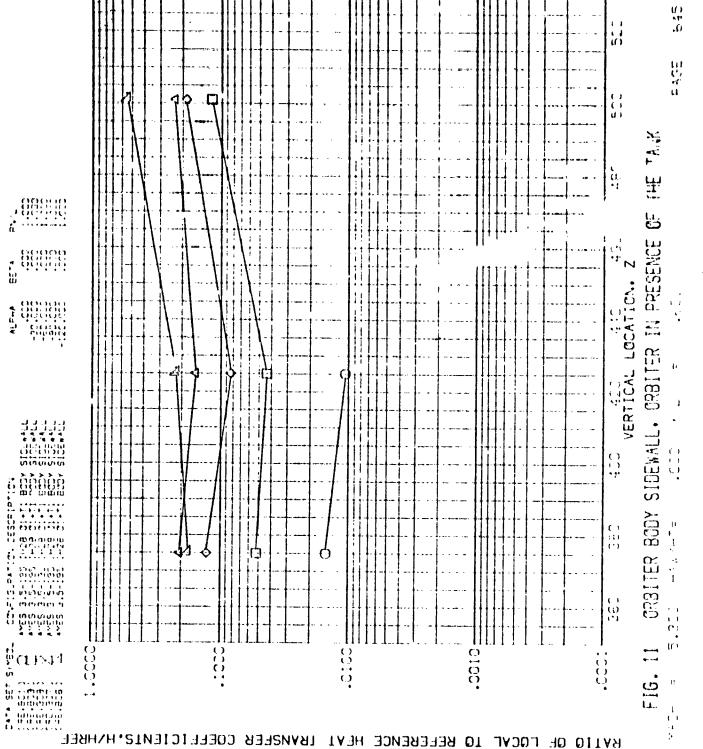
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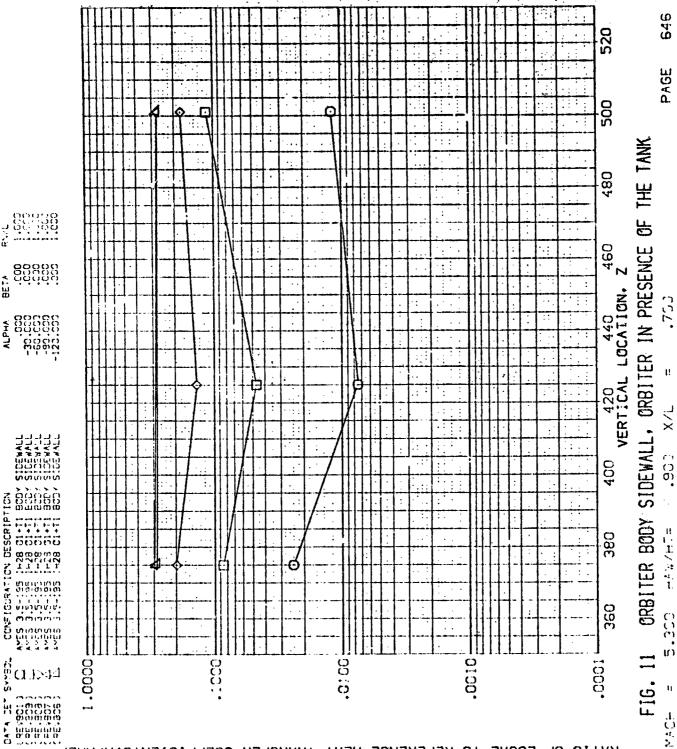
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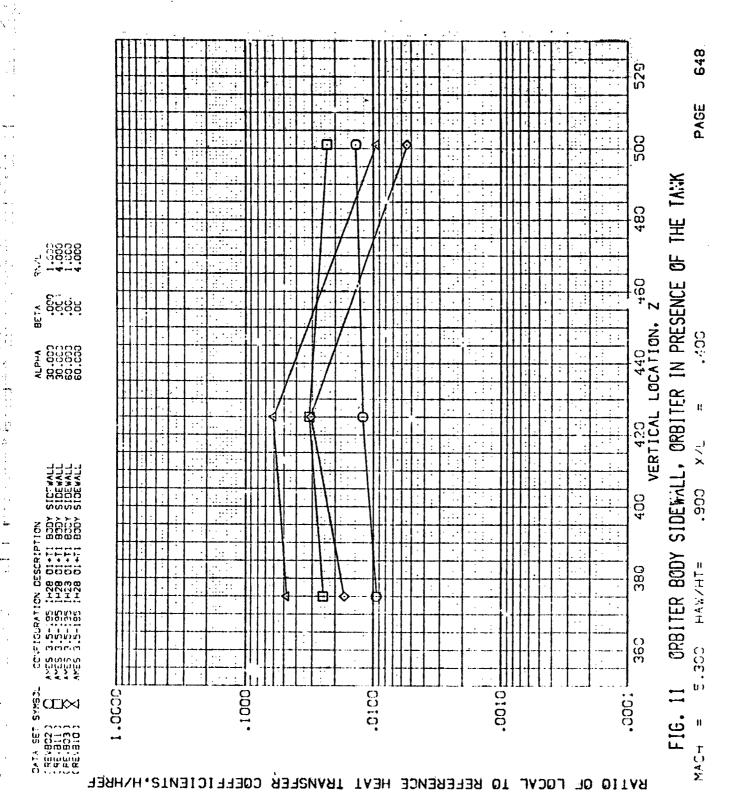
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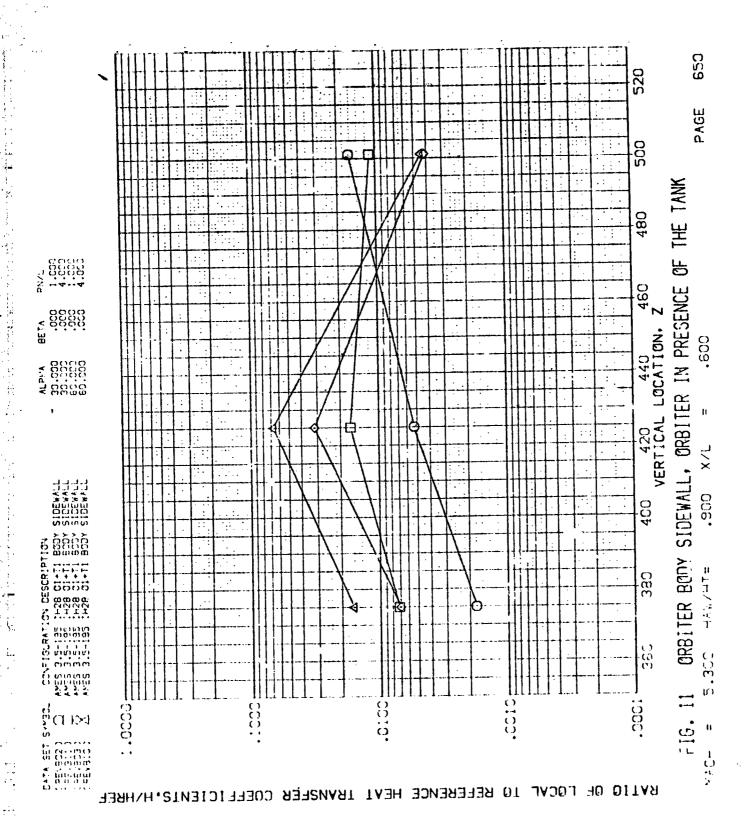
; ; 11: : ; IN PRESENCE OF THE TANK 480 : . 420 440 460 VERTICAL LOCATION, Z ; ; : SIDEWALL, ORBITER ::: 400 : : : : 1111 3000 1111 1111 380 360 .1000 .0100 .0010 1.0000 .0001 $\alpha x = 0$ 0474 SET RATIO OF LOCAL TO REFERENCE HEAT TRANSFER COEFFICIENTS, HAHREF

.900 CRBITER BODY FIG. 11

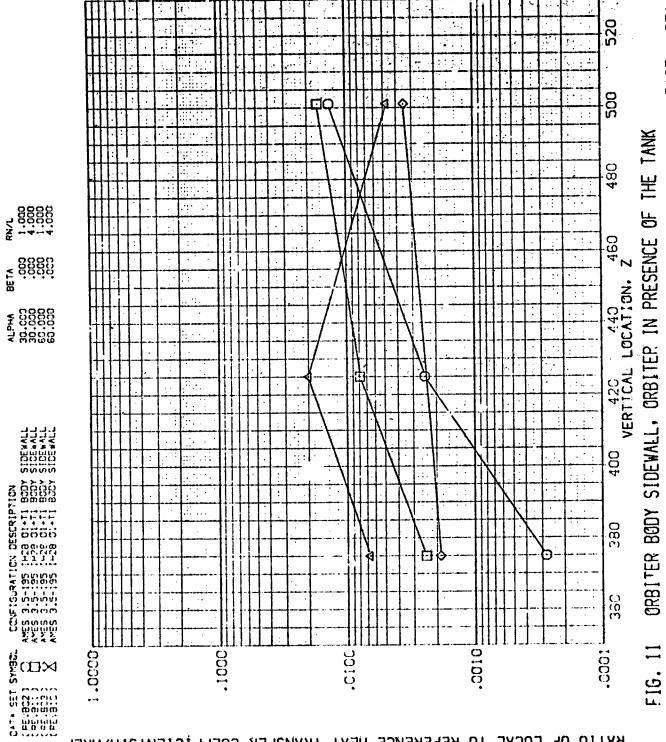


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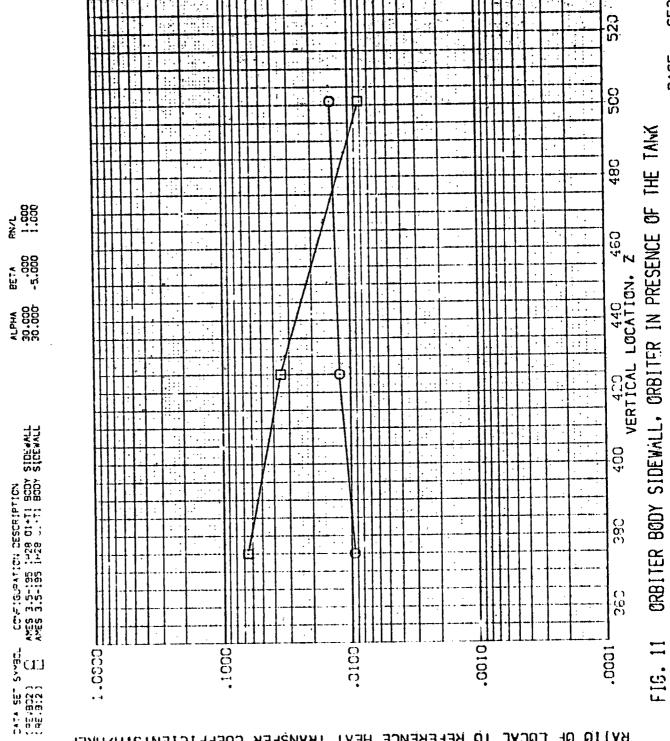
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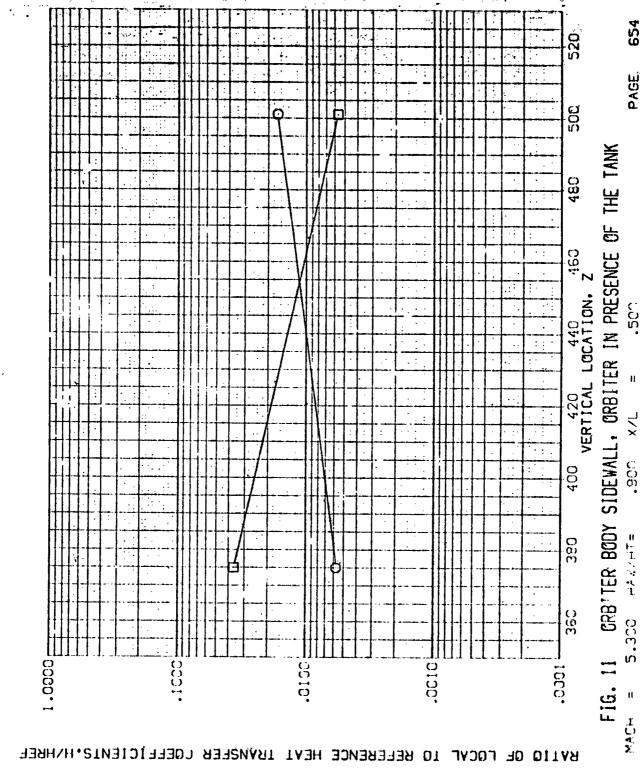
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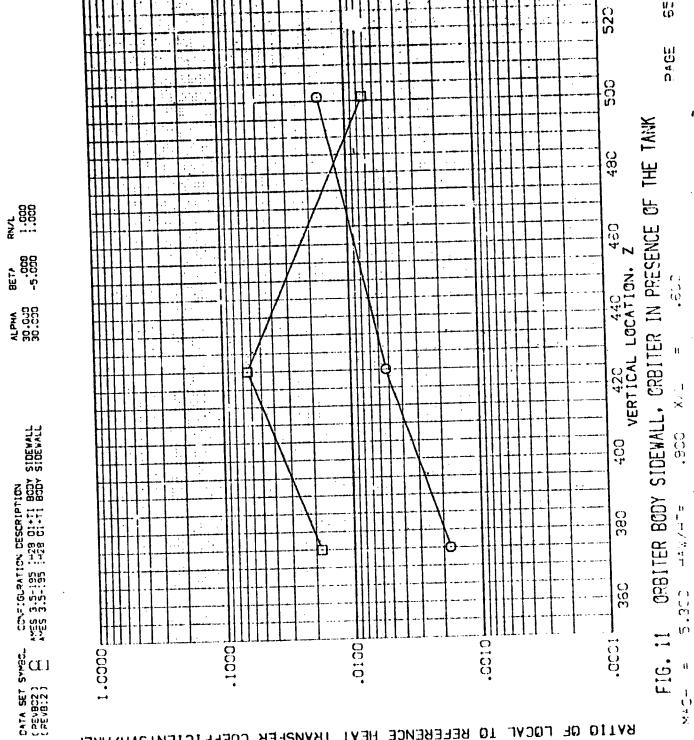
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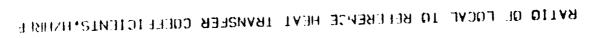


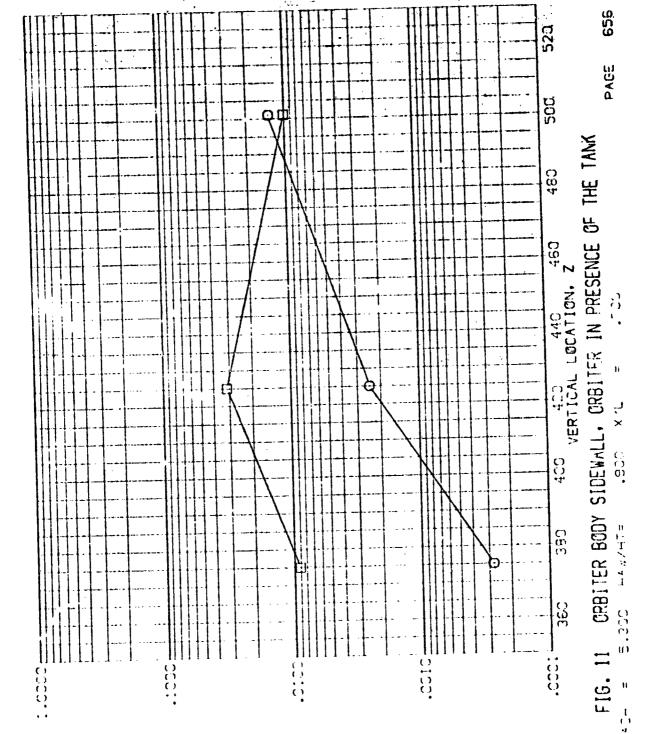
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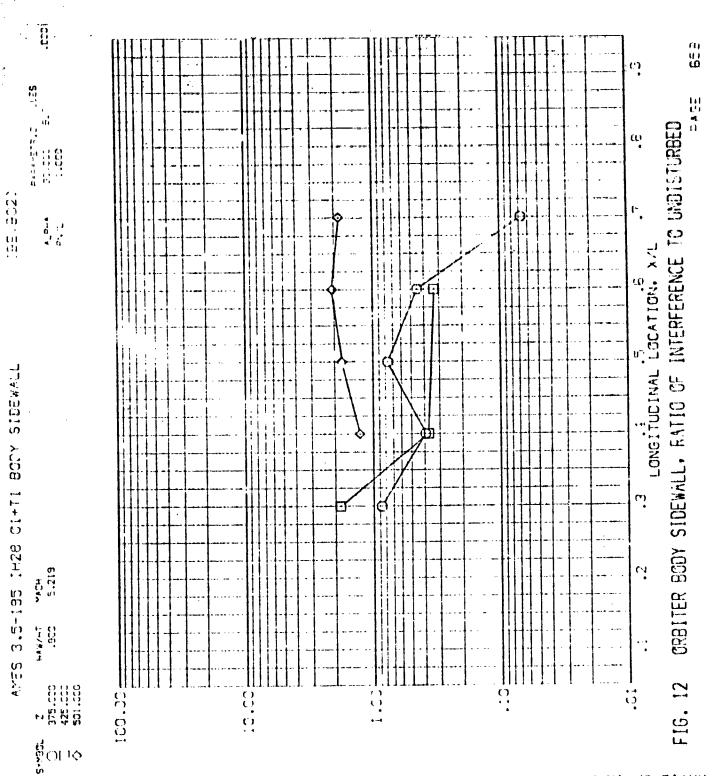
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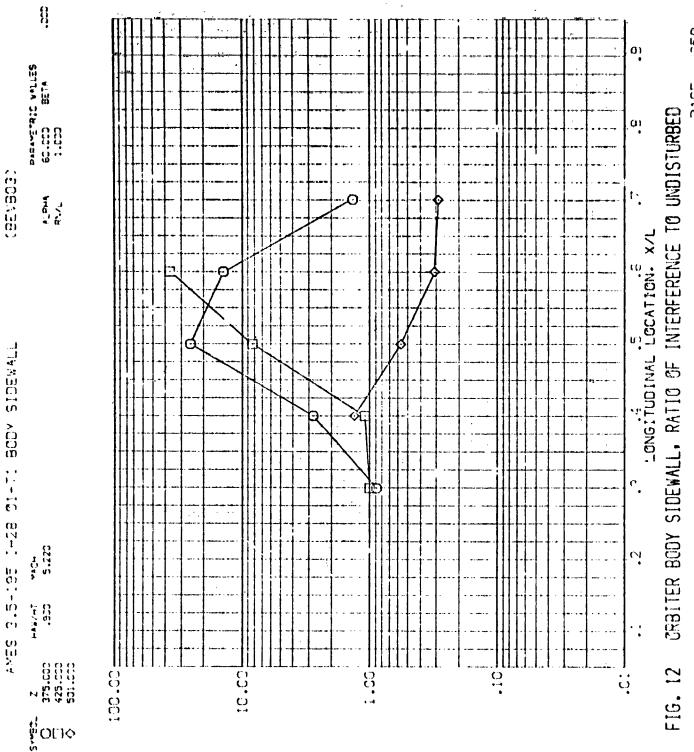


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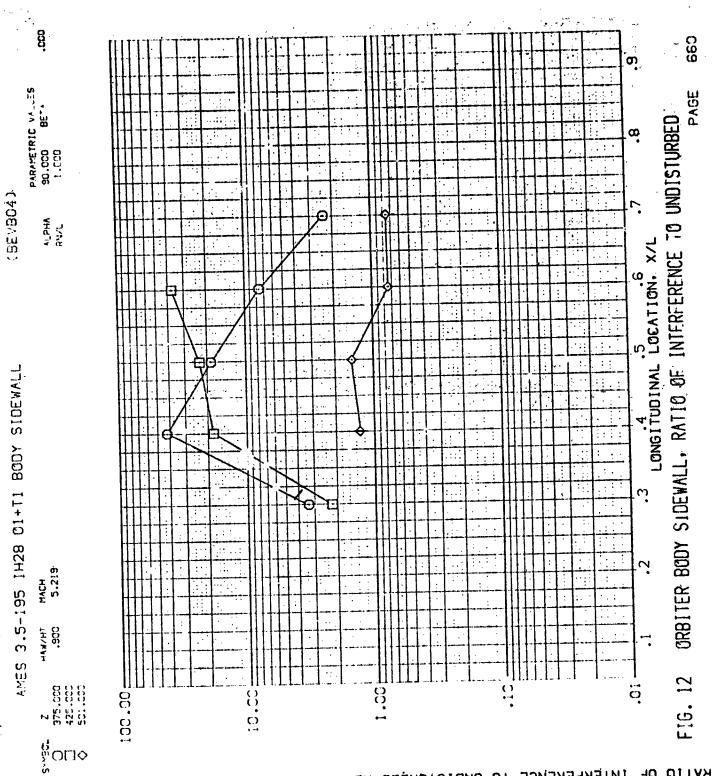
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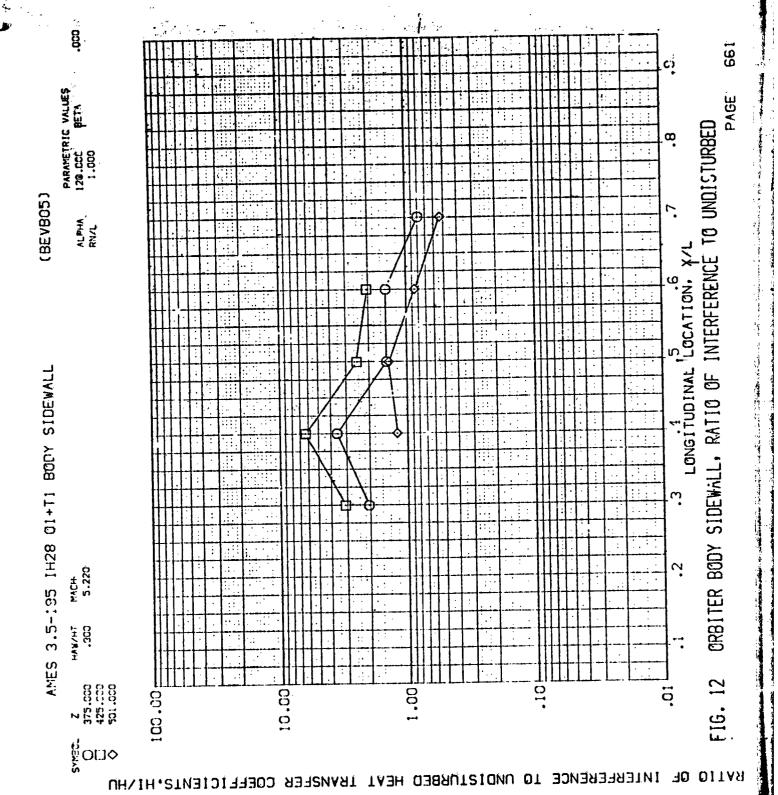
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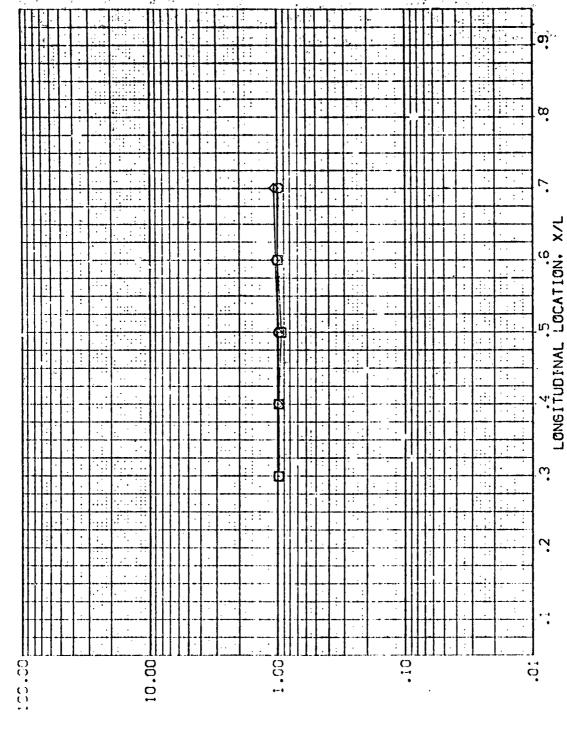
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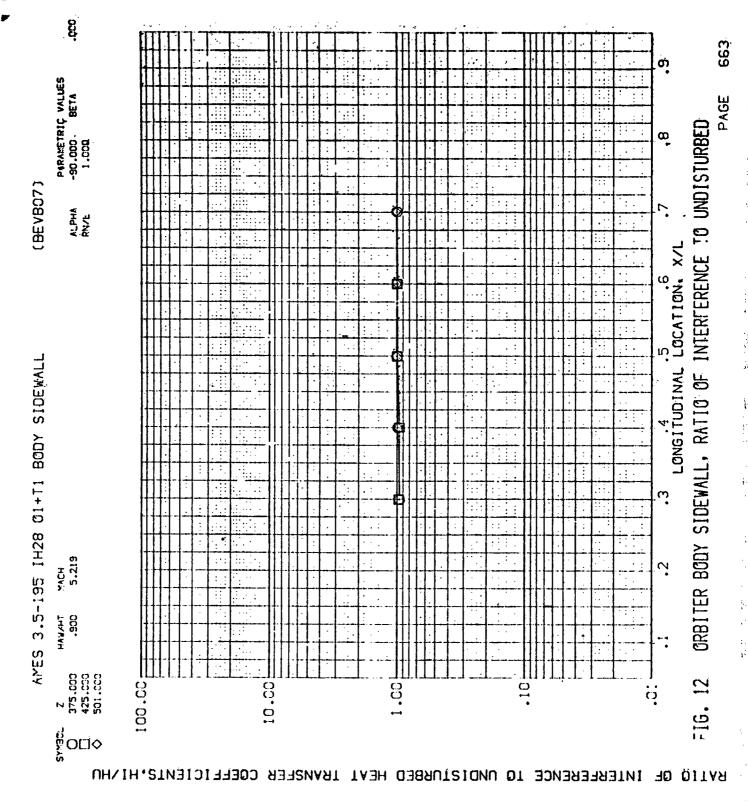
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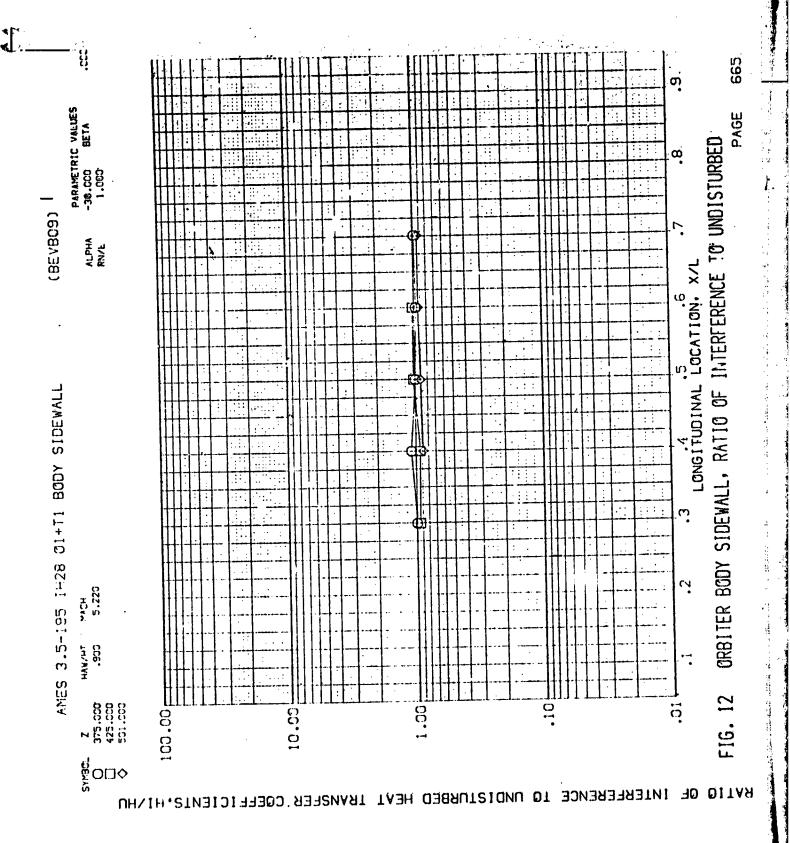
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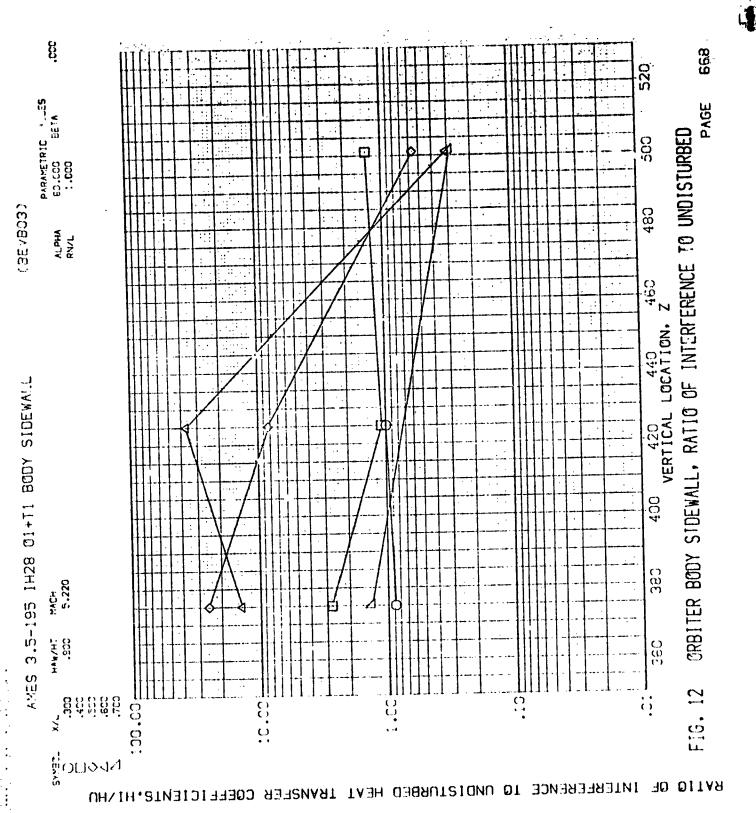
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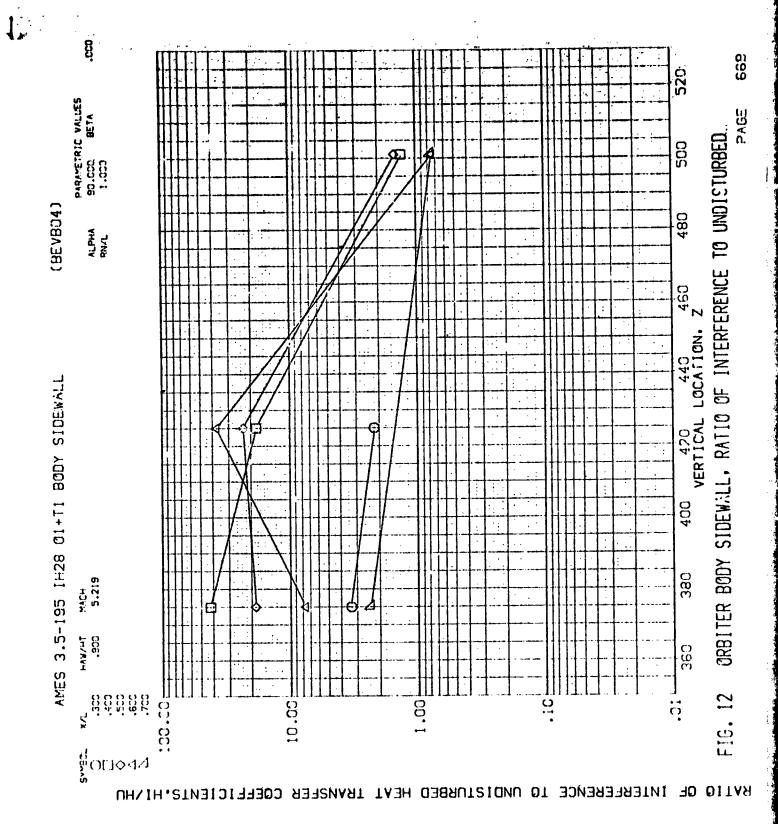
ORBITER BODY SIDEWALL, RATIO OF INTERFERENCE TO UNDISTURBED FIG. 12



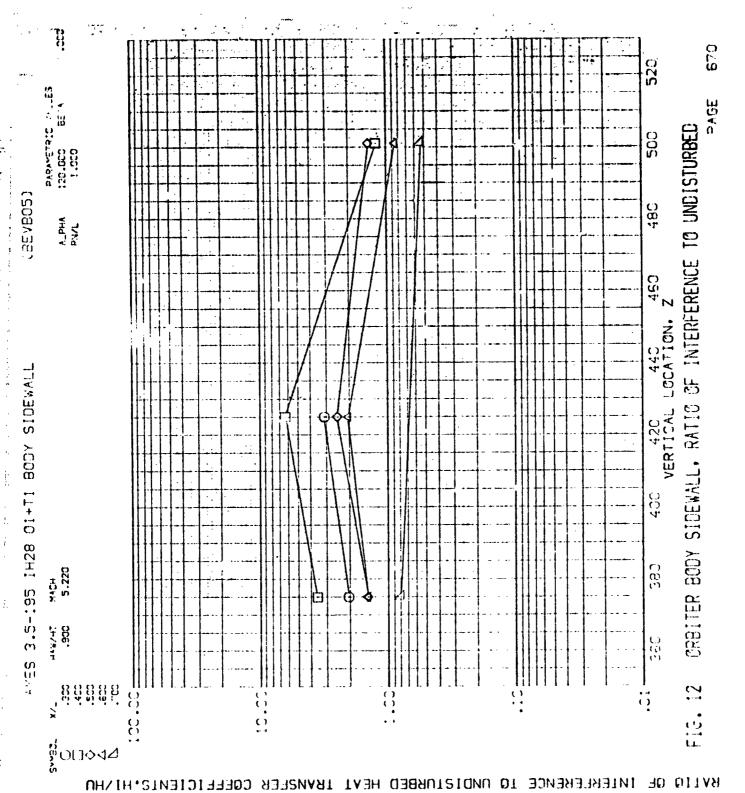


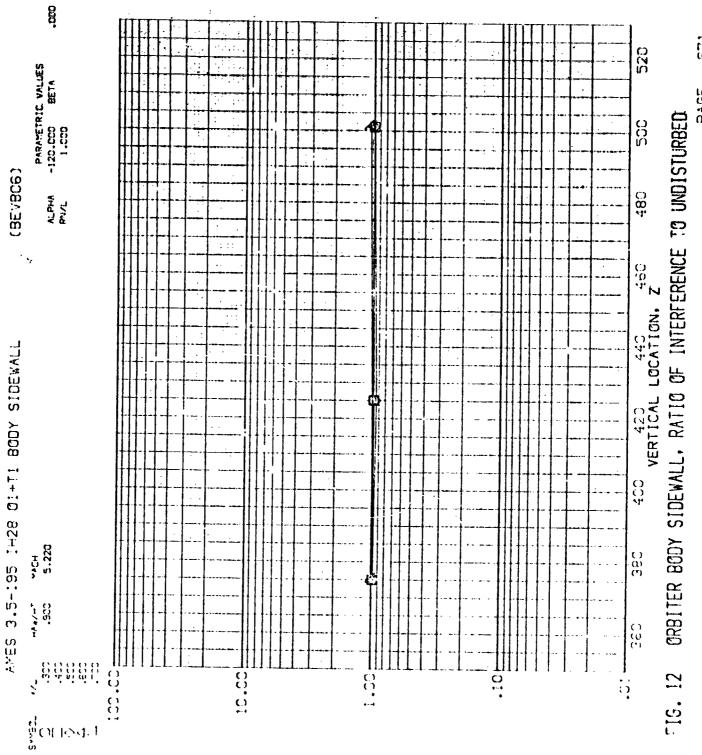
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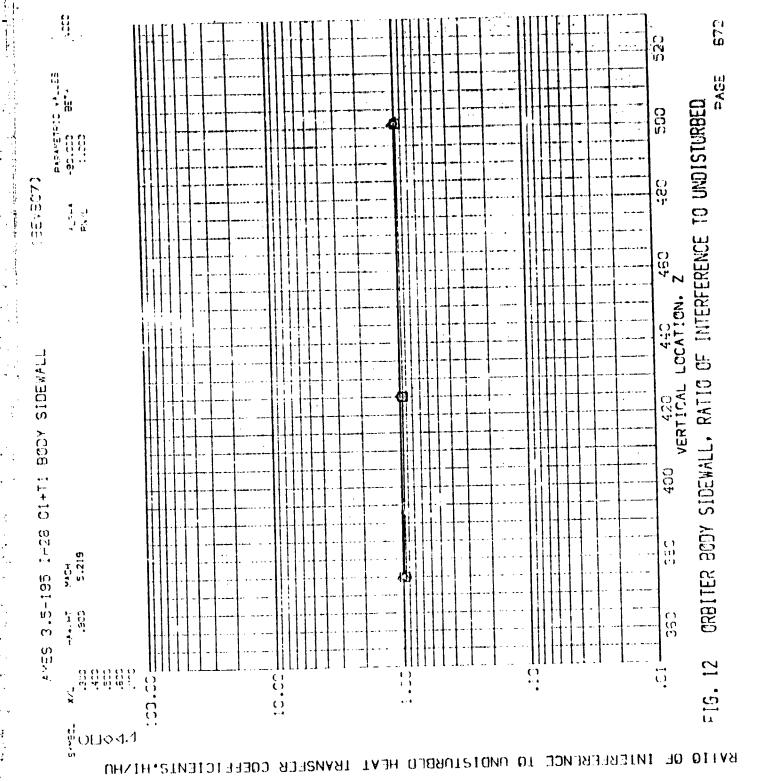
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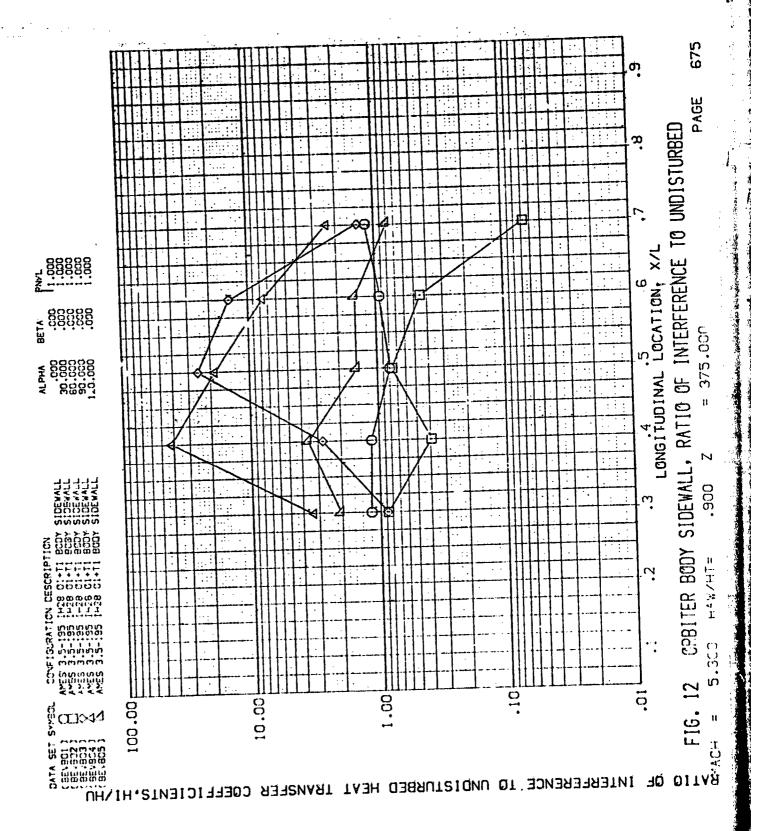
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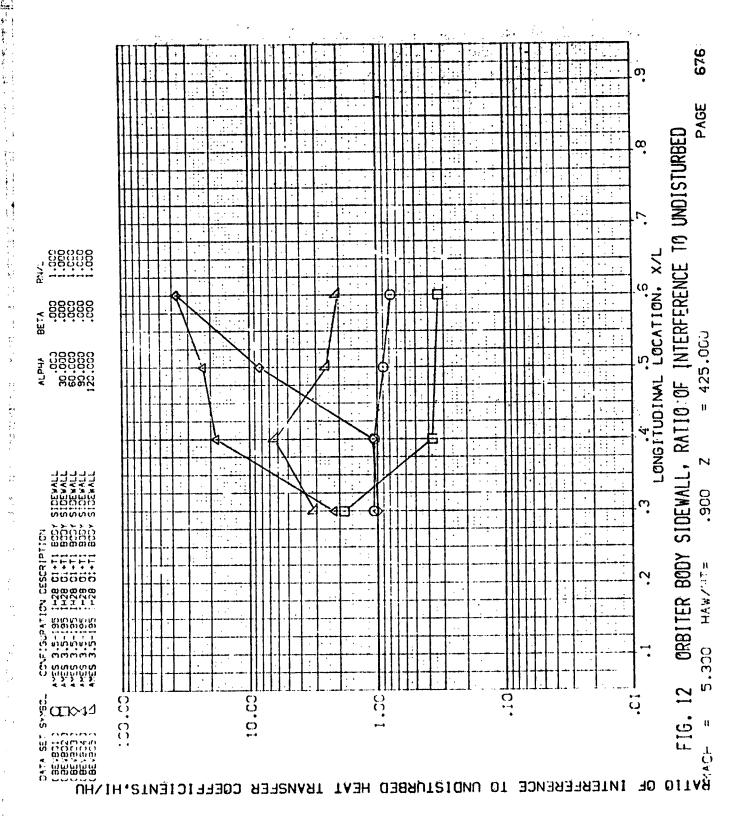
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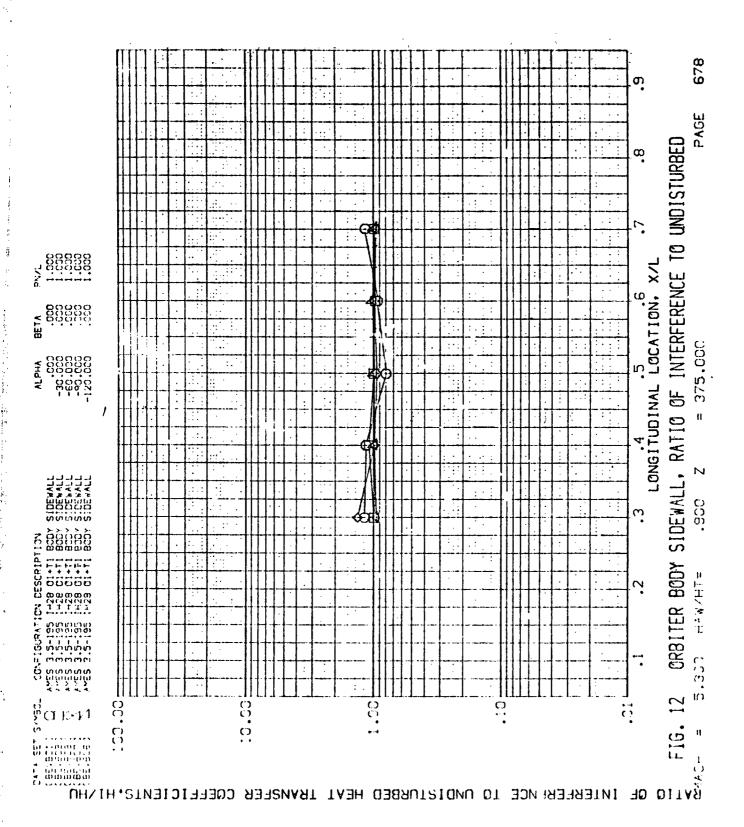
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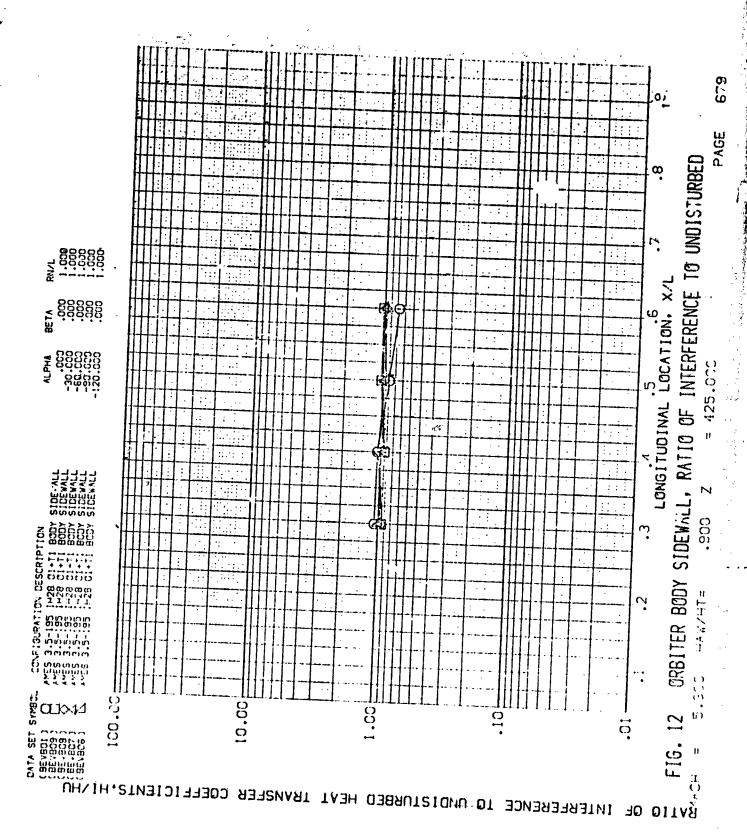
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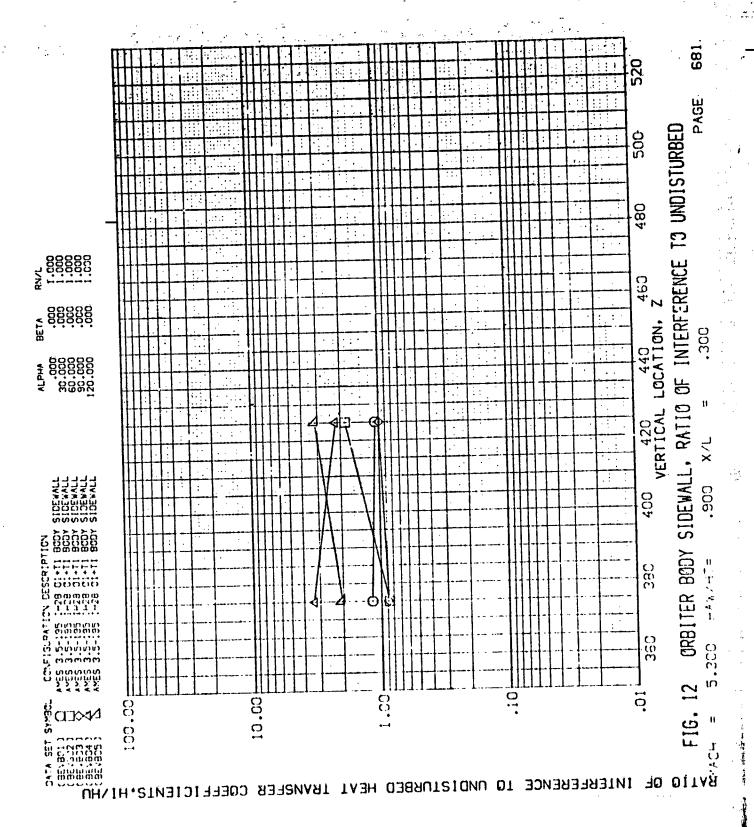
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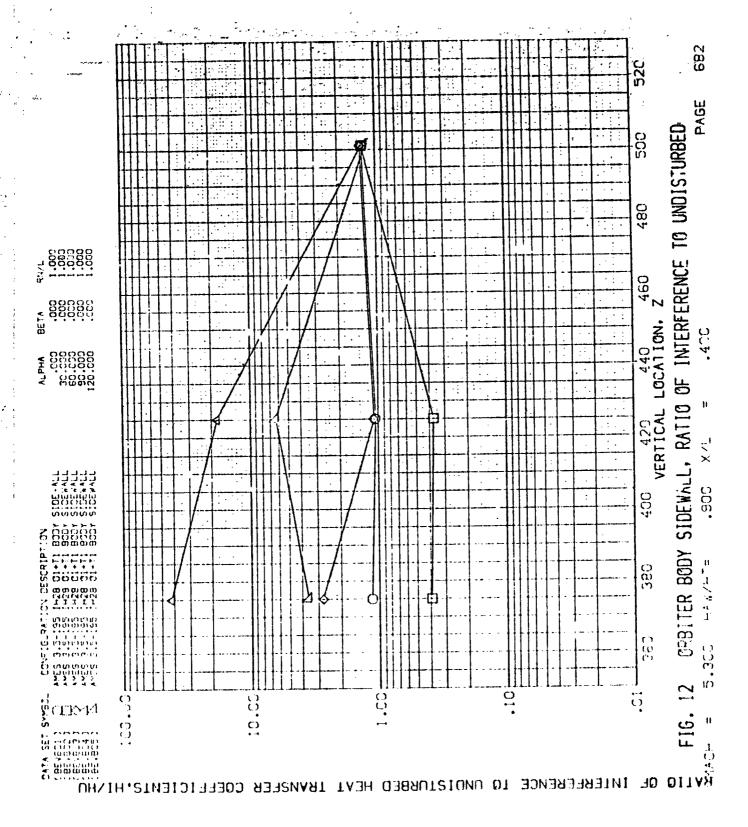
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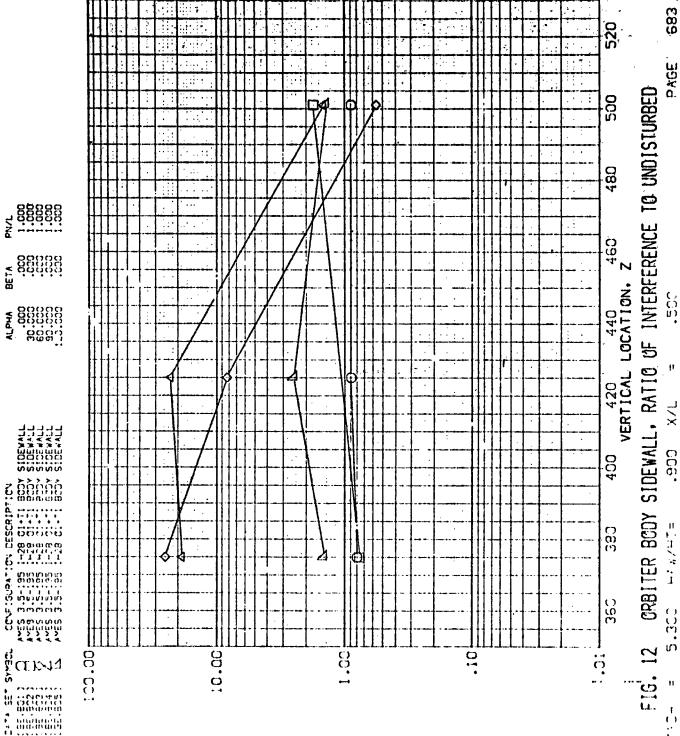
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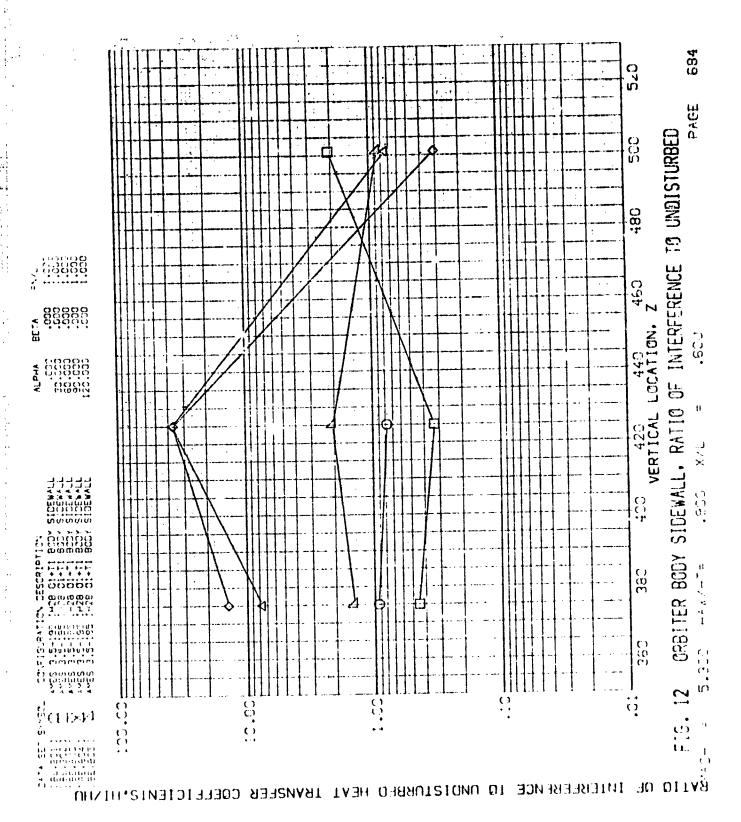
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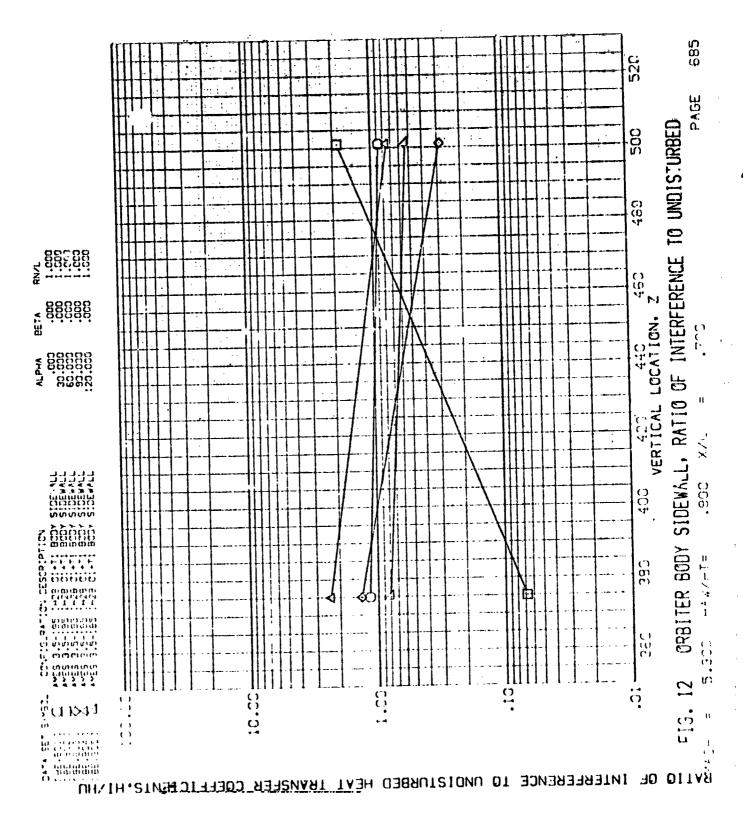


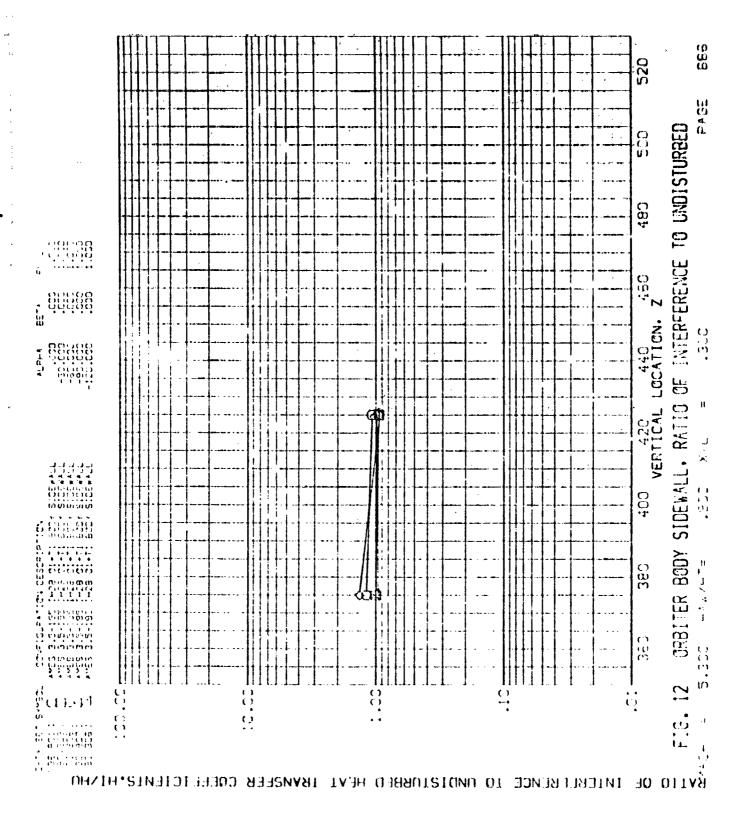


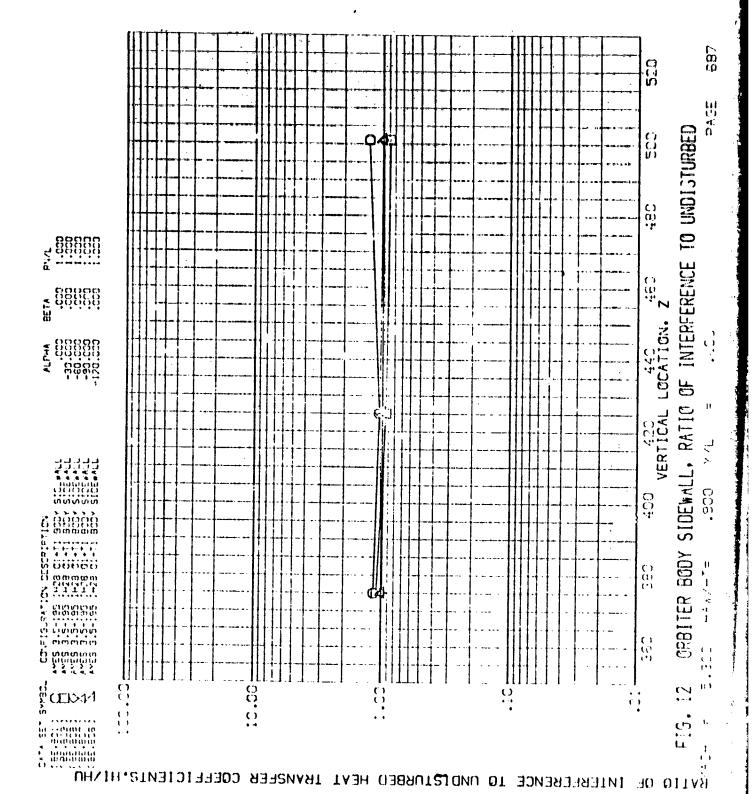


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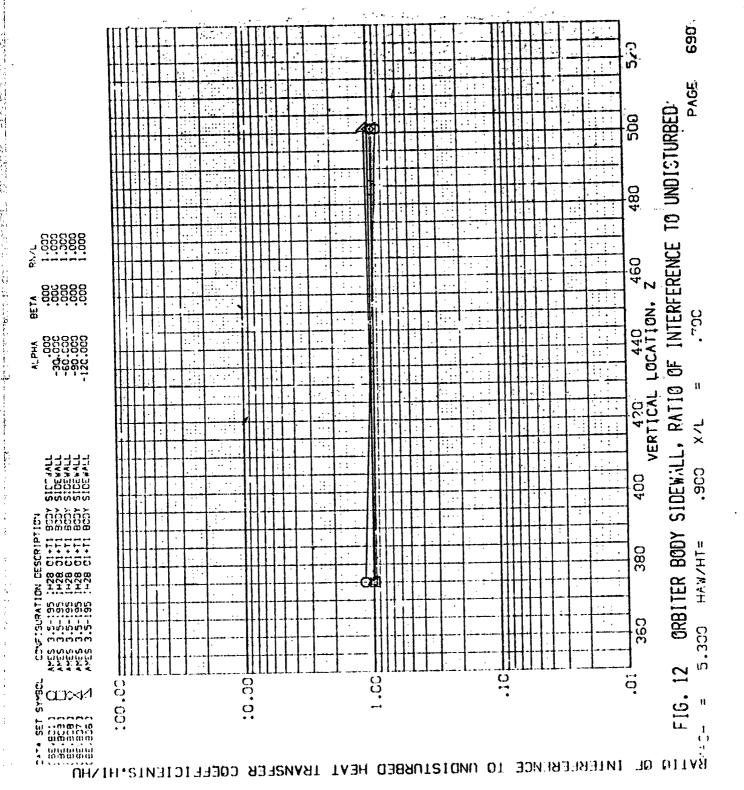




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12 5. FIG. RATIO DE INTERFERENCE TO UNDISTURBED HEAT TRANSFER COEFFICIENTS.HIVHU

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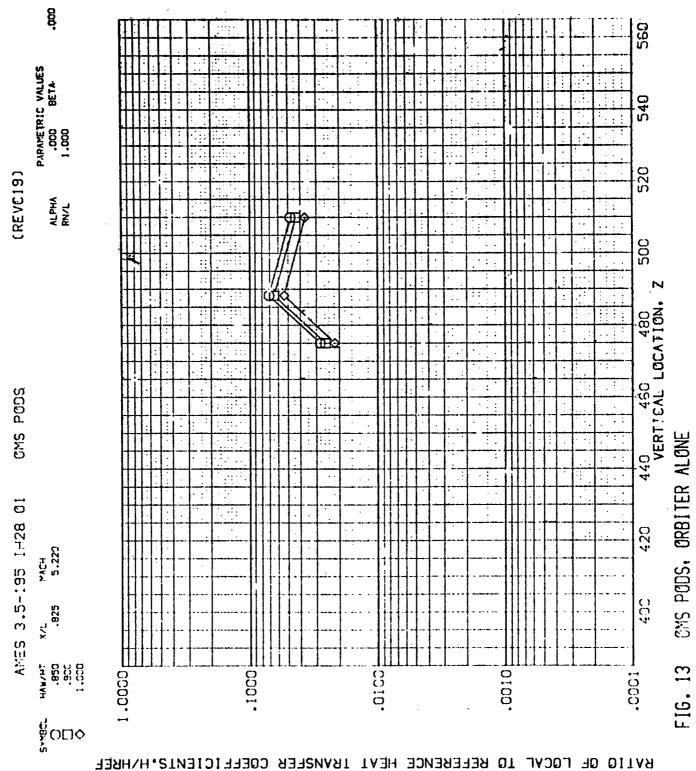
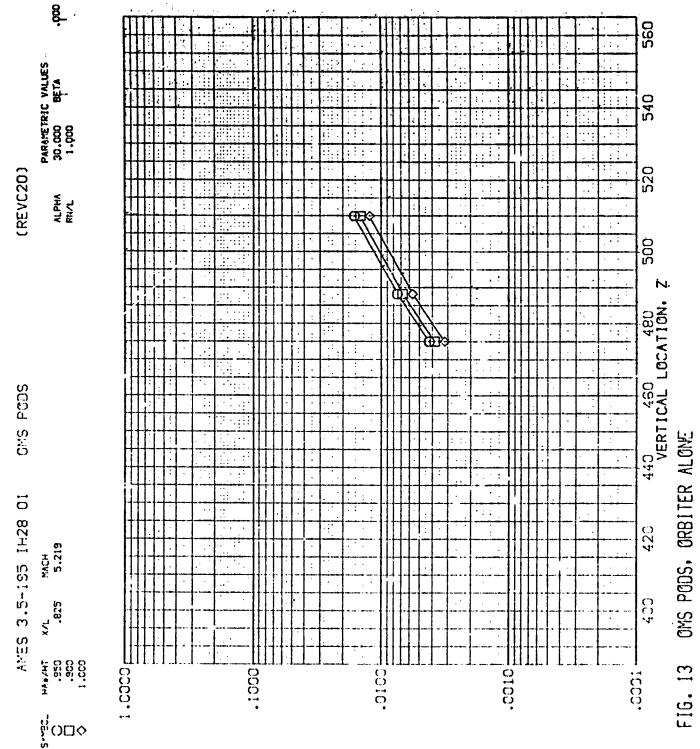


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BATIO OF LOCAL TO REFERENCE HEAT TRANSFER COEFFICIENTS, HAHREF

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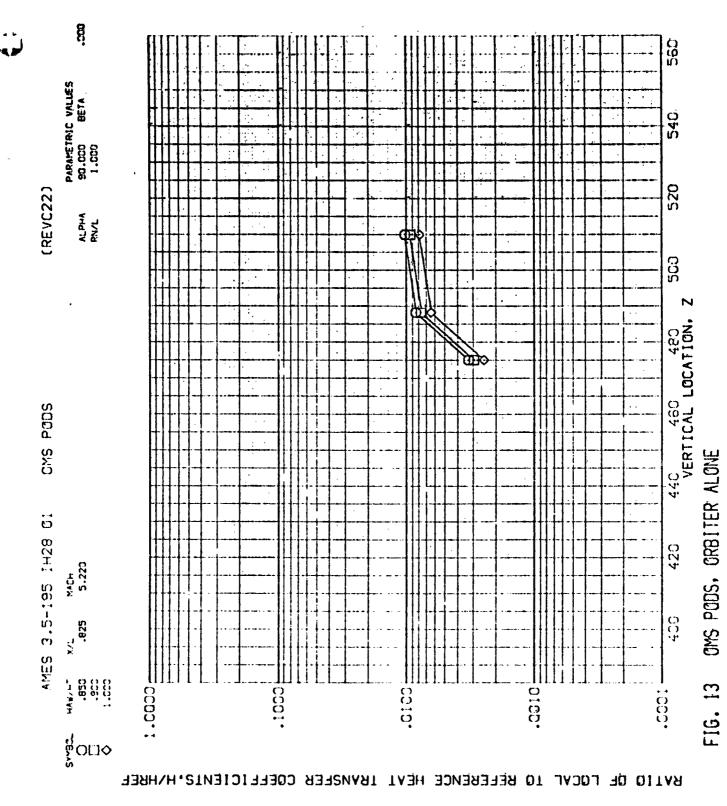
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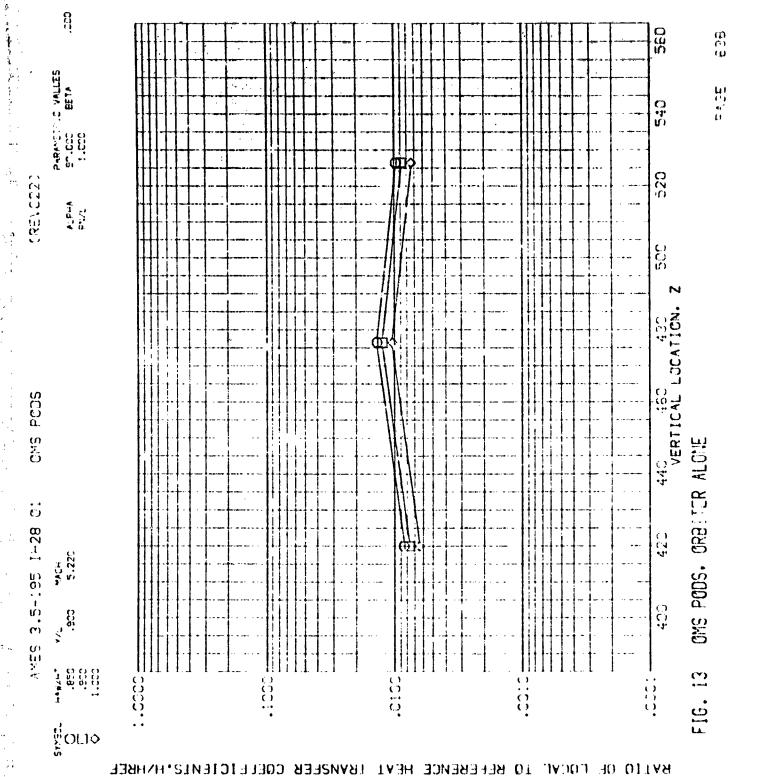
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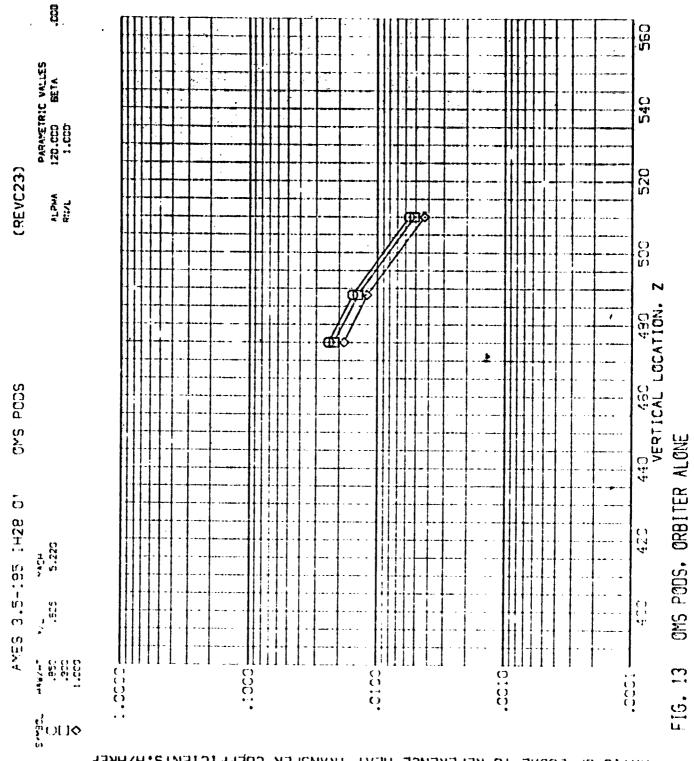
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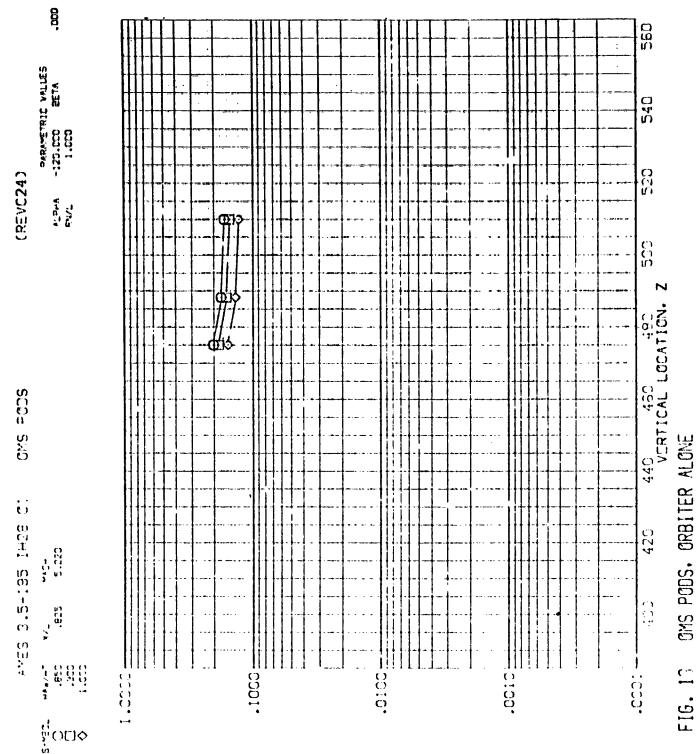


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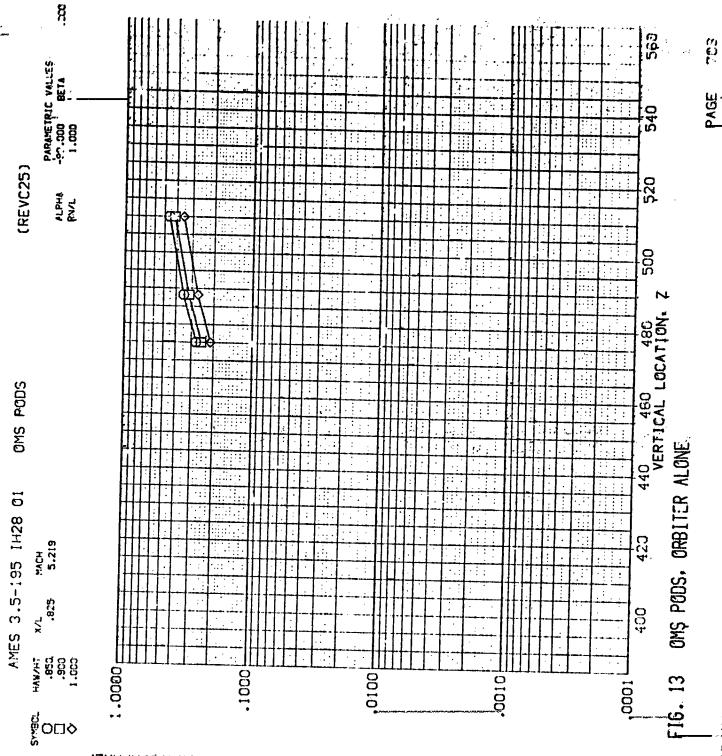
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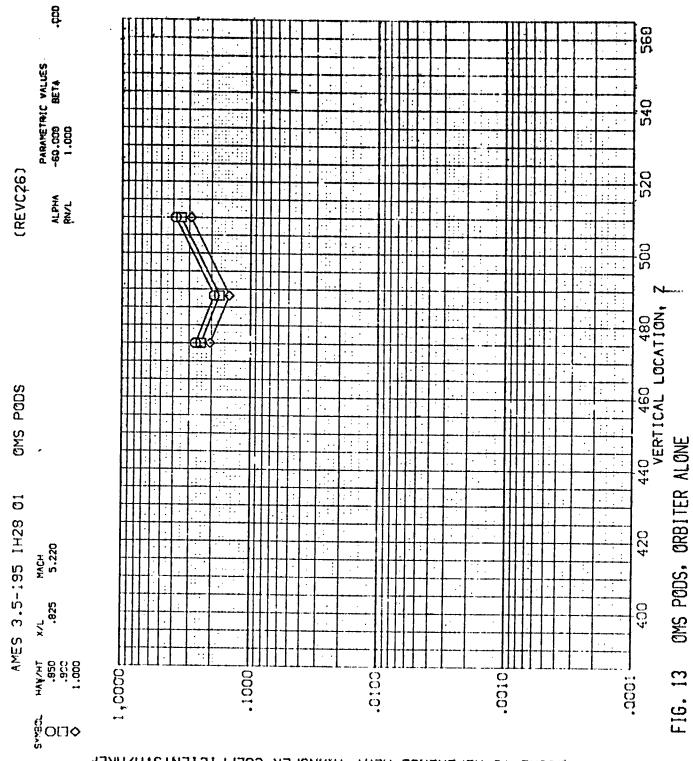


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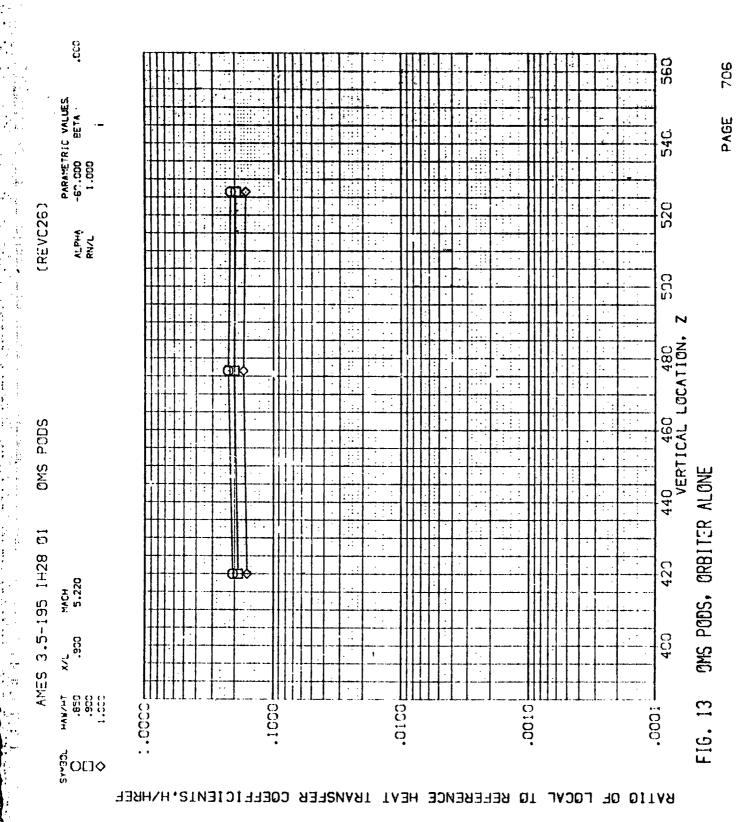
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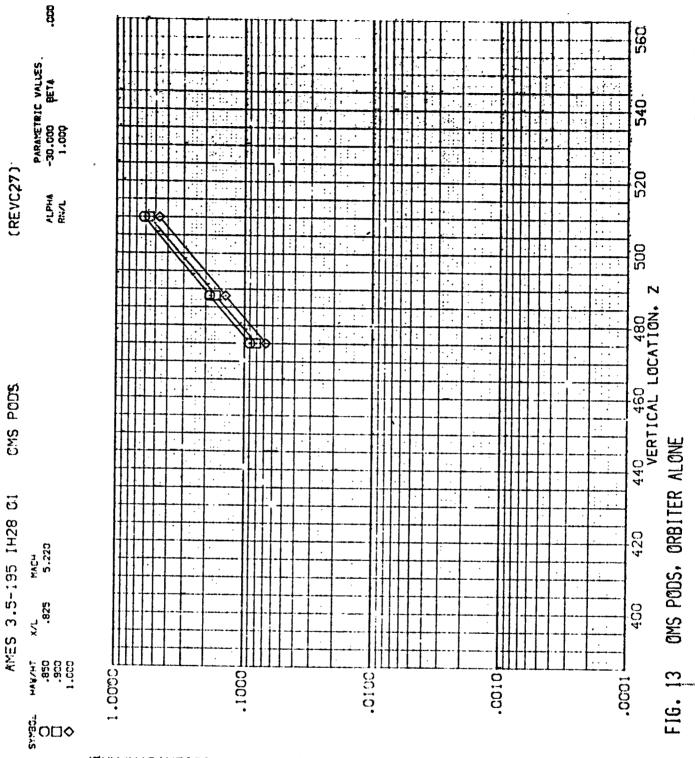


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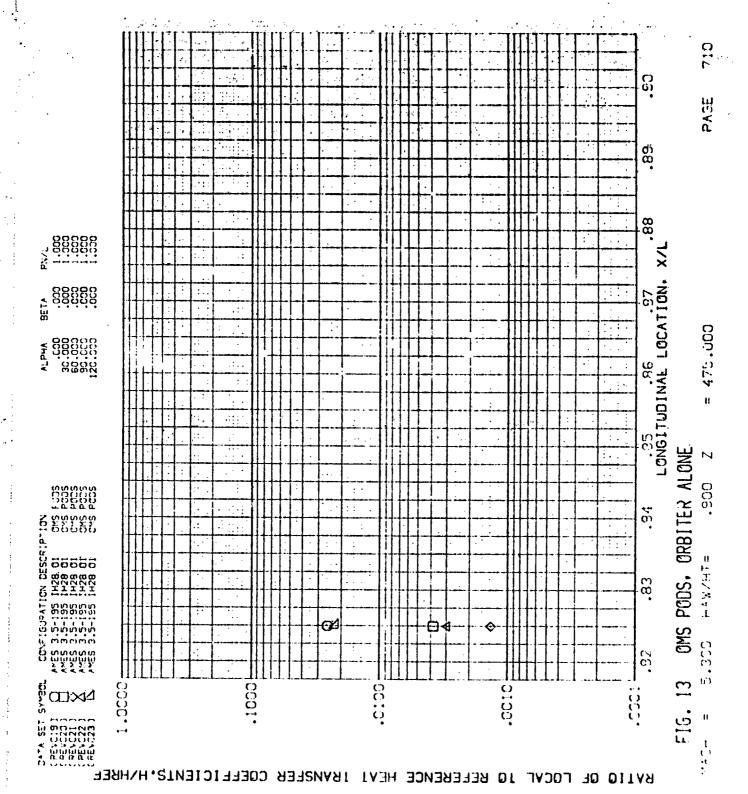
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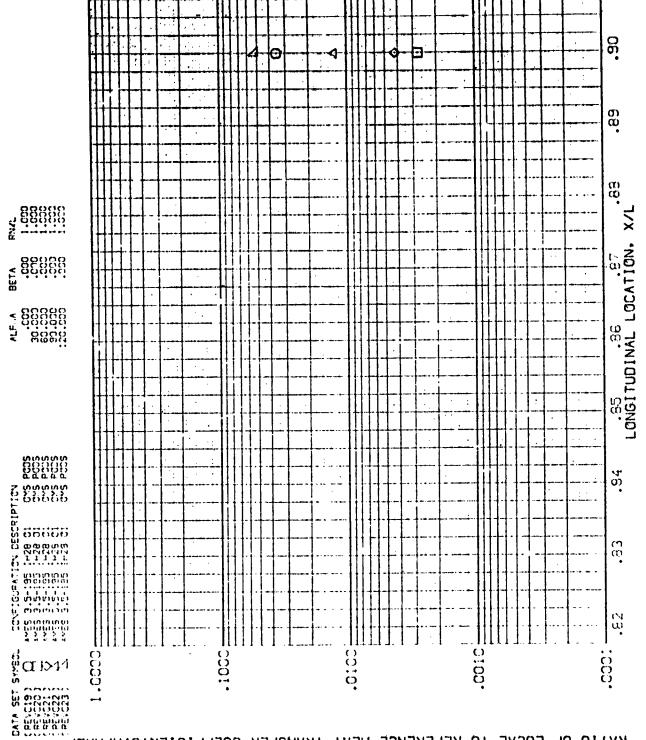
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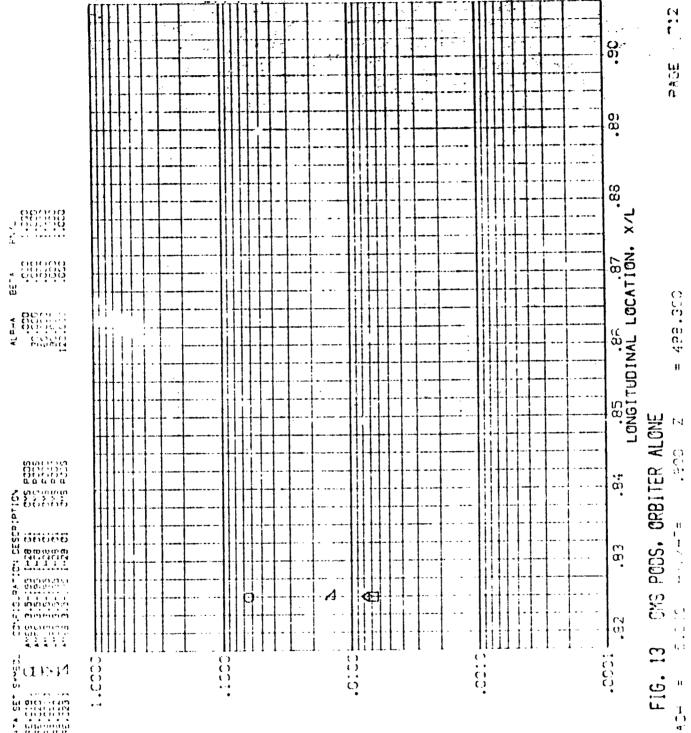
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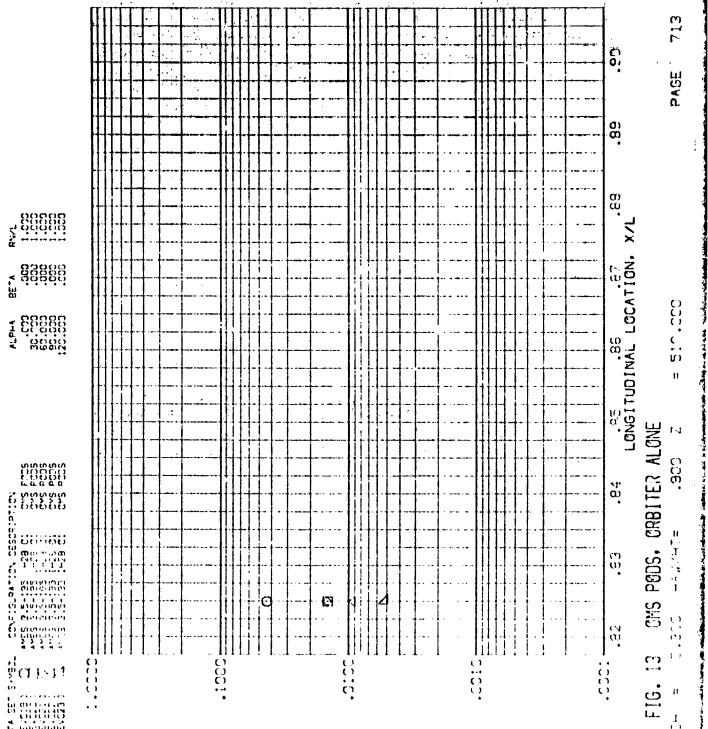


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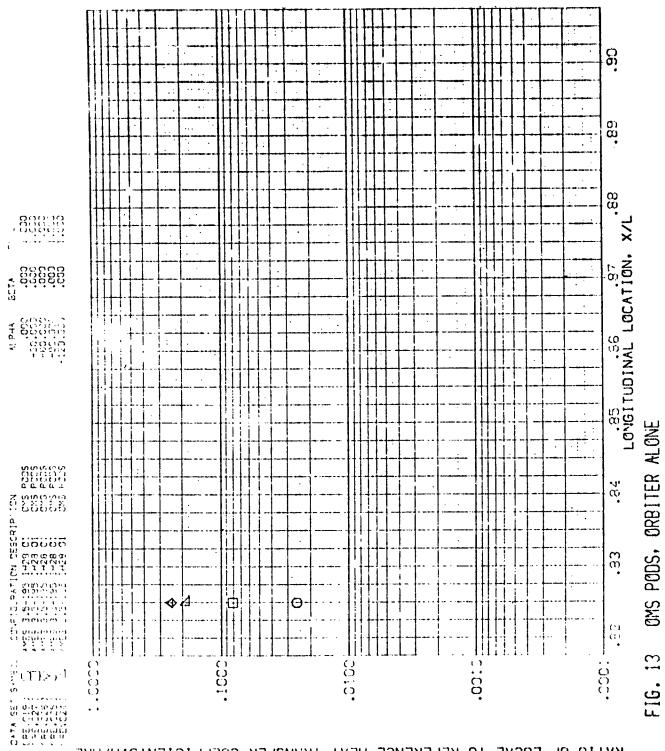
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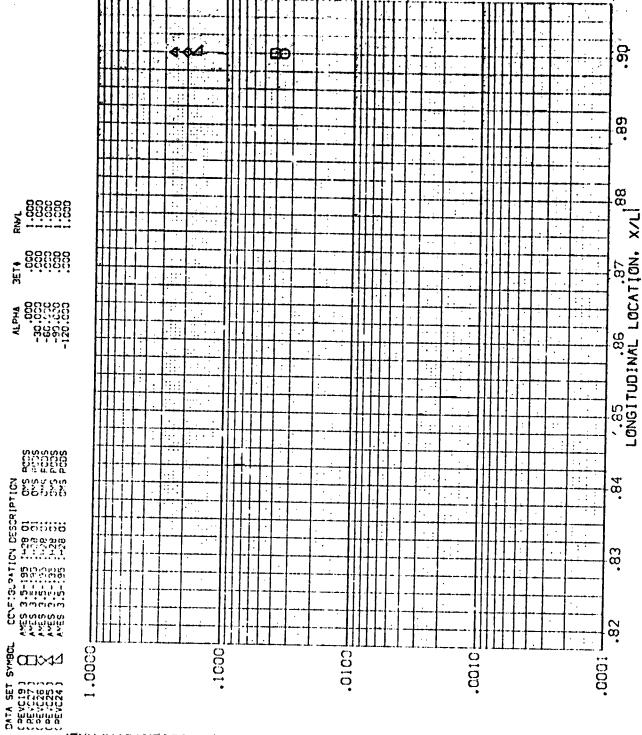
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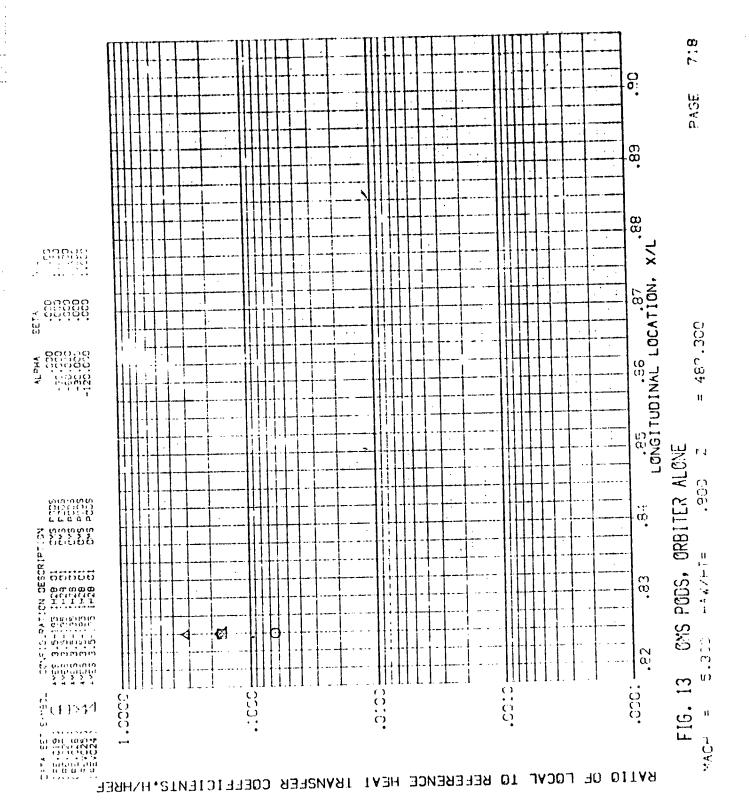
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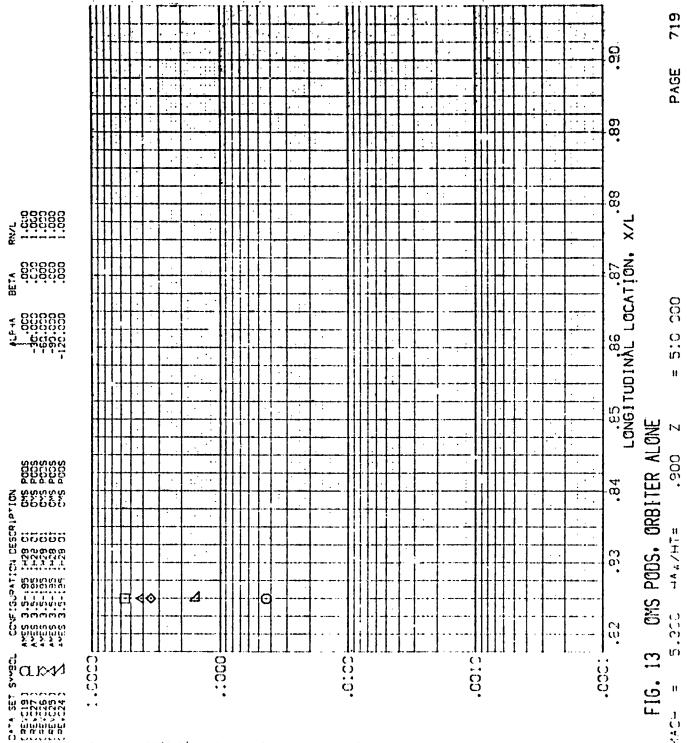
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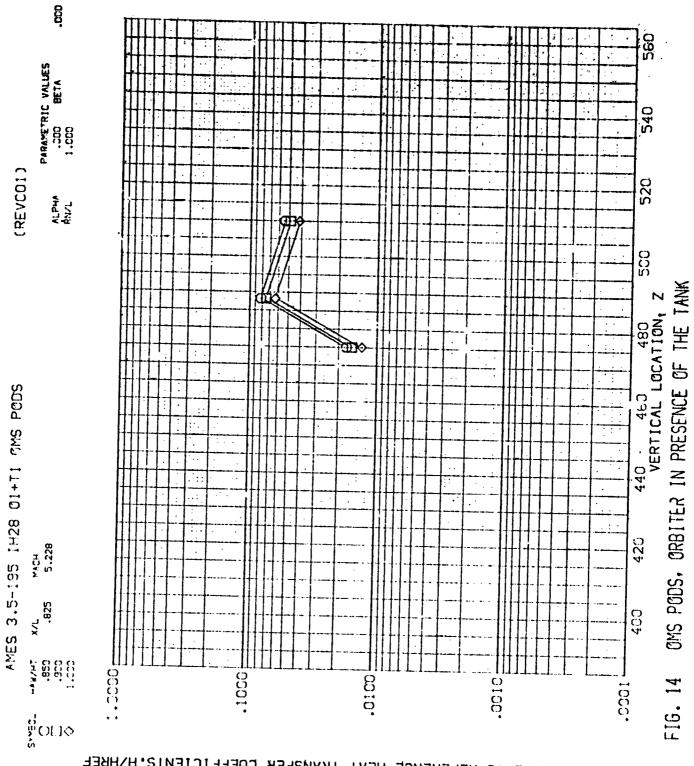
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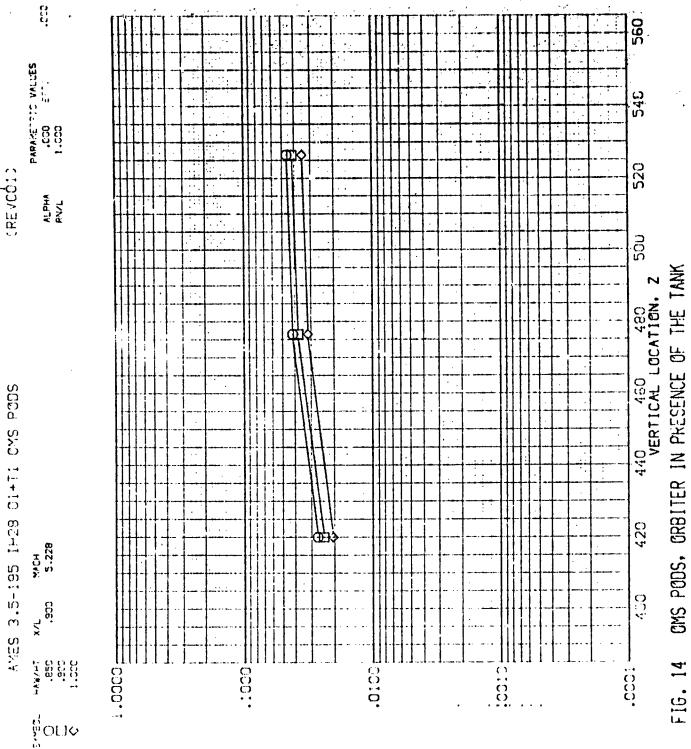
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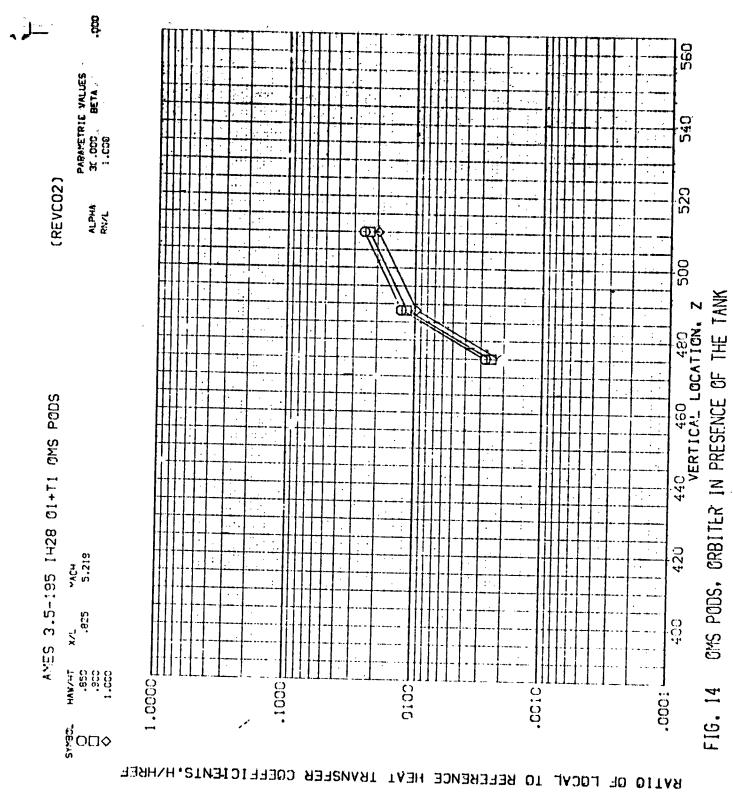
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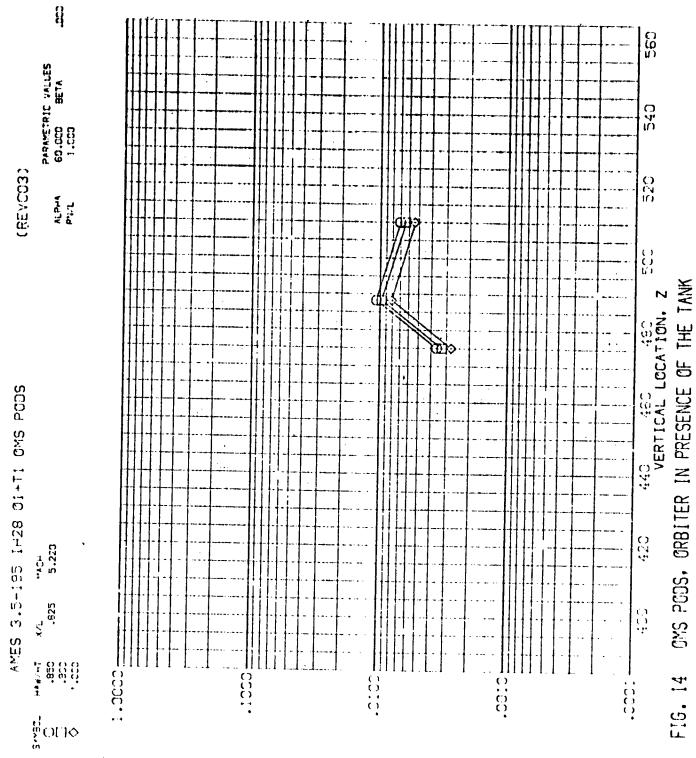
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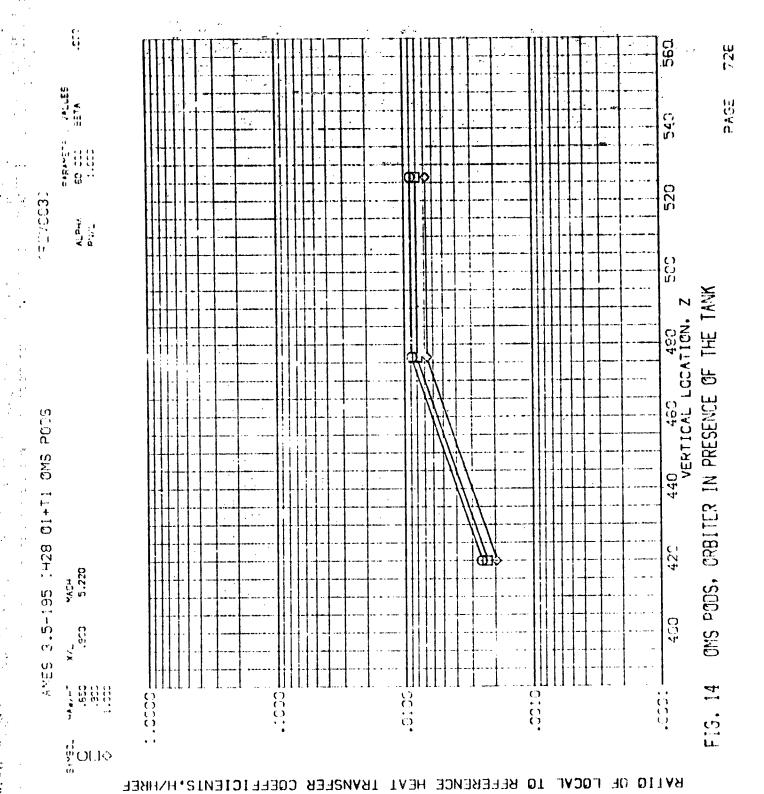
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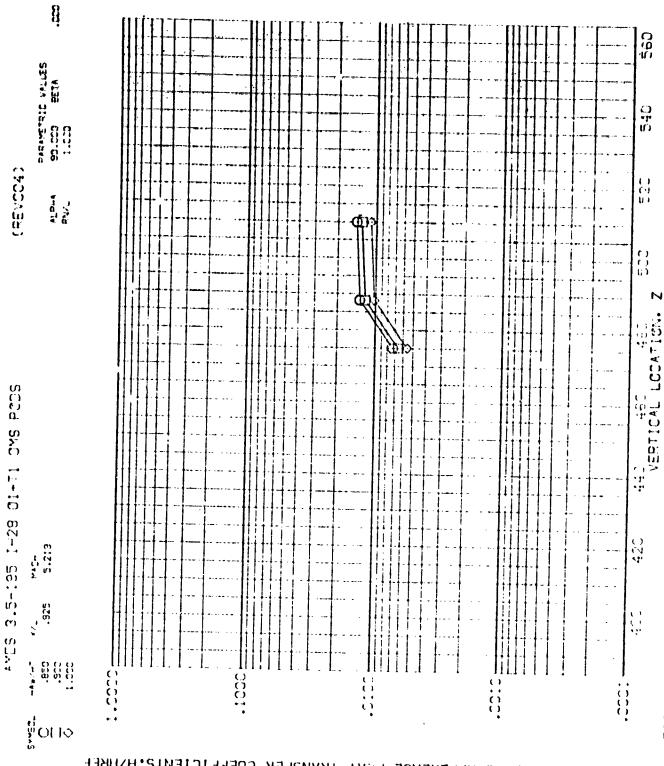


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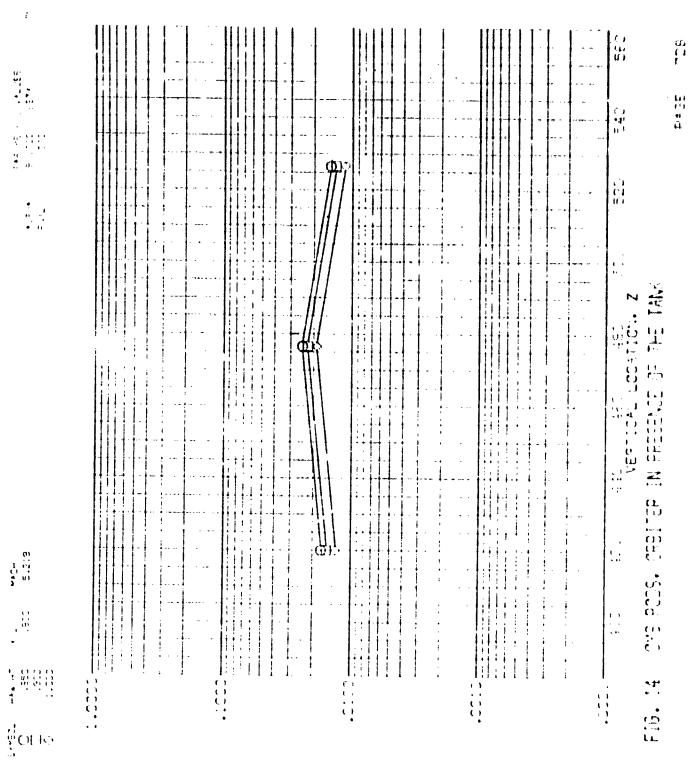


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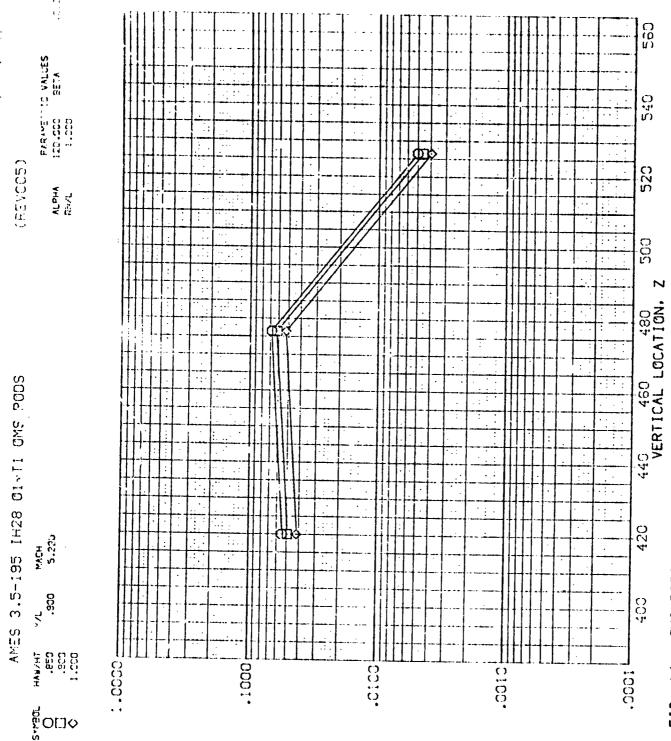
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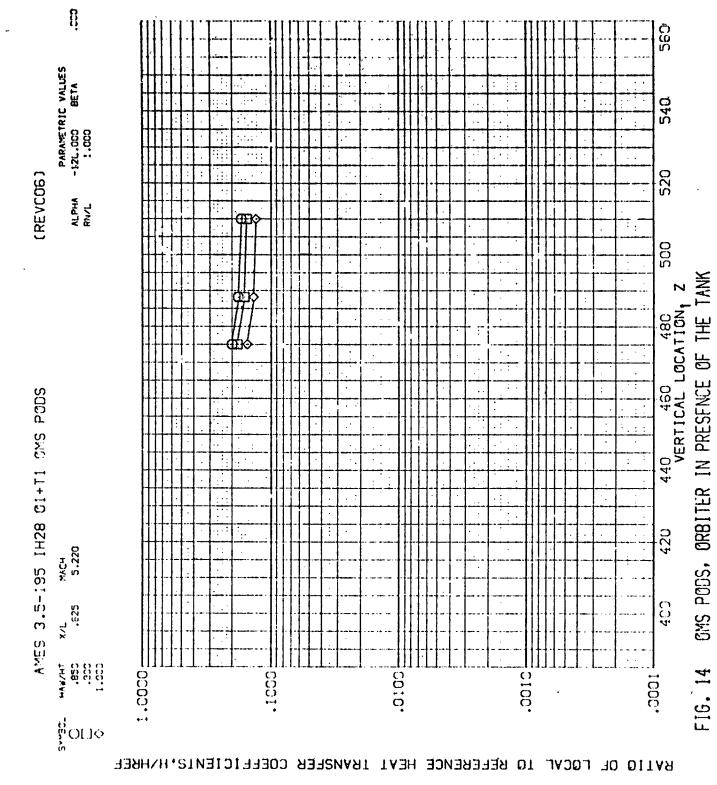
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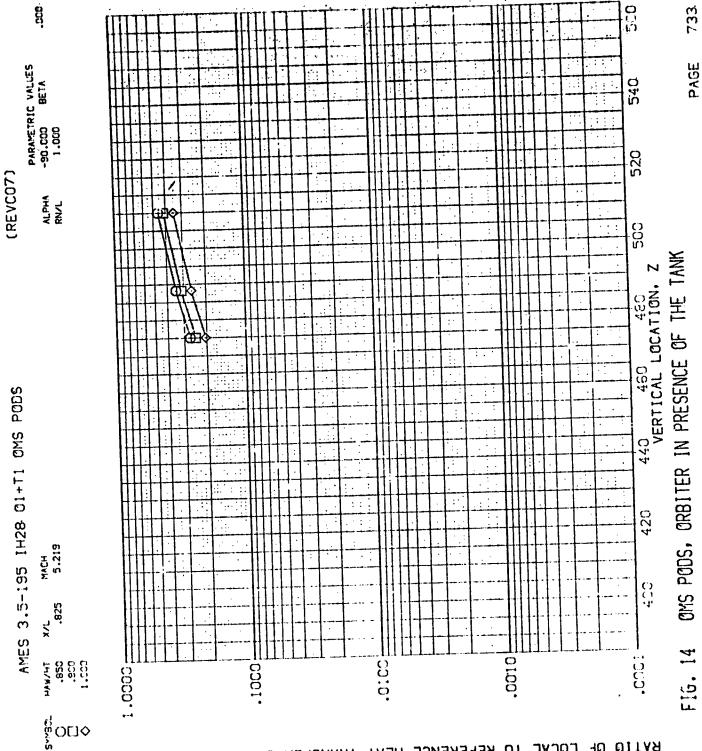
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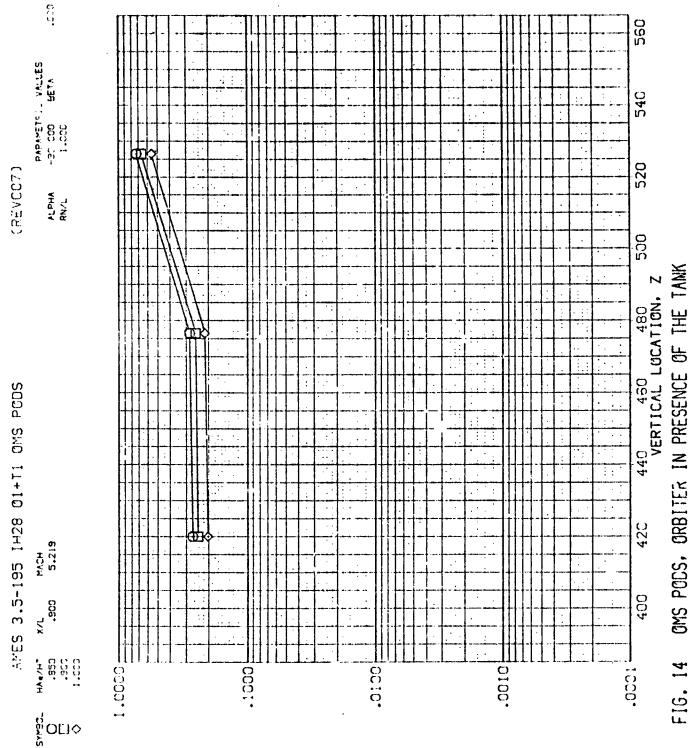
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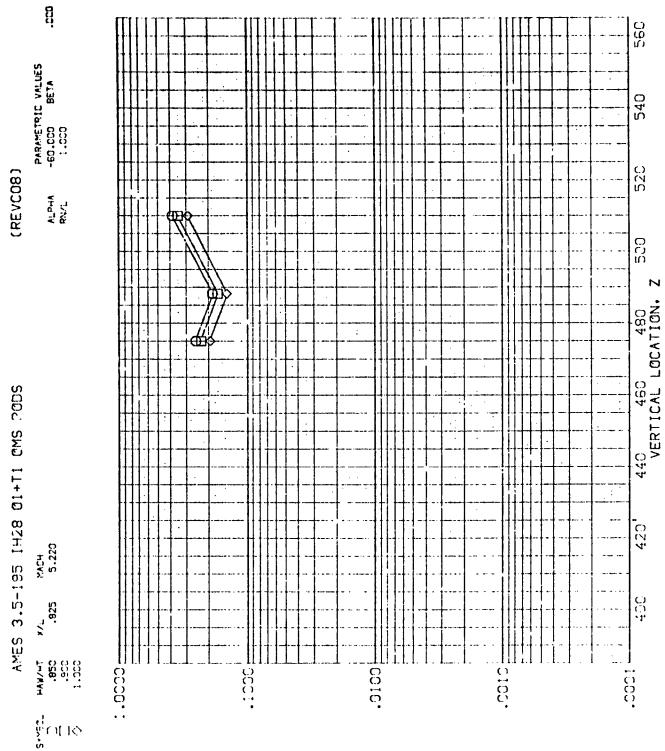
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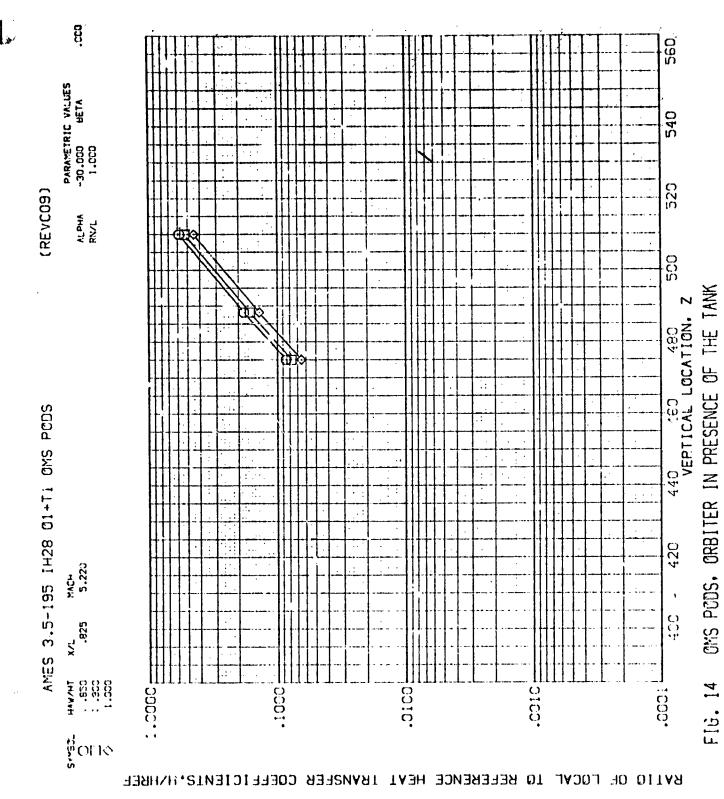
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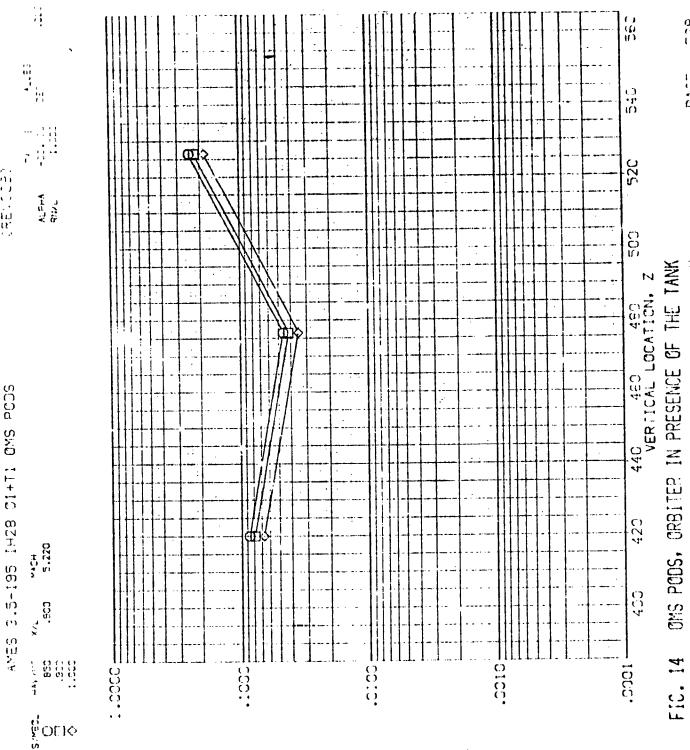


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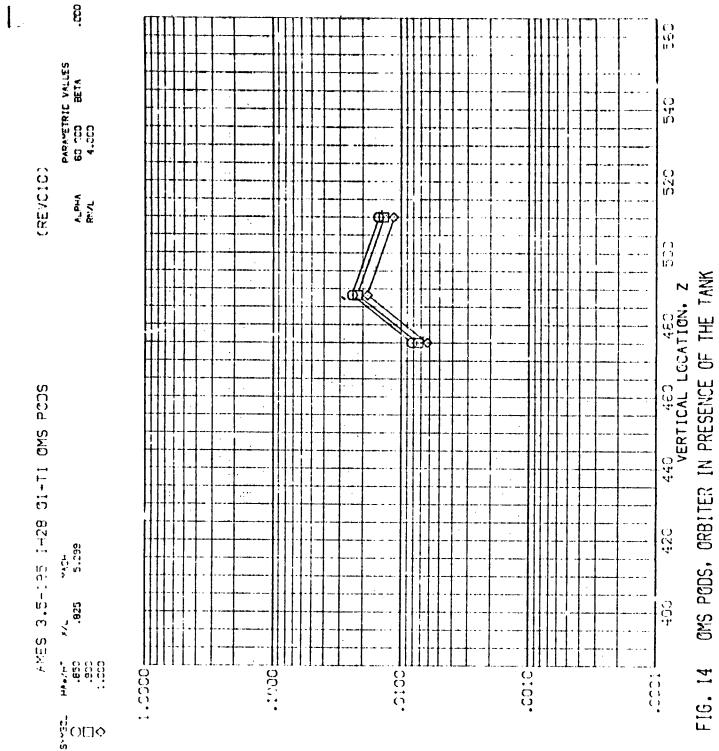
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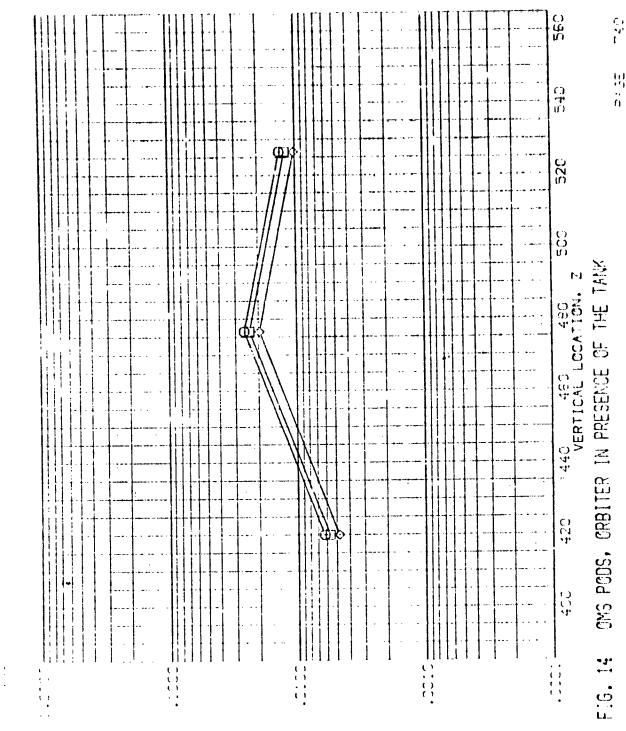
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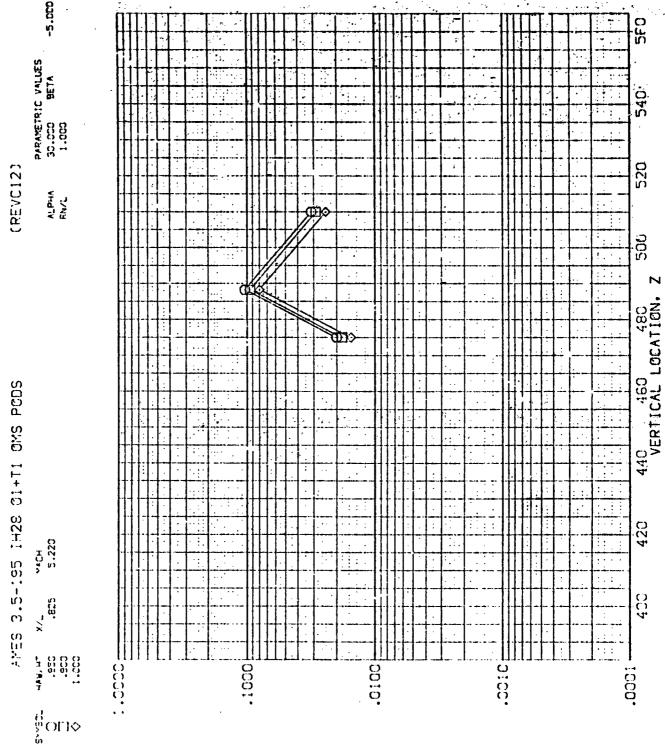
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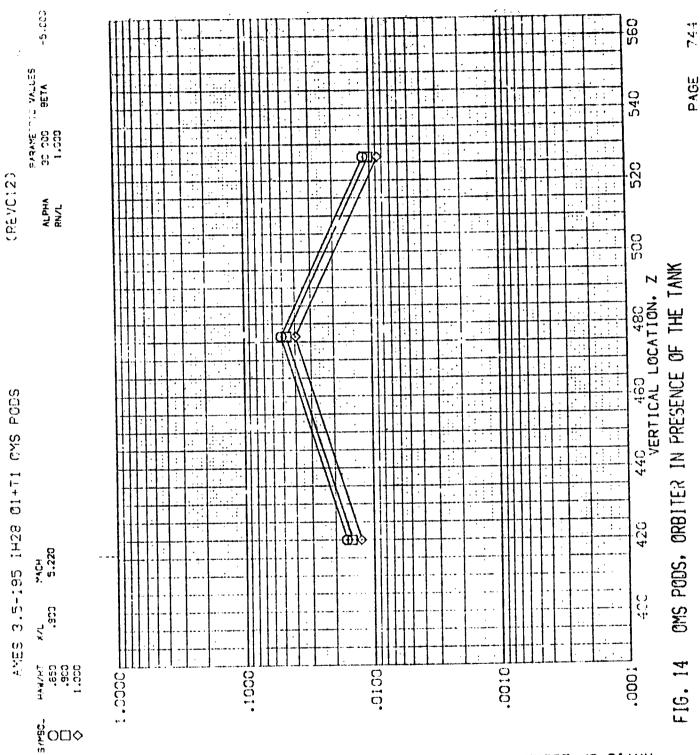
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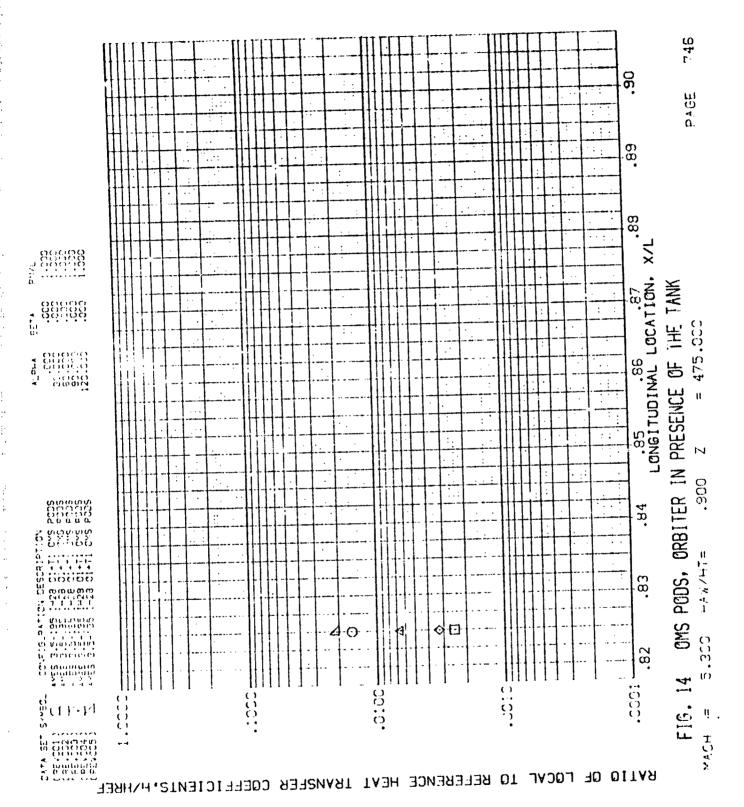
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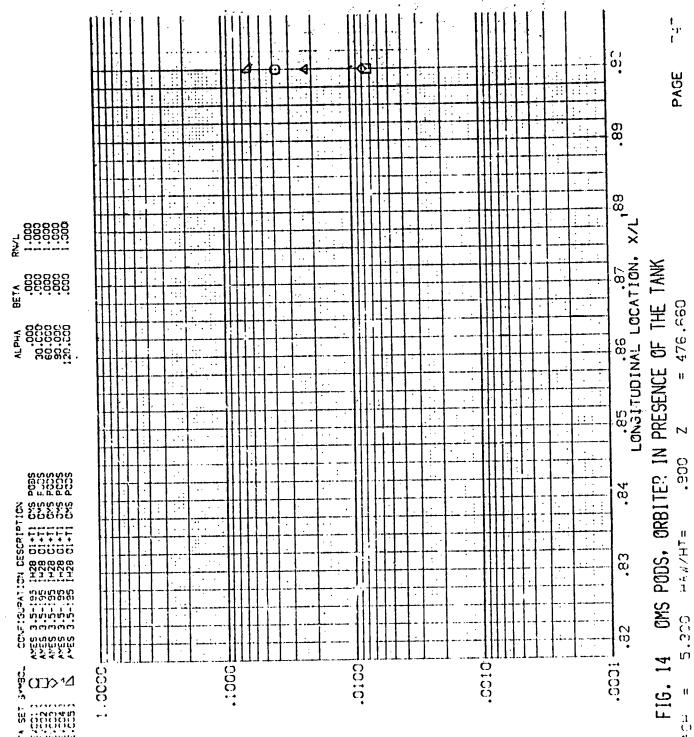
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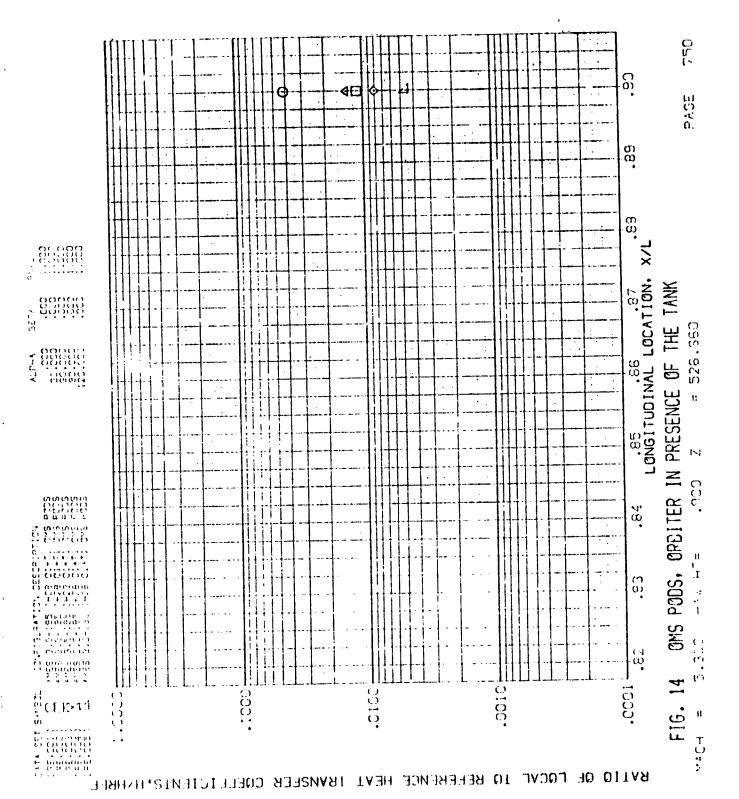
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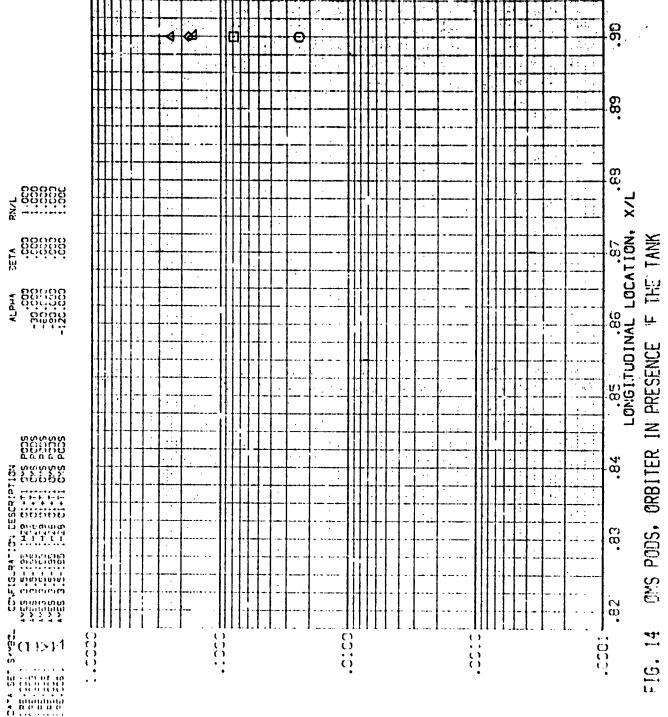
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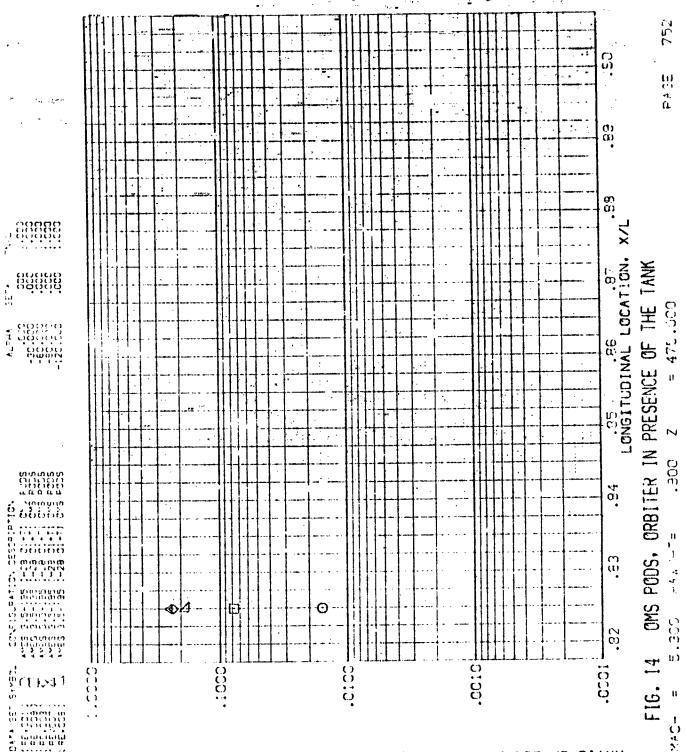
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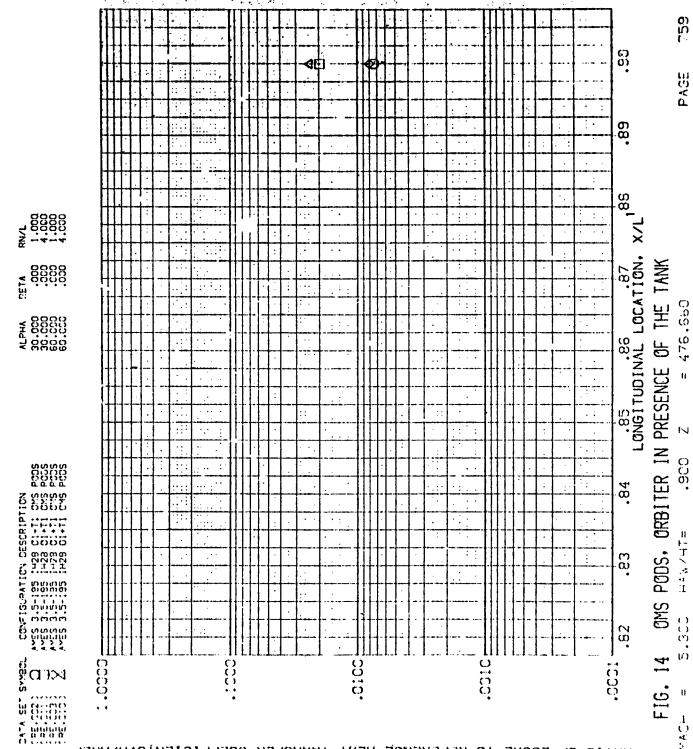
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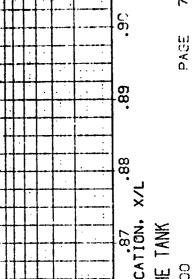
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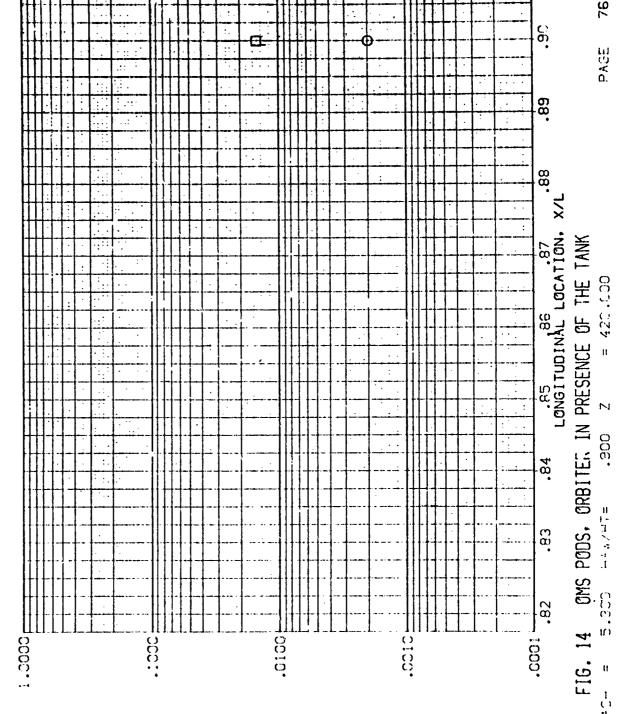
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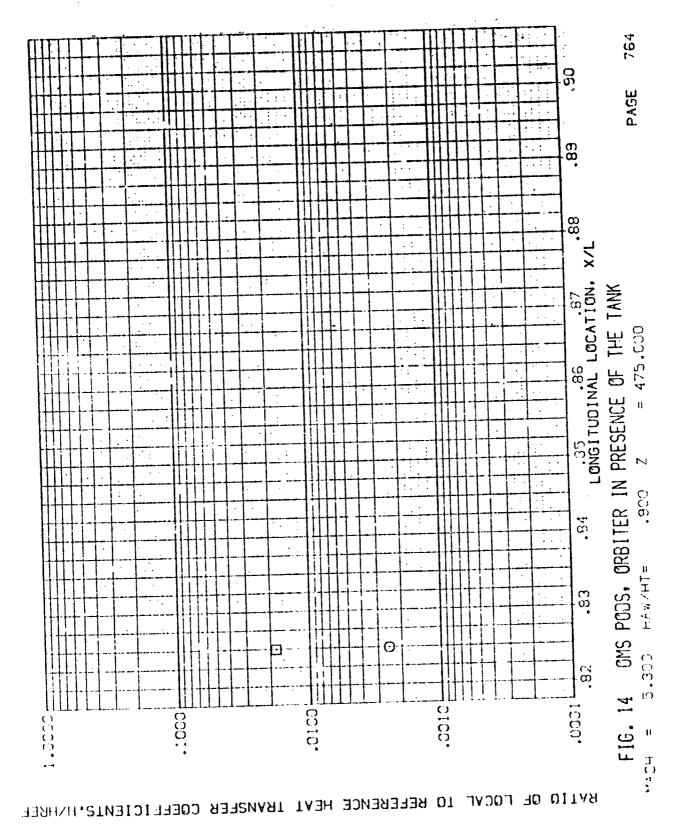


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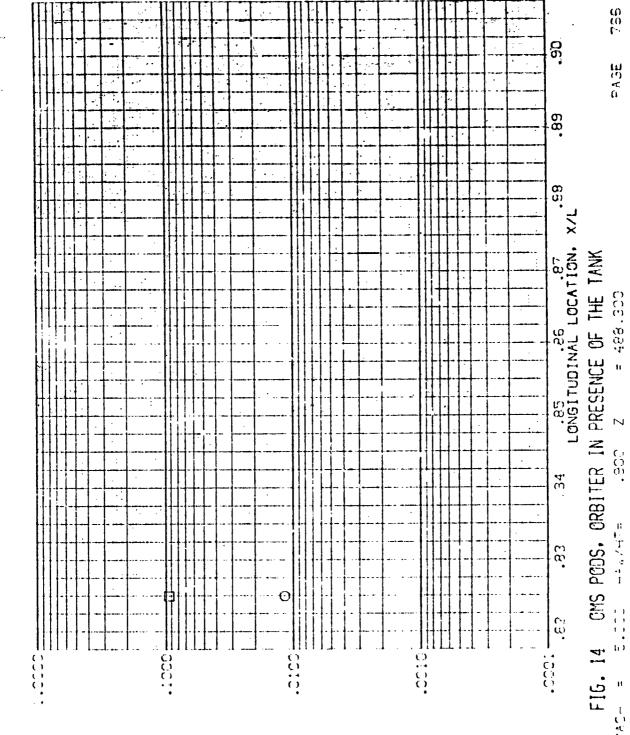
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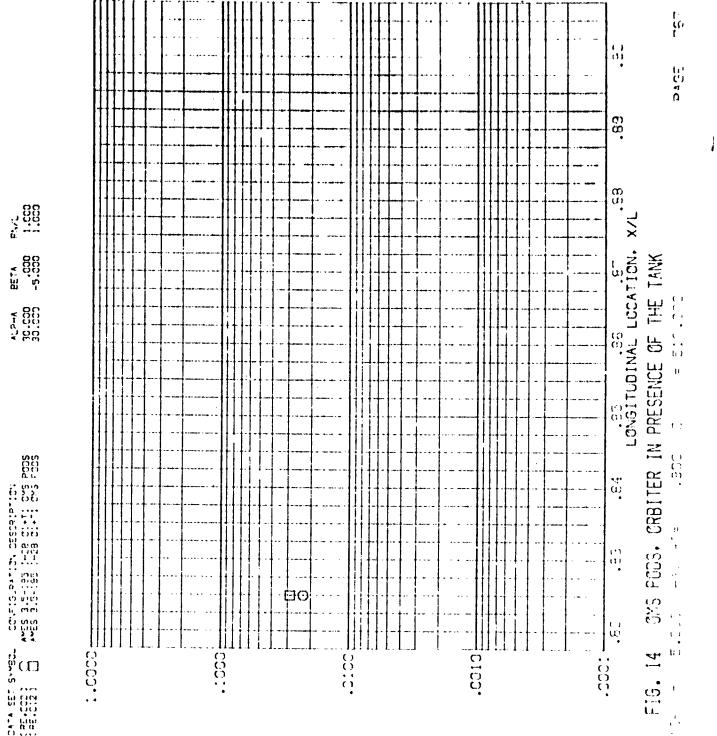
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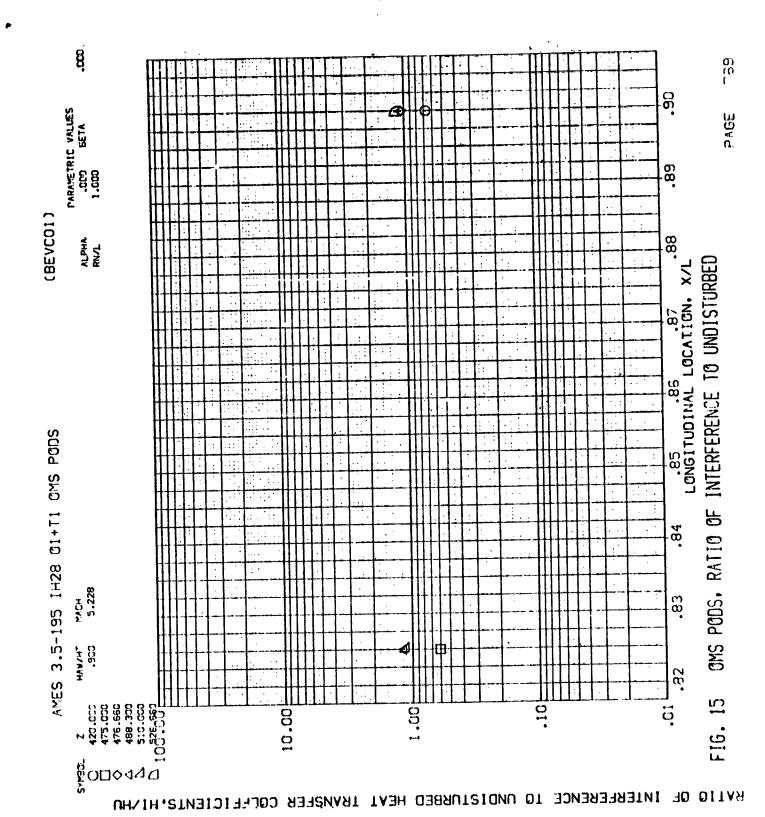
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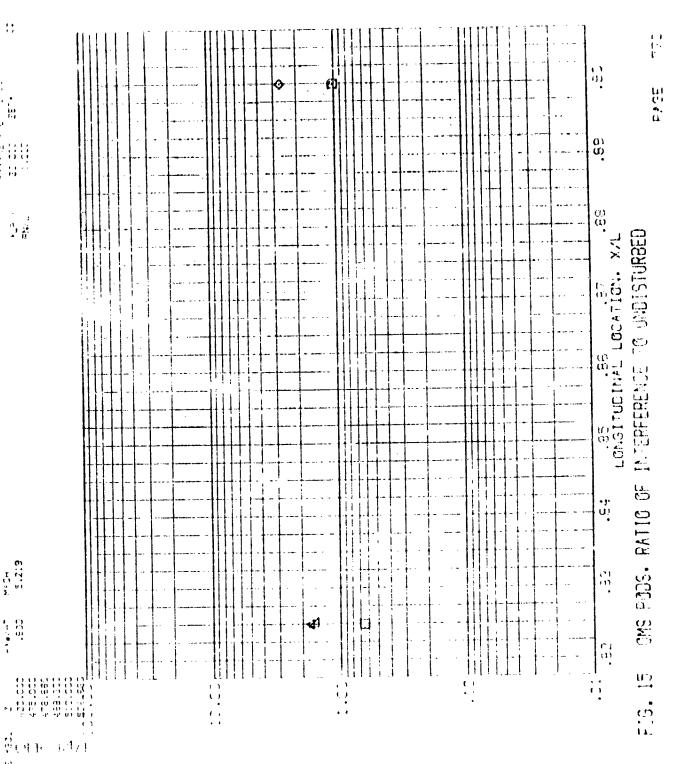
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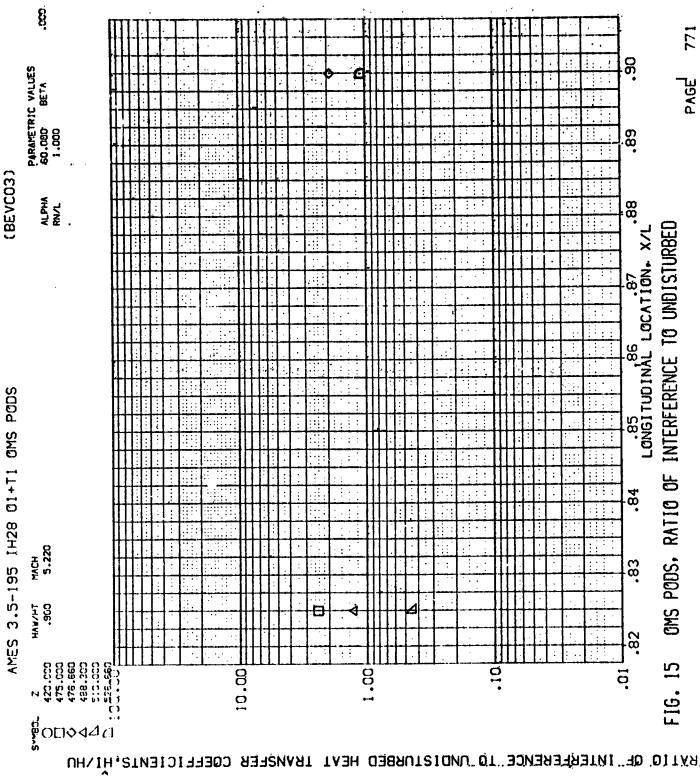
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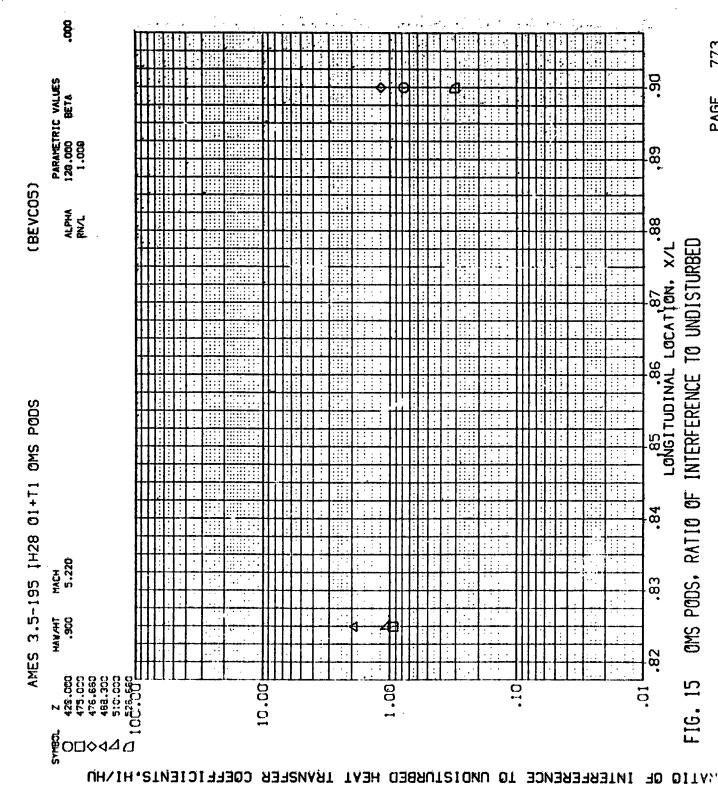
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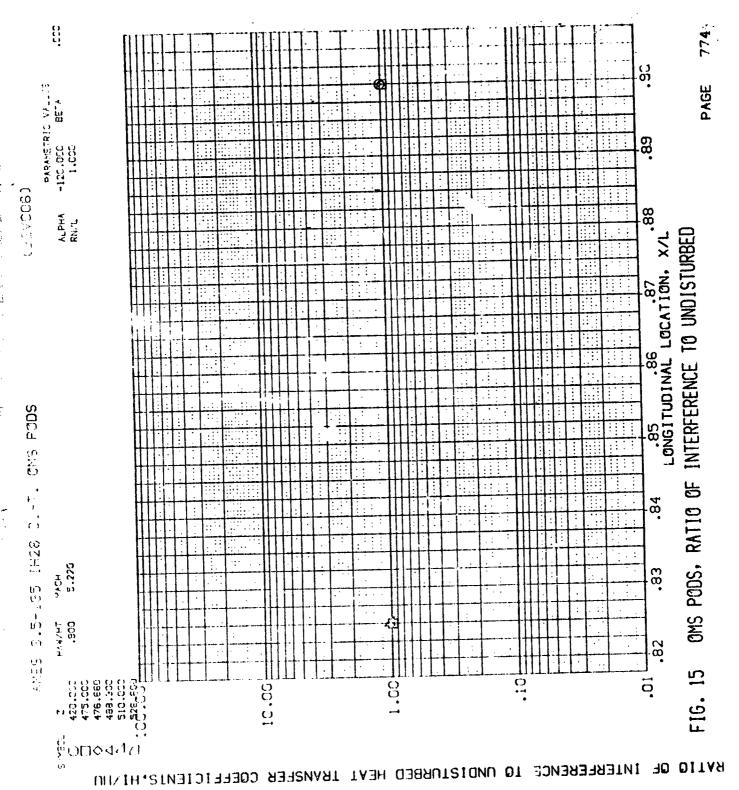
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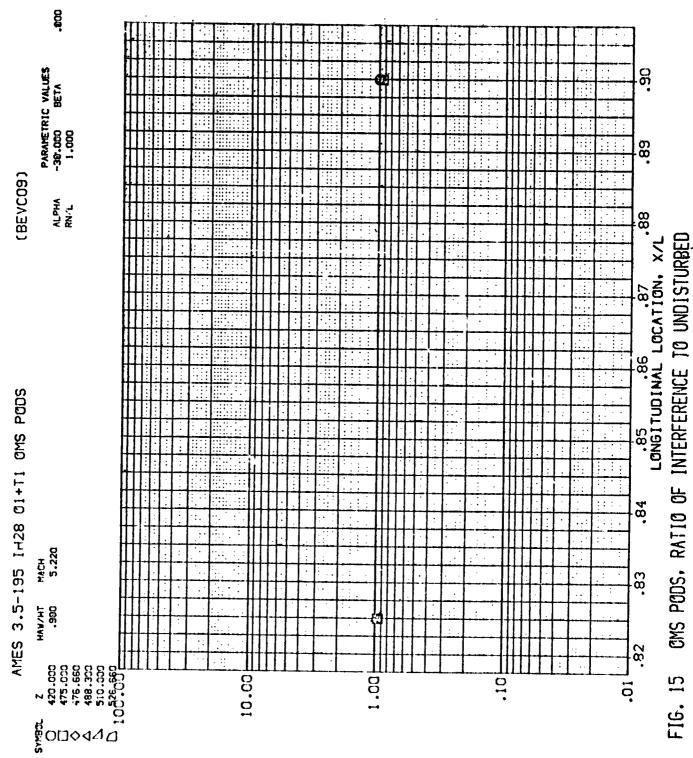




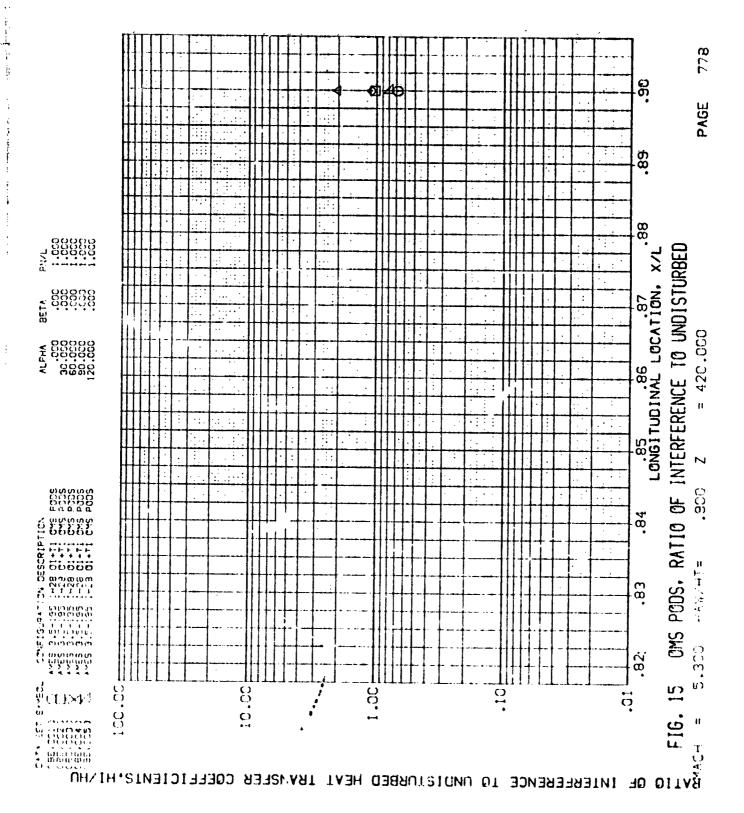
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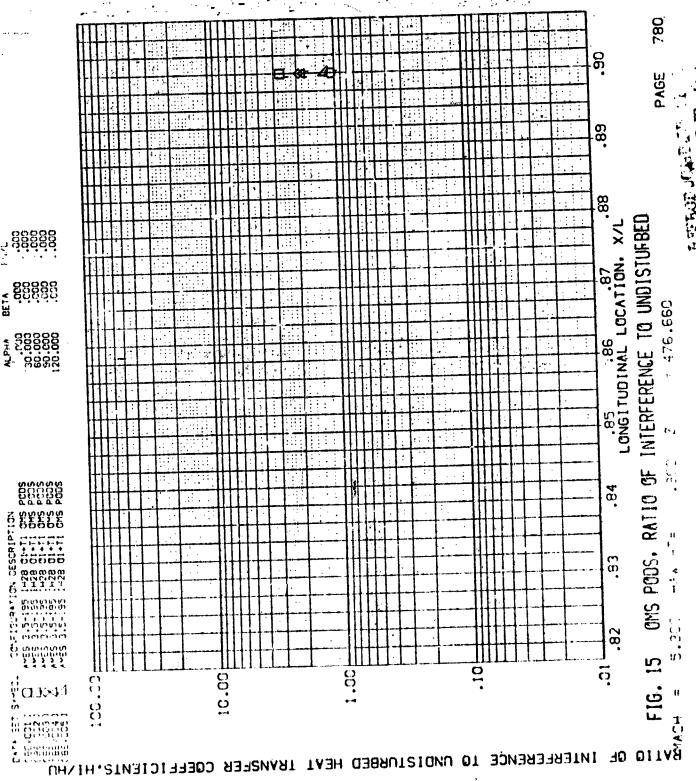
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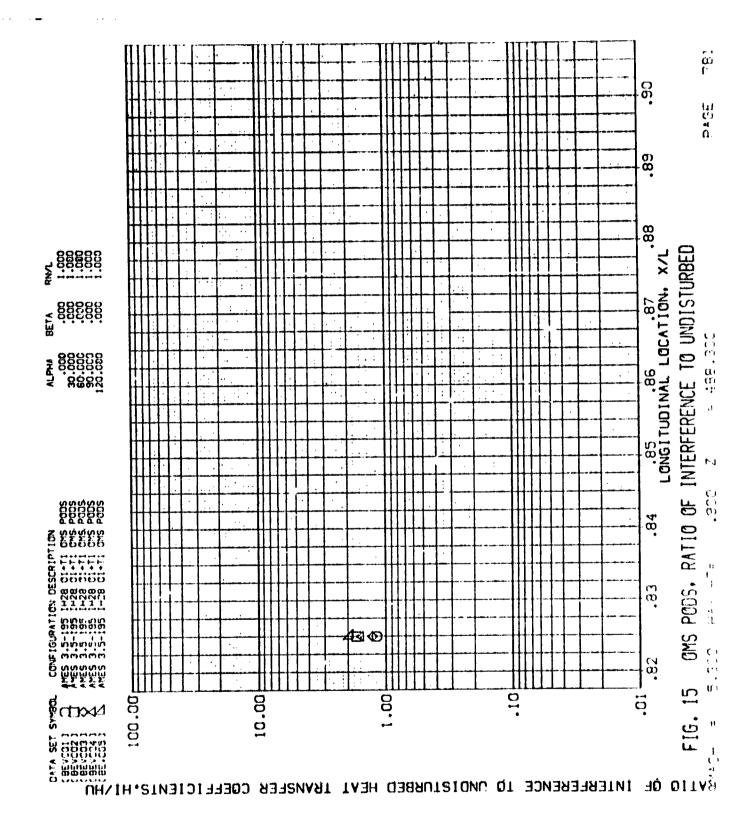
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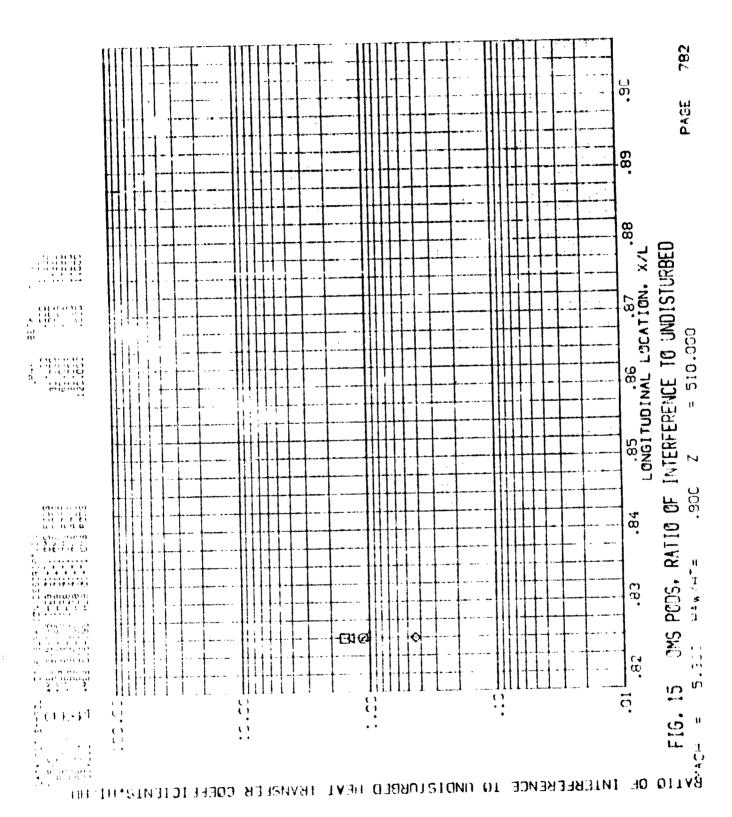
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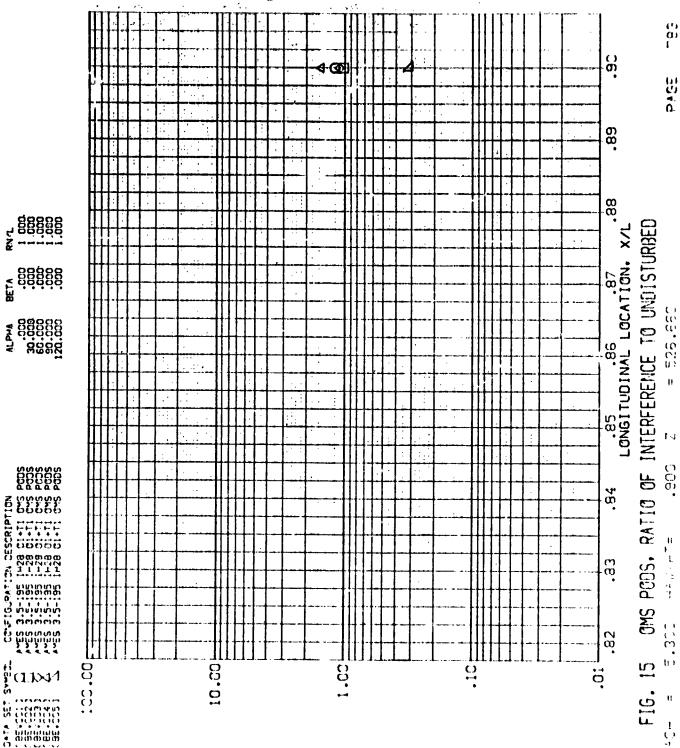
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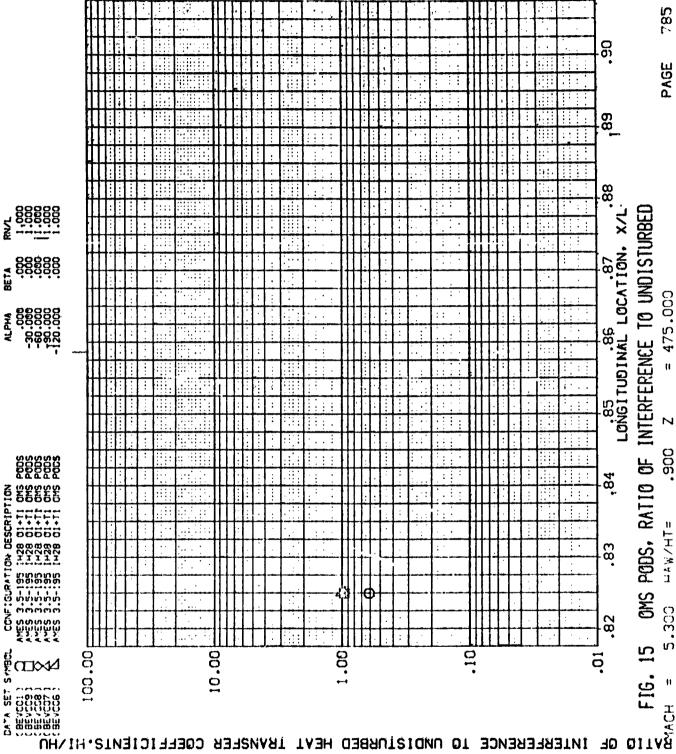






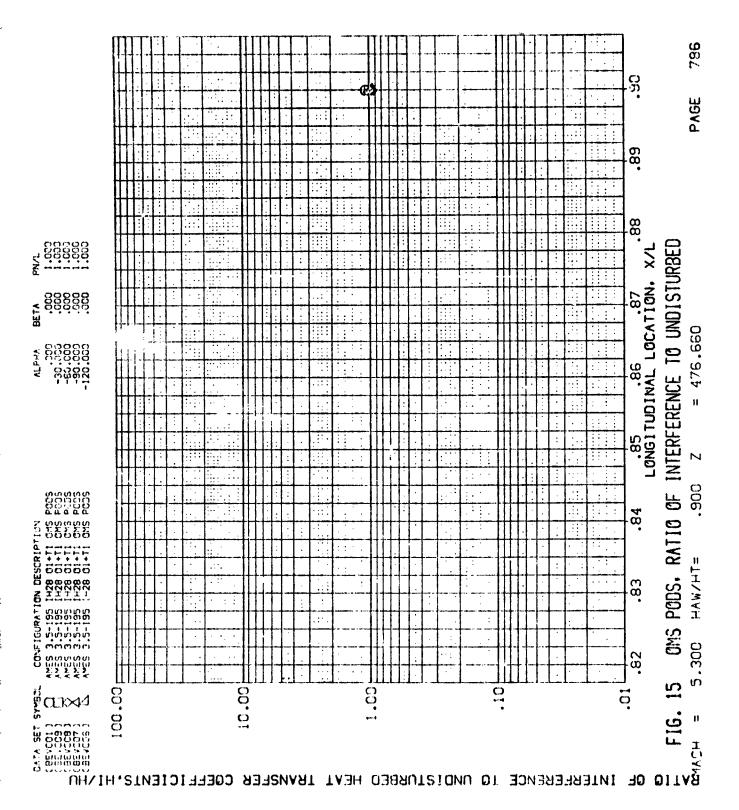
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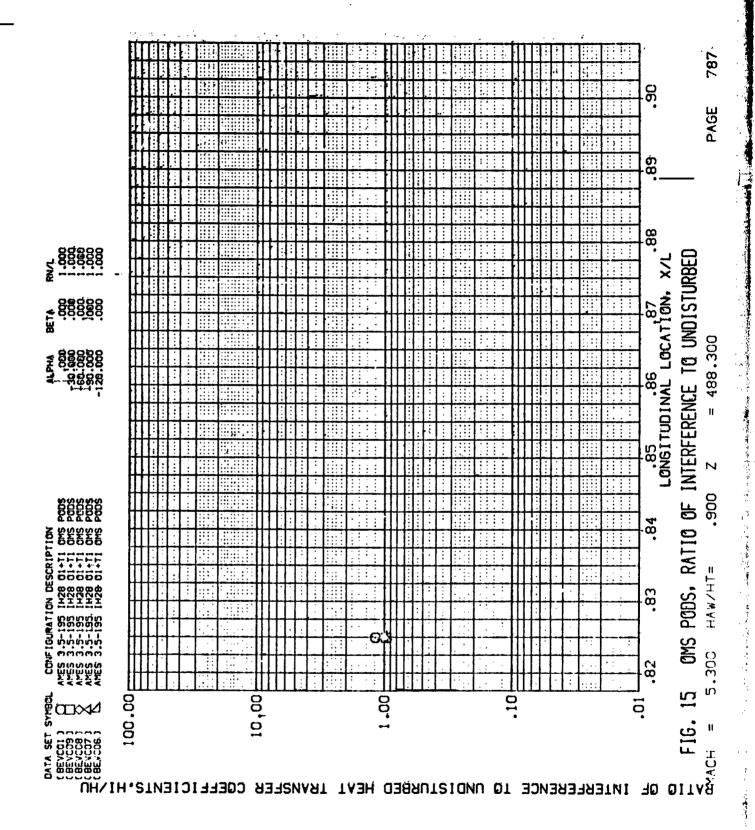
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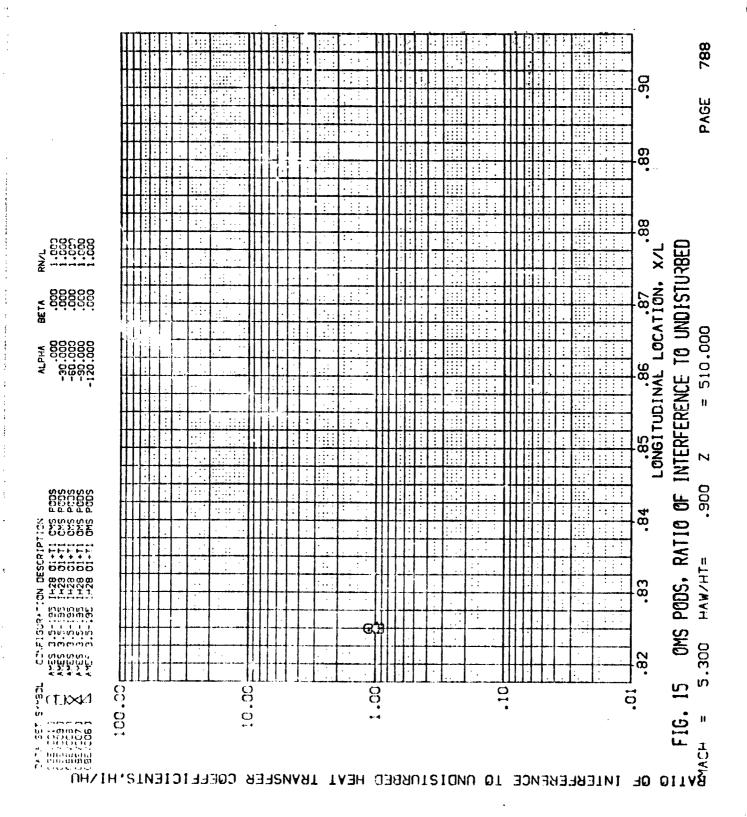


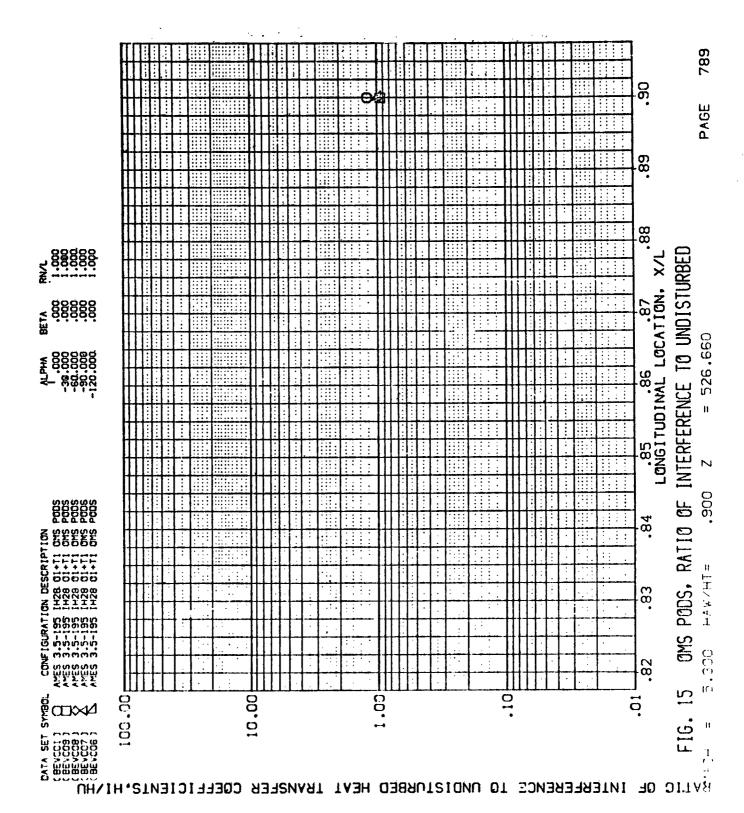
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